

LGHS 43-2-2
COW I

INTERNAL LOADS ANALYSIS

AFT FUSELAGE

VOLUME I

LOCKHEED-GEORGIA COMPANY

A DIVISION OF LOCKHEED CORPORATION

CODE "U"

REPORT NO.: LG1US 46-2-2DATED: 30 September 1968REV. (A) ADP 1 AUG. 69REV. (B) ADP 1-8-71REV. (C) FES 7-31-73REV. (D) KBB 7-31-75REV. (E) JEA 4-3-78**LOCKHEED-GEORGIA COMPANY**

A DIVISION OF LOCKHEED AIRCRAFT CORPORATION

MARIETTA



GEORGIA

TITLE

INTERNAL LOADS ANALYSIS

AFT FUSELAGE

VOLUME I

SUBMITTED UNDER

AIR FORCE CONTRACT NO. AF33(657)15053

MODEL C-5AREFERENCE Form 9 No. US-46-21.0-1 (ASD)PREPARED BY A.D. Pinchot

C-5A Aft Fuselage Structural

A.D. Pinchot

GROUP Integrity GroupCHECKED BY J. HallidayAPPROVED BY P.W. HortonAPPROVED BY L.G. DarbyAPPROVED BY C.R. Bigham

DATE	REV. BY	PAGES AFFECTED	REMARKS
1 Aug. 69	ADP	Volume I:	Rev. A - Revised to incorporate SPO comments
		Title Page, ii, iv,	per LGD 525294 and to include Rev. C loads
		v, 1.2, 2.1-2.9, 4.1,	from LG1US44-1-2.
		5.5, 6.4-6.114, 6.18	
		-6.201, 7.4-7.39,	
		7.145.1-7.145.3,	
		7.146.1, 7.146.2,	

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1 Aug. 69	ADP	7.147.1-7.147.3, 7.148.1, 7.148.2 7.149, 7.150.1- 7.150.3, 7.151.1- 7.151.6, 7.152.1- 7.152.6, 7.153.1- 7.153.4, 7.154, 7.188- 7.209	Rev. A - Revised to incorporate SPO comments per LGD 525294 and to include Rev. C loads from LG1US44-1-2.
		VOLUME II: Title Page, ii, iv, v, 1.2, 8.4-8.14, 8.288-8.299, 9.1- 9.13, 9.13.1, 9.14- 9.18, 10.3, 11.150- 11.156, 14.1-14.3	
8 Jan. 71	ADP	VOLUME I Title Page, v, 2.8, 5.6, 5.6.1, 6.4, 6.9.1, 6.18.1, 6.29.1, 6.29.2, 6.43.1, 6.43.2, 6.50, 6.51, 6.53.1, 6.53.2, 6.64.1, 6.73.1, 6.84.1, 6.84.2, 6.98.1, 6.98.2, 6.105, 6.106, 6.108.1, 6.108.2, 6.165-6.178, 6.190, 6.294, 6.295, 6.335-6.339, 7.143, 7.162-7.187, 7.205, 7.190 VOLUME II Title Page, v, 11.14, 11.16	Rev. B - Revised to incorporate addition of exterior straps around upper lobe doors (F.S. 1844 to 1884), the redesign of troop deck splices at F.S. 1804, 1884, 1964 & 2044, and to incorporate engineering corrections.
31 July 73	FES	Volume I Title Page, ii, iv, v, 2.1. 2.8.1 2.9 4.0 Volume II Title Page ii, iv, v 15.1	Rev. C - Revised to Incorporate LDCS Loads. Revised Revised Added Revised Revised Rev. C Revised to incorporate LDCS Loads. Revised Revised Added
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	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	FORM.
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FOREWORD

The purpose of this report is to present the C-5A Aft Fuselage Internal Loads as authorized under Air Force Contract AF 33(657)-15053. Empennage-Fuselage Interface Loads are also presented. This report is submitted in partial fulfillment of requirements specified in Form 9 No. US 46-21.0-1 (ASD).

Volume I - Sections 1 through 7
 Aft Fuselage Structure from
 F.S. 1744 to F.S. 2538, excluding
 the Aft Pressure Bulkhead and
 Sloping Longerons.

Volume II - Sections 8 through 15
 Aft Fuselage Fin Support
 Structure from F.S. 2538 to
 F.S. 2735, Aft Pressure Bulkhead,
 Sloping Longerons, Aft Ramp, Doors and
 Afterbody Structure from F.S. 2735 to F.S. 2900.5
 LDOS Effect On Internal Loads

(C)

(D)

ALDOS loads are lower than PLDOS loads for all conditions.

(E)

Original design loads are equal to or higher than SLMP loads for all conditions.

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1.0 REFERENCES

1.1 GENERAL REFERENCES

1. Staff: ENGINEERING STRESS MEMO, Lockheed-California Company, Burbank, California; 1966.
2. Anon: STRENGTH OF METAL AIRCRAFT ELEMENTS, Mil-Hdbk-5, Armed Forces Supply Center, Washington, D. C.; 19
3. Staff: C-5A INTERNAL LOADS ANALYSIS - FORWARD FUSELAGE, LGIUS 46-2-4, Lockheed-Georgia Company, Marietta, Georgia; 1968.
4. Staff: C-5A INTERNAL LOADS ANALYSIS - CENTER FUSELAGE, LGIUS 46-2-3, Lockheed-Georgia Company, Marietta, Georgia; 1968.
5. Fiss, L.: M.A. JOB 3893.000 INPUT-OUTPUT EXPLANATION, SMN 81, Lockheed-Georgia Company, Marietta, Georgia; 1966.
6. Roark, R. J.: FORMULAS FOR STRESS AND STRAIN, McGraw-Hill Book Company, New York, N. Y.; 1965.
7. Peery, D. J.: AIRCRAFT STRUCTURES, McGraw-Hill Book Company, New York, 1950.
8. Staff: DEVELOPMENT OF STRUCTURAL MODELS FOR INTERNAL LOADS ANALYSIS BY COMPUTER SOLUTION, LGIUS 46-1-1, Lockheed-Georgia Company, Marietta, Georgia; 1968.
9. Bruhn, E. F.: ANALYSIS AND DESIGN OF FLIGHT VEHICLE STRUCTURES, Tri-State Offset Company, Cincinnati, Ohio; 1965.
10. Staff: C-5A FUSELAGE LOADS, LGIUS 44-1-2, Lockheed-Georgia Company, Marietta, Georgia; 1968.
11. Myers, C. S.: THEORY AND INPUT FOR REDUNDANT FRAME ANALYSIS PROGRAM 3368.101, SMN 154, Lockheed-Georgia Company, Marietta, Georgia; 1964.
12. Staff: UTILIZATION OF FAMAS F IN STRUCTURAL ANALYSIS, SMN 220, Lockheed-Georgia Company, Marietta, Georgia; 1967.
13. Staff: C-5A ALLOWABLE STRESS DATA-FUSELAGE SHELL, LGIUS 46-3-1, Lockheed-Georgia Company, Marietta, Georgia; 1968.
14. Staff: FAMAS BC MANUAL, Lockheed-California, Burbank, California; 1964.
15. Staff: AFT FUSELAGE STRESS ANALYSIS, LGIUS 46-4-2, Lockheed-Georgia Company, Marietta, Georgia; 1968.
16. Timoshenko, S. and Gere, J. M.: THEORY OF ELASTIC STABILITY, McGraw-Hill Book Company, Inc., New York, Toronto, London, 1961.

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17. Staff: Two-Dimensional Computer-Aided Structural Design Theory of Structural Analysis; SMN 229, Lockheed-Georgia Company, Marietta, Georgia, Dec. 1967.
18. Flügge, W: Stress Problems in Pressurized Cabins; NACA TN2612, Stanford University, Feb. 1952.
19. Staff: C-5A Empennage Loads; LG1US 44-1-3, Lockheed-Georgia Company, Marietta, Georgia, 1968.
20. Staff: C-5A Detailed Structural Static Test Program Requirements; LG1US67-1-1, Lockheed-Georgia Company, Marietta, Georgia, April 1968, Rev. C.
21. Staff: C-5 Program - Actual Weight Report for the First C-5A Air Vehicle; LG1US54-5-1, Lockheed-Georgia Company, Marietta, Georgia, 1968.
22. Staff: Structural Design Criteria Report; LG1US43-1-1, Lockheed-Georgia Company, Marietta, Georgia, 1968.
23. Staff: Two-Dimensional Structural Analysis Program (Batch Version) Theory of Calculations and Input Instructions; SMN 252, Lockheed-Georgia Company, Marietta, Georgia, February 1969.
24. Staff: C-5A Center Fuselage Stress Analysis; LG1US46-4-3, Lockheed-Georgia Company, Marietta, Georgia; 1968.
25. Staff: N-Dimensional Enveloping; SMN 321, Lockheed-Georgia Company, Marietta, Georgia, 1968, Rev. A.

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1.2 DRAWING REFERENCES

DRAWING NUMBER

TITLE

4F40000	Aft Fuselage - Complete
4F40002	Side Panel No. 1, Aft Fus., F.S. 1744 to F.S. 1964, Assy of
4F40005	Side Panel No. 2, Upper Aft Fus, F.S. 1964 to F.S. 2361, Assy of
4F40007	Panel Assy - Top, Aft Fus, F.S. 2273 to F.S. 2735
4F40008	Panel No. 3, Troop Compt, Upper Lobe, F.S. 1744 to F.S. 1964, Assy of
4F40009	Panel, Shear-Top, Fus. Sta. 2405 to 2735, Instl. of
4F40010	Panel No. 4, Troop Compt, F.S. 1964 to F.S. 2273, Assy of
4F40018	"Longeron Sloping, Side Panel No. 2, Lower, F.S. 1964.000- 2361.378, Instl. of"
4F40025	Empennage, Instl. of
4F40029	Side Panel No. 1, Aft Fus, F.S. 1744 to F.S. 1964, Instl. of
4F41016	Frame Side - Aft Fus, F.S. 2084, Instl. of
4F41025	Frame- Up. Lobe, F.S. 2273.00, Instl. of
4F41105	Frame Side, Aft Fus, F.S. 1844.00, Instl. of
4F41146	Intercostals Seat Support - Troop Floor
4F41200	Troop Floor Panel, F.S. 1744-1804
4F41201	Troop Floor Panel, F.S. 1804-1884
4F41202	Troop Floor Panel, F.S. 1884-1964
4F41204	Troop Floor Panel, F.S. 1964-2044
4F41206	Troop Floor Panel, F.S. 2044-2100
4F41230	Aft Torque Deck Panels, F.S. 2538-2735
4F41238	Sloping Torque Shelf Panels, F.S. 2100-2273
4F41261	Frame, Side-Aft Fus, Fus Sta 1844.000, Instl of
4F43314	Longeron, Dorsal-Vert. Stab. Attach, F.S. 2637 to 2735
4F43315	Longeron, Dorsal- Vert. Stab. Attach, F.S. 2405 to 2637

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Rev. (A) ADP 1 Aug 69

2.0 SUMMARY OF CRITICAL CONDITIONS

Rev. (C) FES 31 July 73

2.1 Discussion

Rev. (D) JHL, 7-31-75

Rev. (E) JEA, 4-3-78

The Aft Fuselage loads envelopes shown in Section 2.2 were obtained from the "Fuselage Loads" Report, LG1US44-1-2, Rev. "C" (Ref. 10). They describe the maximum loads for each of the six load components (S_x , S_y , S_z , M_x , M_y , and M_z) from F.S. 1744 to F.S. 2735.

- * The critical Aft Fuselage design conditions shown in Section 2.3 were determined by enveloping five load components (S_x not considered) at F.S. 1764 and F.S. 2538 for all fuselage load cases listed in Reference 10. These critical conditions were used to determine the Aft Fuselage Internal Loads by computer solution (FAMAS Program, Ref. 8).

The critical design conditions for the underfin structure, F.S. 2538 to F.S. 2735, and the empennage-fuselage mate joint were determined by enveloping all six load components of empennage swept axis loads as presented in reference 19. Internal loads were determined by computer solution (FAMAS Program, Reference 8).

- (C) LDOS external loads were found to fall within the design load envelopes. For concrete assurance of higher margins of safety, all LDOS cases, with and without ultimate internal pressure, were analyzed for internal loads by the FAMAS redundant analytis model (Ref. 8). Internal loads comparisons were made and it was determined that original design loads exceeded LDOS loads by 4 to 5 percent in the aft fuselage and the empennage-to-fuselage joint structure.

- (D) ALDOS loads are lower than PLDOS loads for all conditions.

- (E) SLMP external loads were found to fall within the design load envelopes. For assurance of positive margins of safety, all SLMP cases, with and without ultimate internal pressure, were analyzed for internal loads by the FAMAS redundant analysis model (Ref. 8). Internal loads comparisons were made and it was determined that original design loads are equal to or higher than SLMP loads.

PREPARED BY A.D. PINCHOT
 DATE 3 JUN. 68
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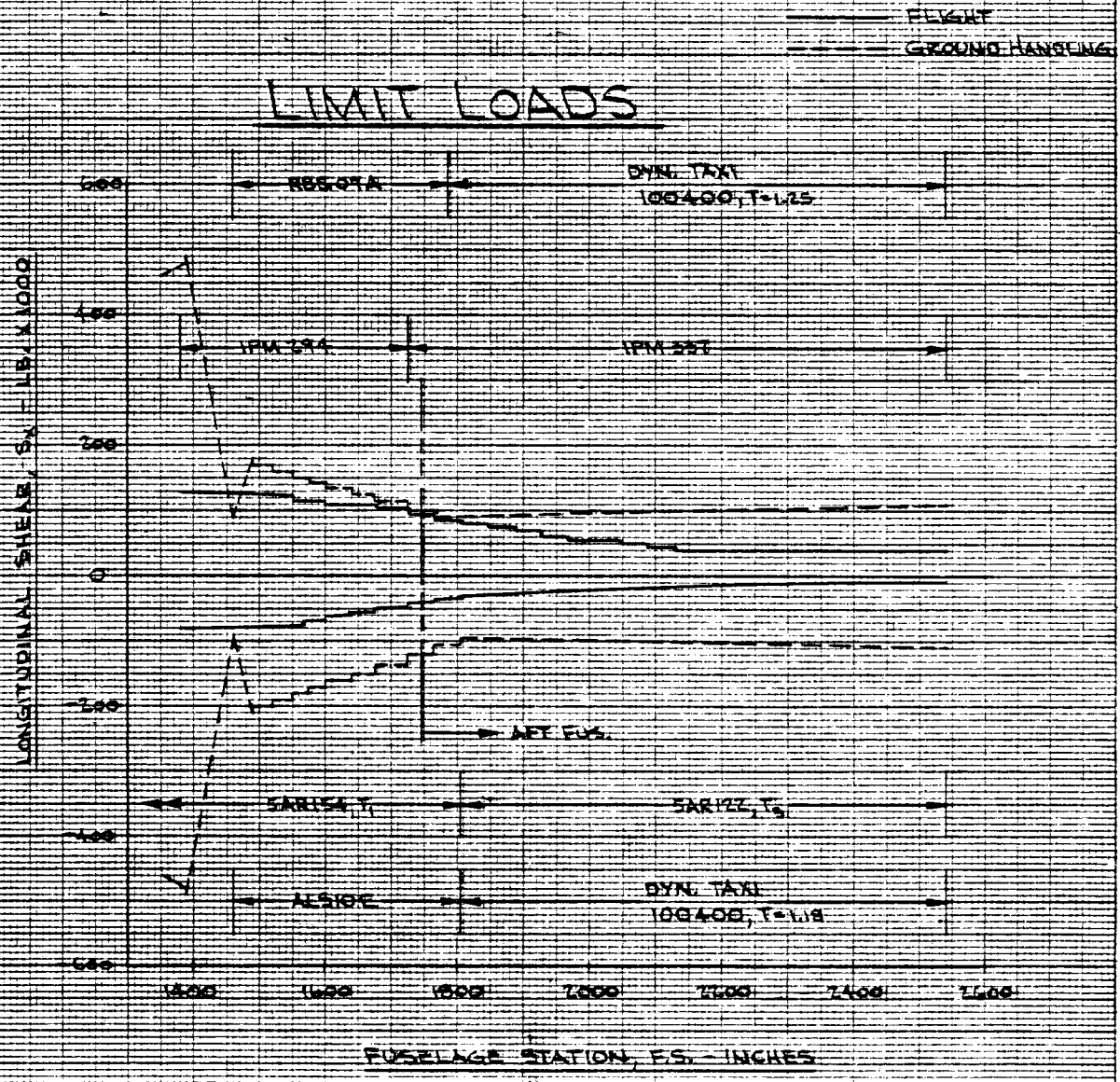
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2.2 AFT FUSELAGE LOADS ENVELOPES

FIGURE 2.11

GROUND HANDLING & FLIGHT CONDITIONS
 LONGITUDINAL ENVELOPE OF LONGITUDINAL SHEAR
 S_x VS. F.S.
 (REF 10, RZ1-005, REV. C)



PREPARED BY L.D. PINCHOT
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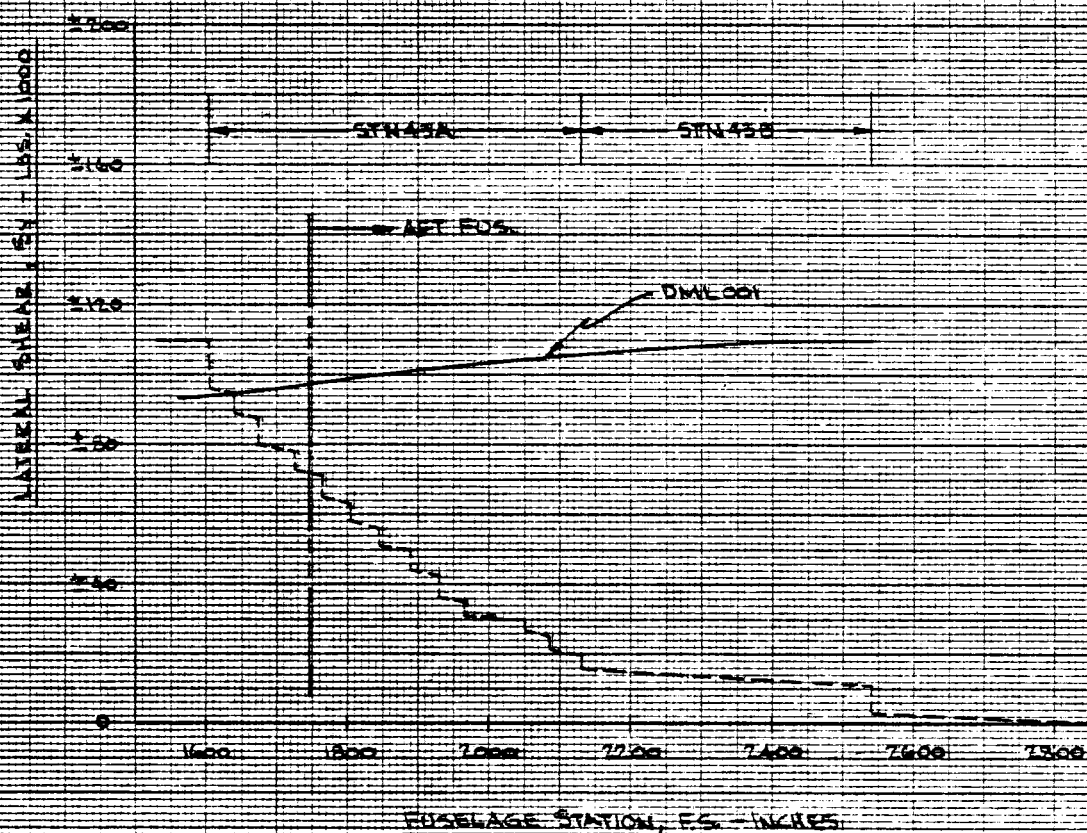
2.2 AFT FUSELAGE LOADS ENVELOPES - CONTINUED

FIGURE 2.1.2

GROUND HANDLING & FLIGHT CONDITIONS
LONGITUDINAL ENVELOPE OF LATERAL SHEAR
SY VS. F.S.
(REF. 10, 821-006, REV. C)

FLIGHT
GROUND HANDLING

LIMIT LOADS



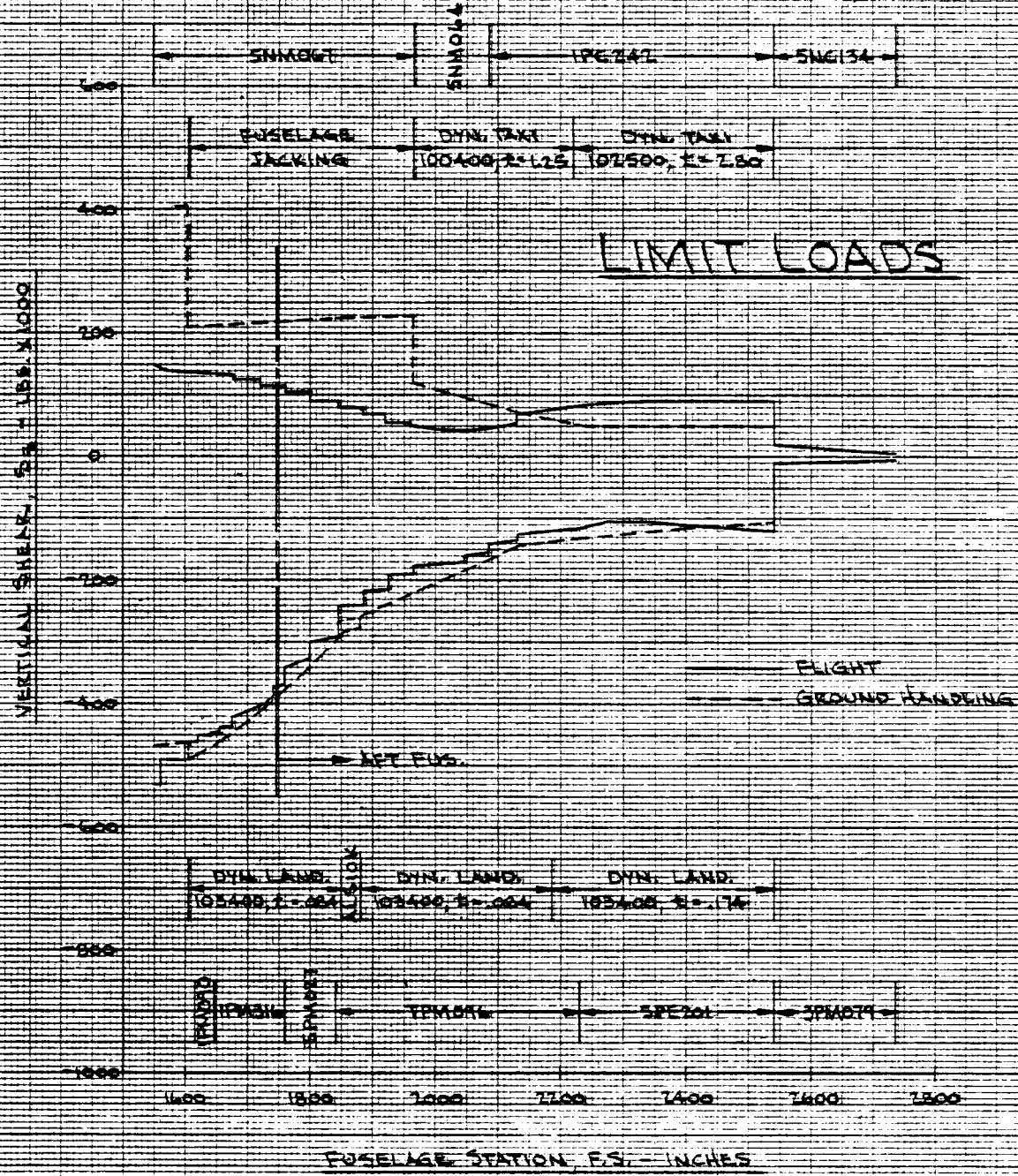
REV. (A) A.D.P. LAUGER

2.2 AFT FUSELAGE LOADS ENVELOPES-CONTINUED

FIGURE 2.13

GROUND HANDLING & FLIGHT CONDITIONS
 LONGITUDINAL VERTICAL SHEAR ENVELOPE

S_z VS. FS. (REF 10, P. 21-001, REV. C)



FUSELAGE STATION, FS - INCHES

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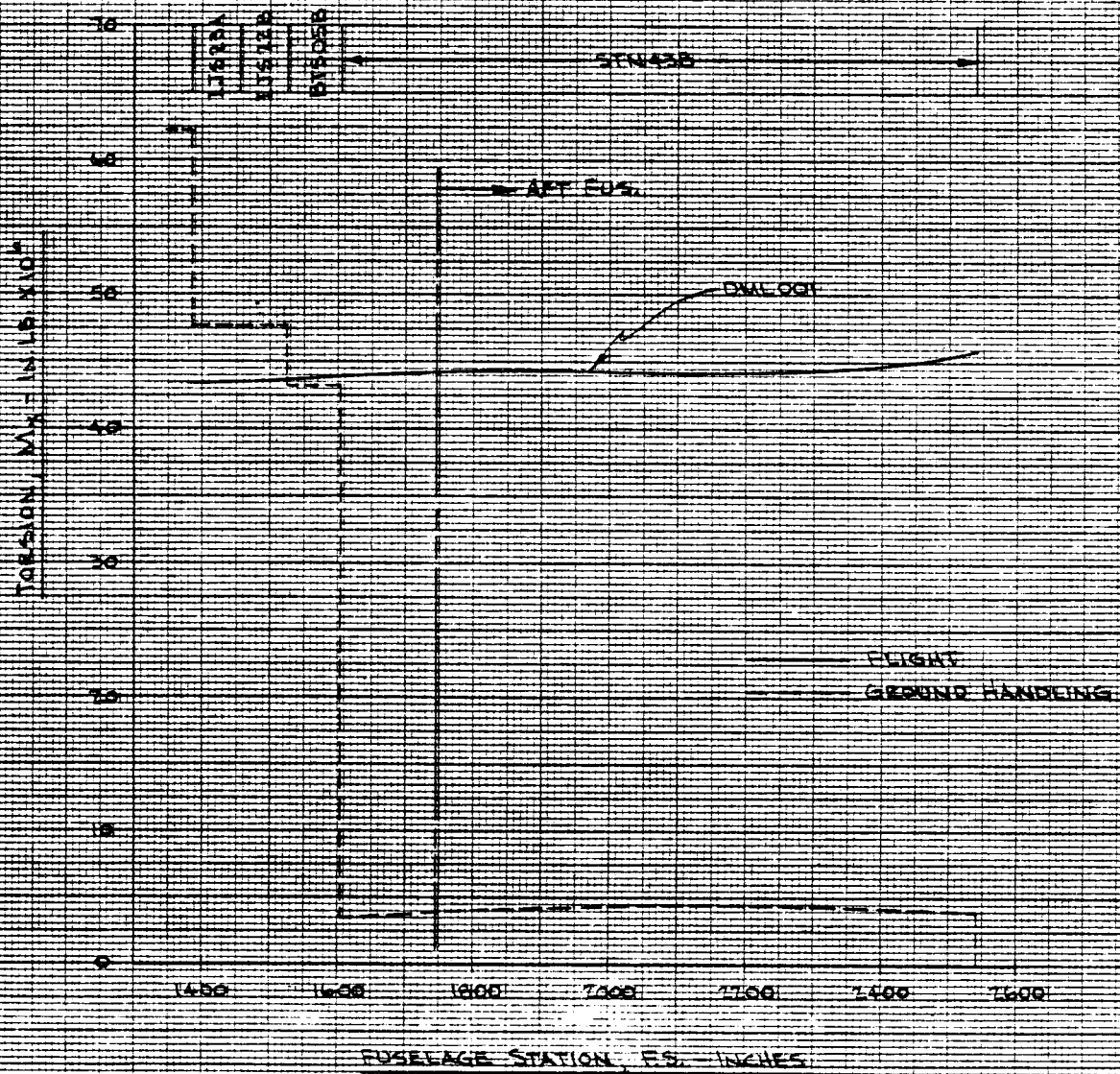
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2.2 AFT FUSELAGE LOADS ENVELOPES - CONTINUED

FIGURE 2.1.4

GROUND HANDLING & FLIGHT CONDITIONS
LONGITUDINAL TORSION ENVELOPE
M_x VS. FS
(REF. 10, PL. I-002, REV. C)

LIMIT LOADS



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2.2 AFT FUSELAGE LOADS ENVELOPES - CONTINUED

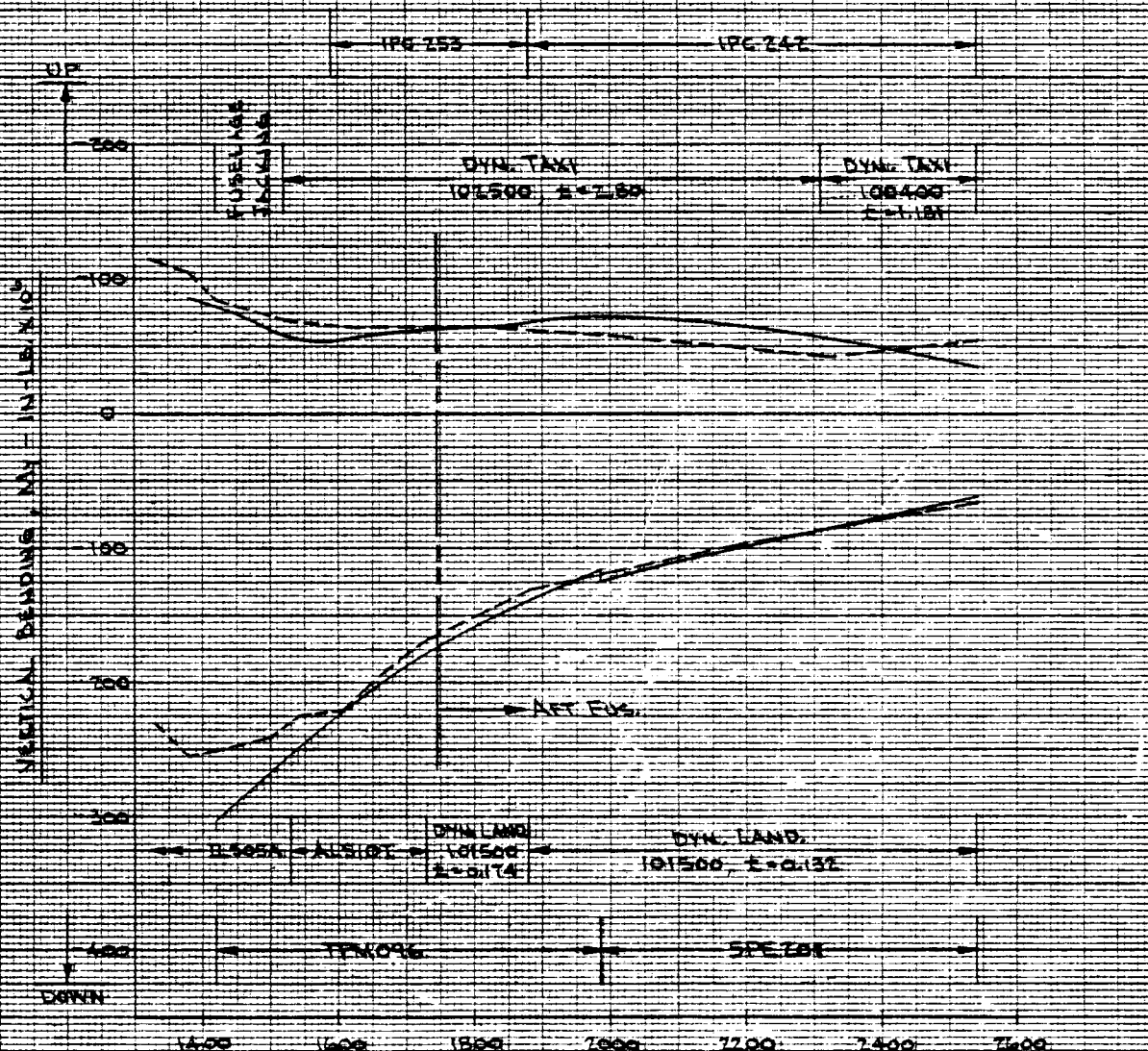
FIGURE 2.1.5

GROUND HANDLING & FLIGHT CONDITIONS
 LONGITUDINAL VERTICAL BENDING ENVELOPE

M_y VS. FS.

(REF. 10, P. 21-005, REV. C)

LIMIT LOADS



FUSELAGE STATION, FS. - INCHES

PREPARED BY A.D. PINCHOT
 DATE 3 JUN. 68
 CHECKED BY J. HALLIDAY

LOCKHEED-GEORGIA COMPANY
 A DIVISION OF LOCKHEED AIRCRAFT CORPORATION

REPORT NO. LG1US46-2-2
 MODEL C-5A
 PAGE 2.7

REV. (A) A.D.P. 1 AUG. 67

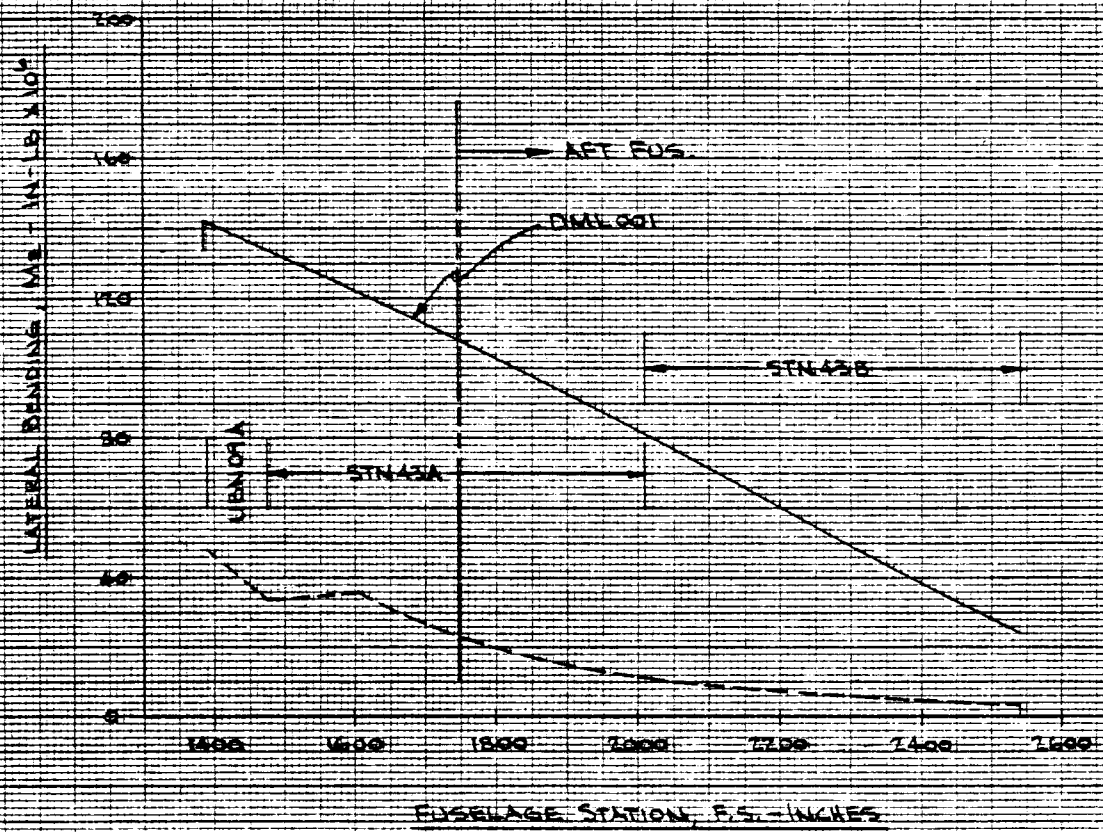
2.2 AFT FUSELAGE LOADS ENVELOPES - CONTINUED

FIGURE 2.1.6

GROUND HANDLING & FLIGHT CONDITIONS
 LONGITUDINAL LATERAL BENDING ENVELOPE
 M_z VS. F.S.
 (REF. 18, P. 21-004, REV. C)

—— FLIGHT
 ---- GROUND HANDLING

LIMIT LOADS



PREPARED	NAME F. E. Schreiber	DATE 7-31-73	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
CHECKED	J. E. Aaron	7-31-73	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	VOL. I 2. 8. 1		
APPROVED	R. W. Smith	7-31-73		MODEL C-5A LGIUS46-2-2 REPORT NO.		

Rev. C , FES ,7-31-73

2. 3. 1. 1 Aft Fuselage LDCS Conditions

The LDCS conditions listed below replace the conditions listed in 2. 3. 1. Conditions listed in 2. 3. 2 are still valid. All of these cases, with and without ultimate pressure where applicable, were analyzed for internal loads by the FAMAS computer program (Ref. 8).

NO.	CONDITION	NO.	CONDITION	NO.	CONDITION
1	5PM237	29	5PM237 + PR	57	100400 (1. 18)
2	2PM497	30	2PM497 + PR	58	100400 (1. 25)
3	3PM374	31	3PM374 + PR	59	101500 (0. 08)
4	1PE094	32	1PE094 + PR	60	101500 (0. 13)
5	5NM520	33	5NM520 + PR	61	101500 (0. 17)
6	KPM274	34	KPM274 + PR	62	101500 (0. 18)
7	KGM274	35	KGM274 + PR	63	102500 (2. 80)
8	1PM276	36	1PM276 + PR	64	103400 (0. 08)
9	1PM278	37	1PM278 + PR	65	ALS11A
10	1AR721	38	1AR721 + PR	66	ALS11I
11	1AR721T2	39	1AR721T2 + PR	67	ALS11K
12	1AR721T3	40	1AR721T3 + PR	68	BTN09A
13	1PM132	41	1PM132 + PR	69	BTS09A
14	1PM133	42	1PM133 + PR	70	BTS12A
15	1PM134	43	1PM134 + PR	71	DML001
16	1PM137	44	1PM137 + PR	72	DML002
17	5PM066	45	5PM066 + PR	73	DML003
18	1PG304	46	1PG304 + PR	74	LAR010T1
19	1GM321	47	1GM321 + PR	75	LAR010T3
20	1GM323	48	1GM323 + PR	76	SBN01A
21	1PG305	49	1PG305 + PR	77	STN43A
22	1PE088	50	1PE088 + PR	78	DML00R
23	1PC663	51	1PC663 + PR	79	Jacking
24	1PM232	52	1PM232 + PR	80	DML00R + PR
25	1NM421	53	1NM421 + PR	81	DML001 + PR
26	1SS314	54	1SS314 + PR	82	DML002 + PR
27	1EY163	55	1EY163 + PR	83	DML003 + PR
28	10Y411	56	10Y411 + PR		

NAME		DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
PREPARED	A. D. Pinchot	1 Aug 69	PAGE VOL. I 2.9			
CHECKED	R. F. Wilkinson	1 Aug 69	TITLE INTERNAL LOADS ANALYSIS		MODEL	C-5A
APPROVED	L. G. Darby	1 Aug 69	AFT FUSELAGE		LG1US46-2-2 REPORT NO.	

REV. A, ADP, 1 Aug 69
REV. C, FES, 31 July 73

2.3.3 The design conditions for the underfin structure, F.S. 2538 to F.S. 2735, and the empennage-fuselage mate joint are listed below. These conditions were used in the FAMAS Computer Program, Ref. 8, to determine the internal loads.

<u>NUMBER</u>	<u>CONDITION</u>	<u>NUMBER</u>	<u>CONDITION</u>
1	1PC242	17	DSB033
2	1PC190	18	DSB056
3	5PE174	19	DML001
4	5PM370	20	DML002
5	5PM368	21	DML003
6	5PM329	22	1OY254
7	1NM172	23	1EY107
8	1NM175	24	1SS078
9	5NC129	25	1EL027
10	5NC134	26	1PM837
11	1NE089	27	(Not Used)
12	1NE092	28	5PE201
13	5PM369	29	101500 (t = .13)
14	5PM371	30	100400 (t = 1.18)
15	DSB031	31	DSB037
16	DSB032	32	DSB038
		33	DSB039

2.3.3.1 LDSCS cases plus unaffected holdover design cases are listed below. These conditions were used in the FAMAS computer model, ref. 8, to determine internal loads for the underfin structure.

<u>NUMBER</u>	<u>CONDITION</u>	<u>NUMBER</u>	<u>CONDITION</u>
1	DML001	11	100400 (t = 1.18)
2	DML002	12	101500 (t = .13)
3	DML003	13	1PE088 (LDSCS)
4	DSB031	14	1NM421 (LDSCS)
5	DSB032	15	1PM232 (LDSCS)
6	DSB033	16	1SS314 (LDSCS)
7	DSB037	17	1EY163 (LDSCS)
8	DSB038	18	1OY411 (LDSCS)
9	DSB039	19	1PC663 (LDSCS)
10	DSB056		

2.3.4 Other aft fuselage critical design conditions that were considered separately were 17.4 psi burst pressure, crash condition, aft side and center cargo door loads, safety of flight items, and vehicle and equipment loading as specified in reference 10, rev. "C".

NAME		DATE	TEMP.	PERM.
PREPARED	A.D. Pinchot	7 Sep 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE 1 3.1
CHECKED	L.G. Darby	7 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A
APPROVED	P.W. Horton	7 Sep 68		LGIUS 46-2-2 REPORT NO.

3.0 SYMBOLS

<u>SYMBOL</u>	<u>DEFINITION</u>	<u>UNIT</u>
A	Area	In ²
Ac	Actual Compression Area	In ²
At	Actual Tension Area	In ²
a	Distance	In
b	Width, Subscript "Bending"	In
br	Subscript "Bearing"	-
C	Circumference	In
c	Fixity Coefficient for Columns; Subscript "Compression"	-
c	Distance from neutral axis to extreme fiber	In
cr	Subscript "Critical"	-
D	Diameter	In.
d	Depth or Height	In.
E	Modulus of Elasticity in Tension	lbs/in ²
e	Unit Deformation or Strain	In/In
e	Eccentricity	In.
e	Subscript "Effective"	-
f	Calculated Stress	Lbs/In ²
G	Modulus of Rigidity	Lbs/In ²
H	Subscript "Hoop"	-
h	Height or Depth	In.
I	Moment of Inertia	In ⁴
J	Polar Moment of Inertia	-
K	A Constant	-
L	Length	In.
M.	Moment	In/Lbs.
N	Cycles	-
n	Subscript "Normal"	-
P	Applied Load	Lbs.
p	Pressure	Lbs/In ²
q	Shear Flow	Lbs./In ²
r	Radius	In.
s	Subscript "shear"	-

	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	PERM. 3.2
PREPARED	A. D. Pinchot	7 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			
CHECKED	L.G. Darby	7 Sep 68		MODEL	C-5A	
APPROVED	P.W. Horton	7 Sep 68		LGJUS	46-2-2	REPORT NO.

3.0 SYMBOLS (Continued)

<u>SYMBOL</u>	<u>DEFINITION</u>	<u>UNIT</u>
T	Torque	In/Lbs.
t	thickness; subscript "Tension"	In
u	Subscript "Ultimate"	-
v	Shear Load	Lbs.
w	Running Load	Lbs/in
X	A/P Longitudinal Axis (FS)	-
x	Value of an Individual Measurement	-
Y	A/P Transverse Axis (BL)	-
$\bar{y}$	Distance of Centroid to Reference Axis	In.
z	A/P Vertical Axis (WL)	-
α	Angle	Degrees
θ	Angle	Degrees
ϕ	Angle	Degrees
δ	Deflection	In.
Δ	Used before symbol means increase/decrease; part of	-

PREPARED	A. D. Pinchot	15 Aug 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I	3.3
CHECKED	J. Halliday	15 Aug 68	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	L.G. Darby	15 Aug 68		REPORT NO.	LGIUS 46-2-2	

3.0 SYMBOLS (Continued)

ABBREVIATIONS

A/P	AIRPLANE
BKH'D	BULKHEAD
BL	BUTTOCK LINE
CASD	COMPUTER AIDED STRUCTURAL DESIGN, REF. 17
CONT.	CONTINUED
DWG.	DRAWING
EFF.	EFFICIENCY
ELEM	ELEMENT
FAMAS	FLUTTER AND MATRIX ALGEBRA SYSTEM, REF. 12
FUS.	FUSELAGE
FS	FUSELAGE STATION
FWD.	FORWARD
G.P.	GRID POINT
LD	LOAD
LH	LEFT HAND
MS	MARGIN OF SAFETY
MATL.	MATERIAL
REF.	REFERENCE
RH	RIGHT HAND
SECT.	SECTION
STRG.	STRINGER
ULT	ULTIMATE
WL	WATER LINE

PREPARED	J. H. Ray	9 Sep 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	VOL. I	4.
CHECKED	J. Halliday	9 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	L.G. Darby	9 Sep 68		REPORT NO.	LGJUS 46-2-2	

REV. (A) A.D.P. 1 AUG. 69
 REV. (C) F.E.S. 31 July 73
 REV. (E) JEA, 4-3-78

4.0 INTRODUCTION

This report contains the internal loads analysis of the C-5A Aft Fuselage from F.S. 1744 to F.S. 2900 including the empennage-fuselage mate joint. Sketches and an introduction to each section adequately describe the structure being analyzed. Methods of analysis are herein referenced. A summary of critical loads and conditions is presented for each section.

(C) The internal loads discussed in this revision (Rev. C) result from static passive LDCS (Lift Distribution Control System) applied external loads on the aircraft as presented in Appendix A of Reference 10.

(C) All LDCS load cases, with and without ultimate internal pressure, were analyzed for internal loads by the FAMAS redundant analysis model (Ref. 8). Internal loads comparisons established the fact that original design loads exceeded LDCS design loads by four to five percent. Refer to Section 15.0 for the LDCS effect on internal loads.

(E) SLMP external loads were found to fall within the design load envelopes. For assurance of positive margins of safety, all SLMP cases, with and without ultimate internal pressure, were analyzed for internal loads by the FAMAS redundant analysis model (Ref. 8). Internal loads comparisons were made and it was determined that original design loads are equal to or higher than SLMP loads.

PREPARED	NAME J.H. Ray	DATE 5 Sep 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	PERM. 5.1
CHECKED	J.Halliday	5 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L.G. Darby	5 Sep 68		GIUS 46-2-2 REPORT NO.		

5.0 AFT FUSELAGE GEOMETRY AND DESIGN

5.1 GEOMETRY

The aft fuselage extends from F.S. 1744 to F.S. 2927. The fuselage is pressurized back to the aft pressure bulkhead at F.S. 2101. The cargo compartment extends back F.S. 1744 to F.S. 1964 and is bounded on the upper side by the troop floor at W.L. 314 and the cargo floor at W.L. 140. The pressure door is supported by the pressure bulkhead at W.L. 303. The lower edge is canted aft and meets the aft ramp at F.S. 2131. The forward end of the aft ramp attaches at F.S. 1964.

The troop compartment extends from F.S. 1744 to F.S. 2101 and is bounded on the lower side by the troop deck at W.L. 314.

The structure aft of the pressure bulkhead is primarily for transfer of loads from the empennage. This section extends from F.S. 2101 to F.S. 2735. The fin-fuselage mate joint is at W.L. 424.65 and extends from F.S. 2538 to F.S. 2735. The front spar is at F.S. 2538 and the rear spar at F.S. 2735.

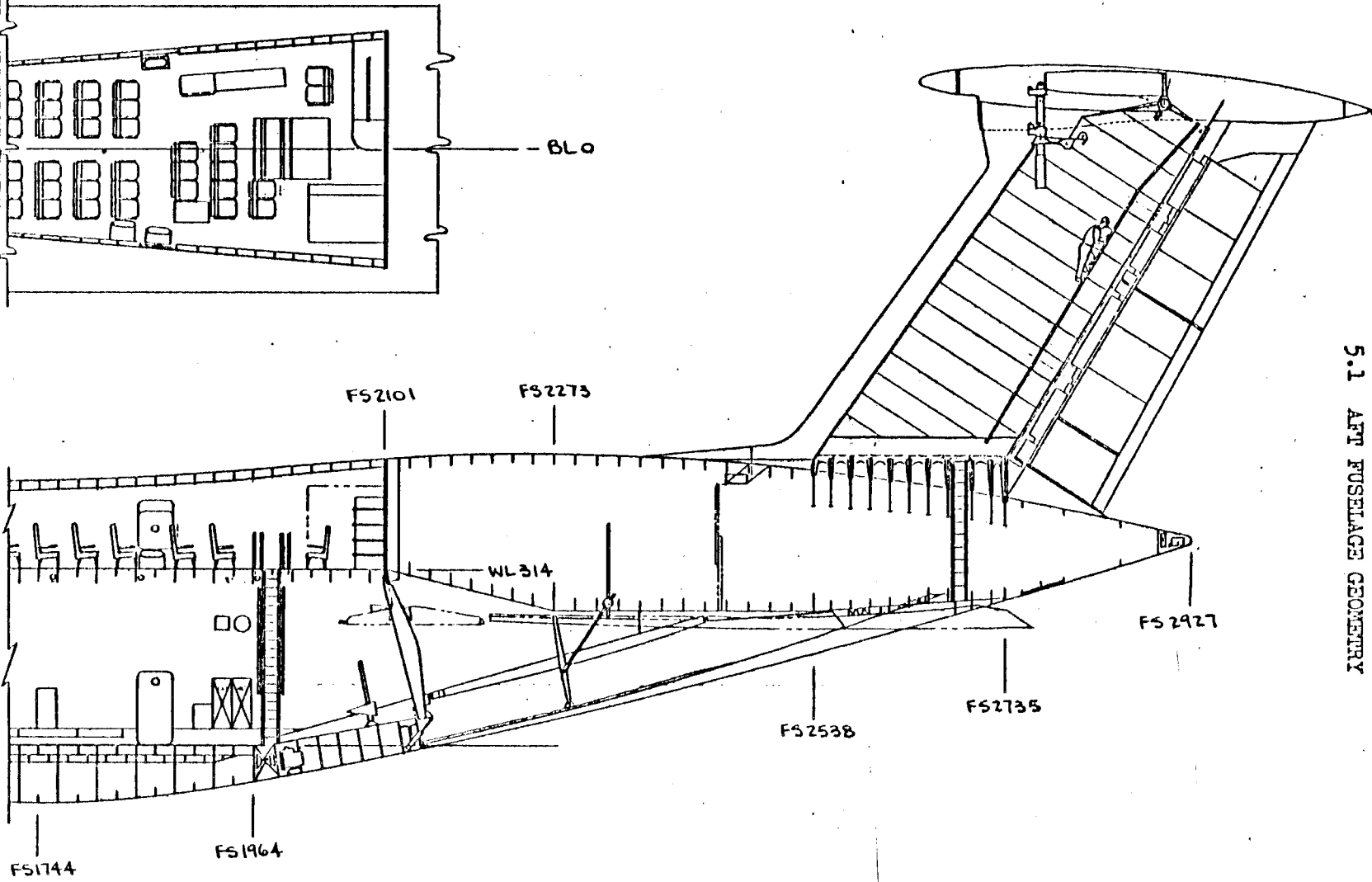
The aft fuselage frames are spaced approximately at 20 inch intervals and are basically for fuselage shell stability purposes. They also serve various other load carrying purposes such as aft ramp latch load reactions and redistributions of shell internal loads.

The afterbody is attached to F.S. 2735 and extends to F.S. 2927. It is primarily a closeout fairing structure.

A more detailed explanation of the structure and the critical load conditions is presented in each section of the report.

PREPARED	A. D. Pinchot	21 Nov 67	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I	5.2
CHECKED	L. G. Darby	21 Nov 67	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	R. W. Smith	21 Nov 67		REPORT NO.	IGIUS 46-2-2	

5.1 AFT FUSELAGE GEOMETRY

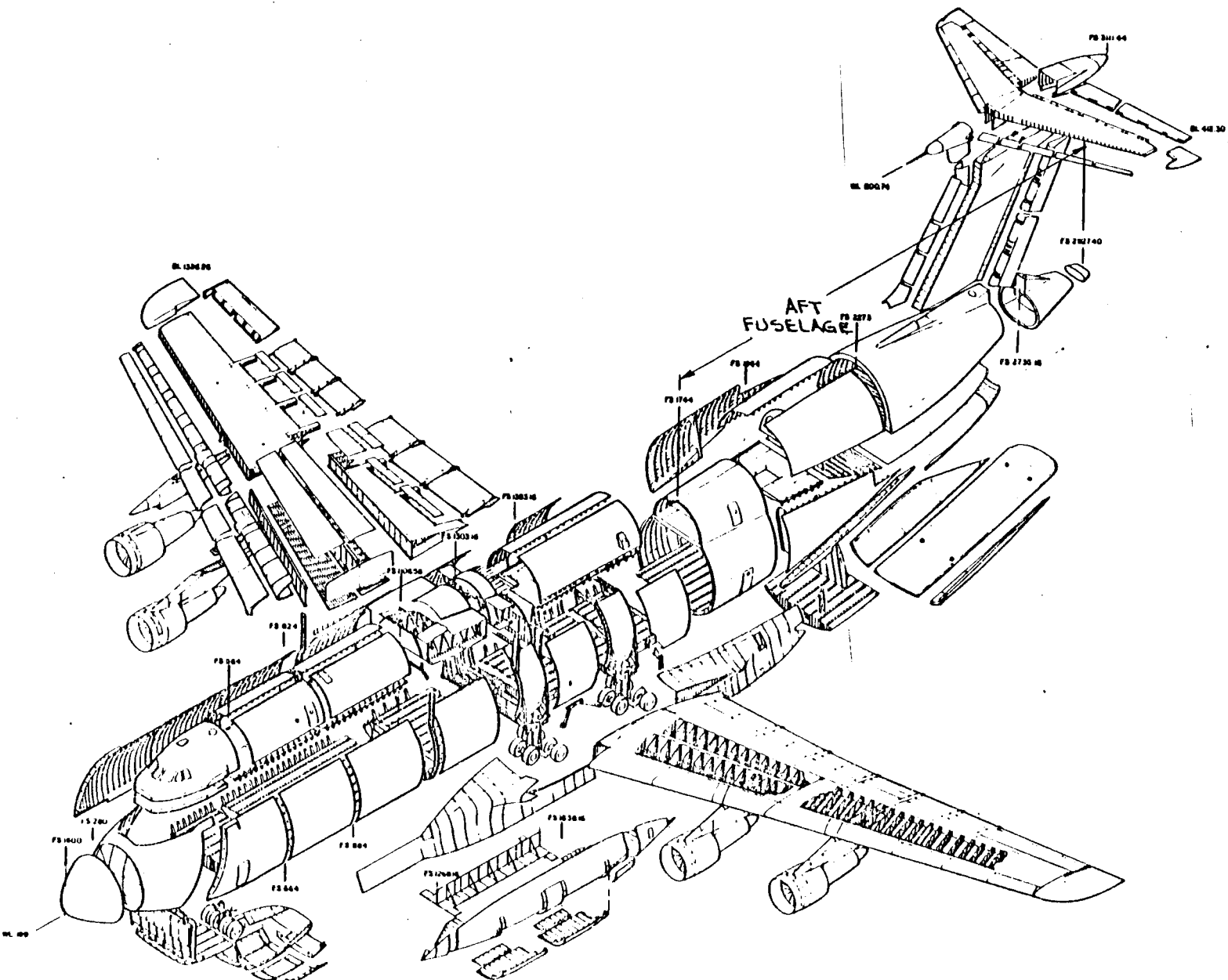


REF. DWG. 4M00005

• **GUARANTEE PAGES**

NAME	DATE	TITLE	PAGE	REV.
PREPARED A.D. Pinchot	22 Jul 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	1	5.4
CHECKED J. Halliday	22 Jul 68	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A	
APPROVED L.G. Darby	22 Jul 68		LAUS 46-2-2 REPORT NO.	

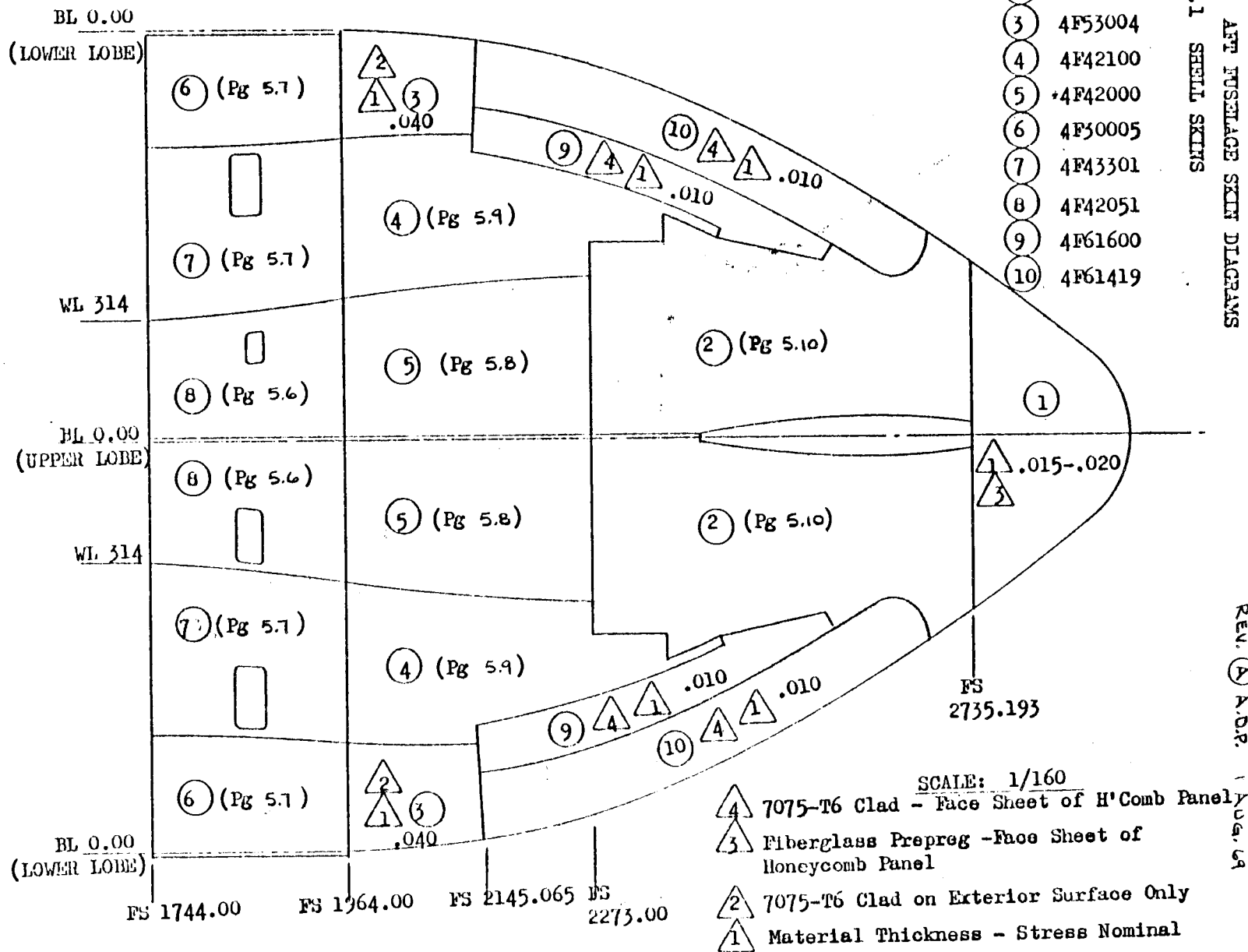
5.3 ISOMETRIC



PREPARED	A. D. Pinchot	17 Nov 67	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	5.5
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5.4 AFT FUSELAGE SKIN DIAGRAMS 5.4.1 SHELL SKINS

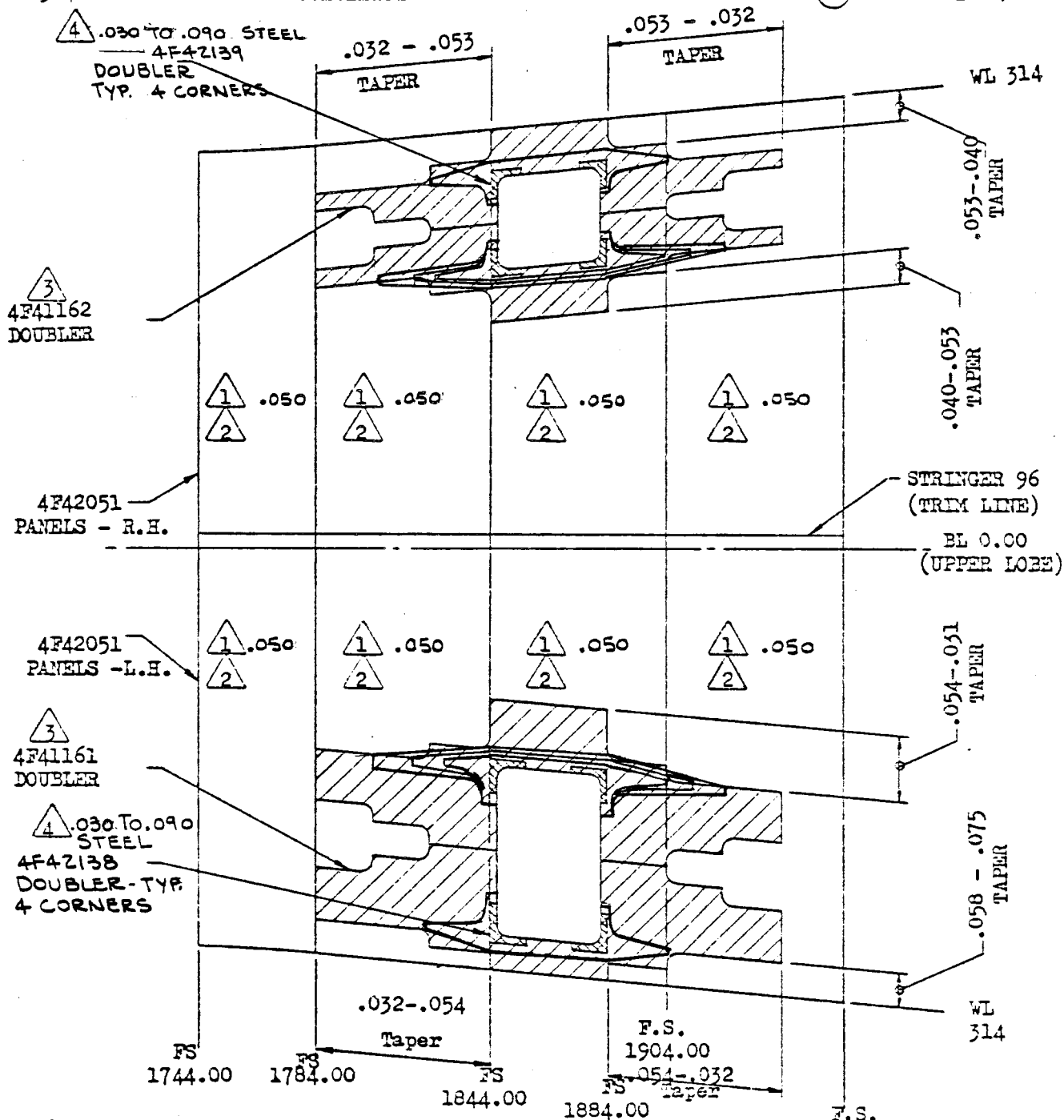
- 1 4F41184
- 2 4F43324
- 3 4F53004
- 4 4F42100
- 5 4F42000
- 6 4F30005
- 7 4F43301
- 8 4F42051
- 9 4F61600
- 10 4F61419



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5.4.1 SHELL SKINS - Continued

REV. (3) M.B. 1-8-71



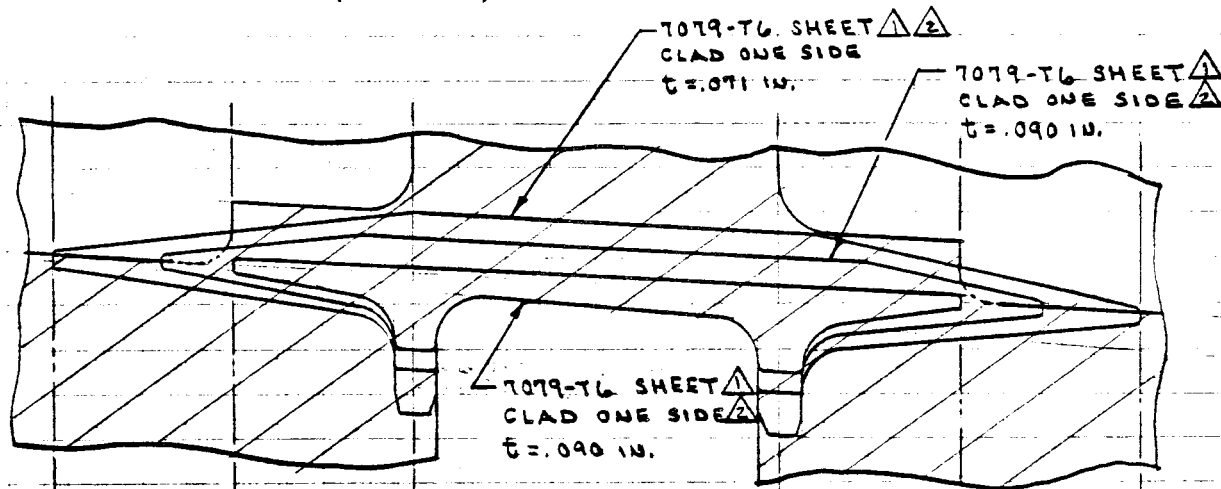
- 4 4130 Steel-Et. Tr. 180-200 KSI
- 3 7079-T6 Clad Both Sides
- 2 7079-T6 Clad on Exterior Surface Only
- 1 Material Thickness - Stress Nominal

SCALE: 1/50

Prepared	NAME A. D. Pinchot	DATE 17 Nov 67	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP VOL. I	PERM. 5.6.
Checked	L. G. Darby	17 Nov 67	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	R. W. Smith	17 Nov 67		Report No.	LGIUS46-2-2	

5.4.1 SHELL SKINS -(Continued)

REV. ⑤ KRW 1-8-71



VIEWS ARE TYP.

FOR L.H. & R.H.

F.S.
1804.

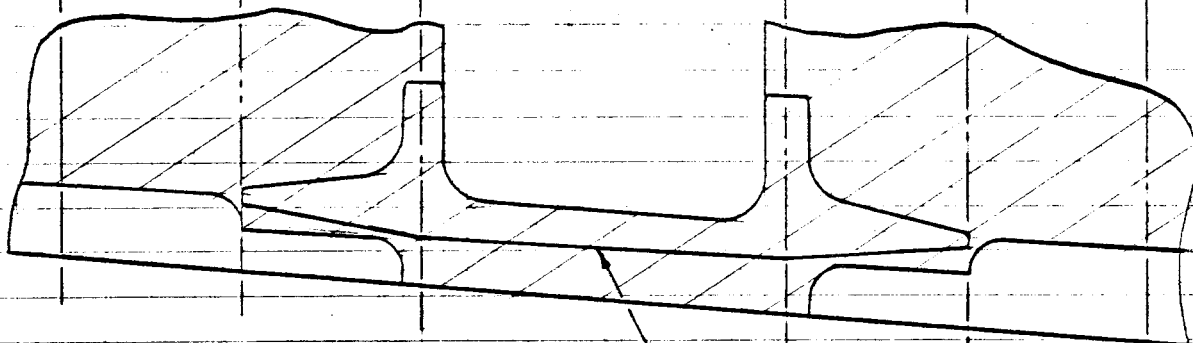
F.S.
1824.

F.S.
1844. (REF P.5.6)

F.S.
1864.

F.S.
1904.

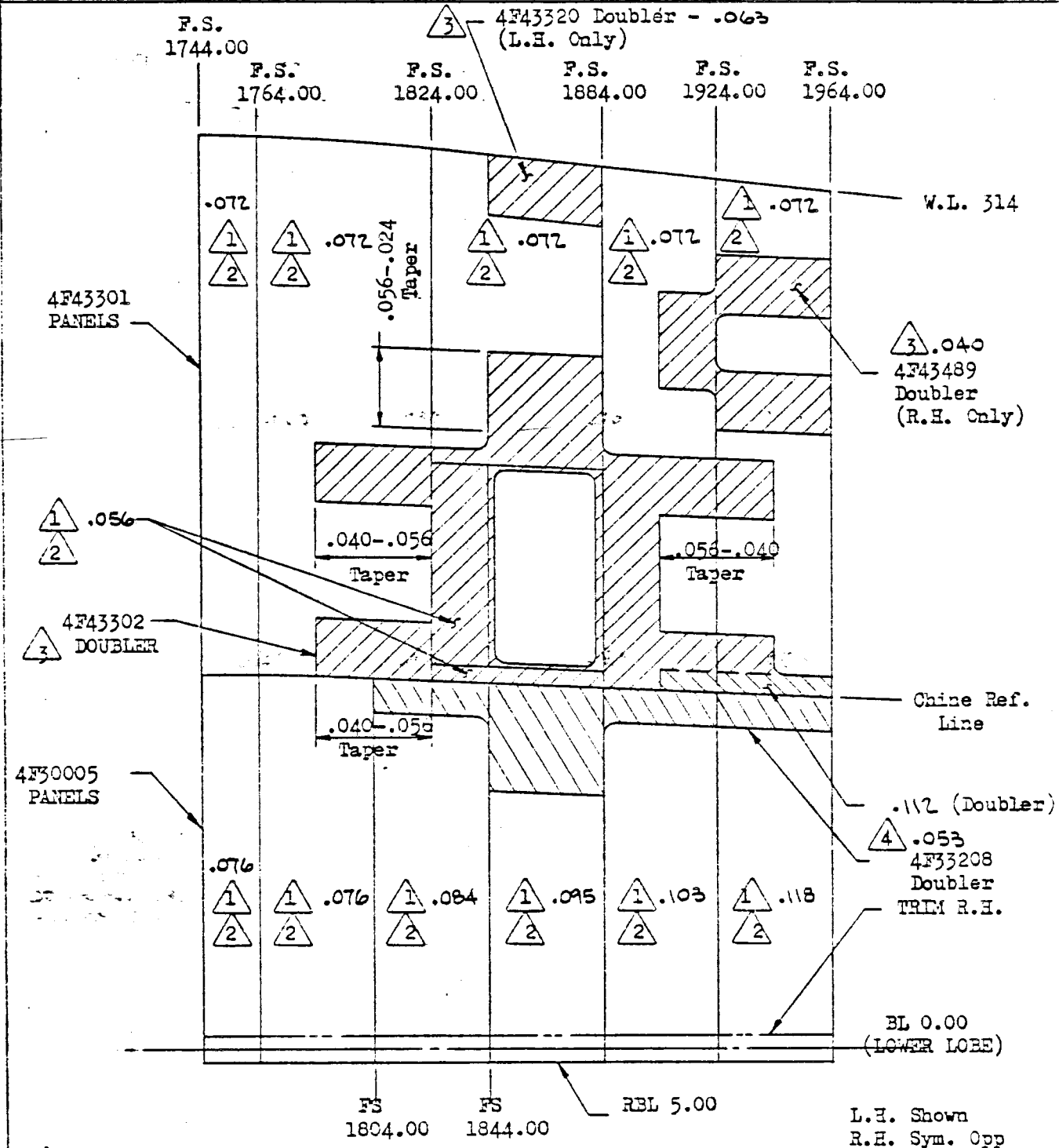
F.S.
1924.



NOTES:

- ① REF. DWG 4V41834 FOR L.H. SIDE,
(AFT TROOP COMP'T SERV. DOOR CUT-OUT). 7079-T6 SHEET $\triangle 2$
CLAD ONE SIDE
 $t = .100$ IN.
- ② REF. DWG 4V41835 FOR R.H. SIDE,
(AFT TROOP COMP'T EMERG. EXIT CUT-OUT).

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APPROVED	R.W. Smith	17 Nov 67		REPORT NO.	LGIUS 46-2-2	



- 4 7079-T6 Clad - Both Sides
- 3 7075-T6 Bare
- 2 7079-T6 Clad on Exterior Surface Only
- 1 Material Thickness - Stress Nominal

Scale: 1/50

PREPARED	NAME	PART	TITLE	PAGE	REV.
A. J. Pinchot	L. G. Darby	17 Nov 67	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	5,8	
CHECKED	R. W. Smith	17 Nov 67	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A
APPROVED				IGIUS 46-2-2	REPORT NO.

5.4.1 SHELL SKINS - Continued

STGR. 96
(Trim Line)

BL 0.00
(UPPER LOBE)

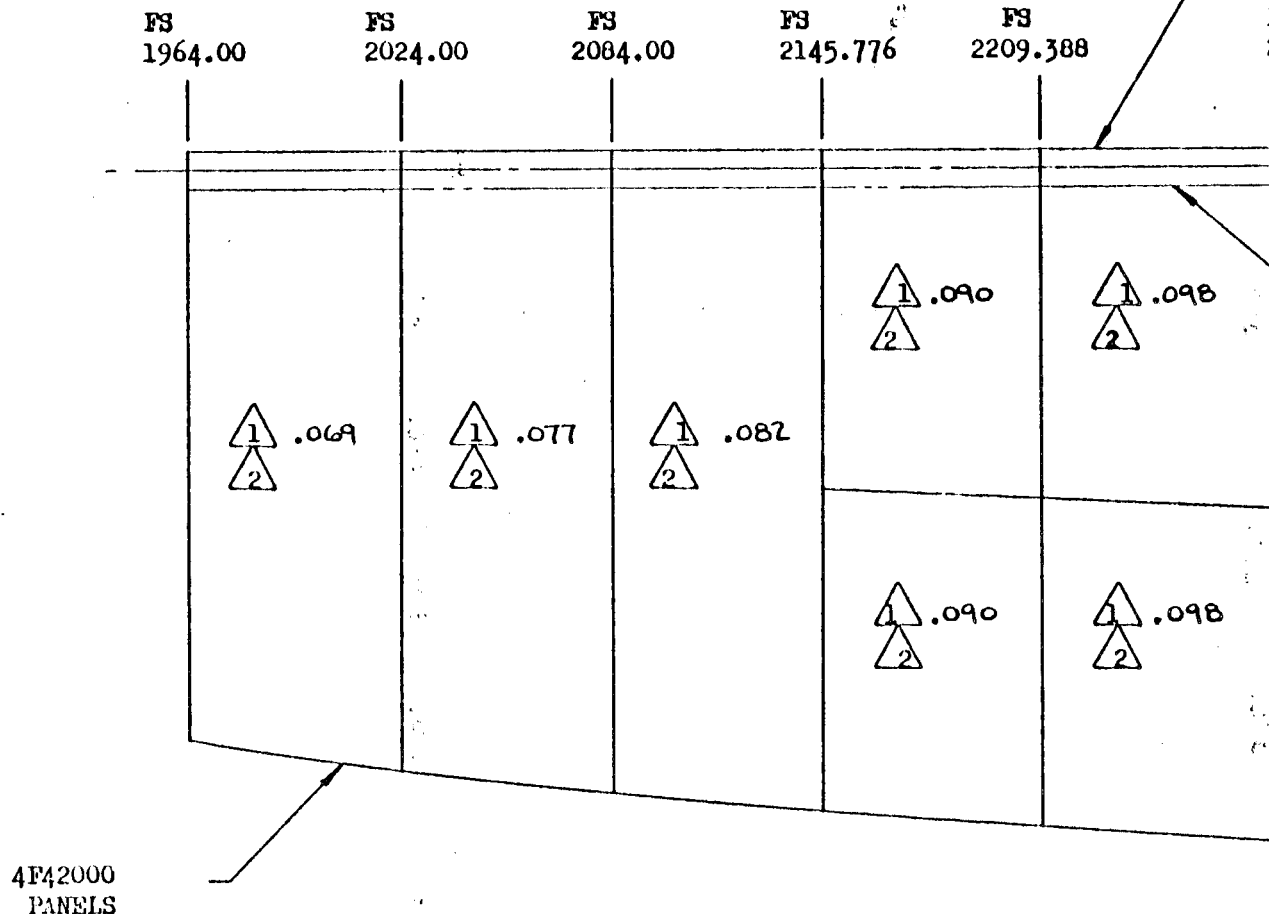
TRIM R.H.

STGR. 71

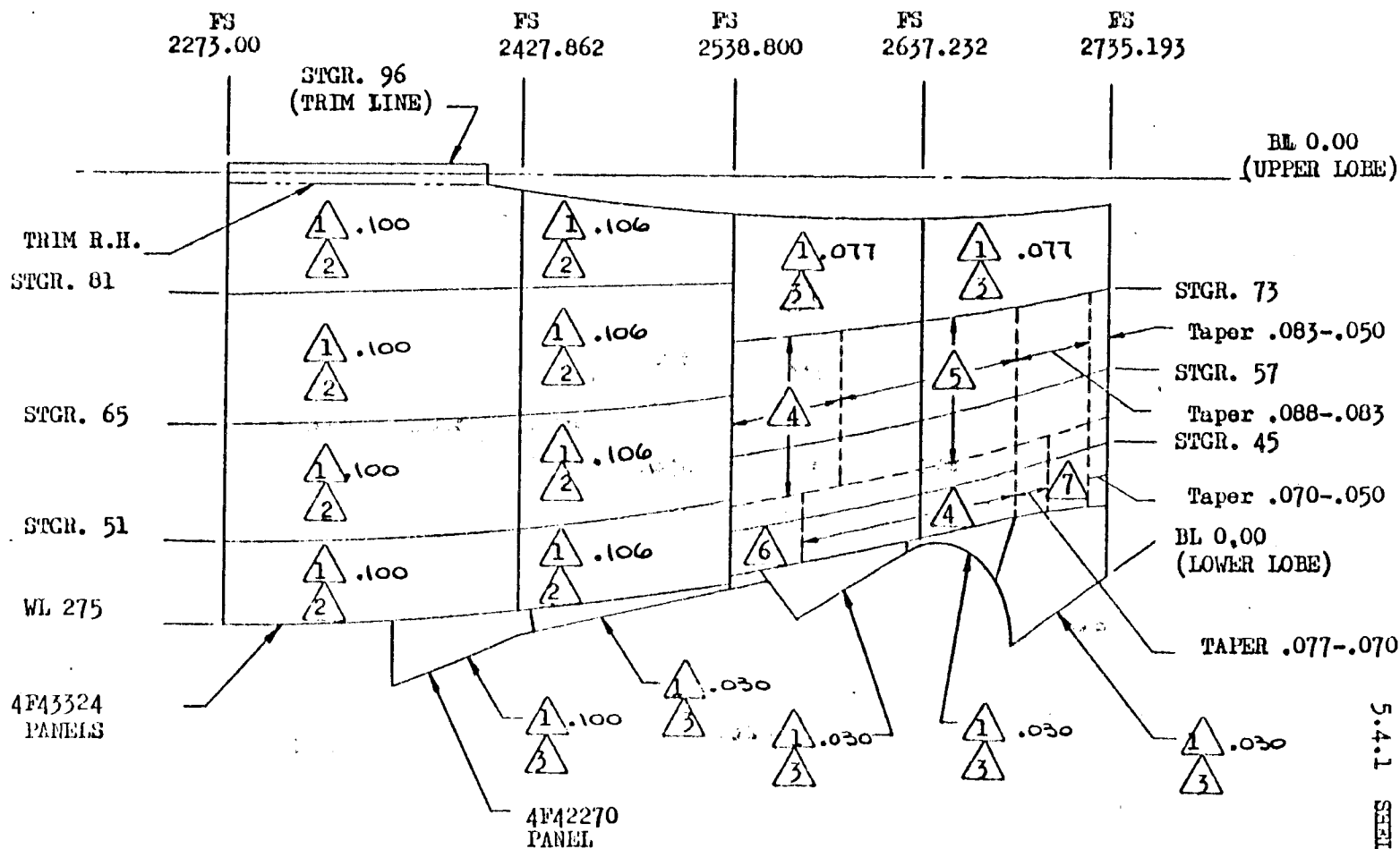
WL 314

L.H. Shown
R.H. Sym. Opp

Scale: 1/50



2 7079-T6 Clad on Exterior Surface Only
1 Material Thickness - Stress Nominal



- 7 .070 1 3
 6 .068 1 3
 5 .085 1 3
 4 .077 1 3
 3 7075-T6 Clad on Exterior Surface Only
 2 7079-T6 Clad on Exterior Surface Only
 1 Material Thickness - Stress Nominal

L.H. SHOWN
R.H. SYM. OPP

SCALE: 1/80

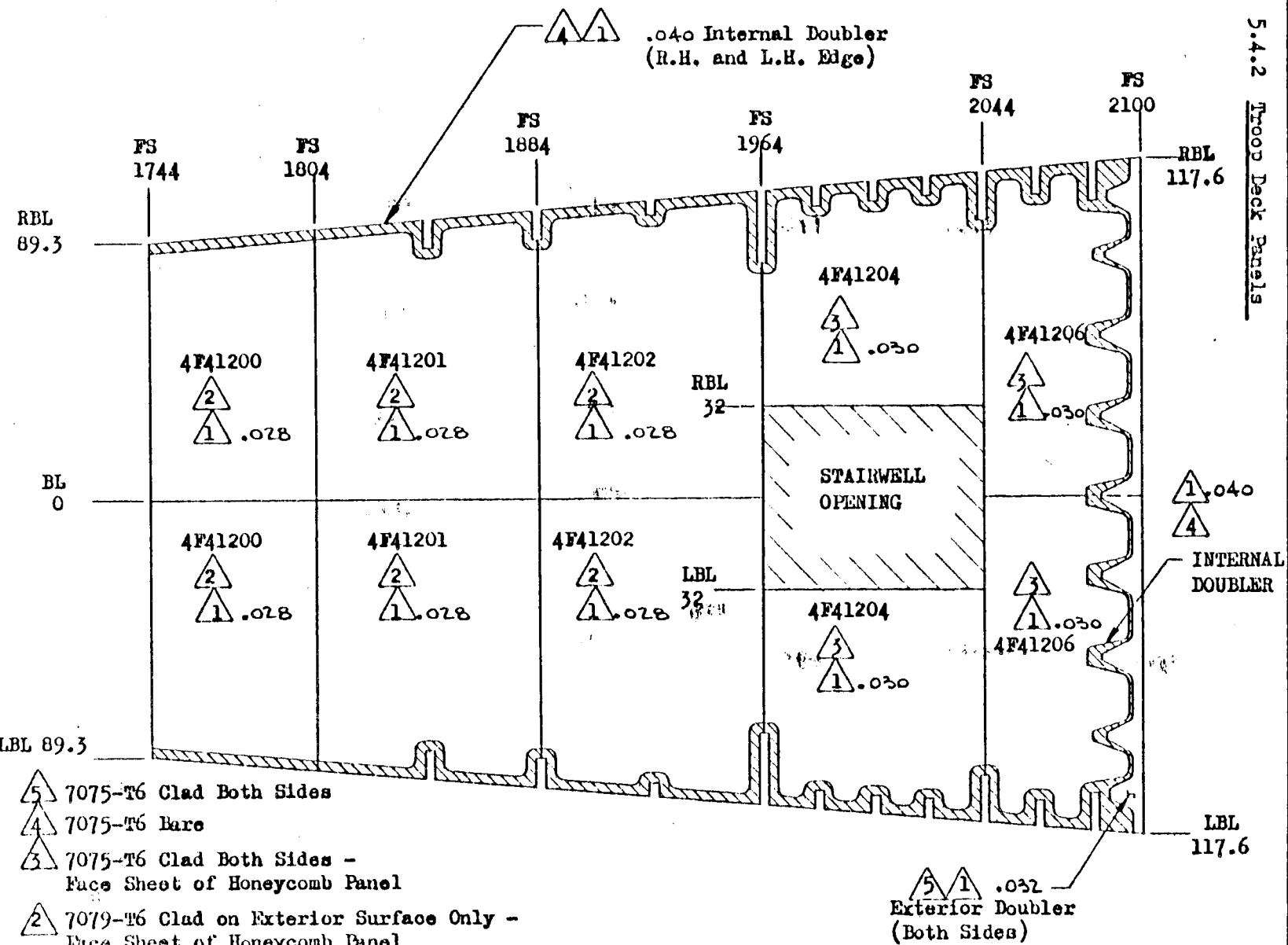
PREPARED	A. D. Bunchot	17 NOV 67	NAME
CHECKED	L. G. Derby	17 NOV 67	DATE
APPROVED	R. W. Smith	17 NOV 67	TITLE
			LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION
			INTERNAL LOADS ANALYSIS APT FUSELAGE
MODEL	C-51	PAGE	Vol. 1
REPORT NO.	10108 46-2-2	5.10	

5.4.1 SHELL SKINS - Continued

PREPARED	A.D. Pinchot	8 Dec 67
CHECKED	L.G. Darby	8 Dec 67
APPROVED	R.W. Smith	8 Dec 67

5.4.2 Troop Deck Panels

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TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	
PAGE	Vol. I 5.11
MODEL	C-5A
REPORT NO.	IGIDS 46-2-2

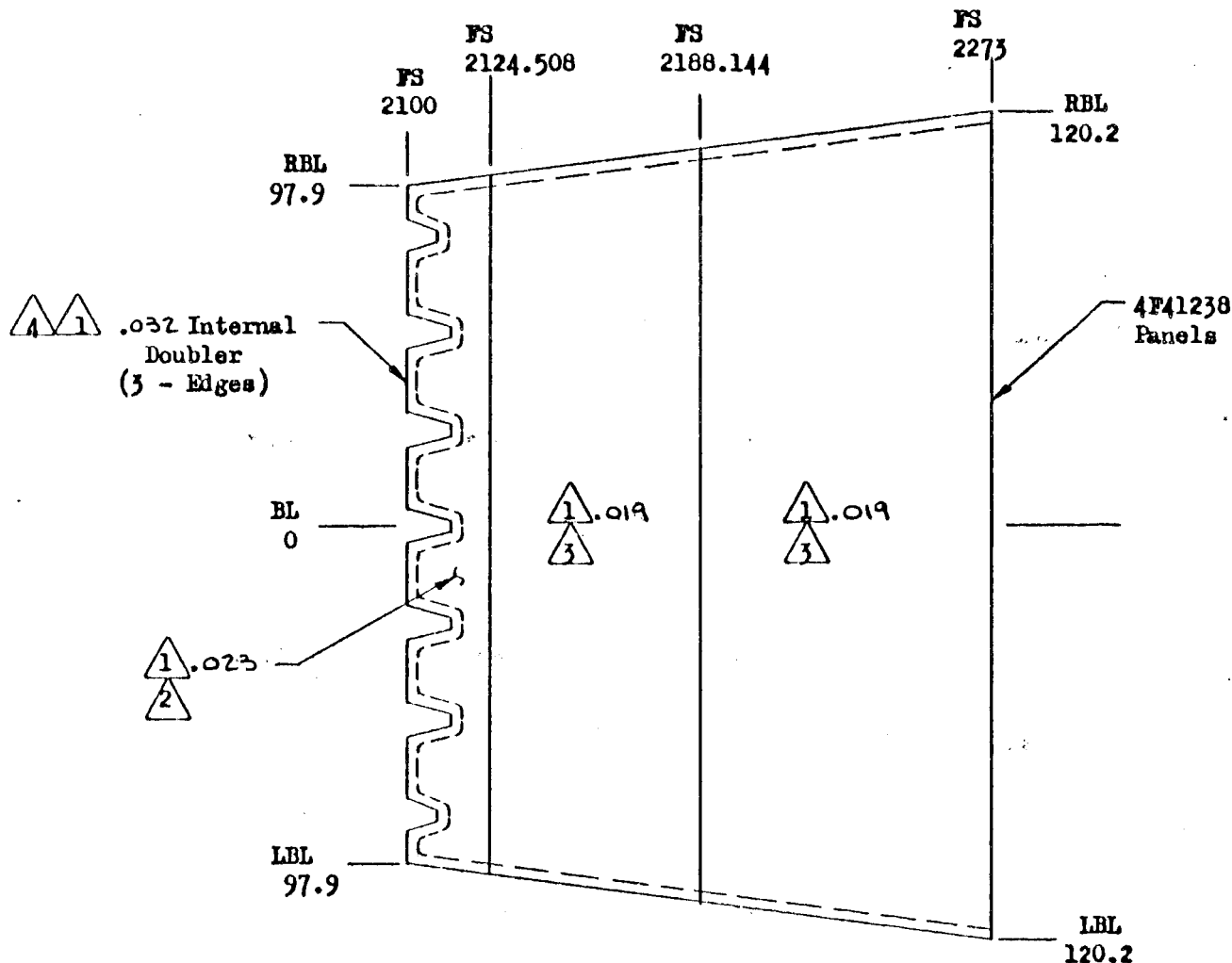


- 5 7075-T6 Clad Both Sides
- 4 7075-T6 Bare
- 3 7075-T6 Clad Both Sides -
Face Sheet of Honeycomb Panel
- 2 7079-T6 Clad on Exterior Surface Only -
Face Sheet of Honeycomb Panel
- 1 Material Thickness - Stress Nominal

1/50 Scale

PREPARED	NAME	DATE	TITLE	PAGE	FIG.	REV.
A.D. Pinchot		8 Dec 67	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Vol. I	5.12	
CHECKED	L.G. Darby	8 Dec 67	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	R.W. Smith	8 Dec 67		IGIUS 46-2-2		

5.4.3 SLOPING TORQUE DECK PANELS

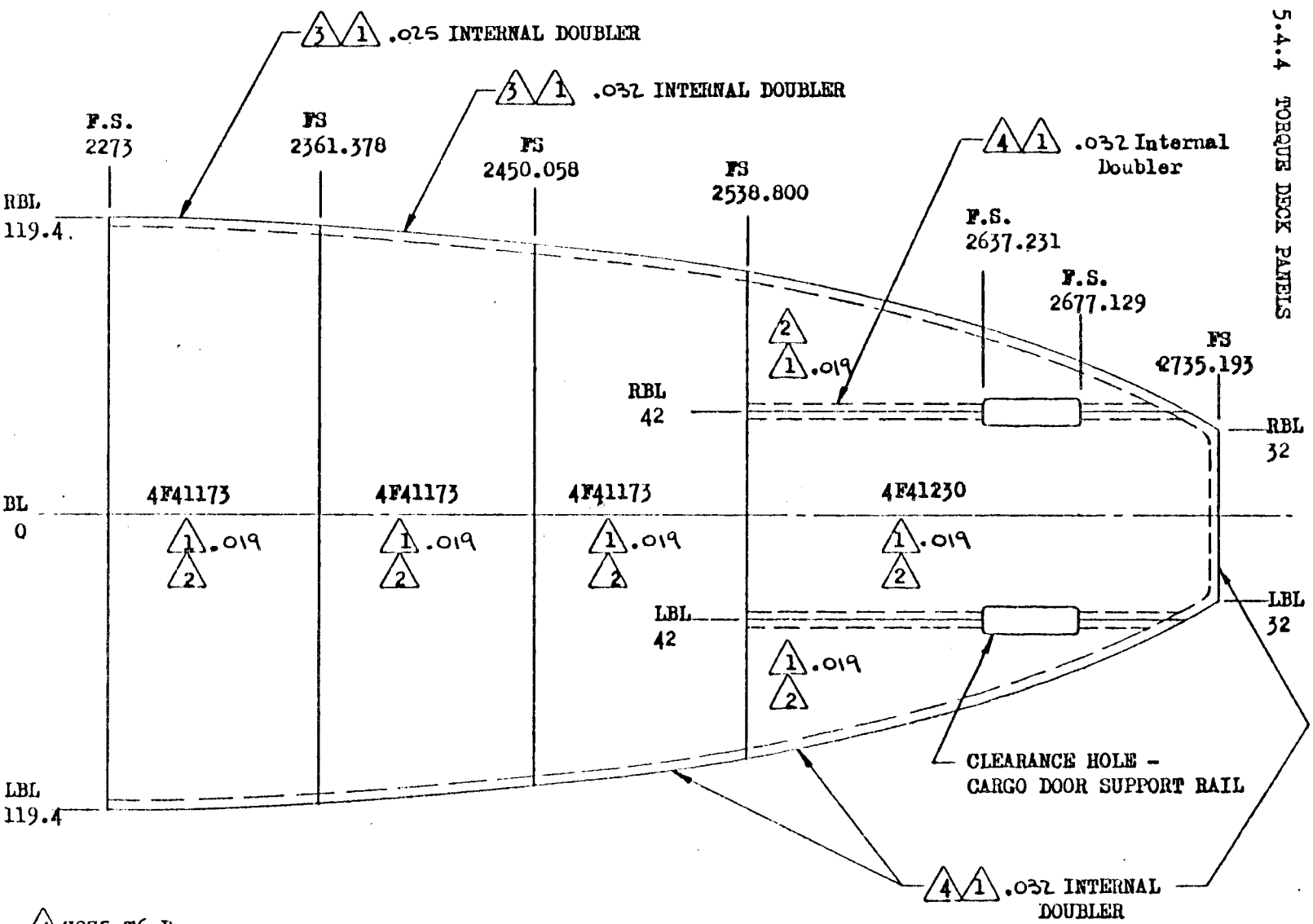


- 4 7075-T6 Bare
- 3 7079-T6 Bare - Face Sheet of Honeycomb Panel
- 2 7075-T6 Bare - Face Sheet of Honeycomb Panel
- 1 Material Thickness - Stress Nominal

1/50 Scale

PREPARED	A.D. Pinchot	8 Dec 67	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. 1 5.13
CHECKED	L.G. Darby	8 Dec 67	INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A
APPROVED	R.W. Smith	8 Dec 67		REPORT NO.	IGIUS 46-2-2

5.4.4 TORQUE DECK PANELS



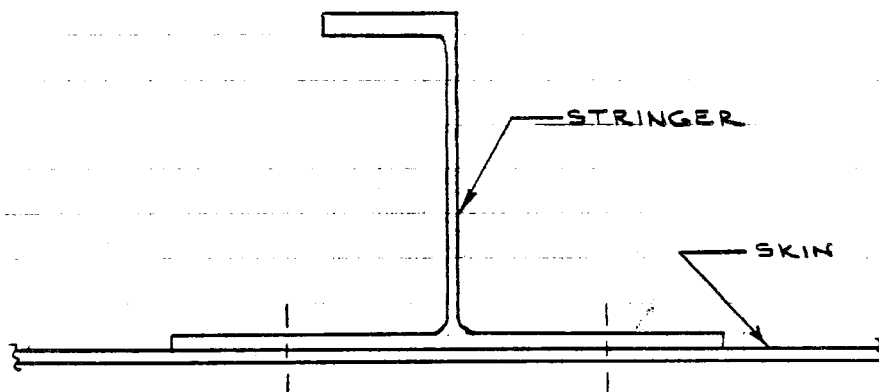
- $\triangle 4$ 7075-T6 Bare
- $\triangle 3$ 7075-T6 Clad Both Sides
- $\triangle 2$ 7079-T6 Bare - Face Sheet of Honeycomb Panel
- $\triangle 1$ Material Thickness - Stress Nominal

1/60 Scale

Prepared	NAME J. H. Ray	DATE 6 Sep 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. S. 1A
Checked	J. Halliday	6 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. Darby	6 Sep 68		Report No.	LGUS 46-2-2	

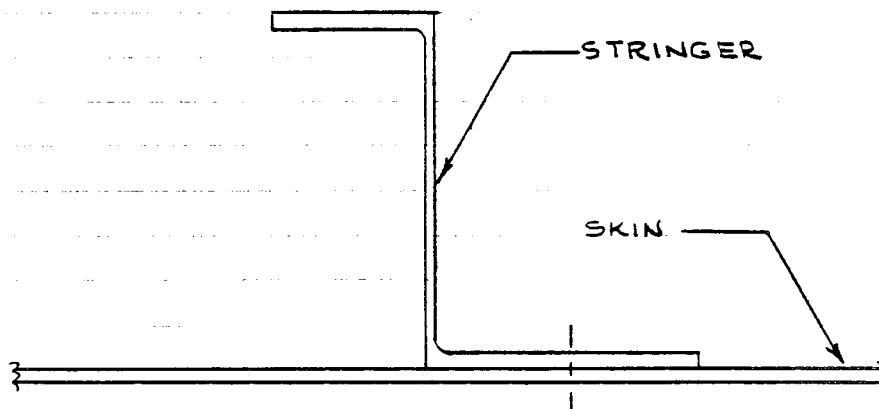
5.4.5 Typical Skin-Stringer Assemblies

"J" SECTION CONFIGURATION USED FOR ALL SKIN-STRINGER COMBINATIONS EXCEPT IN UNDERFLOOR AREA



TYPICAL "J" SECTION STRINGER

"Z" SECTION CONFIGURATION USED ONLY IN UNDERFLOOR AREA F.S. 1744 TO F.S. 1964.



TYPICAL "Z" SECTION STRINGER

PREPARED	J. H. Ray	17 Apr 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I	6.1
CHECKED	L. G. Darby	17 Apr 68	TITLE AFT FUSELAGE INTERNAL LOADS	MODEL	C-5A	
APPROVED	P. W. Horton	17 Apr 68		LGIUS 46-2-2 REPORT NO.		

6.0 SHELL STRUCTURE, F.S. 1744-F.S. 2101

6.1 DISCUSSION

The C-5A fuselage shell between F.S. 1744 and F.S. 2101 is a semi-monocoque structure consisting of a thin aluminum skin supported longitudinally by extruded aluminum alloy stringers spaced on 6 to 8 inch intervals around the circumference. With the exception of the longerons, troop deck and cargo floor, the entire bending, axial load, pressure and shear capability of the fuselage is vested in these skin-stringer elements.

The section between F.S. 1744 and F.S. 1964 is a three-cell torque box structure. One cell consists of the troop compartment shell and is bounded on the lower side by the troop deck at W.L. 314. The second cell consists of the side panel shell structure, the troop deck on the upper side and the cargo floor at W.L. 140. The third cell consists of the underfloor shell structure and is bounded on the upper side by the cargo floor.

The section between F.S. 1964 and F.S. 2101 consists of an upper single cell torque box structure and a lower open cell structure. The upper cell is defined by the troop compartment shell and the W.L. 314 troop deck. The lower cell is defined by the side panel shell structure and the W.L. 314 troop deck. The lower boundary of the side panels in this area consists of the sloping longeron and associated structure.

The troop deck at W.L. 314 from F.S. 1744 to F.S. 2101 consists of honeycomb panels supported by longitudinal and transverse beams. These panels have axial load and shear capability in addition to the normal deck load capability.

This section of the fuselage contains five cutouts. The personnel doors are located between F.S. 1844 and F.S. 1884, W. L. 140 and W.L. 215.9 on both the right and left side. The chine beam serves as the lower boundary member and a built up longeron forms the upper boundary member. The aft emergency exit hatch is located between F.S. 1844 and F.S. 1884, W. L. 334.4 and W.L. 370.5, on the right side only. Built up longerons serve as the upper and lower boundary members. The service door is located on the left side between F.S. 1844 and F.S. 1884, W. L. 327.8 and W. L. 383.1. Built up longerons form the upper and lower boundary members. A thrust recovery valve is located between F. S. 1924 and F. S. 1964, W. L. 252.5 and W. L. 275.2 on the right side. Built up longerons form the upper and lower boundary members. Aluminum alloy doublers are utilized to carry the redistributed loads around the cutout.

Frames are spaced at 20 inch intervals in this section of the fuselage. The door frames at F. S. 1844 and F.S. 1884 distribute the shear around the cutouts. F.S. 1964 frame is a shear redistribution frame and also reacts the sloping longeron kick load. This frame also contains the fuselage jacking points. Alternate frames beginning with F. S. 1764 are major redistribution frames and react the pressure and cargo loads. The intermediate frames between F. S. 1744 and F. S. 2101 are stability frames. F. S. 2004, 2044 and 2084 are major frames and react the pressure loads and redistribute the skin and inner panel shears.

PREPARED	NAME J. H. Ray	DATE 17 Apr 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	PERM. 6.2
CHECKED	L. G. Darby	17 Apr 68	TITLE AFT FUSELAGE INTERNAL LOADS	MODEL	C-5A	
APPROVED	P. W. Horton	17 Apr 68		REPORT NO.	LGIUS 46-2-2	

6.1 DISCUSSION (Continued)

The lower portions of F.S. 2004, 2024, 2044, 2064, and 2084 frames are canted and contain the cargo ramp latch points. These frames react the ramp latch loads and pressure loads on the skin and cargo ramp.

The W. L. 214 longeron extends throughout this section of the fuselage. It acts as an axial load carrying member and splices the troop compartment skin, the side panel skin, and the troop compartment floor.

PREPARED	J.H. Ray	5 Sept 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	VOL I	6.3
CHECKED	J. Halliday	5 Sept 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	L. G. Darby	5-Sept 68		REPORT NO.	LG1US46-2-2	

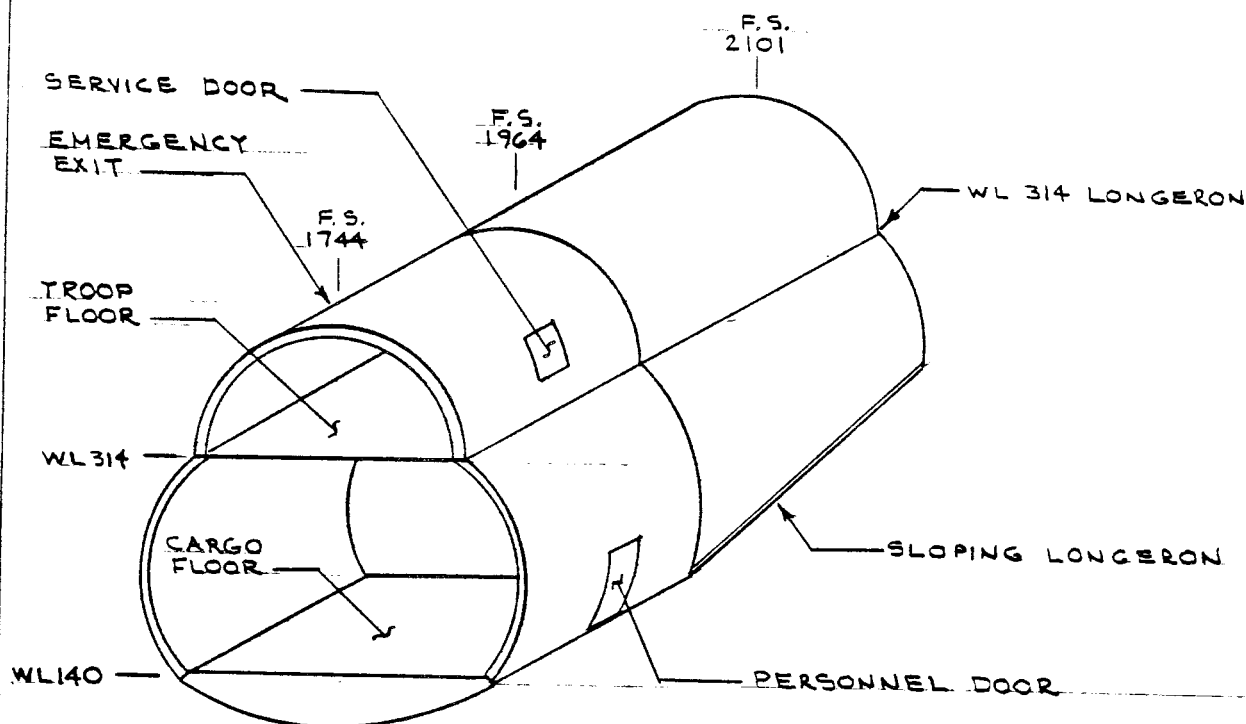
6.2 GEOMETRY AND LOADS

The section of the aft fuselage from F.S. 1744 to F.S. 2101 is basically a semi-monocoque shell structure made up of 7079-T6 aluminum alloy skins with 7075-T6 aluminum alloy longitudinal stringers and longerons. Frames are spaced at 20 inch intervals to provide shell stability and react other loads such as shell loads redistribution and door support loads.

The geometry of the individual elements of the structure is discussed more fully in the respective sections dealing with that particular element. The internal loads for these elements will also be presented in the respective sections.

The basic loads applied to the fuselage are presented in Ref. 17. Moments and shears are applied to the fuselage using the sign convention of Ref. 17, pg. 3.1-001. The various critical load cases will be discussed in detail in each section.

Figure 6.2.1 describes the basic geometry of this section of the fuselage.



SHELL STRUCTURE ~ F.S. 1744 - F.S. 2101

FIGURE 6.2.1

PREPARED	NAME V. Boyd	DATE 5 Sept 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
CHECKED	J. H. Ray	5 Sept 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Vol. I	6.4	
APPROVED	L. G. Darby	5 Sept 68		MODEL C-5A LG1US46-2-2 REPORT NO.		

Rev. (A) JPK 8-1-69

Rev. (B) ADP 1-8-71

6.3 SKINS AND STRINGERS

The internal loads for the skins and stringers are calculated using the aft fuselage FAMAS model as described in Reference 8, Vol. IV. External loads from Reference 10 are applied to the FAMAS model as discrete grid point loads along the entire length of the model using a computerized loading program as described in Reference 8, Vol. I, Section 5.

The aft fuselage FAMAS model is subdivided into modules (or Mods) to simplify calculations. This section gives internal loads for skins and stringers from fuselage station 1744 to fuselage station 2101, which is represented by Mods 1, 2, 3, and a portion of Mod 4 in the FAMAS model. To account for fuselage internal pressurization, a FAMAS pressure model is used to determine skin and stringer loads, Reference 8, Vol. IV. All flight loads are analyzed with and without 13.05 psi ultimate pressure loads, Reference 10, page 9.7-116.

The internal loads presented here are for those conditions with Margins-of-Safety less than 20% that appear in the scan of the Margins-of-Safety (Ref. 15, Section 6.3). These designing conditions are:

5PMO27	Ref. 10, page 9.2-346
TPMO96 + 13.05 psi	Ref. 10, page 9.1-669
ALS11I	Ref. 10, page 9.6-146
DML001	Ref. 10, page 9.3-501
TPMO96	Ref. 10, page 9.1-669
5PE201 + 13.05 psi	Ref. 10, page 9.2-730
5PE201	Ref. 10, page 9.2-730
1PC242	Ref. 10, page 9.2-685
DMLOOR	



Internal loads shown on succeeding pages have been extracted directly from the FAMAS analysis and are loads in FAMAS stringers and FAMAS panels. FAMAS stringer loads are in Section 6.3.1 and FAMAS panel shears are in Section 6.3.2.

All loads are in pounds and shears are in pounds per inch.

A write-up is presented in Reference 15, Section 6.3 explaining the method used to calculate actual stringer loads from FAMAS stringer loads and actual skin shears from FAMAS panel shears.

Shear lag around shell cutouts have been accounted for by eliminating the stringers and panels at these cutouts from the FAMAS model. See pages 6.21, 6.22 and 6.25.

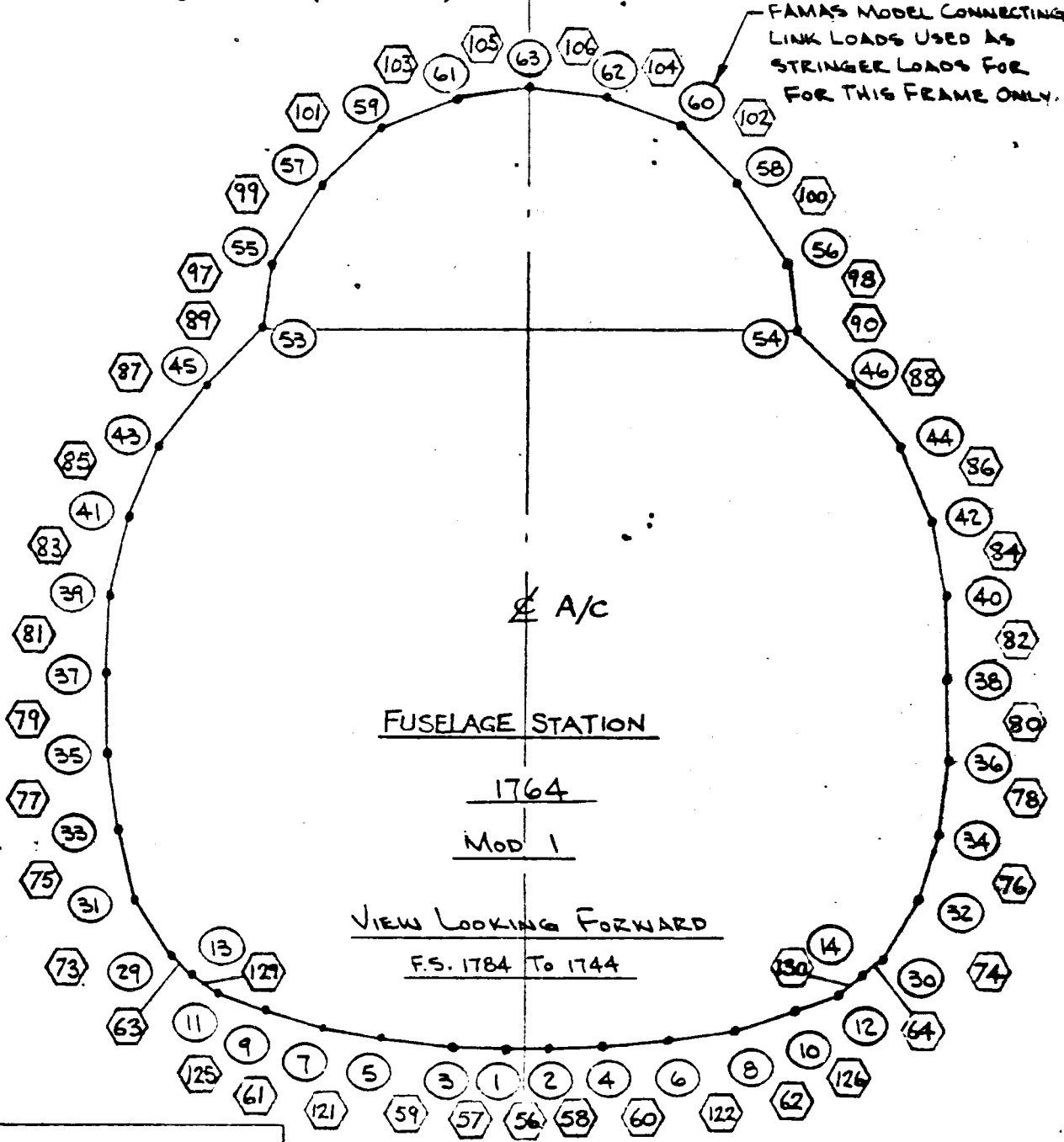


NOTE:

DMLOOR is identical to DML001 except the fuselage is bent laterally in the opposite direction.

REV: A M.B. 1 AUG. 69

6.3.1 Stringer Loads (Continued)



KEY

 GRID POINT

 FAMAS - MODEL CONNECTING LINK

 FAMAS PANEL

FIGURE No 6.3.1.1

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 6.6
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C=SA		
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No. LGIUS46=2=2		

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.1 LINK LOADS (CONDITION SPM027)MOD 1 (REF. P.6.5)

1	*	1	-.389031+05	2	-.388875+05	3	-.391167+05	4	-.390551+05
		5	-.326045+05	6	-.324713+05	7	-.146312+05	8	-.145570+05
		9	-.149568+05	10	-.148757+05	11	-.102599+05	12	-.102052+05
		13	-.693993+05	14	-.689690+05	15	-.217284+05	16	-.217618+05
		17	-.226545+05	18	-.227310+05	19	-.442089+05	20	-.443418+05
		21	-.429231+05	22	-.430453+05	23	-.267456+05	24	-.268342+05
		25	-.211190+05	26	-.211836+05	27	-.333952+05	28	-.333974+05
		29	-.134230+04	30	-.127917+04	31	-.150218+05	32	-.150085+05
		33	-.114134+05	34	-.114071+05	35	-.159471+05	36	-.160292+05
		37	-.160978+05	38	-.167755+05	39	-.149312+05	40	-.162051+05
		41	-.375807+04	42	-.650068+04	43	.110524+05	44	.735642+04
		45	.259837+05	46	.219412+05	47	.376827+05	48	.358578+05
		49	.215387+05	50	.188950+05	51	.114905+05	52	.112034+05
		53	.589974+05	54	.556240+05	55	.416558+05	56	.536765+05
		57	.541802+05	58	.730516+05	59	.880866+05	60	.835655+05
		61	.976299+05	62	.924149+05	63	.996999+05	64	-.238566+04
		65	.195983+04	66	-.122404+05	67	.117823+05	68	-.133710+05
		69	.129431+05	70	-.692156+04	71	.670608+04	72	-.628022+04
		73	.608008+04	74	-.507258+04	75	.489965+04	76	-.765860+04
		77	.762007+04	78	-.133138+04	79	.149690+04	80	-.758229+04
		81	.778109+04	82	-.908399+04	83	.927793+04	84	-.653131+04
		85	.664954+04	86	-.588047+04	87	.599988+04	88	-.384362+04
		89	.393490+04	90	-.434843+04	91	.444247+04	92	-.640472+04
		93	.636587+04	94	-.569653+04	95	.565648+04	96	-.449604+04
		97	.447178+04	98	-.226133+04	99	.225205+04	100	.558512+03
		101	-.574769+03	102	.356704+04	103	-.370211+04	104	.600520+04
		105	-.635881+04	106	.838042+04	107	-.913597+04	108	.123808+05
		109	-.132719+05	110	.266675+04	111	.930835+03	112	.319977+04
		113	-.487281+03	114	.225390+04	115	-.942188+03	116	.128710+05
		117	-.123574+05	118	.667493+04	119	-.642438+04	120	.368624+04
		121	-.752522+04	122	.200049+04	123	-.509542+04	124	.286189+04
		125	-.198003+04	126	.887730+03	127	-.255097+04	128	-.253172+04
		129	-.549989+04	130	-.537574+04	131	-.602672+04	132	-.586475+04
		133	-.443524+04	134	-.430860+04	135	-.465719+04	136	-.451822+04
		137	-.420006+04	138	-.405771+04	139	-.607905+04	140	-.593035+04
		141	-.768809-09	142	-.770987-09	143	-.855826-09	144	-.858308-09
		145	-.231245-08	146	-.231889-08	147	-.257640-08	148	-.257933-08
		149	-.182273-08	150	-.182179-08	151	-.142537-08	152	-.141657-08
		153	.227508+04	154	.232050+04	155	-.136813+05	156	-.135924+05
		157	-.182821+05	158	-.181587+05	159	-.217220+05	160	-.216084+05
		161	-.342967+05	162	-.342359+05	163	-.306832+05	164	-.308173+05
		165	-.176931+05	166	-.183080+05	167	-.175685+05	168	-.186754+05
		169	-.150620+05	170	-.166253+05	171	-.144267+05	172	-.158841+05
		173	-.290585-08	174	-.288714-08	175	-.162763-08	176	-.175868-08
		177	-.117836-08	178	-.103879-08	179	-.181885+05	180	-.141449+05
		181	-.150247+05	182	-.101365+05	183	-.256914+04	184	-.548240+04
		185	.260150+04	186	-.990058+03	187	.174909+04	188	.175005+04
		189	.196248+04						

* LINK (1) REF. P. 6.5 ULTIMATE COMPRESSION LOAD = -38,903. LB.

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Page	Vol. I 6.7
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Model C=5A	LGIUS46=2=2
					Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.1 LINK LOADS (CONDITION TPM096+P)MOD 1 (REF. P.6.5)

1	1	-.242388+05	2	-.242264+05	3	-.259336+05	4	-.258781+05
	5	-.234467+05	6	-.233192+05	7	-.107204+05	8	-.106457+05
	9	-.104001+05	10	-.103199+05	11	-.761828+04	12	-.756340+04
	13	-.585983+05	14	-.581676+05	15	-.583677+04	16	-.590516+04
	17	-.769138+04	18	-.783598+04	19	-.248198+05	20	-.251631+05
	21	-.273459+05	22	-.276752+05	23	-.193013+05	24	-.195179+05
	25	-.161511+05	26	-.163114+05	27	-.188833+05	28	-.190641+05
	29	-.142070+04	30	-.149910+04	31	-.104188+05	32	-.105007+05
	33	-.769126+04	34	-.780621+04	35	-.116316+05	36	-.119911+05
	37	-.808232+04	38	-.944932+04	39	-.132593+04	40	-.349208+04
	41	.148295+05	42	.107337+05	43	.320926+05	44	.264315+05
	45	.512707+05	46	.454447+05	47	.568687+05	48	.548222+05
	49	.462688+05	50	.429063+05	51	.129397+05	52	.136251+05
	53	.103839+06	54	.100189+06	55	.556419+05	56	.767834+05
	57	.692056+05	58	.976036+05	59	.119359+06	60	.112626+06
	61	.136557+06	62	.127601+06	63	.139443+06	64	-.213659+04
	65	.166791+04	66	-.117552+05	67	.112477+05	68	-.129688+05
	69	.124942+05	70	-.641442+04	71	.617436+04	72	-.555143+04
	73	.532824+04	74	-.326401+04	75	.307022+04	76	-.444238+04
	77	.442172+04	78	-.171222+04	79	.172161+04	80	-.938958+04
	81	.943661+04	82	-.111172+05	83	.111909+05	84	-.773034+04
	85	.782511+04	86	-.694558+04	87	.706204+04	88	-.452781+04
	89	.462571+04	90	-.358676+04	91	.370494+04	92	-.297696+04
	93	.294860+04	94	-.103714+04	95	.100893+04	96	.208609+03
	97	-.215788+03	98	.413440+04	99	-.413177+04	100	.667190+04
	101	-.669781+04	102	.827642+04	103	-.847368+04	104	.107600+05
	105	-.113892+05	106	.120857+05	107	-.132943+05	108	.168371+05
	109	-.177586+05	110	.356153+04	111	.189552+04	112	.366560+04
	113	-.151278+03	114	.234440+04	115	-.917146+03	116	.210384+05
	117	-.199387+05	118	.131868+05	119	-.128769+05	120	.729500+04
	121	-.133031+05	122	.541047+04	123	-.100186+05	124	.647990+04
	125	-.464109+04	126	.154226+04	127	-.910728+04	128	-.908701+04
	129	-.135192+05	130	-.133824+05	131	-.138030+05	132	-.136232+05
	133	-.839249+04	134	-.825162+04	135	-.796361+04	136	-.780895+04
	137	-.712182+04	138	-.696196+04	139	-.917755+04	140	-.901583+04
	141	.771375-09	142	.768162-09	143	.559176-09	144	.554731-09
	145	-.732047-09	146	-.743583-09	147	-.143430-08	148	-.144123-08
	149	-.134660-08	150	-.134664-08	151	-.112761-08	152	-.111867-08
1	153	.142357+04	154	.148549+04	155	-.139383+05	156	-.138704+05
	157	-.169393+05	158	-.168583+05	159	-.188540+05	160	-.188288+05
	161	-.293784+05	162	-.295682+05	163	-.269536+05	164	-.274027+05
	165	-.152753+05	166	-.162126+05	167	-.126520+05	168	-.145162+05
	169	-.821763+04	170	-.107599+05	171	-.747953+04	172	-.917019+04
	173	-.494435-08	174	-.488806-08	175	-.335908-08	176	-.339456-08
	177	-.257564-08	178	-.214428-08	179	-.140164+05	180	-.699396+04
	181	-.115525+05	182	-.361273+04	183	.474200+04	184	.112284+03
	185	.110283+05	186	.553179+04	187	.104102+05	188	.105938+05
	189	.113628+05						

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
					Page	Vol. I 6.8
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C=5A	
Approved	L G DARBY	1 JUN 69			Report No. LGIUS46=2=2	

REV. (2) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
6.3 SKINS AND STRINGERS
6.3.1 LINK LOADS (CONDITION ALSIII) MOD 1 (REF. P. 6.5)

1	-.393219+05	2	-.396651+05	3	-.393666+05	4	-.400411+05
5	-.325052+05	6	-.339044+05	7	-.145350+05	8	-.152914+05
9	-.147465+05	10	-.158253+05	11	-.100228+05	12	-.110370+05
13	-.671153+05	14	-.749465+05	15	-.220559+05	16	-.228017+05
17	-.226413+05	18	-.241013+05	19	-.429673+05	20	-.479519+05
21	-.414997+05	22	-.467601+05	23	-.257755+05	24	-.292920+05
25	-.203633+05	26	-.231679+05	27	-.323258+05	28	-.367941+05
29	-.115770+04	30	-.173094+04	31	-.142810+05	32	-.167841+05
33	-.105416+05	34	-.131554+05	35	-.143686+05	36	-.188550+05
37	-.135709+05	38	-.209275+05	39	-.117929+05	40	-.214062+05
41	.118490+03	42	-.136075+05	43	.146645+05	44	-.629970+02
45	.289531+05	46	.143450+05	47	.365616+05	48	.314824+05
49	.215701+05	50	.159496+05	51	.118325+05	52	.927743+04
53	.615531+05	54	.448609+05	55	.424704+05	56	.468028+05
57	.541367+05	58	.661221+05	59	.868184+05	60	.771427+05
61	.949563+05	62	.870941+05	63	.956057+05	64	-.449547+04
65	-.264446+03	66	-.140855+05	67	.923159+04	68	-.148452+05
69	.106195+05	70	-.741952+04	71	.565131+04	72	-.665594+04
73	.513471+04	74	-.532321+04	75	.408040+04	76	-.777990+04
77	.713652+04	78	-.249686+04	79	.326976+03	80	-.872682+04
81	.660294+04	82	-.100191+05	83	.831219+04	84	-.678082+04
85	.641046+04	86	-.599560+04	87	.591351+04	88	-.387382+04
89	.394506+04	90	-.438053+04	91	.457117+04	92	-.632143+04
93	.607829+04	94	-.559798+04	95	.541085+04	96	-.439708+04
97	.435840+04	98	-.220352+04	99	.221333+04	100	.555122+03
101	-.538620+03	102	.347601+04	103	-.362432+04	104	.580712+04
105	-.628539+04	106	.799096+04	107	-.921644+04	108	.115294+05
109	-.137903+05	110	-.216844+04	111	-.392997+04	112	.113753+03
113	-.375668+04	114	.419239+03	115	-.291350+04	116	.121036+05
117	-.125326+05	118	.668282+04	119	-.606118+04	120	.349798+04
121	-.750782+04	122	.143093+04	123	-.580198+04	124	.170455+04
125	-.339411+04	126	-.636517+03	127	-.266176+04	128	-.247448+04
129	-.598318+04	130	-.475278+04	131	-.654669+04	132	-.506615+04
133	-.471732+04	134	-.372283+04	135	-.490613+04	136	-.389575+04
137	-.440289+04	138	-.338688+04	139	-.606853+04	140	-.531798+04
141	-.808138-09	142	-.843247-09	143	-.880763-09	144	-.948617-09
145	-.228790-08	146	-.252248-08	147	-.253887-08	148	-.278603-08
149	-.179424-08	150	-.196107-08	151	-.140297-08	152	-.152395-08
153	.229776+04	154	.241178+04	155	-.134950+05	156	-.129650+05
157	-.179501+05	158	-.174449+05	159	-.212189+05	160	-.211319+05
161	-.332243+05	162	-.340388+05	163	-.296963+05	164	-.306411+05
165	-.171360+05	166	-.181356+05	167	-.168173+05	168	-.188079+05
169	-.141471+05	170	-.171628+05	171	-.131970+05	172	-.169404+05
173	-.274138-08	174	-.254695-08	175	-.157565-08	176	-.146922-08
177	-.117039-08	178	-.813624-09	179	-.175369+05	180	-.146234+05
181	-.149866+05	182	-.101424+05	183	-.231778+04	184	-.597243+04
185	.295376+04	186	-.174963+04	187	.205409+04	188	.108729+04
189	.185590+04						

LOADS ARE IN POUNDS

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.9
Checked	A L PINCHOT	1 JUN 69	TITLE	Model C=5A	
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	LGIUS46=2=2	
			AFT FUSELAGE	Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.1 LINK LOADS (CONDITION DML001)MOD 1 (REF. P.6.5)

1	1	-.641581+04	2	-.170360+05	3	-.118623+04	4	-.223943+05
	5	.897337+04	6	-.287544+05	7	.555175+04	8	-.144404+05
	9	.751226+04	10	-.165853+05	11	.617432+04	12	-.124047+05
	13	.491836+05	14	-.908541+05	15	-.111542+04	16	-.111721+05
	17	.356088+04	18	-.164342+05	19	.224522+05	20	-.480220+05
	21	.250840+05	22	-.500732+05	23	.175007+05	24	-.332172+05
	25	.143028+05	26	-.268300+05	27	.244351+05	28	-.443181+05
	29	.355762+04	30	-.438955+04	31	.130102+05	32	-.217060+05
	33	.127379+05	34	-.190270+05	35	.203055+05	36	-.285918+05
	37	.298897+05	38	-.373860+05	39	.373531+05	40	-.440832+05
	41	.500235+05	42	-.506814+05	43	.525283+05	44	-.444206+05
	45	.511512+05	46	-.365009+05	47	.237577+05	48	.496559+03
	49	.178059+05	50	-.555508+04	51	.123844+05	52	-.616753+04
	53	.667541+05	54	-.355321+05	55	.374302+05	56	-.169791+05
	57	.420617+05	58	-.729241+04	59	.600161+05	60	.120615+04
	61	.497244+05	62	.157355+05	63	.335816+05	64	.834116+04
	65	.951672+04	66	.663662+04	67	.130647+05	68	.526469+04
	69	.122294+05	70	.190359+04	71	.540126+04	72	.155723+04
	73	.467155+04	74	.100375+04	75	.340970+04	76	-.173568+03
	77	.375510+04	78	-.966851+04	79	-.883928+04	80	-.109325+05
	81	-.644168+04	82	-.945978+04	83	-.411575+04	84	-.321050+04
	85	.580842+03	86	-.196370+04	87	.144309+04	88	-.833960+03
	89	.140339+04	90	-.385331+03	91	.215705+04	92	.161245+03
	93	.336568+04	94	.272172+03	95	.308374+04	96	.573030+03
	97	.283069+04	98	.402209+03	99	.154343+04	100	.299587+02
	101	-.265934+03	102	-.668841+03	103	-.258998+04	104	-.168072+04
	105	-.473398+04	106	-.313241+04	107	-.761194+04	108	-.516202+04
	109	-.127274+05	110	-.880794+04	111	-.897915+04	112	-.650236+04
	113	-.634996+04	114	-.395815+04	115	-.384654+04	116	-.103914+04
	117	-.872102+04	118	-.142038+04	119	-.476474+04	120	-.886283+04
	121	-.894173+04	122	-.125323+05	123	-.132461+05	124	-.145604+05
	125	-.180108+05	126	-.181124+05	127	-.503805+02	128	-.146365+04
	129	.161915+04	130	-.466433+04	131	.217403+04	132	-.539736+04
	133	.125954+04	134	-.354473+04	135	.124407+04	136	-.359142+04
	137	.835563+03	138	-.282888+04	139	.110039+04	140	-.398770+04
	141	.328663-10	142	-.446075-09	143	.246807-09	144	-.719619-09
	145	.128690-08	146	-.263743-08	147	.147498-08	148	-.299838-08
	149	.105300-08	150	-.213898-08	151	.848396-09	152	-.169650-08
1	153	.114536+03	154	.121548+04	155	.360859+03	156	-.717618+04
	157	.120368+04	158	-.102664+05	159	.317460+04	160	-.141012+05
	161	.857327+04	162	-.258967+05	163	.800105+04	164	-.235814+05
	165	.462999+04	166	-.140201+05	167	.716161+04	168	-.161695+05
	169	.876906+04	170	-.165689+05	171	.957028+04	172	-.183431+05
	173	-.186695-08	174	-.102200-09	175	-.196742-08	176	.604944-09
	177	-.195188-08	178	.845884-09	179	.683646+04	180	-.182311+05
	181	.867746+04	182	-.170112+05	183	.153368+05	184	-.140141+05
	185	.133819+05	186	-.102910+05	187	.660949+04	188	-.614118+04
	189	.521726+03						

LOADS ARE IN POUNDS

NAME		DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.9.1
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS	C=5A	
Approved	H H WADE	NOV70	AFT FUSELAGE		Model	LGIUS46=2=2
			Report No.			

REV. (B) ADP 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101

6.3 SKINS AND STRINGERS

6.3.1 LINK LOADS (CONDITION DMLOOR) MOD 1 (REF. P.6.5)

1	1	-.167146+05	2	-.608932+04	3	-.220934+05	4	-.858111+03
	5	-.285617+05	6	.922890+04	7	-.143623+05	8	.566403+04
	9	-.165537+05	10	.758020+04	11	-.124003+05	12	.620024+04
	13	-.909095+05	14	.493145+05	15	-.113363+05	16	-.130771+04
	17	-.165895+05	18	.334380+04	19	-.482915+05	20	.220432+05
	21	-.503418+05	22	.246799+05	23	-.333772+05	24	.172484+05
	25	-.269194+05	26	.141443+05	27	-.447074+05	28	.239753+05
	29	-.449561+04	30	.347145+04	31	-.219690+05	32	.127209+05
	33	-.194542+05	34	.122676+05	35	-.294510+05	36	.193092+05
	37	-.386484+05	38	.280816+05	39	-.450987+05	40	.354257+05
	41	-.507723+05	42	.481663+05	43	-.443824+05	44	.502983+05
	45	-.341639+05	46	.512413+05	47	.833320+02	48	.222801+05
	49	-.422806+04	50	.176577+05	51	-.514859+04	52	.133484+05
	53	-.295895+05	54	.712933+05	55	-.114818+05	56	.510615+05
	57	-.793309+04	58	.532200+05	59	-.401644+04	60	.513693+05
	61	.120585+05	62	.423825+05	63	.291889+05	64	-.964574+04
	65	-.843295+04	66	-.132963+05	67	-.664226+04	68	-.124802+05
	69	-.523616+04	70	-.554876+04	71	-.186831+04	72	-.481376+04
	73	-.151815+04	74	-.354486+04	75	-.956623+03	76	-.387887+04
	77	.282224+03	78	.886433+04	79	.969988+04	80	.646325+04
	81	.109863+05	82	.413362+04	83	.952031+04	84	-.570379+03
	85	.325572+04	86	-.142179+04	87	.200220+04	88	-.137248+04
	89	.850568+03	90	-.215942+04	91	.440690+03	92	-.346266+04
	93	-.820216+02	94	-.319320+04	95	-.178453+03	96	-.291798+04
	97	-.492276+03	98	-.158962+04	99	-.356768+03	100	.254003+03
	101	-.277725+02	102	.249762+04	103	.678214+03	104	.475202+04
	105	.144746+04	106	.744818+04	107	.286572+04	108	.116164+05
	109	.585466+04	110	.103760+05	111	.950548+04	112	.832242+04
	113	.603406+04	114	.522518+04	115	.326327+04	116	.813633+04
	117	.200822+04	118	.500996+04	119	.132177+04	120	.102848+05
	121	.502856+04	122	.130592+05	123	.108334+05	124	.160060+05
	125	.173021+05	126	.187092+05	127	-.145236+04	128	-.293046+02
	129	-.471265+04	130	.163489+04	131	-.548561+04	132	.216981+04
	133	-.362745+04	134	.124260+04	135	-.368842+04	136	.121850+04
	137	-.294030+04	138	.796544+03	139	-.409585+04	140	.107073+04
	141	-.456414-09	142	.209648-10	143	-.728424-09	144	.236029-09
	145	-.264839-08	146	.126908-08	147	-.300854-08	148	.145978-08
	149	-.214858-08	150	.104233-08	151	-.171184-08	152	.837176-09
1	153	.123163+04	154	.157460+03	155	-.739654+04	156	.180452+03
	157	-.106161+05	158	.900803+03	159	-.145319+05	160	.277146+04
	161	-.266645+05	162	.775700+04	163	-.241351+05	164	.729335+04
	165	-.137292+05	166	.453288+04	167	-.160917+05	168	.656066+04
	169	-.163839+05	170	.804696+04	171	-.168856+05	172	.103011+05
	173	.199148-10	174	-.173030-08	175	.889224-09	176	-.174064-08
	177	.113133-08	178	-.155143-08	179	-.164085+05	180	.113675+05
	181	-.163136+05	182	.125257+05	183	-.155985+05	184	.118441+05
	185	-.110447+05	186	.103344+05	187	-.533239+04	188	.749805+04
	189	.758278+03						

LOADS ARE IN POUNDS

NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.10
Prepared				Page	
Checked A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C=5A	
Approved L G DARBY	1 JUN 69	AFT FUSELAGE		LGIUS46=2=2 Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.1 LINK LOADS (CONDITION TPM096) MOD 1 (REF. P. 6.5)

1	1	-.417522+05	2	-.417474+05	3	-.420568+05	4	-.420175+05
	5	-.355453+05	6	-.354529+05	7	-.160360+05	8	-.159847+05
	9	-.165060+05	10	-.164539+05	11	-.113953+05	12	-.113664+05
	13	-.764506+05	14	-.761952+05	15	-.227400+05	16	-.227951+05
	17	-.238052+05	18	-.239245+05	19	-.473158+05	20	-.475980+05
	21	-.462925+05	22	-.465716+05	23	-.291273+05	24	-.293187+05
	25	-.231458+05	26	-.232868+05	27	-.372572+05	28	-.374317+05
	29	-.193510+04	30	-.190377+04	31	-.163274+05	32	-.164076+05
	33	-.123091+05	34	-.124223+05	35	-.167221+05	36	-.170294+05
	37	-.161111+05	38	-.171614+05	39	-.144022+05	40	-.161116+05
	41	-.217041+04	42	-.539613+04	43	.136259+05	44	.959558+04
	45	.293789+05	46	.255147+05	47	.420103+05	48	.401486+05
	49	.238530+05	50	.213101+05	51	.126888+05	52	.126606+05
	53	.649128+05	54	.628104+05	55	.457772+05	56	.605775+05
	57	.609104+05	58	.819207+05	59	.100590+06	60	.941181+05
	61	.111098+06	62	.104342+06	63	.112844+06	64	-.231704+04
	65	.194311+04	66	-.118565+05	67	.114564+05	68	-.128272+05
	69	.124529+05	70	-.645463+04	71	.626508+04	72	-.575832+04
	73	.558246+04	74	-.445521+04	75	.430390+04	76	-.703236+04
	77	.701178+04	78	-.147250+04	79	.155046+04	80	-.812444+04
	81	.823334+04	82	-.965899+04	83	.977681+04	84	-.680496+04
	85	.689776+04	86	-.606838+04	87	.617074+04	88	-.393992+04
	89	.402119+04	90	-.452157+04	91	.461486+04	92	-.567385+04
	93	.565035+04	94	-.501842+04	95	.499712+04	96	-.404657+04
	97	.403935+04	98	-.205550+04	99	.205601+04	100	.478115+03
	101	-.495431+03	102	.310738+04	103	-.325722+04	104	.518872+04
	105	-.556482+04	106	.727614+04	107	-.801264+04	108	.109198+05
	109	-.115925+05	110	.272172+04	111	.935994+03	112	.312186+04
	113	-.537535+03	114	.217822+04	115	-.103117+04	116	.121625+05
	117	-.114520+05	118	.609927+04	119	-.584366+04	120	.207754+04
	121	-.657164+04	122	.533830+03	123	-.387570+04	124	.236293+04
	125	-.952624+03	126	.108589+04	127	-.270856+04	128	-.269223+04
	129	-.553886+04	130	-.543117+04	131	-.590840+04	132	-.576740+04
	133	-.420607+04	134	-.409509+04	135	-.433433+04	136	-.421272+04
	137	-.369008+04	138	-.356564+04	139	-.514788+04	140	-.501740+04
	141	-.777888-09	142	-.780983-09	143	-.886155-09	144	-.890419-09
	145	-.251791-08	146	-.253131-08	147	-.283840-08	148	-.284852-08
	149	-.202769-08	150	-.203136-08	151	-.160090-08	152	-.159604-08
1	153	.238736+04	154	.243557+04	155	-.120918+05	156	-.120381+05
	157	-.161905+05	158	-.161287+05	159	-.196233+05	160	-.195954+05
	161	-.314244+05	162	-.315413+05	163	-.279702+05	164	-.282695+05
	165	-.155436+05	166	-.162477+05	167	-.151474+05	168	-.163463+05
	169	-.128495+05	170	-.144117+05	171	-.123355+05	172	-.135332+05
	173	-.326856-08	174	-.324111-08	175	-.191102-08	176	-.200005-08
	177	-.144745-08	178	-.122997-08	179	-.162978+05	180	-.113902+05
	181	-.128081+05	182	-.715398+04	183	.213869+03	184	-.324039+04
	185	.456868+04	186	.445325+03	187	.240352+04	188	.258654+04
	189	.233132+04						

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
					Vol. I	6.11
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C=5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		LGIUS46=2=2	
					Report No.	

REV. (A) RFW 1 AUG. 69

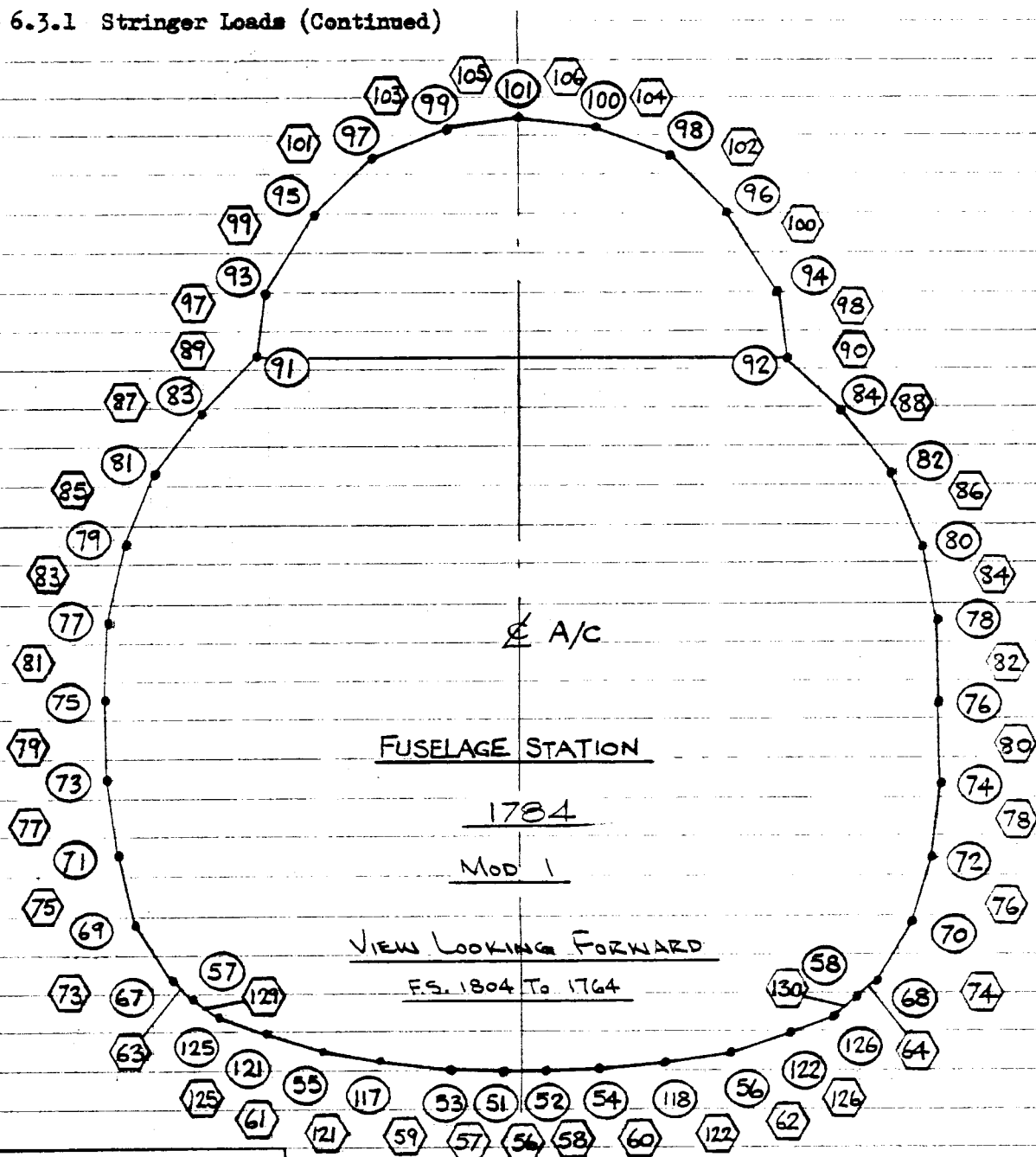
6.0 SHELL STRUCTURE F.S. 1744 TO 2101
6.3 SKINS AND STRINGERS
6.3.1 LINK LOADS (CONDITION IPC242) MOD 1 (REF. 6.5)

1	.100938+05	2	.936454+04	3	.108848+05	4	.946452+04
5	.119007+05	6	.958252+04	7	.581854+04	8	.462197+04
9	.661444+04	10	.528713+04	11	.487312+04	12	.399473+04
13	.315057+05	14	.250968+05	15	.507903+04	16	.475564+04
17	.576185+04	18	.511416+04	19	.146194+05	20	.122889+05
21	.155974+05	22	.130923+05	23	.108316+05	24	.913366+04
25	.900333+04	26	.760374+04	27	.168693+05	28	.145410+05
29	.292854+04	30	.262719+04	31	.571648+04	32	.504180+04
33	.422068+04	34	.407219+04	35	.404499+04	36	.454131+04
37	.137944+04	38	.344446+04	39	-.249097+03	40	.272836+04
41	-.284936+04	42	.168553+04	43	-.553199+04	44	-.810912+03
45	-.765459+04	46	-.419791+04	47	-.869118+04	48	-.724335+04
49	-.493563+04	50	-.403104+04	51	-.268810+04	52	-.246583+04
53	-.131666+05	54	-.109687+05	55	-.102836+05	56	-.148422+05
57	-.190838+05	58	-.206980+05	59	-.370331+05	60	-.270245+05
61	-.395561+05	62	-.324495+05	63	-.374047+05	64	.164671+04
65	.215712+04	66	.158250+03	67	.377193+04	68	-.729025+03
69	.413705+04	70	-.125781+04	71	.264682+04	72	-.156881+04
73	.275093+04	74	-.206383+04	75	.296609+04	76	-.280403+04
77	.304735+04	78	-.443576+03	79	-.105992+04	80	.779105+03
81	-.233478+04	82	.978280+03	83	-.234429+04	84	.460102+03
85	-.103602+04	86	.230785+03	87	-.664856+03	88	.381003+02
89	-.280828+03	90	.212339+03	91	-.476103+03	92	-.310939+04
93	.325944+04	94	-.287779+04	95	.299215+04	96	-.195949+04
97	.203845+04	98	-.904294+03	99	.944515+03	100	.343752+03
101	-.388834+03	102	.193774+04	103	-.214458+04	104	.341972+04
105	-.390173+04	106	.463701+04	107	-.564401+04	108	.618300+04
109	-.836545+04	110	.149456+04	111	.171107+04	112	.123432+04
113	.148634+04	114	.831100+03	115	.109743+04	116	.367293+04
117	-.558746+04	118	.240552+04	119	-.380178+04	120	.559214+04
121	-.565463+04	122	.456034+04	123	-.660432+04	124	.585693+03
125	-.585208+04	126	-.254852+04	127	.631602+03	128	.422782+03
129	.546454+03	130	-.552525+03	131	.556825+02	132	-.124938+04
133	-.560505+03	134	-.139536+04	135	-.927037+03	136	-.176493+04
137	-.170713+04	138	-.244815+04	139	-.345023+04	140	-.415381+04
141	.169532-09	142	.153699-09	143	.257380-09	144	.224230-09
145	.100746-08	146	.856661-09	147	.121804-08	148	.103780-08
149	.940287-09	150	.810616-09	151	.811012-09	152	.707283-09
153	-.174228+03	154	-.318678+03	155	-.673021+04	156	-.705097+04
157	-.890587+04	158	-.927030+04	159	-.915664+04	160	-.953364+04
161	-.125325+05	162	-.130727+05	163	-.117164+05	164	-.123825+05
165	-.900599+04	166	-.982345+04	167	-.995425+04	168	-.110437+05
169	-.898898+04	170	-.105260+05	171	-.845918+04	172	-.109569+05
173	.646064-09	174	.552029-09	175	.640082-09	176	.361108-09
177	.725136-09	178	.334181-09	179	-.772447+04	180	-.128984+05
181	-.837639+04	182	-.140180+05	183	-.945327+04	184	-.105110+05
185	-.615532+04	186	-.652681+04	187	-.169602+04	188	-.352966+04
189	-.115397+04						

LOADS ARE IN POUNDS

REV: (A) M.B. 1 AUG 69.

6.3.1 Stringer Loads (Continued)



KEY

— GRID POINT

○ FAMAS STRINGER

◡ FAMAS PANEL

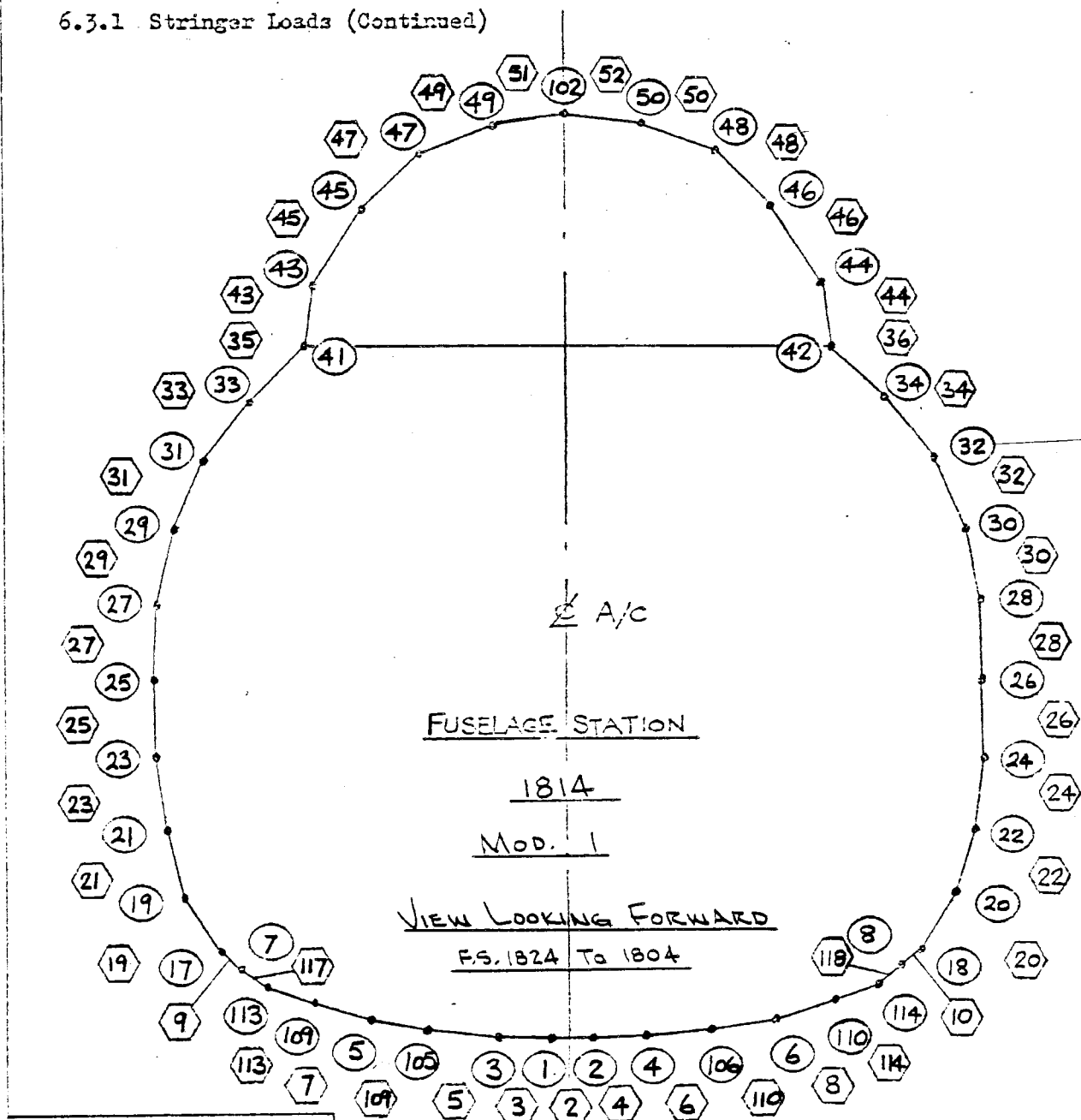
FIGURE No 6.3.1-2

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

Prepared	NAME V.W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Vol. 1	PERM. 6.13
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	Report No.
Approved	L.G. Darby	16 Aug 68			

REV. (A) A.D.P. 1 AUG 69

6.3.1 Stringer Loads (Continued)



KEY

—●— GRID POINT

○ FAMAS STRINGER

⬡ FAMAS PANEL

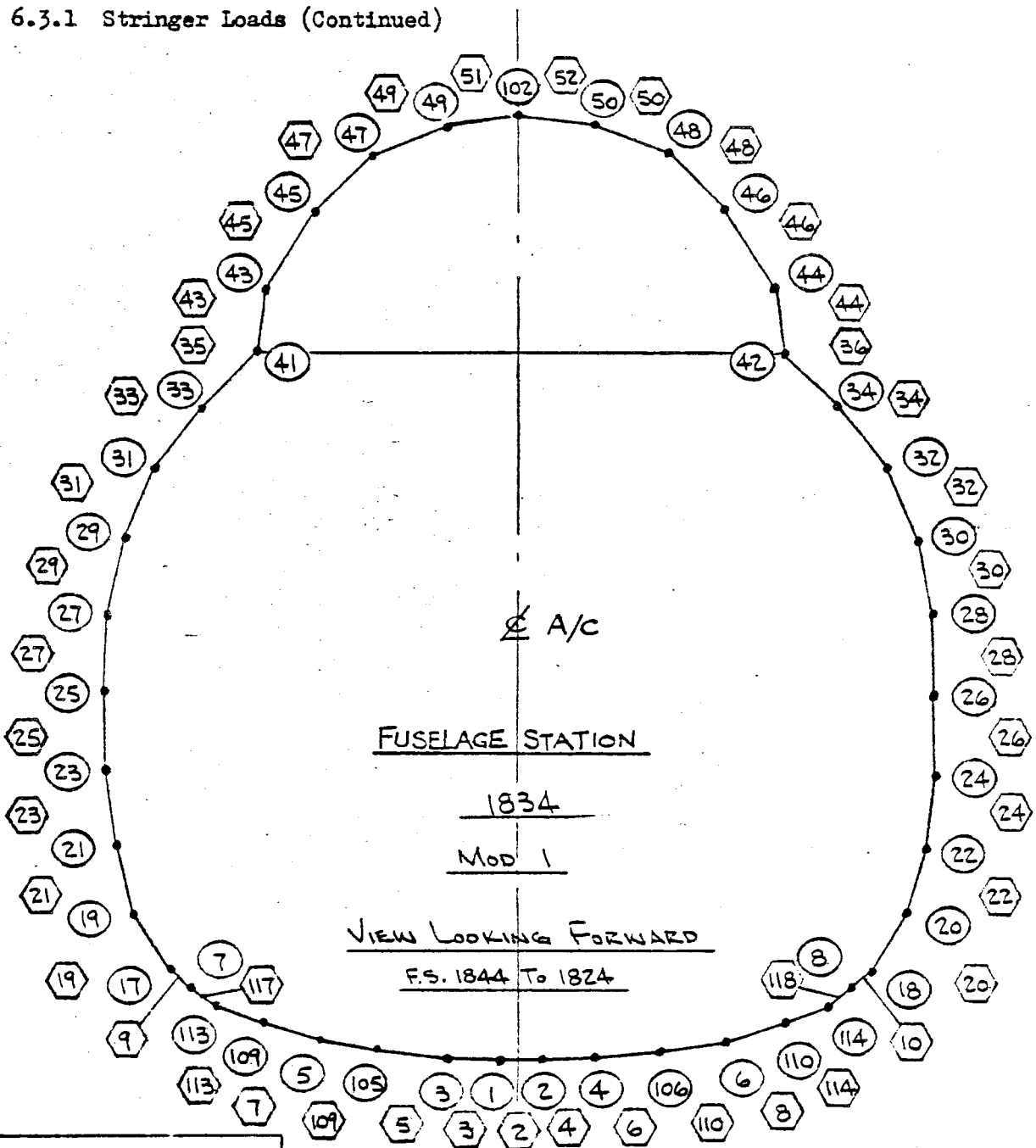
FIGURE No 6.3.1.3

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2.

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
	V.W. Boyd	16 Aug 68		Vol. I	6.14
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	
				LGJUS 45-2-2	
Approved	L.G. Darby	16 Aug 68		Report No.	

REV: (A) M.D. 1 AUG. 69.

6.3.1 Stringer Loads (Continued)



KEY

—●— GRID POINT

○ FAMAS STRINGER

⬡ FAMAS PANEL

FIGURE No 6.3.1.4

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.15
Checked	A L PINCHOT	1JUN69	TITLE	Model	C-5A
Approved	L G DARBY	1JUN69	INTERNAL LOADS ANALYSIS	Report No.	LG1US46-2-
			AFT FUSELAGE		

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON. I)

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS (CONDITION 5PMQ27) 1 MOD 1

* 1 -.237603+05	2 -.237396+05	3 -.245898+05	4 -.245678+05
5 -.147415+05	6 -.147342+05	7 -.926382+05	8 -.825636+05
9 -.124882+05	10 -.125136+05	11 -.137096+05	12 -.137488+05
13 -.348818+05	14 -.349078+05	15 -.311768+05	16 -.309274+05
17 .349589+04	18 .353344+04	19 -.233935+04	20 -.231358+04
21 -.301510+04	22 -.299552+04	23 -.877352+04	24 -.868140+04
25 -.569910+05	26 -.562483+05	27 -.157147+05	28 -.158339+05
29 -.312841+04	30 -.477478+04	31 .692658+04	32 .530152+04
33 .183725+05	34 .162279+05	35 .331009+05	36 .323663+05
37 .194383+05	38 .188509+05	39 .142722+05	40 .114814+05
41 .828154+05	42 .409996+05	43 .117581+05	44 .513391+05
45 .175713+05	46 .607992+05	47 .105203+06	48 .733757+05
49 .980033+05	50 .914834+05	51 -.342691+05	52 -.342397+05
53 -.345947+05	54 -.345447+05	55 -.140626+05	56 -.140060+05
57 -.638487+05	58 -.634609+05	59 -.186732+05	60 -.187018+05
61 -.198204+05	62 -.198825+05	63 -.405061+05	64 -.405752+05
65 -.349248+05	66 -.350216+05	67 -.683153+04	68 -.682356+04
69 -.982623+04	70 -.979380+04	71 -.863969+04	72 -.867696+04
73 -.167899+05	74 -.167389+05	75 -.245913+05	76 -.248939+05
77 -.181718+05	78 -.189578+05	79 -.527358+04	80 -.757750+04
81 .957994+04	82 .639427+04	83 .251522+05	84 .202473+05
85 .363462+05	86 .347986+05	87 .214106+05	88 .188044+05
89 .139444+05	90 .111224+05	91 .621778+05	92 .514432+05
93 .326206+05	94 .513654+05	95 .445965+05	96 .710915+05
97 .906235+05	98 .789096+05	99 .984786+05	100 .920796+05
101 .979554+05	102 .930544+05	103 -.329022+05	104 -.329806+05
105 -.269523+05	106 -.269386+05	107 -.234324+05	108 -.233896+05
109 -.172432+05	110 -.172379+05	111 -.177958+05	112 -.177217+05
113 -.174904+05	114 -.174903+05	115 -.407988+05	116 -.409039+05
117 -.312465+05	118 -.311453+05	119 -.256204+05	120 -.256275+05
121 -.155346+05	122 -.154541+05	123 -.196233+05	124 -.196357+05
125 -.116604+05	126 -.115852+05		

* STRINGER (1) REF. P. 6.13 ULTIMATE COMPRESSION LOAD = -23,760. LB.

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	No. I 6.16
Checked	A L PINCHOT	1JUN69	TITLE	Model	C-5A
Approved	L G DARBY	1JUN69	INTERNAL LOADS ANALYSIS	LG1US46-2-	
			AFT FUSELAGE	Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 *CON.1
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS *CONDITION TPM096+P I MOD 1

1	1	-.904098+04	2	-.902829+04	3	-.100403+05	4	-.100329+05
	5	-.112583+05	6	-.112650+05	7	-.712121+05	8	-.712082+05
	9	.552406+04	10	.547918+04	11	.309746+04	12	.302362+04
	13	-.198422+05	14	-.199553+05	15	-.255002+05	16	-.252653+05
	17	.342352+04	18	.348938+04	19	-.196425+04	20	-.192559+04
	21	-.200762+04	22	-.198833+04	23	-.543800+04	24	-.538305+04
	25	-.413771+05	26	-.411210+05	27	-.597087+04	28	-.656258+04
	29	.118837+05	30	.111342+05	31	.268233+05	32	.238448+05
	33	.460045+05	34	.399343+05	35	.535744+05	36	.523885+05
	37	.409551+05	38	.390898+05	39	.238109+05	40	.196819+05
	41	.140311+06	42	.793626+05	43	.158803+05	44	.810317+05
	45	.231446+05	46	.928350+05	47	.155574+06	48	.102125+06
	49	.137514+06	50	.128390+06	51	-.198637+05	52	-.198392+05
	53	-.203385+05	54	-.202962+05	55	-.101424+05	56	-.100890+05
	57	-.503914+05	58	-.500058+05	59	-.212909+04	60	-.218833+04
	61	-.417312+04	62	-.429182+04	63	-.246346+05	64	-.248697+05
	65	-.275366+05	66	-.277578+05	67	-.590550+04	68	-.581037+04
	69	-.720247+04	70	-.720476+04	71	-.587484+04	72	-.591303+04
	73	-.112774+05	74	-.114022+05	75	-.130926+05	76	-.140172+05
	77	-.570770+04	78	-.734059+04	79	.115841+05	80	.864144+04
	81	.312933+05	82	.259270+05	83	.517396+05	84	.439823+05
	85	.556507+05	86	.537410+05	87	.461637+05	88	.424285+05
	89	.172352+05	90	.144081+05	91	.108394+06	92	.939287+05
	93	.443433+05	94	.763922+05	95	.582662+05	96	.987228+05
	97	.126002+06	98	.107457+06	99	.138459+06	100	.127902+06
	101	.137607+06	102	.131447+06	103	-.131667+05	104	-.133438+05
	105	-.180353+05	106	-.180457+05	107	-.165324+05	108	-.165292+05
	109	-.139622+05	110	-.139763+05	111	-.136131+05	112	-.135613+05
	113	-.149901+05	114	-.150038+05	115	-.210310+05	116	-.213085+05
	117	-.216714+05	118	-.215804+05	119	-.178438+05	120	-.179488+05
	121	-.118344+05	122	-.117540+05	123	-.147013+05	124	-.147848+05
	125	-.928450+04	126	-.920320+04				

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL. I	PERM. 6.17
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model	C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No.	LG1US46-2-	

REV. (2) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 *CON.1

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS *CONDITION ALSIII 1 MOD 1

1	1	-.245628+05	2	-.245987+05	3	-.253899+05	4	-.255191+05
	5	-.149893+05	6	-.153216+05	7	-.815616+05	8	-.882353+05
	9	-.128147+05	10	-.133321+05	11	-.138043+05	12	-.148246+05
	13	-.340379+05	14	-.376620+05	15	-.300935+05	16	-.334094+05
	17	.349271+04	18	.325730+04	19	-.220304+04	20	-.285764+04
	21	-.282199+04	22	-.353847+04	23	-.811739+04	24	-.965472+04
	25	-.524584+05	26	-.622028+05	27	-.129113+05	28	-.197499+05
	29	.187279+03	30	-.108581+05	31	.997665+04	32	-.631286+03
	33	.208343+05	34	.102027+05	35	.317034+05	36	.286367+05
	37	.193221+05	38	.158588+05	39	.146250+05	40	.913302+04
	41	.840368+05	42	.329110+05	43	.117975+05	44	.460670+05
	45	.175175+05	46	.546615+05	47	.103252+06	48	.686164+05
	49	.946358+05	50	.863428+05	51	-.348604+05	52	-.350949+05
	53	-.350871+05	54	-.355993+05	55	-.140790+05	56	-.147567+05
	57	-.624255+05	58	-.693785+05	59	-.190405+05	60	-.197115+05
	61	-.198757+05	62	-.212021+05	63	-.393072+05	64	-.440231+05
	65	-.337706+05	66	-.380731+05	67	-.668021+04	68	-.737380+04
	69	-.930619+04	70	-.112048+05	71	-.805318+04	72	-.100689+05
	73	-.154155+05	74	-.191864+05	75	-.216395+05	76	-.293714+05
	77	-.150792+05	78	-.238439+05	79	-.162448+04	80	-.142973+05
	81	.129642+05	82	-.610672+03	83	.277493+05	84	.132033+05
	85	.350905+05	86	.305224+05	87	.214121+05	88	.157127+05
	89	.143326+05	90	.890863+04	91	.641130+05	92	.416007+05
	93	.331354+05	94	.450396+05	95	.444257+05	96	.643858+05
	97	.890696+05	98	.730235+05	99	.955279+05	100	.867199+05
	101	.937605+05	102	.889011+05	103	-.322241+05	104	-.356830+05
	105	-.275622+05	106	-.280551+05	107	-.227875+05	108	-.251666+05
	109	-.173416+05	110	-.180871+05	111	-.172566+05	112	-.190507+05
	113	-.174165+05	114	-.184500+05	115	-.397755+05	116	-.442728+05
	117	-.314309+05	118	-.325660+05	119	-.247920+05	120	-.278851+05
	121	-.154942+05	122	-.163507+05	123	-.189706+05	124	-.214050+05
	125	-.115678+05	126	-.123926+05				

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. No. I	PERM. 6.13
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON. I)

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS (CONDITION DMLOOI) 1 MOD 1

1	1	-.280887+04	2	-.122674+05	3	.165827+04	4	-.173110+05
	5	.653348+04	6	-.158070+05	7	.541321+05	8	-.105662+06
	9	.116664+03	10	-.698275+04	11	.335601+04	12	-.110227+05
	13	.202165+05	14	-.407150+05	15	.219702+05	16	-.403381+05
	17	.227351+04	18	-.513755+03	19	.396198+04	20	-.541613+04
	21	.355870+04	22	-.521705+04	23	.776138+04	24	-.123253+05
	25	.491878+05	26	-.778042+05	27	.304163+05	28	-.368042+05
	29	.409021+05	30	-.448352+05	31	.463162+05	32	-.389700+05
	33	.531165+05	34	-.348291+05	35	.215719+05	36	.724137+03
	37	.204169+05	38	-.644342+04	39	.176642+05	40	-.910510+04
	41	.740417+05	42	-.367229+05	43	.103927+05	44	-.851820+04
	45	.150131+05	46	-.215722+05	47	.939533+05	48	.385852+04
	49	.428288+05	50	.157816+05	51	-.522072+04	52	-.157242+05
	53	-.118895+03	54	-.210880+05	55	.576554+04	56	-.144232+05
	57	.489349+05	58	-.882242+05	59	-.664210+03	60	-.983035+04
	61	.359014+04	62	-.148103+05	63	.236234+05	64	-.472432+05
	65	.228788+05	66	-.436586+05	67	.424126+04	68	-.836726+04
	69	.104496+05	70	-.162685+05	71	.991276+04	72	-.147286+05
	73	.179026+05	74	-.266137+05	75	.337718+05	76	-.453444+05
	77	.362800+05	78	-.441920+05	79	.468628+05	80	-.494406+05
	81	.518920+05	82	-.432107+05	83	.508505+05	84	-.354716+05
	85	.235506+05	86	.467326+03	87	.188814+05	88	-.596070+04
	89	.140382+05	90	-.739974+04	91	.671430+05	92	-.353043+05
	93	.291337+05	94	-.157293+05	95	.360741+05	96	-.979336+04
	97	.671460+05	98	.141225+04	99	.496078+05	100	.154148+05
	101	.320703+05	102	.292545+05	103	.178879+05	104	-.370362+05
	105	.105515+05	106	-.276335+05	107	.142283+05	108	-.280863+05
	109	.866958+04	110	-.194661+05	111	.112671+05	112	-.217969+05
	113	.956626+04	114	-.204504+05	115	.211417+05	116	-.447736+05
	117	.985561+04	118	-.291128+05	119	.164466+05	120	-.314947+05
	121	.793138+04	122	-.174690+05	123	.131486+05	124	-.247421+05
	125	.723396+04	126	-.144003+05				

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.		PERM.	
Prepared	M BURTON	NOV70		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6:18.1	
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS		C-5A			
Approved	H H WADE	NOV70	AFT FUSELAGE		Model		LGIUS46-2-2		
					Report No.				

REV B M.D 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 1

1	1	-.118077+05	2	-.234520+04	3	-.168411+05	4	.212518+04
	5	-.155997+05	6	.672591+04	7	-.105298+06	8	.544115+05
	9	-.709309+04	10	-.175791+02	11	-.111215+05	12	.321587+04
	13	-.408878+05	14	.199578+05	15	-.408915+05	16	.214602+05
	17	-.579035+03	18	.221178+04	19	-.544765+04	20	.392823+04
	21	-.529913+04	22	.346972+04	23	-.126967+05	24	.739972+04
	25	-.809528+05	26	.461811+05	27	-.387732+05	28	.281868+05
	29	-.415615+05	30	.431274+05	31	-.406820+05	32	.434114+05
	33	-.401006+05	34	.460604+05	35	-.721405+03	36	.196444+05
	37	-.799182+04	38	.183818+05	39	-.849458+04	40	.165685+05
	41	-.226213+05	42	.637750+05	43	-.304251+04	44	.407528+05
	45	-.379283+04	46	.617865+05	47	-.246322+05	48	.428016+05
	49	.192565+05	50	.426352+05	51	-.153658+05	52	-.484955+04
	53	-.207349+05	54	.255149+03	55	-.143391+05	56	.587671+04
	57	-.883306+05	58	.489732+05	59	-.998972+04	60	-.847737+03
	61	-.149651+05	62	.338339+04	63	-.475313+05	64	.232378+05
	65	-.438053+05	66	.226274+05	67	-.835028+04	68	.426229+04
	69	-.164500+05	70	.102674+05	71	-.150400+05	72	.958805+04
	73	-.274273+05	74	.170455+05	75	-.472094+05	76	.315426+05
	77	-.458568+05	78	.339725+05	79	-.484620+05	80	.463451+05
	81	-.445220+05	82	.485578+05	83	-.342432+05	84	.491876+05
	85	-.495415+03	86	.216575+05	87	-.529674+04	88	.180279+05
	89	-.528129+04	90	.146210+05	91	-.280134+05	92	.686317+05
	93	-.874502+04	94	.484967+05	95	-.770330+04	96	.550243+05
	97	-.879256+04	98	.485694+05	99	.129664+05	100	.429087+05
	101	.298966+05	102	.299742+05	103	-.371968+05	104	.176181+05
	105	-.272178+05	106	.109452+05	107	-.282404+05	108	.140460+05
	109	-.192719+05	110	.884199+04	111	-.219686+05	112	.110931+05
	113	-.202954+05	114	.969495+04	115	-.450386+05	116	.207573+05
	117	-.288537+05	118	.101583+05	119	-.317096+05	120	.161868+05
	121	-.174094+05	122	.802895+04	123	-.249046+05	124	.129481+05
	125	-.143917+05	126	.727664+04				

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69		LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I		PERM. 6.19	
Prepared							Page		
Checked	A L PINCHOT		1 JUN 69		TITLE INTERNAL LOADS ANALYSIS		Model	C-5A	
Approved	L G DARBY		1 JUN 69		AFT FUSELAGE		Report No.	LGIUS46-2-	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON.1

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS <CONDITION TPM096

1 MOD 1

I 1	1 -.268030+05	2 -.267927+05	3 -.278684+05	4 -.278666+05
	5 -.167103+05	6 -.167204+05	7 -.952564+05	8 -.953247+05
	9 -.129339+05	10 -.129738+05	11 -.144007+05	12 -.144699+05
	13 -.383245+05	14 -.384633+05	15 -.356474+05	16 -.355561+05
	17 .256158+04	18 .257811+04	19 -.299203+04	20 -.298554+04
	21 -.327776+04	22 -.328354+04	23 -.892410+04	24 -.891260+04
	25 -.572737+05	26 -.570662+05	27 -.156098+05	28 -.160900+05
	29 -.211798+04	30 -.409444+04	31 .931197+04	32 .717027+04
	33 .228683+05	34 .195569+05	35 .371077+05	36 .352498+05
	37 .221923+05	38 .212732+05	39 .163087+05	40 .133000+05
	41 .903596+05	42 .480395+05	43 .129418+05	44 .573024+05
	45 .202648+05	46 .730606+05	47 .126197+06	48 .843088+05
	49 .112363+06	50 .106018+06	51 -.372154+05	52 -.371968+05
	53 -.377414+05	54 -.377120+05	55 -.156110+05	56 -.155753+05
	57 -.729104+05	58 -.727071+05	59 -.194753+05	60 -.195237+05
	61 -.208064+05	62 -.209090+05	63 -.439229+05	64 -.441398+05
	65 -.384936+05	66 -.387107+05	67 -.745777+04	68 -.746046+04
	69 -.111452+05	70 -.111798+05	71 -.342450+04	72 -.948686+04
	73 -.172830+05	74 -.174252+05	75 -.244795+05	76 -.252189+05
	77 -.177020+05	78 -.189309+05	79 -.369343+04	80 -.648897+04
	81 .122601+05	82 .863211+04	83 .287747+05	84 .237180+05
	85 .406686+05	86 .390258+05	87 .238355+05	88 .212314+05
	89 .153038+05	90 .126786+05	91 .684186+05	92 .586064+05
	93 .358450+05	94 .582125+05	95 .506355+05	96 .809000+05
	97 .105169+06	98 .896262+05	99 .112665+06	100 .104944+06
	101 .111600+06	102 .107590+06	103 -.357612+05	104 -.359446+05
	105 -.306895+05	106 -.307085+05	107 -.260618+05	108 -.260969+05
	109 -.195799+05	110 -.195968+05	111 -.199841+05	112 -.199752+05
	113 -.198459+05	114 -.198686+05	115 -.438496+05	116 -.440914+05
	117 -.345517+05	118 -.344893+05	119 -.280820+05	120 -.281902+05
	121 -.173175+05	122 -.172652+05	123 -.216481+05	124 -.217397+05
	125 -.131516+05	126 -.131031+05		

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
RF WILKINSON		1JUN69		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	No. 1 6.20
Prepared			TITLE				
Checked	A L PINCHOT		1JUN69		INTERNAL LOADS ANALYSIS		Model C-5A
Approved	L G DARBY		1JUN69		AFT FUSELAGE		LG10546-2-
						Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS (CONDITION IPC242) 1 MOD 1

1	1	.111712+05	2	.103088+05	3	.124613+05	4	.107827+05
	5	.788424+04	6	.618918+04	7	.525623+05	8	.444809+05
	9	.295972+04	10	.262512+04	11	.389333+04	12	.340448+04
	13	.154949+05	14	.131733+05	15	.201027+05	16	.178043+05
	17	.460429+04	18	.433580+04	19	.335865+04	20	.301694+04
	21	.149731+04	22	.143012+04	23	.681639+03	24	.875139+03
	25	.354057+03	26	.332449+04	27	.778943+03	28	.367089+04
	29	-.130967+04	30	.377063+04	31	-.574719+04	32	.155144+02
	33	-.125850+05	34	-.428923+04	35	-.768970+04	36	-.633914+04
	37	-.629182+04	38	-.375473+04	39	-.531622+04	40	-.314366+04
	41	-.176706+05	42	-.135937+05	43	-.264400+04	44	-.100676+05
	45	-.811188+04	46	-.368592+05	47	-.666153+05	48	-.302207+05
	49	-.436755+05	50	-.425968+05	51	.106927+05	52	.990348+04
	53	.118639+05	54	.103320+05	55	.646176+04	56	.519683+04
	57	.396781+05	58	.331461+05	59	.440030+04	60	.410174+04
	61	.523304+04	62	.462498+04	63	.158215+05	64	.133603+05
	65	.159138+05	66	.135731+05	67	.261321+04	68	.212416+04
	69	.622786+04	70	.554880+04	71	.361394+04	72	.347250+04
	73	.272081+04	74	.316246+04	75	.511363+03	76	.306221+04
	77	-.234703+03	78	.280760+04	79	-.300111+04	80	.184657+04
	81	-.623828+04	82	-.829051+03	83	-.862378+04	84	-.411401+04
	85	-.860451+04	86	-.703123+04	87	-.536168+04	88	-.385620+04
	89	-.300817+04	90	-.250914+04	91	-.140897+05	92	-.117085+05
	93	-.784280+04	94	-.145662+05	95	-.176633+05	96	-.246601+05
	97	-.442604+05	98	-.283741+05	99	-.422862+05	100	-.359864+05
	101	-.395203+05	102	-.436397+05	103	.133041+05	104	.112724+05
	105	.149611+05	106	.119029+05	107	.115923+05	108	.996758+04
	109	.940510+04	110	.757175+04	111	.964369+04	112	.838113+04
	113	.951980+04	114	.780725+04	115	.144425+05	116	.121763+05
	117	.134120+05	118	.108043+05	119	.112515+05	120	.957815+04
	121	.760369+04	122	.605657+04	123	.918666+04	124	.786461+04
	125	.645819+04	126	.522970+04				

LOADS ARE IN POUNDS

6.3.1 Stringer Loads (Continued)
REV (A) M.B. 1 AUG 69.

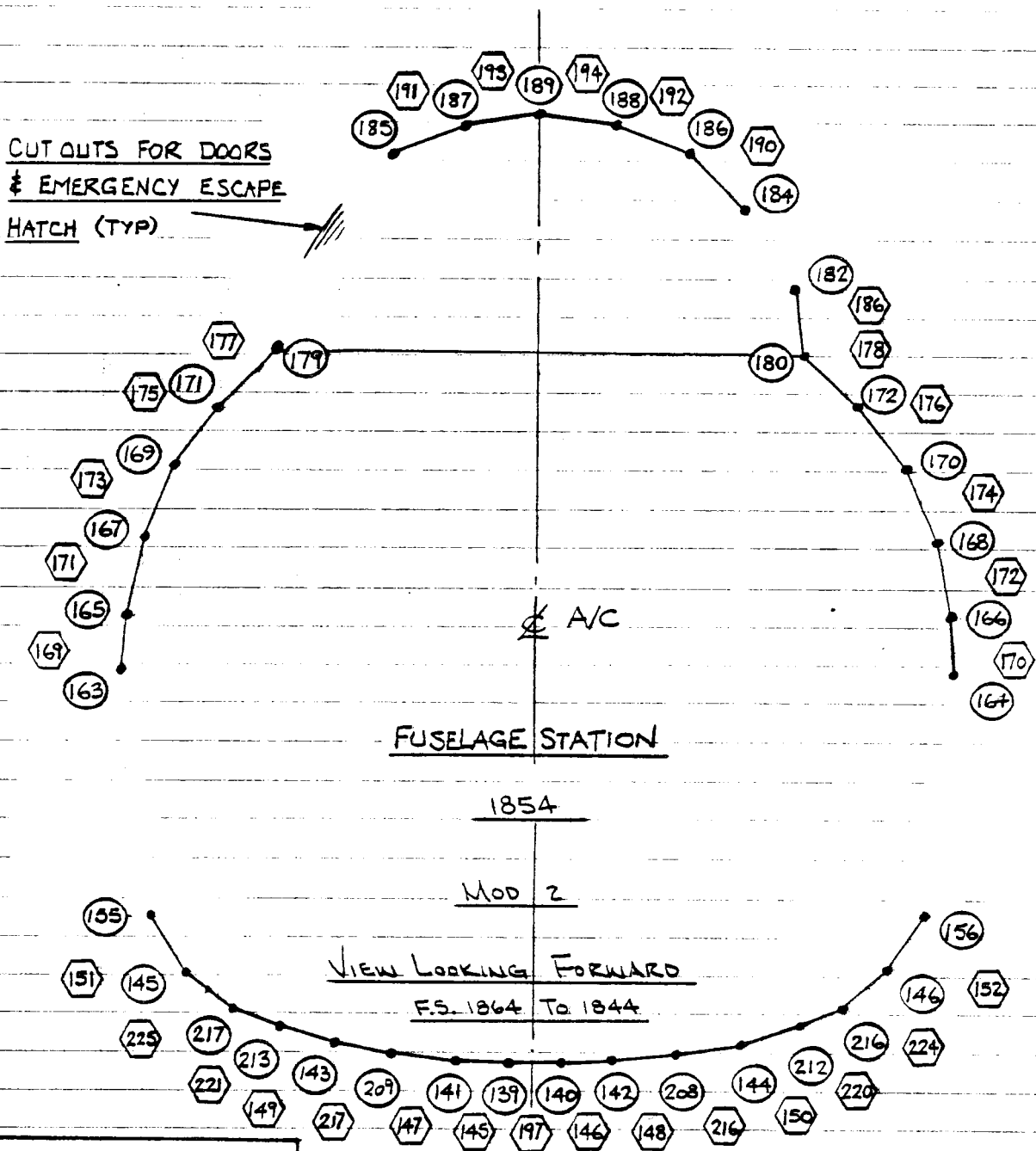


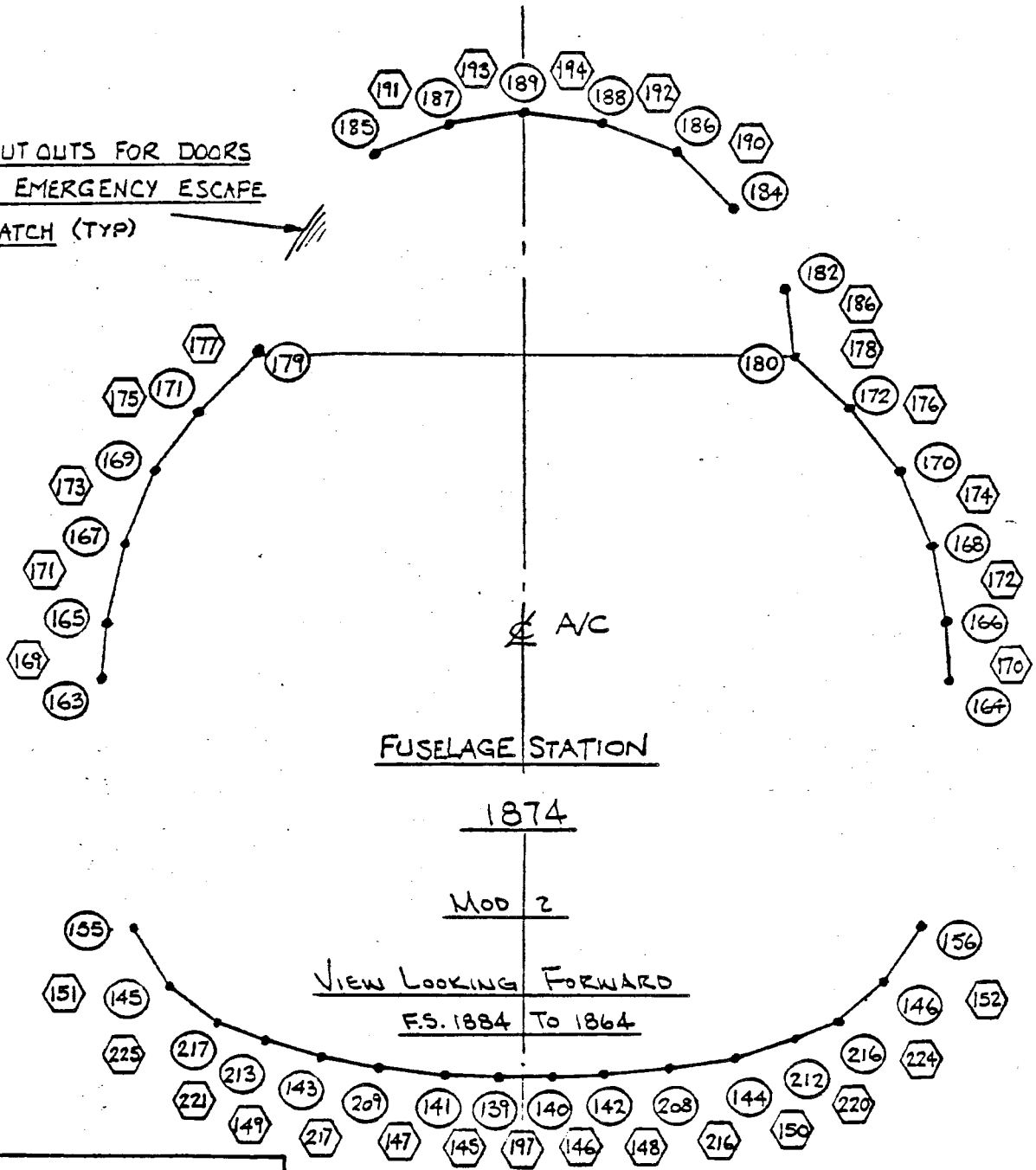
FIGURE No 6.3.1.5

FOR AREAS AND THICKNESSES SEE REF 3
VOLUME IV SECTION 5.2

6.3.1 Stringer Loads (Continued)

REV (A) M.B. 1 AUG 69.

CUT OUTS FOR DOORS
& EMERGENCY ESCAPE
HATCH (TYP)



KEY

—●— GRID POINT

○ FAMAS STRINGER

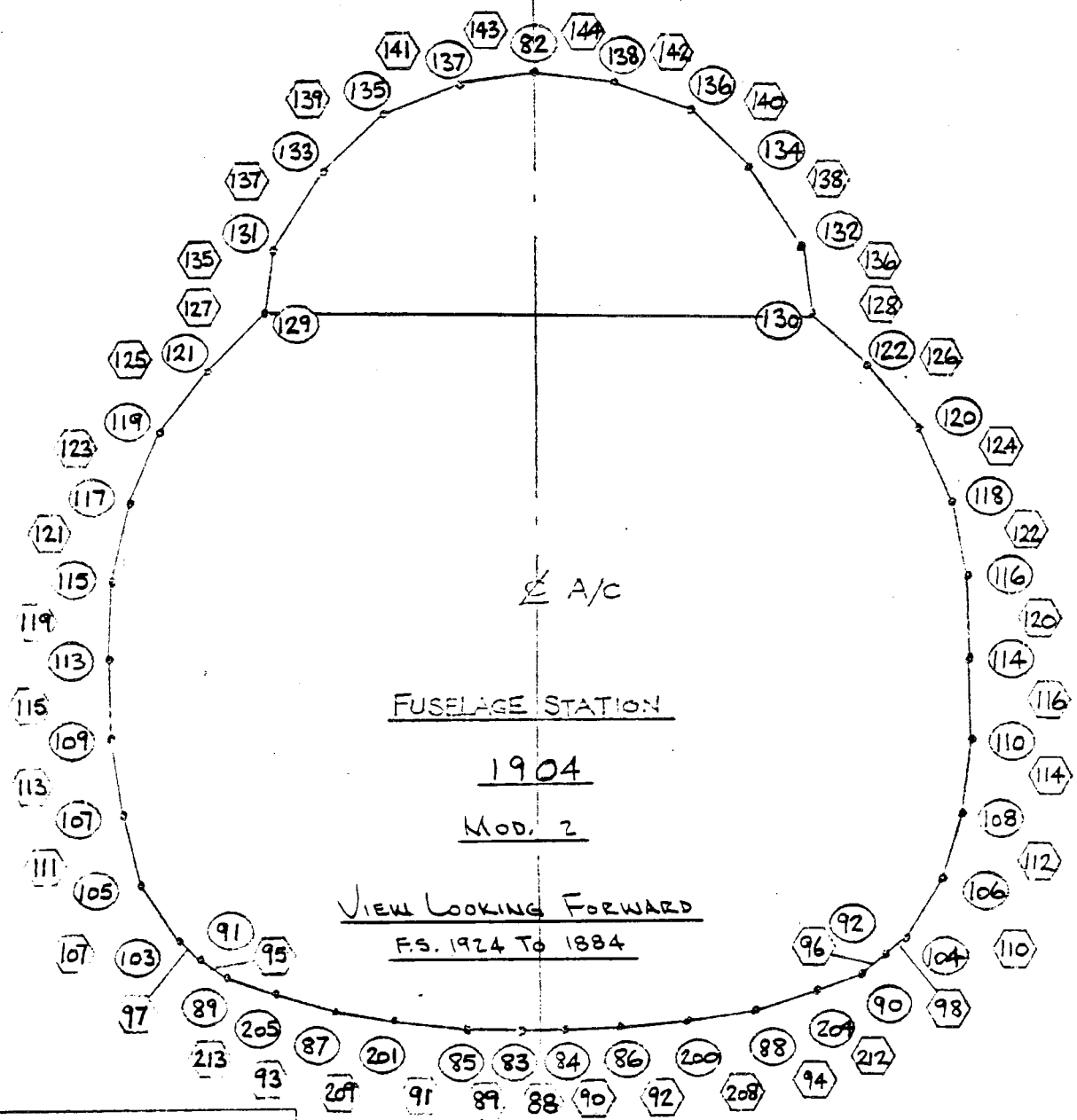
⬡ FAMAS PANEL

FIGURE No 6.3.1.6

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

REV. (A) M.B. 1 AUG 69

6.3.1 Stringer Loads (Continued)



KEY

GRID POINT

FAMAS STRINGER

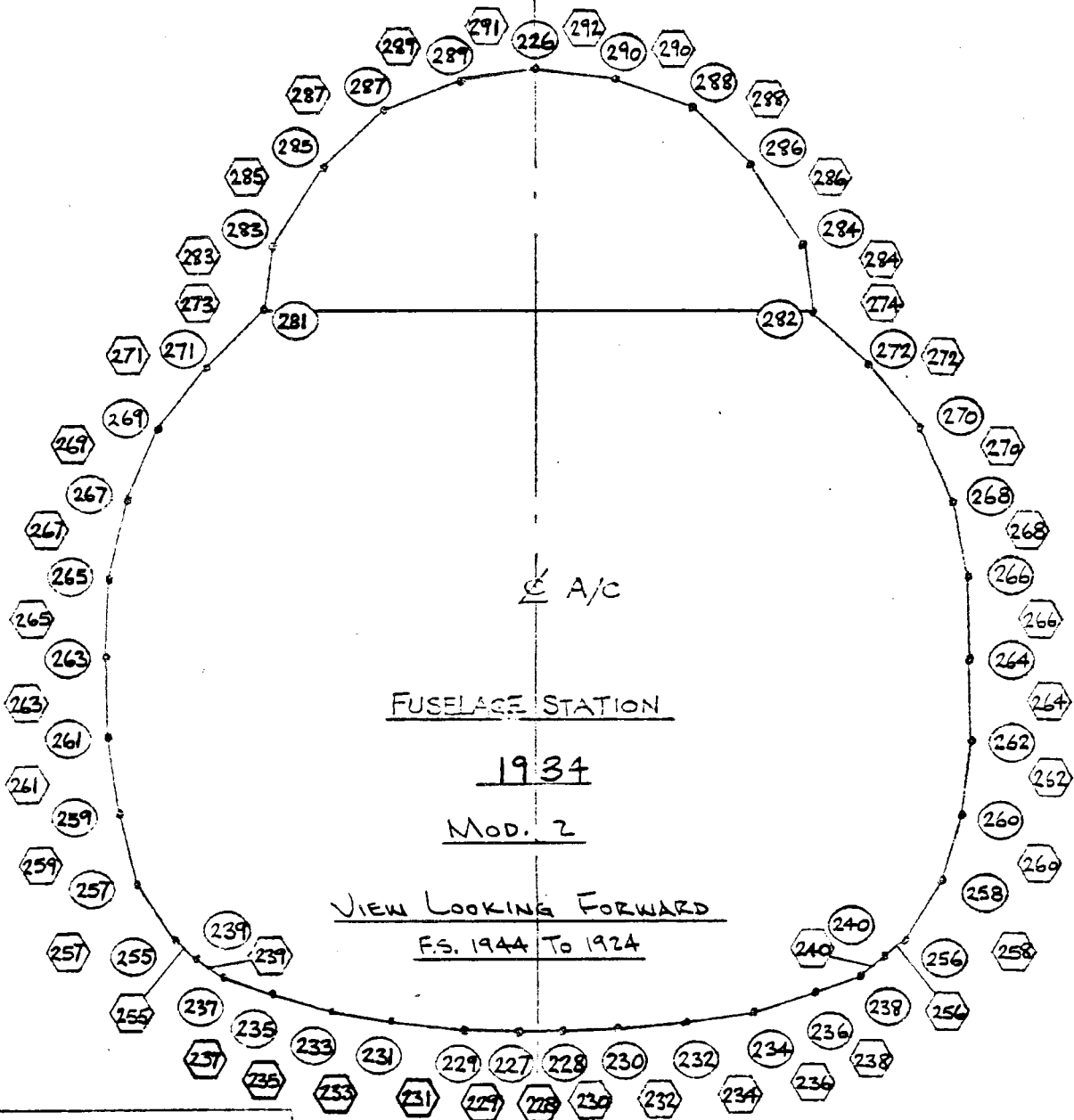
FAMAS PANEL

FIGURE No 6.3.1.7

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

REV (A) M.B 1 AUG 69.

6.3.1 Stringer Loads (Continued)



KEY

—●— GRID POINT

○ FAMA'S STRINGER

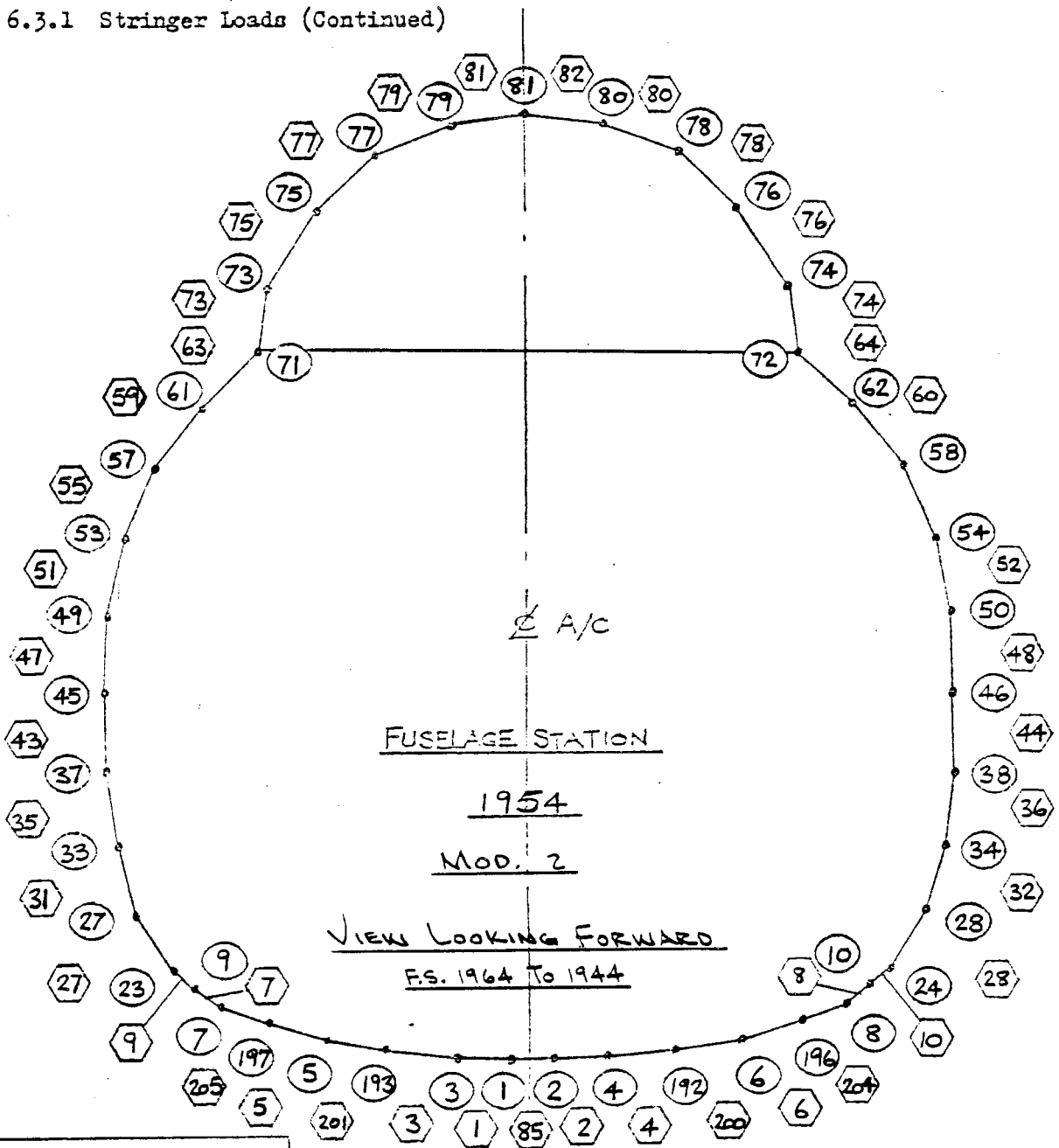
⬡ FAMA'S PANEL

FIGURE No 6.3.1-8

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

REV (A) M.D. 1 AUG 69

6.3.1 Stringer Loads (Continued)



KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FIGURE No 6.3.1.9

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2.

Prepared	NAME FF WILKINSON	DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS	Page	NOL. I 6.26
Approved	L C CAREY	1 JUN 65	AFT FUSELAGE	Model	C-5A LG1LS46-2-2
				Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION 1FMC56+P & MCC 2

* 1	.145500+05	2	.150450+05	3	.135168+05	4	.135662+05
5	.146370+04	6	.141858+04	7	-.257896+04	8	-.252210+04
9	-.334371+05	10	-.331966+05	11	.213155+05	12	.213422+05
13	.209415+05	14	.209436+05	15	.917133+04	16	.917635+04
17	-.572855+04	18	-.573186+04	19	-.456963+05	20	-.454433+05
21	-.706657+05	22	-.700057+05	23	-.104731+06	24	-.104604+06
27	-.245574+05	28	-.253728+05	33	-.137429+05	34	-.143577+05
37	-.134554+05	38	-.146576+05	45	-.267212+04	46	-.516646+04
49	-.705406+03	50	-.296841+04	53	.753703+04	54	.391274+04
57	.201867+05	58	.150275+05	61	.255216+05	62	.306670+05
63	.655151+05	64	.667633+05	65	.253538+05	66	.282355+05
67	.163061+05	68	.156660+05	69	.234506+05	70	.224716+05
71	.665557+05	72	.638038+05	73	.447266+05	74	.681772+05
75	.655606+05	76	.187258+05	77	.563074+05	78	.836047+05
79	.135826+06	80	.126121+06	81	.127157+06	82	.121451+06
83	.122555+05	84	.122453+05	85	.120256+05	86	.120317+05
87	.166085+03	88	.203383+03	89	-.991709+04	90	-.985480+04
91	-.985621+05	92	-.984637+05	93	.187227+05	94	.187072+05
95	.153841+05	96	.153604+05	97	-.396689+04	98	-.394158+04
99	-.176787+05	100	-.176127+05	101	-.719703+05	102	-.719134+05
103	-.276403+05	104	-.276263+05	105	-.131336+05	106	-.131701+05
107	-.705053+04	108	-.735147+04	109	-.354548+04	110	-.423500+04
113	-.291620+04	114	-.471265+04	115	-.426054+04	116	-.511574+04
117	.214045+04	118	.225562+04	119	.200480+05	120	.110056+05
121	.437913+05	122	.330430+05	123	.556616+05	124	.541074+05
125	.343357+05	126	.313115+05	127	.323708+05	128	.275678+05
129	.574328+05	130	.663408+05	131	.148054+05	132	.669145+05
133	.781428+05	134	.100753+06	135	.156452+06	136	.877613+05
137	.133320+06	138	.127548+06	139	.320147+04	140	.320116+04
141	.183842+04	142	.182427+04	143	-.611028+04	144	-.613010+04
145	-.100352+06	146	-.100413+06	147	.126633+05	148	.126562+05
149	.552725+04	150	.587651+04	151	-.144040+05	152	-.144471+05
153	-.425583+05	154	-.421324+05	155	-.404667+04	156	-.403111+04
163	-.333774+05	164	-.330851+05	165	-.462531+04	166	-.498770+04
167	.690925+04	168	.561548+04	169	.205354+05	170	.184803+05
171	.431745+05	172	.347774+05	173	.529893+05	174	.519692+05
175	.374350+05	176	.359001+05	177	.287552+05	178	.252104+05
179	.143885+06	180	.716654+05	182	.755499+05	184	.954540+05

* STRINGER (1) REF. P. 6.25 ULTIMATE TENSION LOAD = 14,998. LB.

NAME RF WILKINSON	DATE 1 JUNE 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.23
Prepared				Page	
Checked A L FINCHOT	1 JUNE 65	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved L C CAFEY	1 JUNE 65	AFT FUSELAGE		LG1LS46-2-2	
				Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SPELL STRUCTURE F.S. 1744 TC 2101 CCM.6
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS CONDITION ENLCC1 6 MCD 2

1	1	.361566+03	2	-.145992+03	3	.371975+03	4	-.119341+04
	5	.122167+04	6	-.196657+04	7	.166991+04	8	-.370672+04
	9	.125001+05	10	-.307556+05	11	.600327+02	12	-.278058+02
	13	-.195518+03	14	.244303+03	15	.112526+04	16	-.366373+04
	17	.317142+04	18	-.726674+04	19	.212850+05	20	-.498630+05
	21	.348183+05	22	-.745655+05	23	.322313+05	24	-.936350+05
	27	.597267+04	28	-.265561+05	33	.767400+04	34	-.181730+05
	37	.123572+05	38	-.234405+05	45	.175015+05	46	-.237321+05
	49	.155137+05	50	-.202370+05	53	.327324+05	54	-.378453+05
	57	.395774+05	58	-.401956+05	61	.465066+05	62	-.378458+05
	63	.425132+04	64	.112902+05	65	.167945+05	66	-.585050+04
	67	.108635+05	68	-.461611+04	69	.172171+05	70	-.885025+04
	71	.617072+05	72	-.351190+05	73	.364247+05	74	-.255210+05
	75	.282836+05	76	-.134136+05	77	.435875+05	78	-.425970+04
	79	.456451+05	80	.150144+05	81	.320096+05	82	.297523+05
	83	.520205+03	84	-.317601+04	85	.233524+04	86	-.515842+04
	87	.492447+04	88	-.617924+04	89	.603642+04	90	-.124436+05
	91	.493605+05	92	-.110922+06	93	.325289+03	94	-.181247+04
	95	.116885+04	96	-.301308+04	97	.111419+05	98	-.222383+05
	99	.670876+04	100	-.206454+05	101	.257213+05	102	-.744759+05
	103	.671605+04	104	-.246223+05	105	.364154+04	106	-.108119+05
	107	.262447+04	108	-.664720+04	109	.405410+04	110	-.763261+04
	113	.301953+05	114	-.390515+05	115	.161464+05	116	-.223624+05
	117	.392146+05	118	-.497664+05	119	.414834+05	120	-.514407+05
	121	.450051+05	122	-.327560+05	123	.127762+05	124	.365439+04
	125	.175538+05	126	-.571541+04	127	.246590+05	128	-.113297+05
	129	.575616+05	130	-.267153+05	131	.128158+05	132	-.328441+05
	133	.525630+04	134	-.270957+04	135	.620140+05	136	-.472404+04
	137	.143733+05	138	.128726+05	139	-.741299+03	140	-.755109+04
	141	.263506+04	142	-.113331+05	143	.615630+04	144	-.127952+05
	145	.584236+05	146	-.122546+06	147	.434690+03	148	-.420220+04
	149	.245216+04	150	-.692361+04	151	.155588+05	152	-.327520+05
	153	.263340+05	154	-.573454+05	155	.240367+04	156	-.511065+04
	163	.494355+05	164	-.741351+05	165	.212025+05	166	-.271653+05
	167	.399164+05	168	-.504558+05	169	.405500+05	170	-.402221+05
	171	.482151+05	172	-.343033+05	173	.176672+05	174	.144019+04
	175	.166830+05	176	-.612570+04	177	.204609+05	178	-.101336+05
	179	.101317+06	180	-.324939+05	182	-.211337+05	184	-.126947+05

LOADS ARE IN POUNDS

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	FF WILKINSON	1 JUNE 65	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.29
Checked	A L FINCHET	1 JUNE 65	TITLE	Model C-5A	
Approved	L C CAFEY	1 JUNE 65	INTERNAL LOADS ANALYSIS	LG1LS46-2-2	
			AFT FUSELAGE	Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION CNLCC1 & MOD 2

1	185	.931576+05	186	-.302701+03	187	.463588+05	188	.145563+05
	189	.282711+05	190	-.199357+04	191	.141719+04	192	-.226523+04
	193	.170752+04	194	-.503941+04	195	.224569+04	196	-.392211+04
	197	.201267+04	198	-.163355+05	199	.705918+04	200	-.110323+05
	201	.714751+04	202	-.205267+05	203	.931618+04	204	-.168910+05
	205	.666426+04	206	-.275316+05	207	.133176+05	208	-.195212+05
	209	.910745+04	210	-.253011+05	211	.120879+05	212	-.193384+05
	213	.925111+04	214	-.215272+05	215	.100718+05	216	-.182671+05
	217	.860725+04	218	.131621-05	219	-.330151-04	220	-.600776-04
	221	.310417-04	222	-.252358-04	223	.717539-04	224	-.406265-03
	225	-.454514-03	226	.313163+05	227	.825236+03	228	-.857772+03
	229	.123018+04	230	-.245027+04	231	.491408+04	232	-.670446+04
	233	.351617+04	234	-.555055+04	235	.610163+04	236	-.117806+05
	237	.445761+04	238	-.101401+05	239	.332374+05	240	-.816290+05
	241	.166472+03	242	-.365308+03	243	.545133+02	244	-.282021+03
	245	.311445+04	246	-.708220+04	247	.542725+04	248	-.116573+05
	249	.558746+04	250	-.138658+05	251	.706085+04	252	-.168573+05
	253	.355633+05	254	-.858543+05	255	.182427+05	256	-.466814+05
	257	.631601+04	258	-.236863+05	259	.553890+04	260	-.146551+05
	261	.100307+05	262	-.191345+05	263	.202795+05	264	-.289258+05
	265	.154612+05	266	-.214803+05	267	.353029+05	268	-.348327+05
	269	.421254+05	270	-.567890+05	271	.488155+05	272	-.328200+05
	273	.836763+04	274	.709134+04	275	.172224+05	276	-.518874+04
	277	.457030+04	278	-.223347+04	279	.233897+05	280	-.105286+05
	281	.696755+05	282	-.323255+05	283	.253624+05	284	-.254640+05
	285	.217359+05	286	-.118058+05	287	.478275+05	288	-.312376+04
	289	.501512+05	290	.137314+05				

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.		PERM.	
Prepared	M BURTON	NOV70		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6.29.1	
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS		C-5A			
Approved	H H WADE	NOV70	AFT FUSELAGE			Model			
						LGIUS46-2-2			
						Report No.			

REV B M.B. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 2

1	1	-.736138+02	2	.453132+03	3	-.111910+04	4	.440890+03
	5	-.199932+04	6	.120980+04	7	-.370723+04	8	.167494+04
	9	-.306399+05	10	.126117+05	11	.168741+01	12	.852510+02
	13	.268752+03	14	-.184654+03	15	-.361308+04	16	.114328+04
	17	-.718009+04	18	.322336+04	19	-.495437+05	20	.216331+05
	21	-.742288+05	22	.353498+05	23	-.926206+05	24	.330243+05
	27	-.264797+05	28	.101608+05	33	-.176558+05	34	.803737+04
	37	-.227138+05	38	.124827+05	45	-.232219+05	46	.171246+05
	49	-.191486+05	50	.155926+05	53	-.362077+05	54	.327813+05
	57	-.383715+05	58	.391658+05	61	-.392347+05	62	.430474+05
	63	.878132+04	64	.136788+04	65	-.879770+04	66	.133575+05
	67	-.711066+04	68	.844093+04	69	-.124131+05	70	.135647+05
	71	-.450979+05	72	.519055+05	73	-.222051+05	74	.493423+05
	75	-.472038+04	76	.466945+05	77	.408018+04	78	.443695+05
	79	.118847+05	80	.402379+05	81	.251415+05	82	.250813+05
	83	-.295779+04	84	.721120+03	85	-.497271+04	86	.256215+04
	87	-.820149+04	88	.491269+04	89	-.124626+05	90	.602138+04
	91	-.111121+06	92	.491963+05	93	-.181571+04	94	.315660+03
	95	-.300685+04	96	.116121+04	97	-.222424+05	98	.111258+05
	99	-.205640+05	100	.879073+04	101	-.738445+05	102	.302356+05
	103	-.244120+05	104	.886998+04	105	-.106219+05	106	.373598+04
	107	-.662766+04	108	.270549+04	109	-.704434+04	110	.437947+04
	113	-.378547+05	114	.306649+05	115	-.223339+05	116	.159373+05
	117	-.476746+05	118	.409737+05	119	-.409898+05	120	.468341+05
	121	-.335278+05	122	.405699+05	123	.369861+04	124	.121866+05
	125	-.536464+04	126	.165848+05	127	-.154117+05	128	.189505+05
	129	-.680548+05	130	.501068+05	131	-.780102+04	132	.568598+05
	133	.220586+03	134	.426659+05	135	.873044+04	136	.459463+05
	137	.348659+04	138	.434734+05	139	-.716483+04	140	-.358578+03
	141	-.109449+05	142	.300958+04	143	-.126694+05	144	.625790+04
	145	-.122817+06	146	.583777+05	147	-.426094+04	148	.360452+03
	149	-.696754+04	150	.237764+04	151	-.328154+05	152	.158332+05
	153	-.576521+05	154	.261700+05	155	-.513998+04	156	.238217+04
	163	-.757523+05	164	.479558+05	165	-.277867+05	166	.203832+05
	167	-.445886+05	168	.442318+05	169	-.380997+05	170	.414691+05
	171	-.366976+05	172	.429272+05	173	.102470+04	174	.168806+05
	175	-.607430+04	176	.182439+05	177	-.114305+05	178	.179057+05
	179	-.519537+05	180	.555370+05	182	.489455+05	184	.525414+05

LOADS ARE IN POUNDS

NAME		DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 629.2
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS	C-5A	
Approved	H H WADE	NOV70	AFT FUSELAGE		Model	LGIUS46-2-2
			Report No.			

REV(B) M.B. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 2

1	185	-.120381+05	186	.444495+05	187	.120659+05	188	.429577+05
	189	.275871+05	190	.143991+04	191	-.198181+04	192	.172370+04
	193	-.226411+04	194	.229889+04	195	-.497306+04	196	.203193+04
	197	-.392211+04	198	.700265+04	199	-.163348+05	200	.716605+04
	201	-.110386+05	202	.934412+04	203	-.208985+05	204	.882305+04
	205	-.169454+05	206	.131714+05	207	-.275932+05	208	.933630+04
	209	-.192586+05	210	.120183+05	211	-.253545+05	212	.935656+04
	213	-.191888+05	214	.100125+05	215	-.215993+05	216	.866330+04
	217	-.181659+05	218	-.157159+05	219	.322804+04	220	.626067+04
	221	-.274016+04	222	.377393+04	223	.287832+04	224	.688796+03
	225	.356289+03	226	.246521+05	227	-.744337+03	228	.921593+03
	229	-.236561+04	230	.136154+04	231	-.674056+04	232	.492032+04
	233	-.604010+04	234	.349653+04	235	-.118589+05	236	.606910+04
	237	-.101571+05	238	.450887+04	239	-.814412+05	240	.334997+05
	241	-.363301+03	242	.179388+03	243	-.260981+03	244	.665427+02
	245	-.705190+04	246	.312634+04	247	-.116087+05	248	.546096+04
	249	-.137608+05	250	.608047+04	251	-.167103+05	252	.718467+04
	253	-.852180+05	254	.364850+05	255	-.463127+05	256	.185056+05
	257	-.232744+05	258	.850645+04	259	-.141825+05	260	.612177+04
	261	-.181644+05	262	.104231+05	263	-.276015+05	264	.205162+05
	265	-.196319+05	266	.159383+05	267	-.407845+05	268	.322839+05
	269	-.413097+05	270	.488409+05	271	-.395696+05	272	.409865+05
	273	.586612+04	274	.654505+04	275	-.781561+04	276	.137024+05
	277	-.333536+04	278	.377980+04	279	-.164533+05	280	.168206+05
	281	-.488914+05	282	.520156+05	283	-.177193+05	284	.505712+05
	285	-.114244+04	286	.468840+05	287	.535431+04	288	.433408+05
	289	.907778+04	290	.421124+05				

LOADS ARE IN POUNDS

Prepared	NAME FF WILKINSON	DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL. I	PERM. 6.30
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LEADS ANALYSIS		Model C-5A		
Approved	L C DAFEY	1 JUN 65	AFT FUSELAGE		Report No.		

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCA.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LEADS CONDITIGN TPMC96 6 MOD 2

1	1	.471815+03	2	.477655+03	3	-.152236+04	4	-.152219+04
	5	-.156600+04	6	-.154315+04	7	-.384730+04	8	-.381619+04
	9	-.338206+05	10	-.337821+05	11	.663003+02	12	.638241+02
	13	.808610+02	14	.695327+02	15	-.465448+04	16	-.472005+04
	17	-.757428+04	18	-.760908+04	19	-.534925+05	20	-.534314+05
	21	-.754715+05	22	-.751722+05	23	-.112112+06	24	-.112419+06
	27	-.306525+05	28	-.311066+05	33	-.184161+05	34	-.189866+05
	37	-.197670+05	38	-.207478+05	45	-.116641+05	46	-.133055+05
	49	-.758458+04	50	-.533262+04	53	-.760445+04	54	-.104756+05
	57	.127922+04	58	-.272831+04	61	.127446+05	62	.902024+04
	63	.233410+05	64	.227218+05	65	.144061+05	66	.134485+05
	67	.681704+04	68	.651251+04	69	.865809+04	70	.840831+04
	71	.259236+05	72	.259715+05	73	.261011+05	74	.434033+05
	75	.432066+05	76	.610051+05	77	.872568+05	78	.739375+05
	79	.106181+06	80	.101866+06	81	.105110+06	82	.100721+06
	83	-.473154+04	84	-.476022+04	85	-.524120+04	86	-.523485+04
	87	-.648825+04	88	-.646526+04	89	-.121601+05	90	-.121465+05
	91	-.115390+06	92	-.115302+06	93	-.281343+04	94	-.282553+04
	95	-.346918+04	96	-.348454+04	97	-.207216+05	98	-.207195+05
	99	-.220110+05	100	-.215507+05	101	-.815091+05	102	-.816358+05
	103	-.287574+05	104	-.288355+05	105	-.128318+05	106	-.129842+05
	107	-.741803+04	108	-.764626+04	109	-.585567+04	110	-.634683+04
	113	-.164352+05	114	-.175235+05	115	-.118357+05	116	-.121986+05
	117	-.161256+05	118	-.168635+05	119	.155570+03	120	-.858545+04
	121	.201242+05	122	.135403+05	123	.255017+05	124	.284302+05
	125	.218567+05	126	.154905+05	127	.167026+05	128	.136412+05
	129	.538410+05	130	.375640+05	131	.918253+04	132	.439691+05
	133	.172575+05	134	.721035+05	135	.127518+06	136	.754149+05
	137	.106311+06	138	.103265+06	139	-.147317+05	140	-.147353+05
	141	-.155406+05	142	-.155556+05	143	-.122668+05	144	-.123014+05
	145	-.120154+06	146	-.120355+06	147	-.715285+04	148	-.717627+04
	149	-.643650+04	150	-.648455+04	151	-.313744+05	152	-.314632+05
	153	-.581155+05	154	-.578155+05	155	-.504913+04	156	-.503205+04
	163	-.491061+05	164	-.487773+05	165	-.124299+05	166	-.127555+05
	167	-.925324+04	168	-.120614+05	169	.366824+04	170	.156838+04
	171	.200720+05	172	.145485+05	173	.333929+05	174	.327575+05
	175	.225220+05	176	.216752+05	177	.161538+05	178	.138716+05
	179	.882058+05	180	.405833+05	182	.501026+05	184	.721553+05

LOADS ARE IN POUNDS

	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. VOL. I	PERM. 6,31
Prepared					Page	
Checked	A L FINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L E DAFNEY	1JUN69			LG10S46-2-2	
					Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCM.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION TYPICAL & MCC 2

1	185	.147977+06	186	.802103+05	187	.106622+06	188	.104993+06
	189	.102146+06	190	-.109840+04	191	-.112692+04	192	-.121975+04
	193	-.124671+04	194	-.515688+04	195	-.514351+04	196	-.360821+04
	197	-.365012+04	198	-.173927+05	199	-.173080+05	200	-.782829+04
	201	-.787966+04	202	-.215437+05	203	-.215632+05	204	-.154287+05
	205	-.154606+05	206	-.266990+05	207	-.265668+05	208	-.191673+05
	209	-.191209+05	210	-.247044+05	211	-.246939+05	212	-.187341+05
	213	-.186695+05	214	-.213910+05	215	-.214291+05	216	-.179103+05
	217	-.178430+05	218	-.556237-06	219	-.116222-05	220	.375046-05
	221	.651588-05	222	-.280875-04	223	.201069-03	224	.525562-03
	225	-.265069-03	226	.102966+06	227	.231052+02	228	-.432065+01
	229	-.234699+04	230	-.233158+04	231	-.366898+04	232	-.380483+04
	233	-.488513+04	234	-.482546+04	235	-.109276+05	236	-.108349+05
	237	-.105332+05	238	-.104792+05	239	-.896010+05	240	-.894136+05
	241	-.403069+03	242	-.413473+03	243	-.426848+03	244	-.433288+03
	245	-.735419+04	246	-.740732+04	247	-.115694+05	248	-.115672+05
	249	-.145196+05	250	-.145276+05	251	-.180449+05	252	-.180683+05
	253	-.526464+05	254	-.527937+05	255	-.526396+05	256	-.527822+05
	257	-.276137+05	258	-.279652+05	259	-.153865+05	260	-.158640+05
	261	-.157574+05	262	-.166877+05	263	-.148761+05	264	-.166331+05
	265	-.850211+04	266	-.107924+05	267	-.111466+05	268	-.640403+04
	269	.601497+03	270	-.144285+05	271	.162551+05	272	.141036+05
	273	.255075+05	274	.244925+05	275	.169080+05	276	.152480+05
	277	.251386+04	278	.271816+04	279	.126040+05	280	.112312+05
	281	.374610+05	282	.345853+05	283	.212314+05	284	.455830+05
	285	.375919+05	286	.640458+05	287	.969124+05	288	.738866+05
	289	.108662+06	290	.102701+06				

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. VOL. I	PERM. 6.32
Prepared					Page	
Checked	A L FINCHET	1 JUN 69	TITLE INTERNAL LEADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUN 69			LG10S46-2-2	
			AFT FLSEELAGE		Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CCA.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPE201+P & MOD 2

1	1	.145216+05	2	.146524+05	3	.136477+05	4	.140467+05
	5	.581815+04	6	.626257+04	7	-.302879+04	8	-.249764+04
	9	-.355625+05	10	-.325423+05	11	.213955+05	12	.213956+05
	13	.210500+05	14	.209942+05	15	.939967+04	16	.954647+04
	17	-.625544+04	18	-.632377+04	19	-.507796+05	20	-.440965+05
	21	-.818550+05	22	-.673127+05	23	-.105932+06	24	-.105793+06
	27	-.246683+05	28	-.257577+05	33	-.135956+05	34	-.153406+05
	37	-.134655+05	38	-.166264+05	45	-.370803+04	46	-.900206+04
	49	-.563419+03	50	-.669811+04	53	.819153+04	54	-.347577+03
	57	.213501+05	58	.112964+05	61	.370080+05	62	.275159+05
	63	.643000+05	64	.677135+05	65	.283162+05	66	.262035+05
	67	.163462+05	68	.148360+05	69	.239285+05	70	.211073+05
	71	.683836+05	72	.894860+05	73	.463825+05	74	.665019+05
	75	.672587+05	76	.880561+05	77	.100734+06	78	.861272+05
	79	.141771+06	80	.133216+06	81	.123986+06	82	.124190+06
	83	.111341+05	84	.116991+05	85	.102574+05	86	.114747+05
	87	-.224931+04	88	-.623462+03	89	-.115472+05	90	-.101364+05
	91	-.106206+06	92	-.969324+05	93	.192578+05	94	.192831+05
	95	.158871+05	96	.159469+05	97	-.414321+04	98	-.250990+04
	99	-.176714+05	100	-.162044+05	101	-.702781+05	102	-.656460+05
	103	-.265815+05	104	-.253281+05	105	-.124838+05	106	-.125601+05
	107	-.666717+04	108	-.730059+04	109	-.406434+04	110	-.551021+04
	113	-.724031+04	114	-.123011+05	115	-.335161+04	116	-.593251+04
	117	.531427+04	118	.728760+03	119	.217972+05	120	.106643+05
	121	.435254+05	122	.290097+05	123	.507143+05	124	.503661+05
	125	.315258+05	126	.281809+05	127	.322630+05	128	.254227+05
	129	.102240+06	130	.613084+05	131	.152440+05	132	.681672+05
	133	.281455+05	134	.643673+05	135	.154138+06	136	.880799+05
	137	.137305+06	138	.128505+06	139	.276409+04	140	.346470+04
	141	.792554+03	142	.221808+04	143	-.755192+04	144	-.605854+04
	145	-.105508+06	146	-.574674+05	147	.133756+05	148	.134788+05
	149	.104927+05	150	.107069+05	151	-.144263+05	152	-.126657+05
	153	-.426022+05	154	-.392210+05	155	-.394342+04	156	-.372624+04
	163	-.299005+05	164	-.327427+05	165	-.316575+04	166	-.525988+04
	167	.692880+04	168	.407788+04	169	.215667+05	170	.156871+05
	171	.433646+05	172	.306944+05	173	.475669+05	174	.467726+05
	175	.346445+05	176	.321628+05	177	.282344+05	178	.227766+05
	179	.141447+06	180	.647686+05	182	.716816+05	184	.921118+05

LOADS ARE IN POUNDS

	NAME FF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.33
Checked	A L FINCHET	1 JUN 69	TITLE INTERNAL LCADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L C CAFEY	1 JUN 69			LCILS46-2-2	
					Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCN.6
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS CONDITION SPE2C1+P 6 MOD 2

1	185	.185307+06	186	.537678+05	187	.125610+06	188	.124827+06
	189	.122700+06	190	.283661+05	191	.281001+05	192	.157060+05
	193	.152410+05	194	-.206355+04	195	-.265648+04	196	-.163185+04
	197	-.225723+04	198	.708365+04	199	.650571+04	200	.597834+04
	201	.359112+04	202	-.130381+05	203	-.145578+05	204	-.590464+04
	205	-.122412+05	206	-.274522+04	207	-.402251+04	208	-.623779+04
	209	-.874441+04	210	-.154006+05	211	-.168720+05	212	-.123853+05
	213	-.143030+05	214	-.147310+05	215	-.155792+05	216	-.133515+05
	217	-.147738+05	218	.410859+05	219	.411661+05	220	.241033+05
	221	.243945+05	222	.503916+05	223	.505450+05	224	.329265+05
	225	.322407+05	226	.128080+06	227	.156723+05	228	.160246+05
	229	.145362+05	230	.154106+05	231	.541906+04	232	.111625+05
	233	.378427+03	234	.156152+04	235	-.755710+04	236	-.619323+04
	237	-.527188+04	238	-.808833+04	239	-.868381+05	240	-.790798+05
	241	.211483+05	242	.211268+05	243	.153858+05	244	.192701+05
	245	.188948+05	246	.151782+05	247	.353153+04	248	.483661+04
	249	-.924536+04	250	-.752641+04	251	-.153899+05	252	-.136271+05
	253	-.885377+05	254	-.785732+05	255	-.546764+05	256	-.516220+05
	257	-.251555+05	258	-.256266+05	259	-.125795+05	260	-.139383+05
	261	-.107677+05	262	-.136514+05	263	-.463572+04	264	-.102149+05
	265	-.718555+03	266	-.556136+04	267	.786605+04	268	.609943+04
	269	.231426+05	270	.601255+04	271	.412532+05	272	.305827+05
	273	.648557+05	274	.655581+05	275	.302348+05	276	.272987+05
	277	.664021+04	278	.581536+04	279	.301046+05	280	.250689+05
	281	.757263+05	282	.612638+05	283	.350067+05	284	.663164+05
	285	.551570+05	286	.511188+05	287	.114204+06	288	.867853+05
	289	.143920+06	290	.132284+06				

LOADS ARE IN POUNDS

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	FF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.34
Checked	A L FINCHET	1JUN69	TITLE		Model C-5A	
			INTERNAL LOADS ANALYSIS		LG1LS46-2-2	
Approved	L C CAFEY	1JUN69	AFT FUSELAGE		Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 2

1	1	.109694+04	2	.971772+03	3	.130171+04	4	.917177+03
	5	.203316+04	6	.154715+04	7	.285062+04	8	.228682+04
	9	.210152+05	10	.176968+05	11	-.355969+03	12	-.321748+03
	13	-.815993+03	14	-.407103+03	15	.134479+04	16	.115601+04
	17	.436876+04	18	.327537+04	19	.386617+05	20	.280843+05
	21	.567530+05	22	.405593+05	23	.691176+05	24	.639065+05
	27	.155231+05	28	.161201+05	33	.958685+04	34	.107462+05
	37	.109488+05	38	.130455+05	45	.890547+04	46	.122202+05
	49	.497488+04	50	.806072+04	53	.282916+04	54	.924737+04
	57	-.308657+04	58	.386960+04	61	-.100053+05	62	-.259196+04
	63	-.401381+04	64	-.899082+04	65	-.620314+04	66	-.388683+04
	67	-.418991+04	68	-.233257+04	69	-.614366+04	70	-.272428+04
	71	-.200355+05	72	-.101014+05	73	-.171602+05	74	-.218812+05
	75	-.241783+05	76	-.329644+05	77	-.482851+05	78	-.459385+05
	79	-.676722+05	80	-.656054+05	81	-.700333+05	82	-.587631+05
	83	.619137+04	84	.851482+04	85	.766666+04	86	.629580+04
	87	.909077+04	88	.755791+04	89	.961688+04	90	.808164+04
	91	.711411+05	92	.609648+05	93	-.330177+03	94	-.390284+03
	95	.329180+04	96	-.729125+02	97	.900382+04	98	.710607+04
	99	.977307+04	100	.802745+04	101	.326646+05	102	.267459+05
	103	.111077+05	104	.951322+04	105	.486400+04	106	.483482+04
	107	.292188+04	108	.328836+04	109	.443051+04	110	.518632+04
	113	.182602+05	114	.213090+05	115	.328180+04	116	.505302+04
	117	-.305746+03	118	.499150+04	119	-.447043+04	120	-.141446+04
	121	-.694918+04	122	-.274304+04	123	-.545780+04	124	-.693480+04
	125	-.503317+04	126	-.423557+04	127	-.666242+04	128	-.321898+04
	129	-.413507+05	130	-.639029+04	131	-.577398+04	132	-.291587+05
	133	-.631225+04	134	-.258210+05	135	-.557077+05	136	-.416096+05
	137	-.645411+05	138	-.574421+05	139	.975293+04	140	.892908+04
	141	.112518+05	142	.957390+04	143	.955546+04	144	.779895+04
	145	.671982+05	146	.575768+05	147	.107118+04	148	.916746+03
	149	.199037+04	150	.168629+04	151	.134726+05	152	.113584+05
	153	.254901+05	154	.218511+05	155	.199732+04	156	.174091+04
	163	.132654+05	164	.165245+05	165	.172420+04	166	.380944+04
	167	-.103593+04	168	.313709+04	169	-.468090+04	170	-.784401+02
	171	-.107922+05	172	-.383916+04	173	-.624687+04	174	-.603374+04
	175	-.501168+04	176	-.385491+04	177	-.705119+04	178	-.339196+04
	179	-.381127+05	180	-.110087+05	182	-.201601+05	184	-.324952+05

LOADS ARE IN POUNDS

Prepared	NAME R F WILKINSON	DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL. I	PERM. 6.35
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A LG10546-2-2		
Approved	L C CAFEY	1 JUN 65			Report No.		

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CEN. 8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION 1PC242 6 MOD 2

1	185	-.7114527+05	186	-.363118+05	187	-.523311+05	188	-.495225+05
	189	-.497225+05	190	-.206774+03	191	.410349+02	192	.202323+04
	193	.273325+04	194	.149772+04	195	.221024+04	196	.264977+04
	197	.334820+04	198	.529527+04	199	.605059+04	200	.111474+05
	201	.137391+05	202	.802525+04	203	.975653+04	204	.123717+05
	205	.149217+05	206	.178389+04	207	.103677+05	208	.123372+05
	209	.152804+05	210	.951089+04	211	.112110+05	212	.110429+05
	213	.133271+05	214	.825365+04	215	.964526+04	216	.914708+04
	217	.109069+05	218	.787013+05	219	-.237792+05	220	-.435859+04
	221	.190453+05	222	-.429117+04	223	-.327576+03	224	-.544169+03
	225	.272828+03	226	-.657072+05	227	.308783+04	228	.271054+04
	229	.380208+04	230	.286922+04	231	.674867+04	232	.676663+04
	233	.595387+04	234	.465722+04	235	.101034+05	236	.817728+04
	237	.709591+04	238	.582955+04	239	.536495+05	240	.450113+05
	241	-.726809+03	242	-.657027+03	243	-.900150+03	244	-.799290+03
	245	.194191+04	246	.159126+04	247	.438080+04	248	.329066+04
	249	.660200+04	250	.500951+04	251	.948937+04	252	.739557+04
	253	.554941+05	254	.445087+05	255	.364592+05	256	.330047+05
	257	.124256+05	258	.125560+05	259	.726418+04	260	.812735+04
	261	.976599+04	262	.115053+05	263	.966537+04	264	.126639+05
	265	.496640+04	266	.690937+04	267	.227515+04	268	.104435+05
	269	-.399078+04	270	-.976989+03	271	-.114776+05	272	-.237928+04
	273	-.538597+04	274	-.841079+04	275	-.668748+04	276	-.454848+04
	277	-.210575+04	278	-.113000+04	279	-.880527+04	280	-.334424+04
	281	-.241506+05	282	-.971418+04	283	-.135233+05	284	-.229501+05
	285	-.192022+05	286	-.321502+05	287	-.494841+05	288	-.428249+05
	289	-.670020+05	290	-.628145+05				

LOADS ARE IN POUNDS

Prepared	NAME V W BOYD	DATE 9 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
					Vol. I	6.36
Checked	J HALLIDAY	9 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A		
Approved	LC DARBY	9 SEPT 68		LGIDS 46-2-2 Report No.		

6.3.1 STRINGER LOADS (CONTINUED)

REV. (A) MB. 1 AUG 69

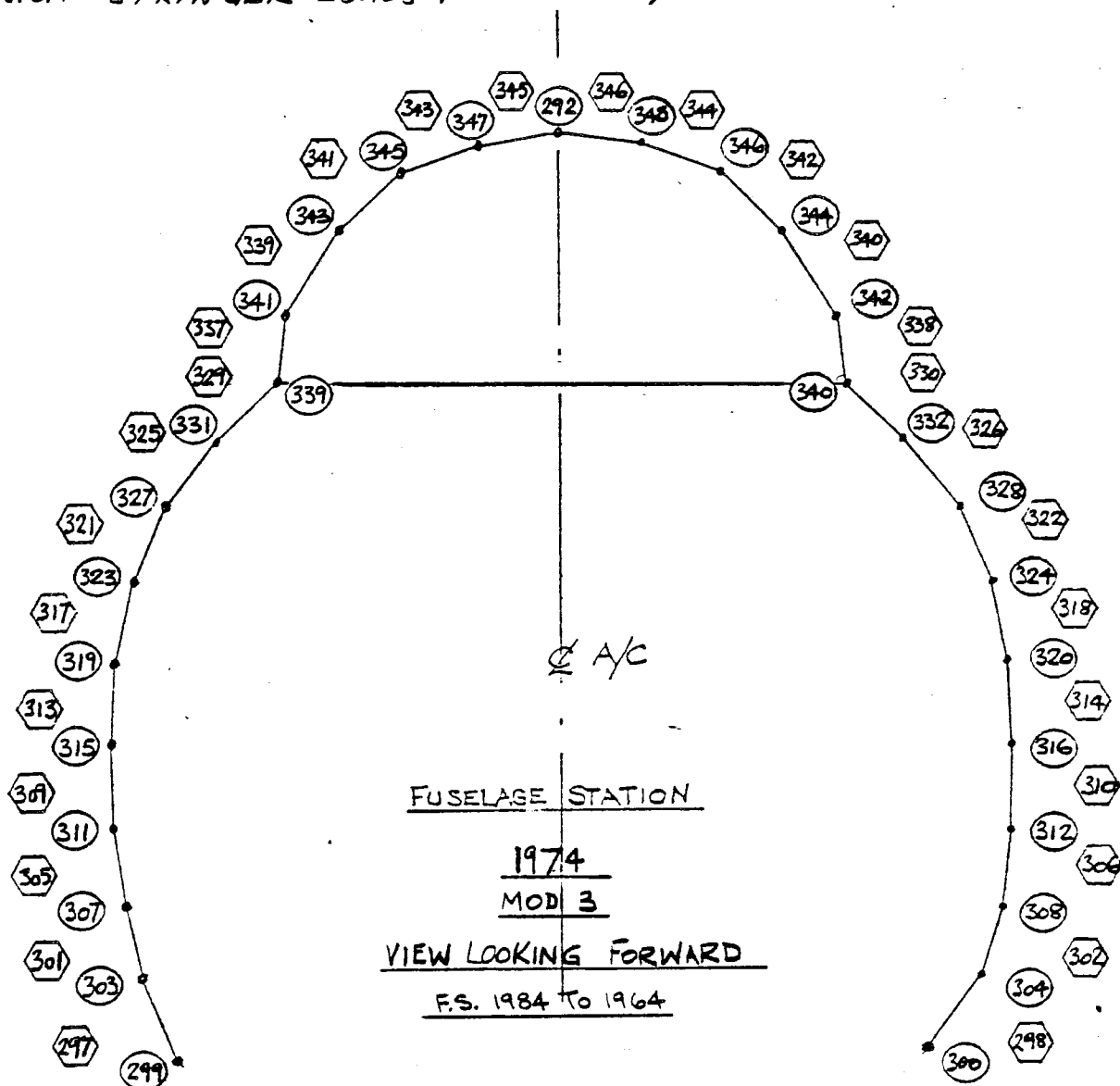


FIGURE No. 6.3.1.10

KEY

GRID POINT

○ FAMES STRINGER

⬡ FAMES PANEL

FOR AREAS AND THICKNESS SEE REF 8.
VOLUME IV SECTION 5.2.

Prepared	NAME K.W. BOYD	DATE 10 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I 6, 37
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	
Approved	L. G. DARBY	10 SEPT 68		Report No. LGIUS 46-2-2	

6.3.1 STRINGER LOADS (CONTINUED)

REV A M.B. 1 AUG 69

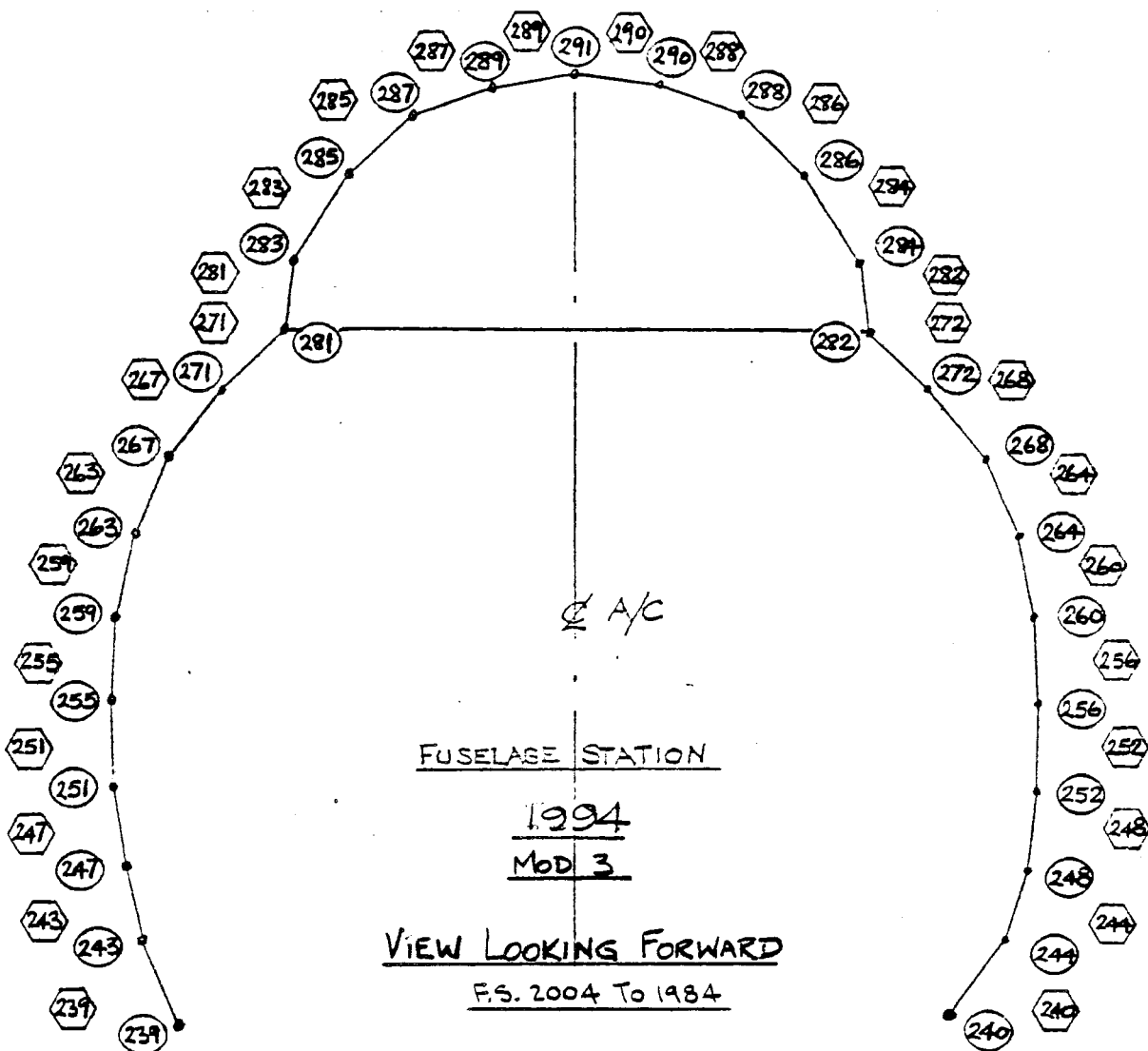


FIGURE No. 6.3.1.11

KEY

—●— GRID POINT

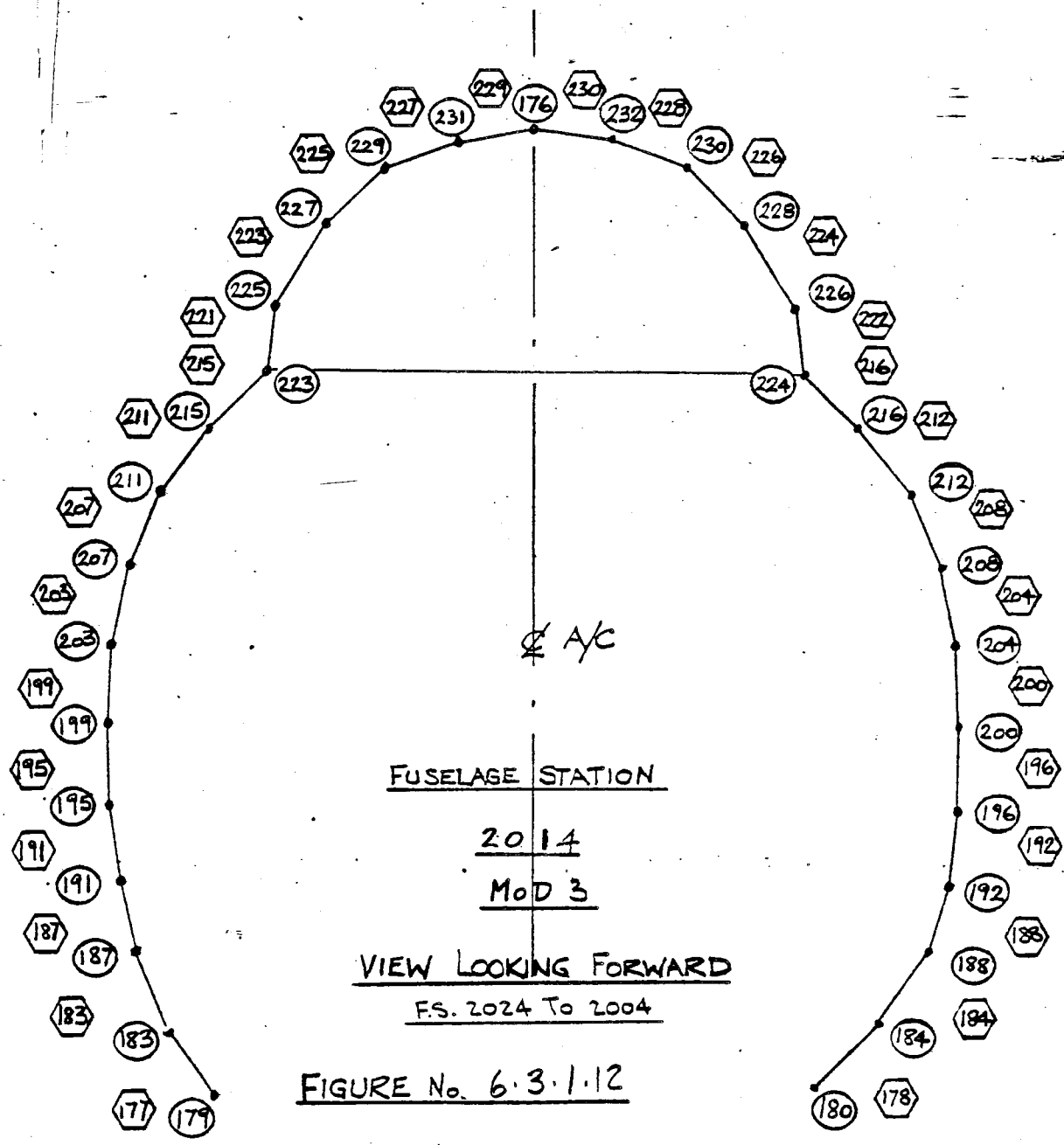
○ FAMES STRINGER

⬡ FAMES PANEL

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 52

6.3.1 STRINGER LOADS (CONTINUED)

REV A MB 1 AUG 69



KEY

GRID POINT

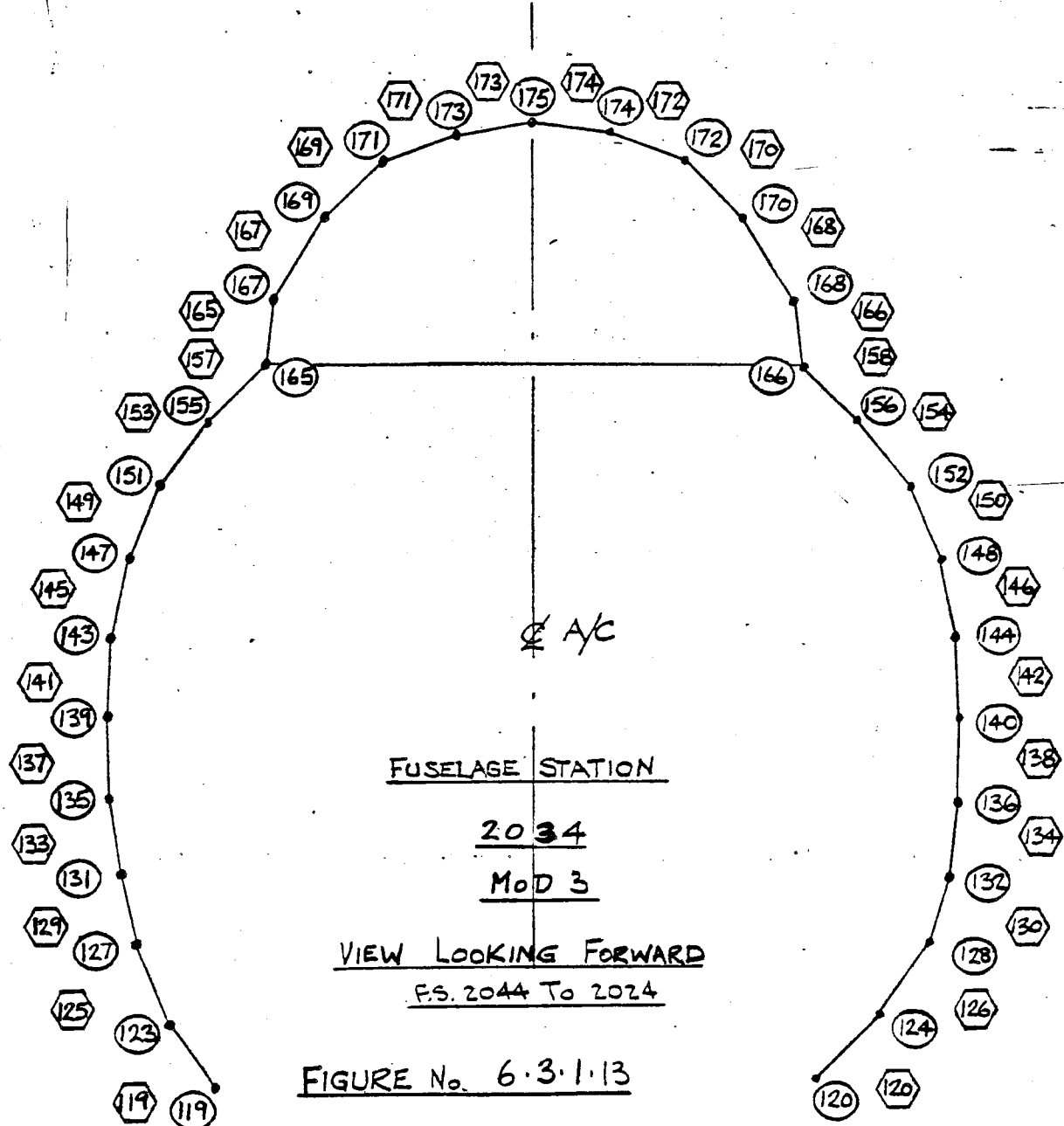
FAMAS STRINGER

FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 3.2

REV. A MB. 1 ALY 69

6.3.1 STRINGER LOADS (CONTINUED)



KEY

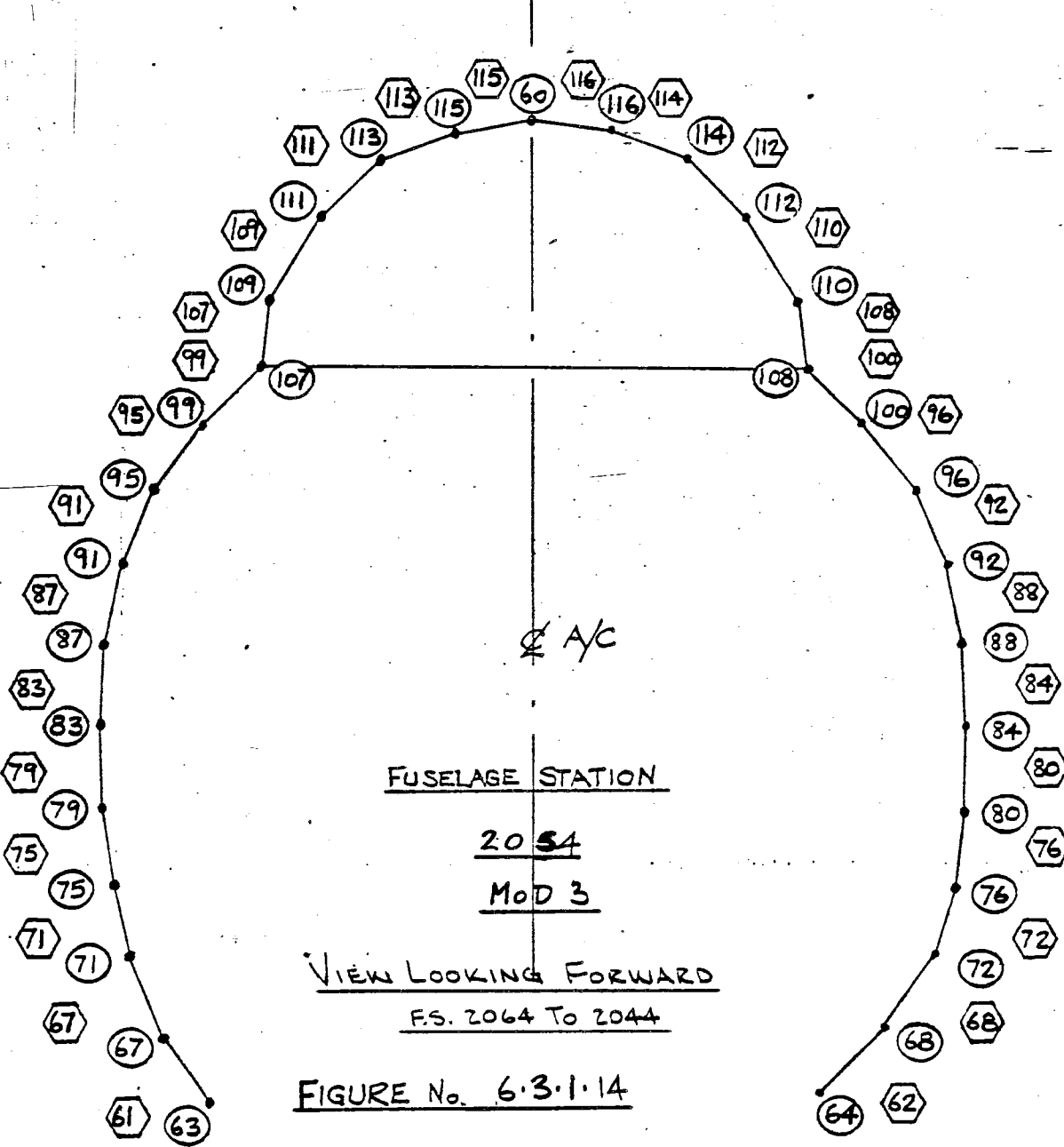

GRID POINT


FAMAS STRINGER


FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF8
VOLUME IV SECTION 5.2

6.3.1 STRINGER LOADS (CONTINUED)
REV. (A) MB 1 AUG 69



KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF8
VOLUME IV SECTION 5.2

Prepared	NAME V.W. BOYD	DATE 10 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 6.41
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L. G. DARBY	10 SEPT 68		Report No.	LG105 46-2-2	

REV. (A) M.B. 1 AUG 69.

6.3.1 STRINGER LOADS (CON'T)

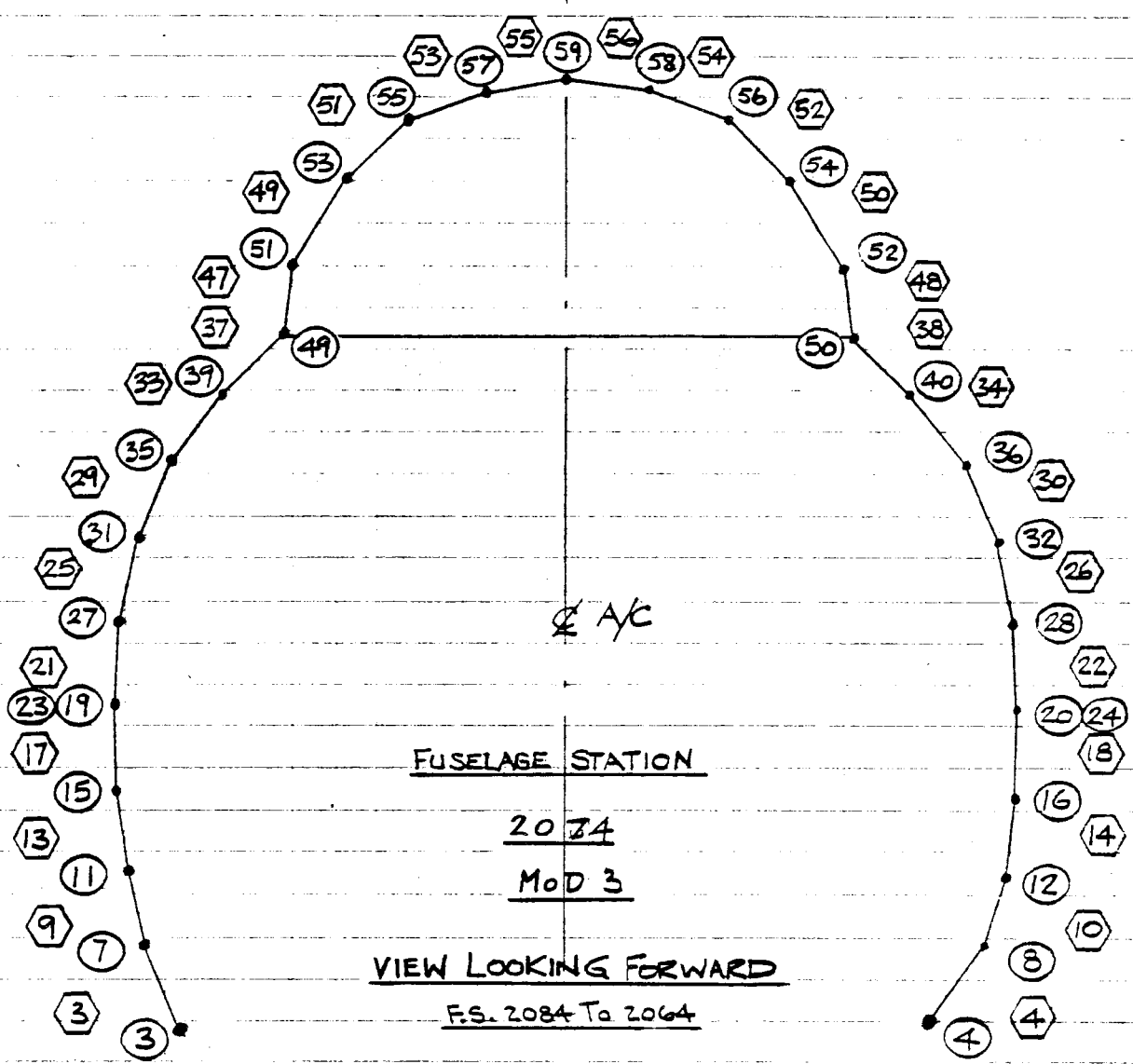


FIGURE No. 6.3.1.15

KEY

 GRID POINT

 FAMAS STRINGER

 FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2.

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	FF WILKINSON	1JUN65	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.42
Checked	A L FINCHET	1JUN65	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1JUN65			LC1LS46-2-2	
			AFT FLSELAGE		Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION CML001 6 NCC 3

1	.405212+04	2	-.118925+05	*3	.381388+04	4	-.866619+04
5	.512009+05	6	-.147131+06	7	.367801+05	8	-.802480+05
9	.101549+04	10	-.318440+04	11	.140805+05	12	-.282502+05
13	.253705+03	14	-.111039+04	15	.124799+05	16	-.220370+05
17	.143999+02	18	.232193+03	19	.502895+04	20	-.793690+04
23	.106773+05	24	-.159166+05	25	-.866219+02	26	-.166627+03
27	.122003+05	28	-.160312+05	29	-.555591+02	30	.239444+03
31	.202950+05	32	-.232723+05	33	-.234435+02	34	.173663+03
35	.236326+05	36	-.237531+05	37	.885602+03	38	-.477689+03
39	.246107+05	40	-.213497+05	41	.153171+05	42	-.937970+04
43	.146205+05	44	-.113028+05	45	.120288+05	46	-.935672+04
47	.126244+05	48	-.840383+04	49	.450306+05	50	-.337009+05
51	.444490+05	52	-.255430+05	53	.397460+05	54	-.131710+05
55	.358820+05	56	-.556882+03	57	.385351+05	58	.169197+05
59	.325444+05	60	.327476+05	61	.582736+04	62	-.156838+05
63	.339704+04	64	-.797149+04	65	.517898+05	66	-.146842+06
67	.378207+05	68	-.836268+05	71	.134170+05	72	-.263849+05
75	.116583+05	76	-.205765+05	79	.656223+04	80	-.134546+05
83	.105875+05	84	-.153164+05	87	.127687+05	88	-.158991+05
91	.216184+05	92	-.240088+05	95	.255969+05	96	-.249008+05
99	.289787+05	100	-.249147+05	101	.141653+05	102	-.739643+04
103	.128877+05	104	-.542613+04	105	.111718+05	106	-.675120+04
107	.450767+05	108	-.332577+05	109	.443446+05	110	-.262590+05
111	.388674+05	112	-.138464+05	113	.389348+05	114	-.145668+04
115	.383223+05	116	.170287+05	117	.684922+04	118	-.176306+05
119	.340770+04	120	-.766288+04	121	.539778+05	122	-.150551+06
123	.384738+05	124	-.893145+05	127	.121464+05	128	-.245629+05
131	.111864+05	132	-.200433+05	135	.102448+05	136	-.163988+05
139	.111905+05	140	-.192695+05	143	.138918+05	144	-.170871+05
147	.230847+05	148	-.250401+05	151	.277165+05	152	-.263287+05
155	.340091+05	156	-.290104+05	157	.189493+05	158	-.796724+04
159	.176045+05	160	-.117083+05	161	.135529+05	162	-.940490+04
163	.114576+05	164	-.607812+04	165	.459571+05	166	-.331955+05
167	.436139+05	168	-.264242+05	169	.384393+05	170	-.150046+05
171	.384173+05	172	-.200030+04	173	.383622+05	174	.171748+05
175	.329825+05	176	.329388+05	177	.771551+04	178	-.178594+05
179	.288777+04	180	-.779745+04	181	.564854+05	182	-.150134+06
183	.396146+05	184	-.579790+05	187	.106500+05	188	-.223714+05

* STRINGER (3) REF. P.6.41 ULTIMATE TENSION LOAD = 3814. LB.

	NAME RF WILKINSON	DATE 1 JUNE 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.43
Prepared					Page	
Checked	A L FINCHET	1 JUNE 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUNE 69	AFT FUSELAGE		LG10S46-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON. 8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION DMLOC1 6 MOD 3

1	191	.106779+05	192	-.196115+05	195	.111056+05	196	-.182323+05
	195	.123176+05	200	-.166310+05	203	.148706+05	204	-.180526+05
	207	.248888+05	208	-.271473+05	211	.300382+05	212	-.276847+05
	215	.390304+05	216	-.332199+05	217	.141573+05	218	-.439212+04
	219	.117452+05	220	-.606720+04	221	.132196+05	222	-.885000+04
	223	.477688+05	224	-.338118+05	225	.430465+05	226	-.266444+05
	227	.374215+05	228	-.156532+05	229	.362760+05	230	-.241001+04
	231	.391573+05	232	.170383+05	237	.685442+05	238	-.168190+06
	239	.420946+05	240	-.111671+06	243	.972316+04	244	-.220471+05
	247	.103841+05	248	-.203436+05	251	.114951+05	252	-.200441+05
	255	.137195+05	256	-.165077+05	259	.155558+05	260	-.189101+05
	263	.270568+05	264	-.316319+05	267	.329801+05	268	-.284536+05
	271	.434213+05	272	-.370217+05	273	.760601+04	274	.672554+04
	275	.204770+05	276	-.114727+05	277	.113075+05	278	-.550633+04
	279	.141024+05	280	-.683288+04	281	.501500+05	282	-.347084+05
	283	.418790+05	284	-.264967+05	285	.355200+05	286	-.152488+05
	287	.392501+05	288	-.331819+04	289	.407093+05	290	.164363+05
	291	.327313+05	292	.325215+05	297	.707961+05	298	-.160275+06
	299	.416315+05	300	-.119439+06	303	.101330+05	304	-.260205+05
	307	.979115+04	308	-.212629+05	311	.124957+05	312	-.229414+05
	315	.156711+05	316	-.206385+05	319	.156435+05	320	-.182972+05
	323	.299137+05	324	-.404851+05	327	.364040+05	328	-.269693+05
	331	.482795+05	332	-.404773+05	333	.150132+05	334	-.373523+04
	335	.113951+05	336	-.509381+04	337	.151290+05	338	-.882280+04
	339	.544539+05	340	-.358563+05	341	.401683+05	342	-.259187+05
	343	.330970+05	344	-.144268+05	345	.410082+05	346	-.448817+04
	347	.425815+05	348	.158477+05	349	.768207+03	350	-.137787+04
	351	.153284+04	352	-.368108+04	353	.140568+05	354	-.194842+05
	355	.169013+04	356	-.430901+04	357	.136312+04	358	-.320852+04
	359	-.216878+02	360	-.329998+03	361	.479088+03	362	-.137395+04

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.		PERM.	
Prepared	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6.43.1		
Checked	A L PINCHOT	NOV70	TITLE		INTERNAL LOADS ANALYSIS		C-5A		
Approved	H H WADE	NOV70			AFT FUSELAGE		Model		LGIUS46-2-2
						Report No.			

REV (E) M.S. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
6.3 SKINS AND STRINGERS
6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 3

1	1	-0.117769+05	2	0.417659+04	3	-0.849768+04	4	0.388497+04
	5	-0.145678+06	6	0.524481+05	7	-0.787961+05	8	0.372027+05
	9	-0.315297+04	10	0.103769+04	11	-0.277457+05	12	0.141258+05
	13	-0.110232+04	14	0.261580+03	15	-0.216977+05	16	0.124231+05
	17	0.208664+03	18	-0.205476+01	19	-0.784489+04	20	0.497894+04
	23	-0.156901+05	24	0.104817+05	25	-0.158593+03	26	-0.774762+02
	27	-0.159291+05	28	0.118937+05	29	0.224532+03	30	-0.664385+02
	31	-0.233089+05	32	0.196749+05	33	-0.174166+03	34	-0.270800+02
	35	-0.241423+05	36	0.228847+05	37	-0.537906+03	38	0.840536+03
	39	-0.220485+05	40	0.238877+05	41	-0.114579+05	42	0.131847+05
	43	-0.125100+05	44	0.134759+05	45	-0.101298+05	46	0.113698+05
	47	-0.930661+04	48	0.118665+05	49	-0.353670+05	50	0.439649+05
	51	-0.258320+05	52	0.458614+05	53	-0.120936+05	54	0.425632+05
	55	0.597257+03	56	0.422965+05	57	0.173095+05	58	0.395407+05
	59	0.320503+05	60	0.312101+05	61	-0.156493+05	62	0.593247+04
	63	-0.774737+04	64	0.346040+04	65	-0.145543+06	66	0.530550+05
	67	-0.820085+05	68	0.384418+05	71	-0.260043+05	72	0.134396+05
	75	-0.202647+05	76	0.116039+05	79	-0.132375+05	80	0.845751+04
	83	-0.150485+05	84	0.103643+05	87	-0.157068+05	88	0.124408+05
	91	-0.239572+05	92	0.209725+05	95	-0.254128+05	96	0.246626+05
	99	-0.258247+05	100	0.279808+05	101	-0.925386+04	102	0.123302+05
	103	-0.103135+05	104	0.121287+05	105	-0.780749+04	106	0.102373+05
	107	-0.350958+05	108	0.439215+05	109	-0.262812+05	110	0.465134+05
	111	-0.120246+05	112	0.428163+05	113	0.914877+03	114	0.421144+05
	115	0.172624+05	116	0.388383+05	117	-0.175763+05	118	0.701100+04
	119	-0.755741+04	120	0.347243+04	121	-0.149433+06	122	0.551950+05
	123	-0.874961+05	124	0.393619+05	127	-0.242523+05	128	0.122247+05
	131	-0.197002+05	132	0.111090+05	135	-0.160738+05	136	0.101180+05
	139	-0.149199+05	140	0.109332+05	143	-0.167100+05	144	0.135179+05
	147	-0.247383+05	148	0.224514+05	151	-0.270307+05	152	0.265763+05
	155	-0.301555+05	156	0.327026+05	157	-0.106600+05	158	0.156904+05
	159	-0.136038+05	160	0.157364+05	161	-0.104271+05	162	0.126760+05
	163	-0.742507+04	164	0.101990+05	165	-0.354045+05	166	0.445065+05
	167	-0.261641+05	168	0.467023+05	169	-0.123105+05	170	0.438954+05
	171	0.139989+04	172	0.421419+05	173	0.171435+05	174	0.381497+05
	175	0.303025+05	176	0.290572+05	177	-0.179645+05	178	0.787543+04
	179	-0.763167+04	180	0.297291+04	181	-0.148854+06	182	0.575497+05
	183	-0.964929+05	184	0.407683+05	187	-0.218246+05	188	0.107461+05

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.		PERM.	
Prepared	M BURTON	NOV70		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6:43.2	
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS		C-5A			
Approved	H H WADE	NOV70	AFT FUSELAGE			Model	LGIUS46-2-2		
						Report No.			

REV. (B) M.S. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 3

1	191	-.191767+05	192	.105955+05	195	-.178670+05	196	.109479+05
	199	-.162092+05	200	.120267+05	203	-.174525+05	204	.145707+05
	207	-.260908+05	208	.245113+05	211	-.289879+05	212	.284979+05
	215	-.344927+05	216	.373982+05	217	-.696301+04	218	.115328+05
	219	-.761446+04	220	.102496+05	221	-.100457+05	222	.121336+05
	223	-.365978+05	224	.457812+05	225	-.261359+05	226	.472881+05
	227	-.119182+05	228	.448418+05	229	.221414+04	230	.423948+05
	231	.165769+05	232	.379976+05	237	-.167104+06	238	.696889+05
	239	-.110210+06	240	.433986+05	243	-.214261+05	244	.985084+04
	247	-.198205+05	248	.103825+05	251	-.196138+05	252	.113710+05
	255	-.182585+05	256	.132651+05	259	-.181739+05	260	.153018+05
	263	-.284001+05	264	.276292+05	267	-.316529+05	268	.304299+05
	271	-.379013+05	272	.416308+05	273	.377654+04	274	.463859+04
	275	-.142877+05	276	.175421+05	277	-.722886+04	278	.959602+04
	279	-.104946+05	280	.125134+05	281	-.383688+05	282	.471801+05
	283	-.257097+05	284	.476997+05	285	-.101756+05	286	.454132+05
	287	.266297+04	288	.434049+05	289	.153786+05	290	.384848+05
	291	.275806+05	292	.261587+05	297	-.159470+06	298	.719163+05
	299	-.117942+06	300	.428341+05	303	-.255506+05	304	.103098+05
	307	-.207231+05	308	.988722+04	311	-.223344+05	312	.124852+05
	315	-.205366+05	316	.150579+05	319	-.188373+05	320	.148879+05
	323	-.319814+05	324	.329351+05	327	-.349485+05	328	.314453+05
	331	-.390314+05	332	.442362+05	333	-.673958+04	334	.117477+05
	335	-.720752+04	336	.923594+04	337	-.111342+05	338	.128573+05
	339	-.413936+05	340	.494280+05	341	-.246549+05	342	.482697+05
	343	-.787480+04	344	.462762+05	345	.304205+04	346	.445757+05
	347	.139375+05	348	.390418+05	349	-.135255+04	350	.798420+03
	351	-.354230+04	352	.157368+04	353	-.163880+05	354	.171956+05
	355	-.412547+04	356	.170837+04	357	-.341128+04	358	.137254+04
	359	-.144273+03	360	.103565+01	361	-.147867+04	362	.487615+03

LOADS ARE IN POUNDS

	NAME FF WILKINSON	DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.44
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L C CAFEY	1 JUN 65			LG 10546-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TC 2101 CCM.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPEC1+P 6 MOD 3

1	1	-.135346+05	2	-.122756+05	3	-.101611+05	4	-.975462+04
	5	-.163786+06	6	-.173324+06	7	-.815264+05	8	-.791878+05
	9	-.408889+04	10	-.390121+04	11	-.289834+05	12	-.292954+05
	13	-.115703+04	14	-.115582+04	15	-.175320+05	16	-.183678+05
	17	-.260506+03	18	-.175309+03	19	-.525840+04	20	-.555451+04
	23	-.587963+04	24	-.800760+04	25	.407429+03	26	.344523+03
	27	-.442920+04	28	-.688973+04	29	.460014+03	30	.527515+03
	31	.173176+03	32	-.351859+04	33	-.113780+04	34	-.113115+04
	35	.555586+04	36	.155761+04	37	.161531+04	38	.156558+04
	39	.128563+05	40	.995134+04	41	.634652+05	42	.606589+05
	43	.435668+05	44	.411047+05	45	.256564+05	46	.243674+05
	47	.165746+05	48	.171456+05	49	.376077+05	50	.364381+05
	51	.506946+05	52	.512812+05	53	.716587+05	54	.725277+05
	55	.100522+06	56	.100658+06	57	.138901+06	58	.138578+06
	59	.159242+06	60	.157436+06	61	-.175789+05	62	-.157616+05
	63	-.626046+04	64	-.785745+04	65	-.182584+06	66	-.170509+06
	67	-.883306+05	68	-.845266+05	71	-.260554+05	72	-.260199+05
	75	-.150404+05	76	-.156513+05	79	-.670636+04	80	-.781670+04
	83	-.476322+04	84	-.734014+04	87	-.154067+04	88	-.453677+04
	91	.351379+04	92	-.128100+04	95	.988244+04	96	.514248+04
	99	.185583+05	100	.148540+05	101	.353836+05	102	.343556+05
	103	.239566+05	104	.228819+05	105	.157827+05	106	.156844+05
	107	.414103+05	108	.356186+05	109	.538207+05	110	.555608+05
	111	.705861+05	112	.732753+05	113	.986112+05	114	.988042+05
	115	.137863+06	116	.136654+06	117	-.201857+05	118	-.178292+05
	119	-.901557+04	120	-.637525+04	121	-.182497+06	122	-.167516+06
	123	-.971486+05	124	-.929162+05	127	-.208403+05	128	-.204710+05
	131	-.135558+05	132	-.148026+05	135	-.802673+04	136	-.981630+04
	139	-.253462+04	140	-.575258+04	143	.144354+02	144	-.384961+04
	147	.553355+04	148	-.662389+02	151	.126962+05	152	.746525+04
	155	.240857+05	156	.156018+05	157	.501021+05	158	.468401+05
	159	.422865+05	160	.396957+05	161	.225231+05	162	.215765+05
	163	.152082+05	164	.150632+05	165	.460871+05	166	.436981+05
	167	.883013+05	168	.593938+05	169	.658783+05	170	.744116+05
	171	.168196+05	172	.565429+05	173	.137396+06	174	.135207+06
	175	.155464+06	176	.151656+06	177	-.235107+05	178	-.205500+05
	179	-.814453+04	180	-.752421+04	181	-.183214+06	182	-.167660+06
	183	-.103373+06	184	-.561526+05	187	-.188673+05	188	-.198169+05

LOADS ARE IN POUNDS

	NAME R F WILKINSON	DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.45
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUN 65	AFT FUSELAGE		LC 10546-2-2 Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPE201+P 6 MCC 3

1	191	-.125920+05	192	-.145830+05	195	-.765192+04	196	-.101722+05
	195	-.125157+04	200	-.527166+04	203	.131252+04	204	-.329044+04
	207	.788155+04	208	.104112+04	211	.145542+05	212	.943655+04
	215	.282224+05	216	.225016+05	217	.333469+05	218	.333419+05
	219	.174421+05	220	.166541+05	221	.195591+05	222	.184223+05
	223	.507873+05	224	.476516+05	225	.577795+05	226	.629177+05
	227	.691349+05	228	.762731+05	229	.556510+05	230	.942009+05
	231	.137467+06	232	.134057+06	237	-.204305+06	238	-.182075+06
	239	-.123912+06	240	-.115799+06	243	-.155869+05	244	-.176029+05
	247	-.108110+05	248	-.132515+05	251	-.702526+04	252	-.100508+05
	255	-.160191+04	256	-.621665+04	259	.248603+04	260	-.270248+04
	263	.925456+04	264	.142540+03	267	.170069+05	268	.121272+05
	271	.305555+05	272	.242134+05	273	.462593+05	274	.484388+05
	275	.311950+05	276	.280128+05	277	.176686+05	278	.169548+05
	279	.196244+05	280	.181099+05	281	.542436+05	282	.499753+05
	283	.573870+05	284	.654071+05	285	.702468+05	286	.801065+05
	287	.545156+05	288	.511162+05	289	.137525+06	290	.133463+06
	291	.147186+06	292	.141825+06	297	-.182005+06	298	-.154782+06
	299	-.141767+06	300	-.135542+06	303	-.155573+05	304	-.211589+05
	307	-.126021+05	308	-.149404+05	311	-.102281+05	312	-.136297+05
	315	-.250228+04	316	-.748580+04	319	.541367+03	320	-.357312+04
	323	.686337+04	324	-.483436+04	327	.189068+05	328	.160051+05
	331	.327466+05	332	.242335+05	333	.282126+05	334	.274724+05
	335	.173000+05	336	.162277+05	337	.210424+05	338	.190418+05
	339	.554661+05	340	.532757+05	341	.547192+05	342	.669683+05
	343	.715185+05	344	.853505+05	345	.928849+05	346	.860400+05
	347	.139592+06	348	.133757+06	349	.685534+03	350	.106017+04
	351	-.223185+04	352	-.240563+04	353	.126189-04	354	.120138-04
	355	-.651457+04	356	-.652484+04	357	-.141646+04	358	-.623715+03
	359	-.115061+04	360	-.153317+04	361	.605653+03	362	.106111+04

LOADS ARE IN POUNDS

NAME FF WILKINSON		DATE 1JUN65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.46
Checked	A L FINCHET	1JUN65	TITLE INTERNAL LCADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1JUN65	AFT FUSELAGE		LC1US46-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LCADS CCNDITION 1PC242 & MCD 3

1	1	.557483+04	2	.820503+04	3	.750403+04	4	.686515+04
	5	.123013+06	6	.110346+06	7	.650000+05	8	.598524+05
	9	.248555+04	10	.225573+04	11	.196129+05	12	.191554+05
	13	.726477+03	14	.690676+03	15	.134329+05	16	.135547+05
	17	-.311613+03	18	-.405050+03	19	.385747+04	20	.393064+04
	23	.811543+04	24	.570663+04	25	.155674+03	26	.286004+03
	27	.711011+04	28	.511557+04	29	-.158739+03	30	-.256159+03
	31	.611517+04	32	.565668+04	33	-.341573+03	34	-.307565+03
	35	.294827+04	36	.683676+04	37	.767388+02	38	.170619+03
	39	-.870888+03	40	.233734+04	41	-.171051+04	42	.175652+04
	43	-.970052+02	44	.337300+04	45	-.706400+03	46	.951650+03
	47	-.151113+04	48	-.142302+04	49	-.753353+04	50	-.536891+04
	51	-.242758+05	52	-.227743+05	53	-.385254+05	54	-.384313+05
	55	-.576655+05	56	-.578806+05	57	-.757847+05	58	-.800562+05
	59	-.511740+05	60	-.880519+05	61	.128345+05	62	.105451+05
	63	.682779+04	64	.603666+04	65	.120615+06	66	.106183+06
	67	.665135+05	68	.559535+05	71	.184840+05	72	.177150+05
	75	.122657+05	76	.124355+05	79	.645255+04	80	.703143+04
	83	.754750+04	84	.546355+04	87	.625512+04	88	.868268+04
	91	.535026+04	92	.544845+04	95	.210088+04	96	.671104+04
	99	-.250177+04	100	.148423+04	101	-.143409+04	102	-.627552+02
	103	-.182854+04	104	-.333540+03	105	-.212911+04	106	-.184102+04
	107	-.651865+04	108	-.586044+04	109	-.236770+05	110	-.227293+05
	111	-.374625+05	112	-.378182+05	113	-.557292+05	114	-.563514+05
	115	-.774614+05	116	-.776442+05	117	.142845+05	118	.118548+05
	119	.616829+04	120	.528635+04	121	.121490+06	122	.105234+06
	123	.665458+05	124	.609355+05	127	.165117+05	128	.161478+05
	131	.116588+05	132	.121656+05	135	.777631+04	136	.880492+04
	139	.701954+04	140	.542170+04	143	.588740+04	144	.881480+04
	147	.470755+04	148	.525450+04	151	.132887+04	152	.650069+04
	155	-.385554+04	156	.515158+03	157	-.357582+04	158	.410344+03
	159	-.301735+04	160	.286388+03	161	-.210566+04	162	-.704272+03
	163	-.272110+04	164	-.235535+04	165	-.107570+05	166	-.689710+04
	167	-.233180+05	168	-.226416+05	169	-.354043+05	170	-.371209+05
	171	-.536305+05	172	-.544731+05	173	-.750122+05	174	-.746587+05
	175	-.852011+05	176	-.617630+05	177	.144467+05	178	.113653+05
	179	.561748+04	180	.522627+04	181	.120025+06	182	.101467+06
	183	.722502+05	184	.633625+05	187	.147793+05	188	.145600+05

LOADS ARE IN POUNDS

	NAME	DATE	LOCKHEED-GEORGIA COMPANY			TEMP.	PERM.
Prepared	FF WILKINSON	1JUN65	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6.47
Checked	A L FINCHET	1JUN65	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A		
Approved	L C CAFEY	1JUN65			LG1US46-2-2		
					Report No.		

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CEN.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION IPC242 & MOD 3

1	151	.111452+05	152	.121138+05	155	.646891+04	156	.999582+04
	155	.704914+04	200	.596934+04	203	.566602+04	204	.898162+04
	207	.413807+04	208	.510542+04	211	.471642+03	212	.625006+04
	215	-.533014+04	216	.215332+03	217	-.338364+04	218	-.323844+04
	219	-.335587+04	220	-.249307+04	221	-.258515+04	222	-.137978+04
	223	-.126237+05	224	-.758245+04	225	-.227771+05	226	-.226275+05
	227	-.331546+05	228	-.382556+05	229	-.517127+05	230	-.525066+05
	231	-.727680+05	232	-.723629+05	237	.133095+06	238	.107319+06
	239	.804456+05	240	.713146+05	243	.144341+05	244	.147830+05
	247	.113720+05	248	.127265+05	251	.522428+04	252	.111516+05
	255	.745302+04	256	.108825+05	259	.553517+04	260	.922520+04
	263	.363791+04	264	.870746+04	267	-.470701+03	268	.643471+04
	271	-.683604+04	272	-.853808+03	273	-.244557+04	274	-.535523+04
	275	-.523800+04	276	-.109198+04	277	-.382317+04	279	-.282873+04
	279	-.385438+04	280	-.185325+04	281	-.146701+05	282	-.898882+04
	283	-.215373+05	284	-.225450+05	285	-.305837+05	296	-.351812+05
	287	-.455855+05	288	-.507011+05	289	-.707103+05	290	-.702977+05
	291	-.775835+05	292	-.741302+05	297	.123810+06	298	.931974+05
	299	.874877+05	300	.802243+05	303	.157309+05	304	.165367+05
	307	.112655+05	308	.127402+05	311	.106485+05	312	.126793+05
	315	.834057+04	316	.119700+05	319	.512464+04	320	.923057+04
	323	.317784+04	324	.754307+04	327	-.176124+04	328	.715351+04
	331	-.831514+04	332	-.227514+04	333	-.584937+04	334	-.488641+04
	335	-.436444+04	336	-.251251+04	337	-.475814+04	338	-.219531+04
	339	-.172831+05	340	-.558672+04	341	-.202740+05	342	-.220175+05
	343	-.280838+05	344	-.338918+05	345	-.483763+05	346	-.485510+05
	347	-.687162+05	348	-.680137+05	349	.143141+04	350	.118783+04
	351	.286529+04	352	.271172+04	353	.212587-05	354	.260355-05
	355	.285824+04	356	.280518+04	357	.261583+04	358	.234822+04
	359	.162204+02	360	.667430+02	361	.118255+04	362	.105146+04

LOADS ARE IN POUNDS

NAME FF WILKINSON		DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.43
Prepared					Page	
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUN 65	AFT FUSELAGE		LG1US46-2-2 Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CCN.8

6.2 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPE201

6 MOD 3

1	1	-.155151+05	2	-.147520+05	3	-.107809+05	4	-.104120+05
	5	-.193505+06	6	-.184040+06	7	-.935182+05	8	-.909247+05
	9	-.423136+04	10	-.406251+04	11	-.296054+05	12	-.298138+05
	13	-.154280+04	14	-.152053+04	15	-.155882+05	16	-.207644+05
	17	.417571+03	18	.519933+03	19	-.604566+04	20	-.629254+04
	23	-.111557+05	24	-.131557+05	25	-.405137+03	26	-.483757+03
	27	-.661758+04	28	-.111272+05	29	.307607+03	30	.371677+03
	31	-.732855+04	32	-.111391+05	33	-.123077+03	34	-.145675+03
	35	-.225373+04	36	-.604736+04	37	.422665+03	38	.366303+03
	39	.375752+04	40	.101169+04	41	.539032+04	42	.275562+04
	43	.273934+04	44	.175841+02	45	.271847+04	46	.150955+04
	47	.470961+04	48	.453277+04	49	.164549+05	50	.152392+05
	51	.373725+05	52	.373329+05	53	.573831+05	54	.577104+05
	55	.642713+05	56	.641331+05	57	.116721+06	58	.116449+06
	59	.133855+06	60	.131183+06	61	-.202581+05	62	-.186561+05
	63	-.598021+04	64	-.951434+04	65	-.151323+06	66	-.179647+06
	67	-.583548+05	68	-.945874+05	71	-.274515+05	72	-.274240+05
	75	-.184263+05	76	-.151772+05	79	-.951605+04	80	-.108120+05
	83	-.100595+05	84	-.124565+05	87	-.722130+04	88	-.100077+05
	91	-.580650+04	92	-.102537+05	95	-.756254+03	96	-.519986+04
	99	.547375+04	100	.200023+04	101	.674516+04	102	.583635+04
	103	.411229+04	104	.311235+04	105	.526408+04	106	.525211+04
	107	.176368+05	108	.158355+05	109	.363354+05	110	.373447+05
	111	.555434+05	112	.569330+05	113	.826855+05	114	.824443+05
	115	.115047+06	116	.114200+06	117	-.225116+05	118	-.203460+05
	119	-.929255+04	120	-.863185+04	121	-.153825+06	122	-.180318+06
	123	-.106536+06	124	-.102226+06	127	-.260462+05	128	-.258694+05
	131	-.175416+05	132	-.190452+05	135	-.120388+05	136	-.136354+05
	139	-.675337+04	140	-.117242+05	143	-.687441+04	144	-.104035+05
	147	-.453754+04	148	-.968953+04	151	.267073+03	152	-.458501+04
	155	.703501+04	156	.288523+04	157	.115613+05	158	.888086+04
	159	.642128+04	160	.400484+04	161	.456108+04	162	.408563+04
	163	.637054+04	164	.638304+04	165	.152569+05	166	.170236+05
	167	.353466+05	168	.373185+05	169	.536170+05	170	.567785+05
	171	.816100+05	172	.806158+05	173	.113565+06	174	.111838+06
	175	.128555+06	176	.124875+06	177	-.220215+05	178	-.190511+05
	179	-.590811+04	180	-.957425+04	181	-.188759+06	182	-.173525+06
	183	-.115514+06	184	-.113458+06	187	-.234276+05	188	-.240181+05

LOADS ARE IN POUNDS

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	FF WILKINSON	1 JUNE 65	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.49
Checked	A L FINCHET	1 JUNE 65	TITLE	Model C-5A	
Approved	L C CAFEY	1 JUNE 65	INTERNAL LOADS ANALYSIS	LC10S46-2-2	
			AFT FUSELAGE	Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

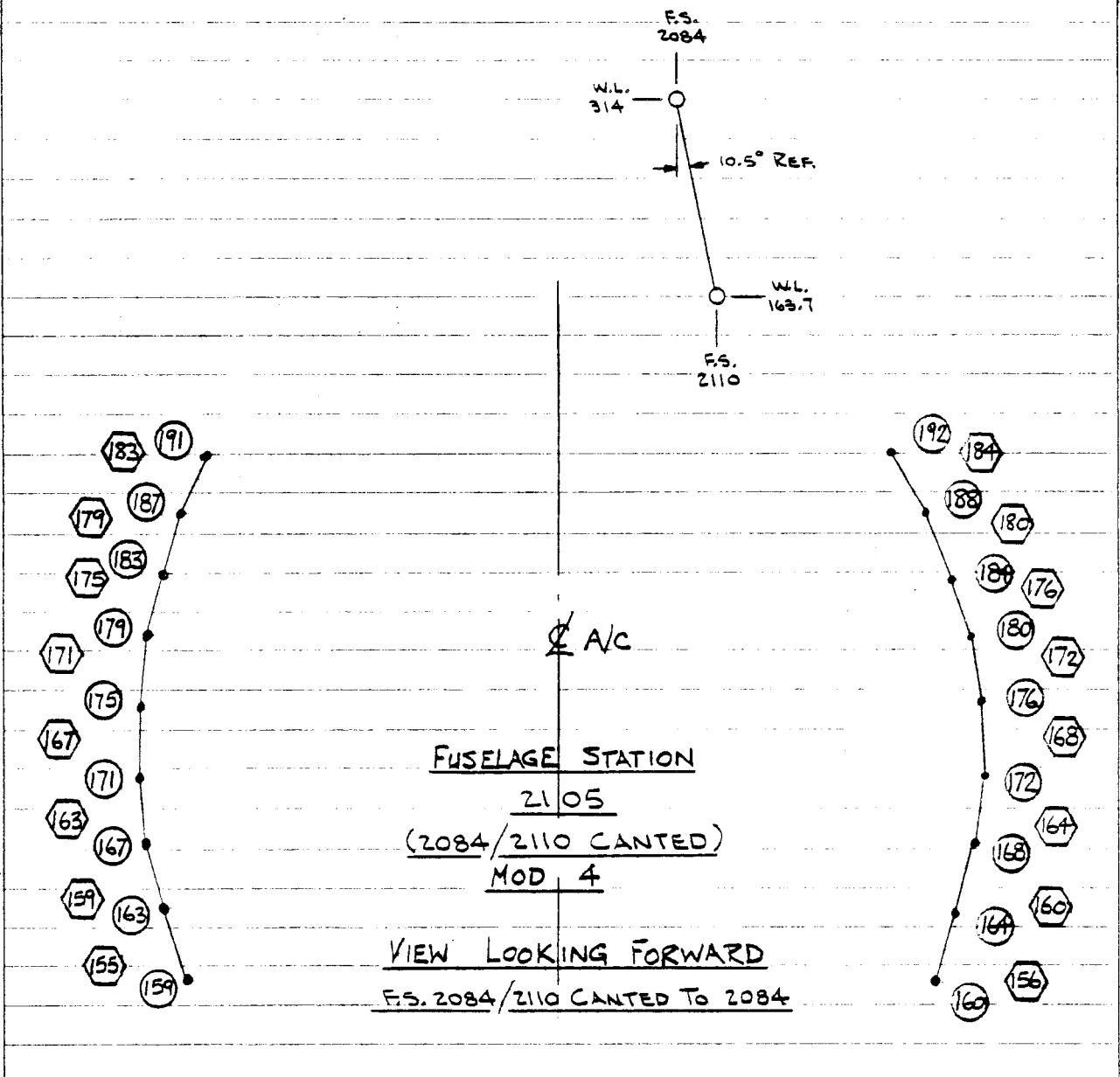
6.3.1 STRINGER LOADS CONDITION SPE201 & MOD 3

1	191	-.174018+05	192	-.190990+05	195	-.134846+05	196	-.156573+05
	199	-.894018+04	200	-.125419+05	203	-.630209+04	204	-.104541+05
	207	-.386529+04	208	-.990296+04	211	.108472+04	212	-.394520+04
	215	.845704+04	216	.352012+04	217	.105080+05	218	.106260+05
	219	.687403+04	220	.644657+04	221	.547912+04	222	.446184+04
	223	.209479+05	224	.182297+05	225	.344220+05	226	.379014+05
	227	.517520+05	228	.569391+05	229	.816015+05	230	.796113+05
	231	.112553+06	232	.105953+06	237	-.205642+06	238	-.183689+06
	239	-.138720+06	240	-.131874+06	243	-.237012+05	244	-.248716+05
	247	-.180008+05	248	-.206994+05	251	-.155733+05	252	-.182100+05
	255	-.100058+05	256	-.140876+05	259	-.635637+04	260	-.109905+05
	263	-.375854+04	264	-.119783+05	267	.171684+04	268	-.277542+04
	271	.103473+05	272	.434461+04	273	.156182+05	274	.178613+05
	275	.104916+05	276	.750230+04	277	.699460+04	278	.645429+04
	279	.648571+04	280	.518900+04	281	.226072+05	282	.191300+05
	283	.329561+05	284	.388309+05	285	.505271+05	286	.577230+05
	287	.826637+05	288	.789021+05	289	.111732+06	290	.108698+06
	291	.120097+06	292	.116099+06	297	-.190182+06	298	-.163555+06
	299	-.150114+06	300	-.145013+06	303	-.290738+05	304	-.304353+05
	307	-.206463+05	308	-.227917+05	311	-.186565+05	312	-.217271+05
	315	-.109208+05	316	-.152768+05	319	-.711027+04	320	-.109551+05
	323	-.485488+04	324	-.167891+05	327	.231740+04	328	-.120260+03
	331	.115499+05	332	.364776+04	333	.133412+05	334	.126900+05
	335	.744807+04	336	.658384+04	337	.744638+04	338	.580929+04
	339	.254716+05	340	.201878+05	341	.311370+05	342	.398291+05
	343	.469673+05	344	.592366+05	345	.848403+05	346	.779363+05
	347	.111172+06	348	.107575+06	349	-.165238+04	350	-.144542+04
	351	-.435171+04	352	-.442348+04	353	-.150094-05	354	-.216823-05
	355	-.490075+04	356	-.508542+04	357	-.427449+04	358	-.378689+04
	359	-.247644+03	360	-.518083+03	361	-.197871+04	362	-.175551+04

LOADS ARE IN POUNDS

REV. (A) MS 1 AUG 69
REV. (B) ADP 1-8-71

6.3.1. STRINGER LOADS (CONTINUED)



KEY

GRID POINT

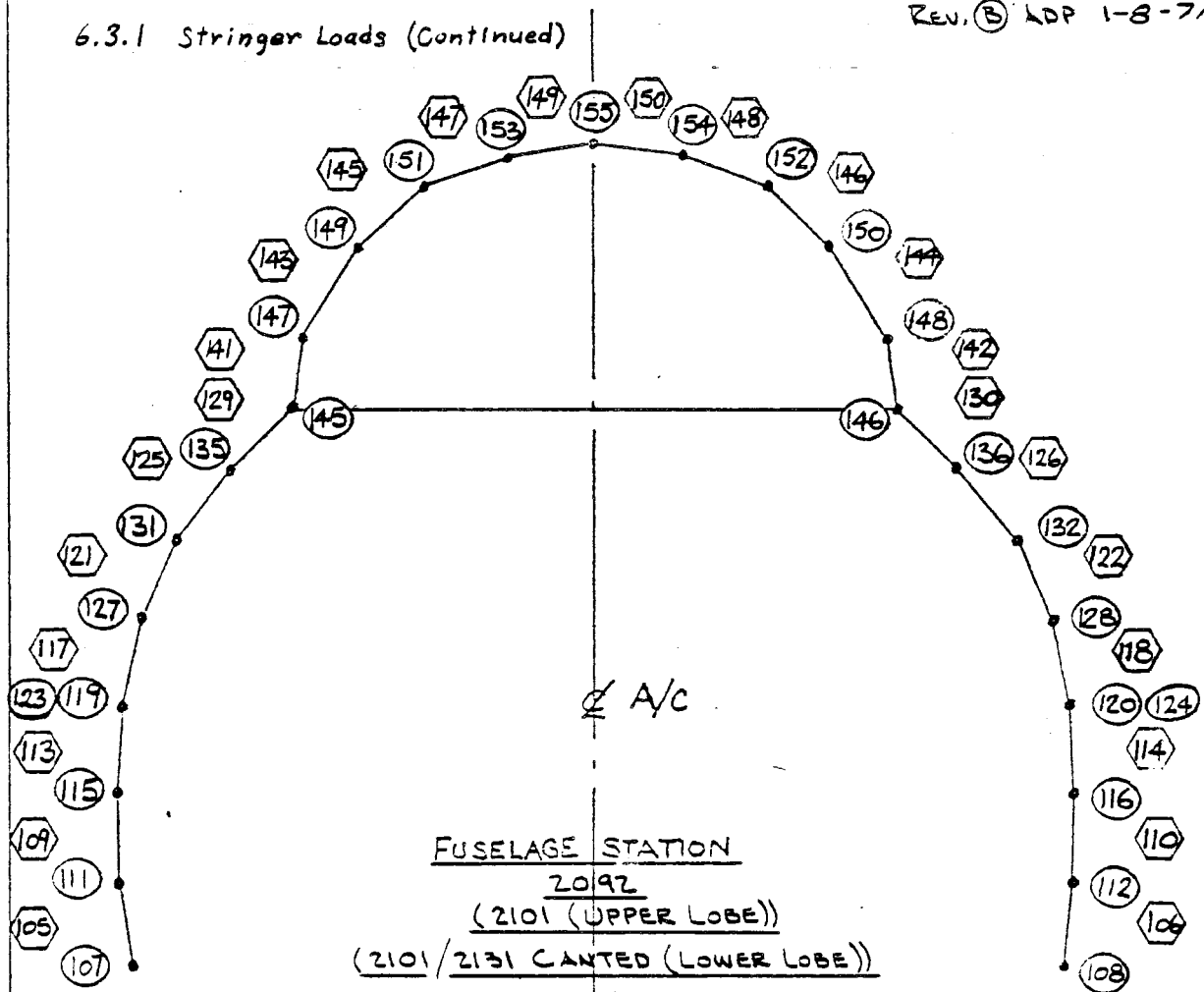
FAMAS STRINGER
 FAMAS PANEL

FIGURE No 6.3.1.16
FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

NAME V.W. BOYD	DATE 9 Sept 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. NOL. I	FORM 6.51
Checked J. HALLIDAY	9 Sept 68	TITLE AFT FUSELAGE	Model C-5A	
Approved L.G. DARBY	9 Sept 68	INTERNAL LOADS ANALYSIS	LGJUS 46-2-2 Report No.	

REV. (A) MB 1 AUG. 69
REV. (B) LDP 1-8-71

6.3.1 Stringer Loads (Continued)



FUSELAGE STATION
2092
(2101 (UPPER LOBE))
(2101 / 2131 CANTED (LOWER LOBE))
MOD. 4
VIEW LOOKING FORWARD
F.S. 2101 TO 2084 (UPPER LOBE)
F.S. 2101 / 2131 CANTED TO F.S. 2084 / 2110 CANTED (LOWER LOBE)

FIGURE NO. 6.3.1.17

KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 8. VOLUME IV
SECTION 5.2

F.S. 2101

W.L. 423.5

W.L. 314

11.5° REF.

W.L. 167.4

F.S. 2131

NAME RF WILKINSON		DATE 1 JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.52
Checked	A L FINCHET	1 JUN 65	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUN 65	AFT FUSELAGE		LG1US46-2-2 Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION DMLCC1 & MOD 4

1	1	.451756+05	2	-.140263+06	3	.351139+05	4	-.903112+05
	7	.132563+05	8	-.264514+05	11	.112310+05	12	-.195510+05
	15	.131475+05	16	-.196256+05	19	.155012+05	20	-.214753+05
	23	.201424+05	24	-.235543+05	25	.658280+04	26	-.794035+04
	27	.811067+04	28	-.502653+04	29	.835655+04	30	-.980550+04
	33	.210867+05	34	-.220372+05	35	.115325+05	36	-.124144+05
	37	.100477+05	38	-.772636+04	39	.265215+05	40	-.246870+05
	41	.422814+05	42	-.238977+05	43	.362522+05	44	-.652526+04
	45	.345135+05	46	.537852+04	47	.357659+05	48	.252574+05
	49	.347345+05	50	.148205+03	51	.456291+05	52	-.139533+06
	53	.414365+05	54	-.576748+05	55	.750970+03	56	-.222539+04
	57	.134233+05	58	-.264774+05	59	.257686+03	60	-.116531+04
	61	.128248+05	62	-.218253+05	63	.525542+02	64	-.472759+03
	65	.194019+05	66	-.292191+05	67	-.142841+03	68	-.128792+03
	69	.187565+05	70	-.231554+05	71	-.221564+03	72	-.142475+03
	73	.234286+05	74	-.260133+05	75	.665583+03	76	-.778067+03
	77	.187265+05	78	-.165625+05	79	.106002+05	80	-.941075+04
	81	.581541+04	82	-.855712+04	83	.504868+04	84	-.894841+04
	85	.122050+05	86	-.115453+05	87	.500667+04	88	-.492094+04
	89	.392476+05	90	-.311387+05	91	.435118+05	92	-.240188+05
	93	.391680+05	94	-.101543+05	95	.392223+05	96	.280750+04
	97	.375803+05	98	.195023+05	99	.330430+05	100	.132090+03
	101	.140935+03	102	-.465253+03	103	.155016+04	104	-.381212+04
	105	.494565+05	106	-.140460+06	107	.388303+05	108	-.894588+05
	109	.157556+04	110	-.501447+04	111	.146548+05	112	-.289631+05
	113	.511350+03	114	-.227324+04	115	.153326+05	116	-.260328+05
	117	-.538651+03	118	-.745323+03	119	.111033+05	120	-.165157+05
	121	.886957+02	122	.378401+03	123	.114851+05	124	-.106326+05
	125	-.153147+03	126	.122742+03	127	.223623+05	128	-.265144+05
	129	-.244951+03	130	-.311055+03	131	.241935+05	132	-.256157+05
	133	.154032+04	134	-.121287+04	135	.252892+05	136	-.236407+05
	137	.100555+05	138	-.685191+04	139	.122931+05	140	-.965567+04
	141	.105071+05	142	-.851035+04	143	.113633+05	144	-.891575+04
	145	.453941+05	146	-.344278+05	147	.444751+05	148	-.247515+05
	149	.402350+05	150	-.124053+05	151	.412637+05	152	-.106867+04
	153	.353176+05	154	.165521+05	155	.322516+05	157	.219206+04
	156	-.680761+04	159	.374906+04	160	-.851770+04	161	.506413+05
	162	-.142536+06	163	.363565+05	164	-.805528+05	167	.150430+05

* STRINGER (107) REF. P. 6.51 ULTIMATE TENSION LOAD = 38,830. LB.

Prepared	NAME FF WILKINSON	DATE 1 JUL 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.53
	Checked A L FINCHET	1 JUL 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1 JUL 69	AFT FUSELAGE		Report No. LG10546-2-2	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCN.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION CMLOO1 6 MGD 4

1	168	- .255696+05	171	.146308+05	172	- .253504+05	175	.112685+05
	176	- .170980+05	179	.134963+05	180	- .189069+05	183	.208605+05
	184	- .245428+05	187	.234761+05	188	- .242536+05	191	.245506+05
	192	- .121674+05	193	.240800-01	194	- .766601-01	195	.709369-01
	196	.113912+00	197	- .182552-00	198	- .558039-00	199	.535428+03
	200	- .177250+04	201	- .137713+03	202	.456109+03	203	.817160+03
	204	- .270340+04	205	.494554+05	206	- .140476+06		

LOADS ARE IN POUNDS

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	M BURTON	NOV70		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 653.7
Checked	A L PINCHOT	NOV70	TITLE	INTERNAL LOADS ANALYSIS		C-5A	
Approved	H H WADE	NOV70		AFT FUSELAGE		Model	LGIUS46-2-2
						Report No.	

REV. ⑤ M.D. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 4

1	1	-.138585+06	2	.499193+05	3	-.889791+05	4	.392758+05
	7	-.261252+05	8	.132874+05	11	-.193702+05	12	.111778+05
	15	-.195352+05	16	.129878+05	19	-.214462+05	20	.153196+05
	23	-.237107+05	24	.198258+05	25	-.894144+04	26	.598778+04
	27	-.963375+04	28	.753968+04	29	-.101840+05	30	.807717+04
	33	-.223343+05	34	.207939+05	35	-.128823+05	36	.111813+05
	37	-.819422+04	38	.968925+04	39	-.254435+05	40	.281064+05
	41	-.246829+05	42	.424565+05	43	-.681315+04	44	.371057+05
	45	.945217+04	46	.357127+05	47	.256469+05	48	.369594+05
	49	.355293+05	50	-.520812+02	51	-.138366+06	52	.505030+05
	53	-.961267+05	54	.417391+05	55	-.219367+04	56	.772409+03
	57	-.261123+05	58	.134362+05	59	-.114060+04	60	.277128+03
	61	-.215832+05	62	.127615+05	63	-.462573+03	64	.632073+02
	65	-.290059+05	66	.192111+05	67	-.125451+03	68	-.136936+03
	69	-.231889+05	70	.184477+05	71	-.123934+03	72	-.201132+03
	73	-.262191+05	74	.230050+05	75	-.814384+03	76	.643799+03
	77	-.173854+05	78	.183100+05	79	-.106101+05	80	.939116+04
	81	-.988765+04	82	.901647+04	83	-.943876+04	84	.862574+04
	85	-.121202+05	86	.117906+05	87	-.495827+04	88	.505523+04
	89	-.325411+05	90	.381975+05	91	-.247257+05	92	.443001+05
	93	-.100724+05	94	.405555+05	95	.321761+04	96	.408266+05
	97	.203395+05	98	.393026+05	99	.338365+05	100	-.545917+02
	101	-.458670+03	102	.144039+03	103	-.374307+04	104	.161757+04
	105	-.138887+06	106	.505048+05	107	-.880093+05	108	.391390+05
	109	-.496630+04	110	.161251+04	111	-.285291+05	112	.146921+05
	113	-.224041+04	114	.538755+03	115	-.257084+05	116	.152932+05
	117	-.710190+03	118	-.633869+02	119	-.163608+05	120	.110723+05
	121	.334851+03	122	.591960+02	123	-.165238+05	124	.112323+05
	125	.110586+03	126	-.198536+03	127	-.269389+05	128	.219072+05
	129	-.293008+03	130	-.227655+03	131	-.258851+05	132	.236398+05
	133	-.130931+04	134	.147280+04	135	-.241443+05	136	.246872+05
	137	-.836314+04	138	.858873+04	139	-.107693+05	140	.112248+05
	141	-.919103+04	142	.990705+04	143	-.953330+04	144	.108902+05
	145	-.360535+05	146	.442917+05	147	-.253214+05	148	.452894+05
	149	-.118598+05	150	.422821+05	151	-.152697+03	152	.433613+05
	153	.170001+05	154	.406131+05	155	.326441+05	157	-.671440+04
	158	.226933+04	159	-.842536+04	160	.381532+04	161	-.141069+06
	162	.517482+05	163	-.792116+05	164	.366476+05	167	-.294643+05

LOADS ARE IN POUNDS

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.53.2
Checked	A L PINCHOT	NOV70	TITLE		C-5A	
			INTERNAL LOADS ANALYSIS		Model	
Approved	H H WADE	NOV70	AFT FUSELAGE		LGIUS46-2-2	
					Report No.	

REV (B) M.B. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS (CONDITION DMLOOR) MOD 4

1	168	.151060+05	171	-.249915+05	172	.145843+05	175	-.169060+05
	176	.111086+05	179	-.187789+05	180	.132391+05	183	-.245903+05
	184	.203471+05	187	-.245655+05	188	.228507+05	191	-.227499+05
	192	.239162+05	193	-.752918-01	194	.246756-01	195	.110688+00
	196	.655110-01	197	-.582977-00	198	-.162087-00	199	-.174732+04
	200	.547216+03	201	.449629+03	202	-.140750+03	203	-.266505+04
	204	.835172+03	205	-.138882+06	206	.505033+05		

Prepared	NAME FF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 6.54
Checked	A L FINCHET	1JUN69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A		
Approved	L C CAREY	1JUN69	AFT FUSELAGE		Report No. LG1LS46-2-2		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPE2014P 8 MOD 4

1	1	- .185948+06	2	- .186953+06	3	- .103356+06	4	- .104280+06
	7	- .269314+06	8	- .272736+06	11	- .176418+05	12	- .179271+05
	15	- .144140+05	16	- .155117+05	19	- .147916+05	20	- .156905+05
	23	- .925020+04	24	- .110458+05	25	- .838309+04	26	- .101220+05
	27	- .421280+04	28	- .547543+04	29	- .421207+04	30	- .518467+04
	33	- .516985+04	34	- .542875+04	35	- .622461+04	36	- .666420+04
	37	- .125562+04	38	- .474643+03	39	- .516762+04	40	- .315756+04
	41	- .372265+05	42	- .376576+05	43	- .640853+05	44	- .636458+05
	45	- .936783+05	46	- .924961+05	47	- .128676+06	48	- .127638+06
	49	- .146450+06	50	- .146104+06	51	- .167677+06	52	- .182070+06
	53	- .106624+06	54	- .106181+06	55	- .307629+04	56	- .298558+04
	57	- .265822+05	58	- .267572+05	59	- .150478+04	60	- .151481+04
	61	- .155836+05	62	- .155327+05	63	- .115976+04	64	- .116353+04
	65	- .201816+05	66	- .210862+05	67	- .150682+04	68	- .185894+04
	69	- .132627+05	70	- .153605+05	71	- .263783+04	72	- .270665+04
	73	- .770220+04	74	- .543259+04	75	- .998556+03	76	- .105697+04
	77	- .135250+04	78	- .507772+03	79	- .260812+04	80	- .551098+04
	81	- .257771+04	82	- .388820+04	83	- .415776+04	84	- .537146+04
	85	- .440235+04	86	- .460826+04	87	- .147308+04	88	- .179179+02
	89	- .484602+04	90	- .573836+04	91	- .448644+05	92	- .443032+05
	93	- .663626+05	94	- .660865+05	95	- .924164+05	96	- .918757+05
	97	- .123454+06	98	- .123213+06	99	- .140687+06	100	- .139596+05
	101	- .826811+03	102	- .862349+03	103	- .141241+04	104	- .113985+04
	105	- .181140+06	106	- .174243+06	107	- .518522+05	108	- .904720+05
	109	- .938491+04	110	- .511944+04	111	- .281163+05	112	- .279817+05
	113	- .124866+04	114	- .122218+04	115	- .221170+05	116	- .222133+05
	117	- .626365+03	118	- .468376+03	119	- .115354+05	120	- .109343+05
	121	- .325019+03	122	- .515489+03	123	- .776314+04	124	- .101412+05
	125	- .938235+03	126	- .105488+04	127	- .753831+04	128	- .104791+05
	129	- .207560+04	130	- .214219+04	131	- .102658+04	132	- .329540+04
	133	- .155144+04	134	- .184545+04	135	- .663604+04	136	- .427444+04
	137	- .639134+05	138	- .623926+05	139	- .434061+05	140	- .413373+05
	141	- .274151+05	142	- .258307+05	143	- .181997+05	144	- .181917+05
	145	- .329092+05	146	- .326406+05	147	- .463141+05	148	- .455945+05
	149	- .721331+05	150	- .720925+05	151	- .103597+06	152	- .103494+06
	153	- .140755+06	154	- .141014+06	155	- .155593+06	156	- .654034+04
	158	- .587195+04	159	- .743422+04	160	- .703625+04	161	- .183870+06
	162	- .174852+06	163	- .816182+05	164	- .799200+05	165	- .298856+05

LOADS ARE IN POUNDS

	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION			TEMP. Vol. I 6.55	PERM.
Prepared					Page		
Checked	A L FINCHET	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A		
Approved	L G CAFEY	1 JUN 69	AFT FUSELAGE		LG10S46-2-2		
					Report No.		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8
 6.3 SKINS AND STRINGERS
 6.3.1 STRINGER LOADS CONDITION SPE201+P : 8 MOD 4

1 166 -.258792+05 171 -.213616+05 172 -.219135+05 175 -.100889+05
 176 -.115325+05 179 -.921369+04 180 -.111433+05 183 -.332290+04
 184 -.658606+04 187 .291100+04 188 -.238558+03 191 .867977+04
 192 .627125+04 193 -.446636+04 194 -.449048+04 195 -.364029+02
 196 -.366877+02 197 -.330116+02 198 -.328456+02 199 -.226015+04
 200 -.216140+04 201 .368933+04 202 .367502+04 203 -.123958+04
 204 -.106991+04 205 -.181007+06 206 -.174109+06

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I 6.56	PERM.
Prepared					Page	
Checked	A L FINCHET	1 JUN 69	TITLE INTERNAL LEADS ANALYSIS AFT FUSELAGE		Model	C-5A
Approved	L C CAPEY	1 JUN 69			Report No.	LG10S45-2-2

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LEADS CONDITION IPC242 & MOD 4

1	1	.125710+06	2	.115744+06	3	.738406+05	4	.725073+05
	7	.199117+05	8	.196486+05	11	.132721+05	12	.131318+05
	15	.114202+05	16	.122466+05	19	.110669+05	20	.116708+05
	23	.642543+04	24	.101615+05	25	.513511+04	26	.722995+04
	27	.375281+04	28	.529508+04	29	.384044+04	30	.504871+04
	33	.462936+04	34	.479357+04	35	.400723+04	36	.467680+04
	37	-.106080+04	38	-.180219+04	39	-.638033+03	40	.182838+04
	41	-.235978+05	42	-.235158+05	43	-.436770+05	44	-.427012+05
	45	-.656314+05	46	-.640550+05	47	-.892956+05	48	-.880446+05
	49	-.556250+05	50	.471586+03	51	.124177+06	52	.115647+06
	53	.783690+05	54	.754560+05	55	.212889+04	56	.200669+04
	57	.195365+05	58	.190090+05	59	.105861+04	60	.111214+04
	61	.143217+05	62	.141080+05	63	.655213+03	64	.669528+03
	65	.165272+05	66	.167307+05	67	.683871+02	68	.588784+02
	69	.551862+04	70	.112761+05	71	.114714+03	72	.285264+03
	73	.761671+04	74	.510627+04	75	.400785+02	76	.185226+03
	77	.432847+03	78	.242988+04	79	.156611+04	80	.533618+04
	81	.222264+04	82	.356426+04	83	.201941+04	84	.347716+04
	85	.192191+04	86	.246522+04	87	.717007+03	88	.250130+04
	89	-.459507+04	90	-.528677+04	91	-.245213+05	92	-.235351+05
	93	-.421626+05	94	-.411751+05	95	-.624078+05	96	-.616334+05
	97	-.654989+05	98	-.651970+05	99	-.972055+05	100	.434465+03
	101	.381572+03	102	.353247+03	103	.342812+04	104	.307518+04
	105	.124483+06	106	.114832+06	107	.708477+05	108	.670553+05
	109	.400430+04	110	.368816+04	111	.211928+05	112	.202032+05
	113	.182832+04	114	.180688+04	115	.168351+05	116	.160590+05
	117	.683257+03	118	.689261+03	119	.535591+04	120	.814607+04
	121	-.180140+03	122	-.399613+03	123	.655415+04	124	.106253+05
	125	-.149258+03	126	-.218037+03	127	.976833+04	128	.117778+05
	129	-.217545+03	130	-.141037+02	131	.555834+04	132	.754719+04
	133	.935720+02	134	.319753+03	135	.834529+03	136	.322171+04
	137	.512672+03	138	.242292+04	139	.127818+03	140	.264212+04
	141	.668147+02	142	.196907+04	143	-.544922+03	144	-.274489+03
	145	-.667195+04	146	-.571543+04	147	-.251655+05	148	-.231395+05
	149	-.406828+05	150	-.396369+05	151	-.598141+05	152	-.596170+05
	153	-.624814+05	154	-.829075+05	155	-.950213+05	156	.610493+04
	158	.536293+04	159	.806055+04	160	.748355+04	161	.122947+06
	162	.111274+06	163	.653192+05	164	.611782+05	167	.219496+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 6.57
Checked	A L FINCHOT	1 JUN 69	TITLE INTERNAL LCADS ANALYSIS AFT FUSELAGE		Model C-5A		
Approved	L C CAFEY	1 JUN 69			LG10S46-2-2 Report No.		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON. &

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 4

1	166	.209790+C5	171	.161919+05	172	.156703+05	175	.922879+04
	176	.101447+05	179	.536057+C4	180	.107705+C5	183	.768452+04
	184	.104734+C5	187	.421410+C4	188	.719663+C4	191	.122156+C2
	192	.260738+C4	193	.426813-01	194	.388683-01	195	-.316779-00
	196	-.221366-C0	197	.119733+C1	198	.856309-00	199	.145084+04
	200	.134343+C4	201	-.373467+C3	202	-.345771+C3	203	.221481+04
	204	.205050+C4	205	.124479+C6	206	.114828+06		

LOADS ARE IN POUNDS

	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.58
Checked	A L FINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-5A
Approved	L C DAREY	1JUN69			LC10S46-2-2	Report No.

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCN.6

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LOADS CONDITION SPE201

6 MOD 4

1	1	-.188126+06	2	-.185115+06	3	-.106000+06	4	-.107001+06
	7	-.278055+05	8	-.281554+05	11	-.180011+05	12	-.182663+05
	15	-.148523+05	16	-.155631+05	19	-.135419+05	20	-.148095+05
	23	-.958334+04	24	-.113428+05	25	-.611087+04	26	-.783134+04
	27	-.445566+04	28	-.574678+04	29	-.479779+04	30	-.577839+04
	33	-.441150+04	34	-.460556+04	35	-.445059+04	36	-.497831+04
	37	.235570+04	38	.313551+04	39	.355419+04	40	.186538+04
	41	.352815+05	42	.357156+05	43	.621543+05	44	.617136+05
	45	.534647+05	46	.522750+05	47	.129229+06	48	.128179+06
	49	.145651+06	50	-.272441+03	51	-.187014+06	52	-.181449+06
	53	-.114800+06	54	-.114534+06	55	-.306760+04	56	-.299481+04
	57	-.275676+05	58	-.277359+05	59	-.173781+04	60	-.177090+04
	61	-.196402+05	62	-.199336+05	63	-.164967+03	64	-.885051+03
	65	-.219122+05	66	-.226572+05	67	-.388669+03	68	-.374199+03
	69	-.112071+05	70	-.131700+05	71	-.534384+03	72	-.675397+03
	73	-.603347+04	74	-.963645+04	75	-.257751+03	76	-.362720+03
	77	.105207+04	78	-.712948+03	79	-.147826+04	80	-.429450+04
	81	-.170872+04	82	-.297906+04	83	-.185961+04	84	-.302100+04
	85	-.136523+04	86	-.163505+04	87	-.462571+03	88	-.197368+04
	89	.107965+05	90	.116689+05	91	.361382+05	92	.375273+05
	93	.607493+05	94	.603695+05	95	.855566+05	96	.889981+05
	97	.122954+06	98	.122703+06	99	.135999+06	100	-.258222+03
	101	-.626716+03	102	-.605474+03	103	-.471886+04	104	-.448184+04
	105	-.187757+06	106	-.181046+06	107	-.103873+06	108	-.102614+06
	109	-.674074+04	110	-.651702+04	111	-.301028+05	112	-.299156+05
	113	-.327841+04	114	-.325028+04	115	-.231200+05	116	-.230282+05
	117	-.136084+04	118	-.156444+04	119	-.122440+05	120	-.114830+05
	121	.554471+03	122	.618564+03	123	-.112705+05	124	-.135526+05
	125	-.462840+02	126	.323526+02	127	-.113657+05	128	-.136940+05
	129	-.653835+03	130	-.617045+03	131	-.532120+04	132	-.741110+04
	133	.219826+03	134	.693411+02	135	.987525+03	136	-.122126+04
	137	.131142+04	138	-.124112+03	139	.157864+04	140	-.378280+03
	141	.154557+04	142	.425111+02	143	.225109+04	144	.229666+04
	145	.154140+05	146	.150854+05	147	.383023+05	148	.373940+05
	149	.589332+05	150	.585952+05	151	.861466+05	152	.859104+05
	153	.118339+06	154	.118565+06	155	.135918+06	156	-.929206+04
	158	-.673345+04	159	-.102908+05	160	-.997063+04	161	-.188240+06
	162	-.179477+06	163	-.930455+05	164	-.913242+05	165	-.313151+05

LOADS ARE IN POUNDS

Prepared	NAME FF WILKINSON	DATE 1 JUNE 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
				Page	Vol. I 6.59
Checked	A L FINCHET	1 JUNE 69	TITLE INTERNAL LCADS ANALYSIS AFT FUSELAGE	Model C-5A	
				LG 10546-2-2	
Approved	L C CAFEY	1 JUNE 69		Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.1 STRINGER LCADS CONDITION SPE2C1 8 MOD 4

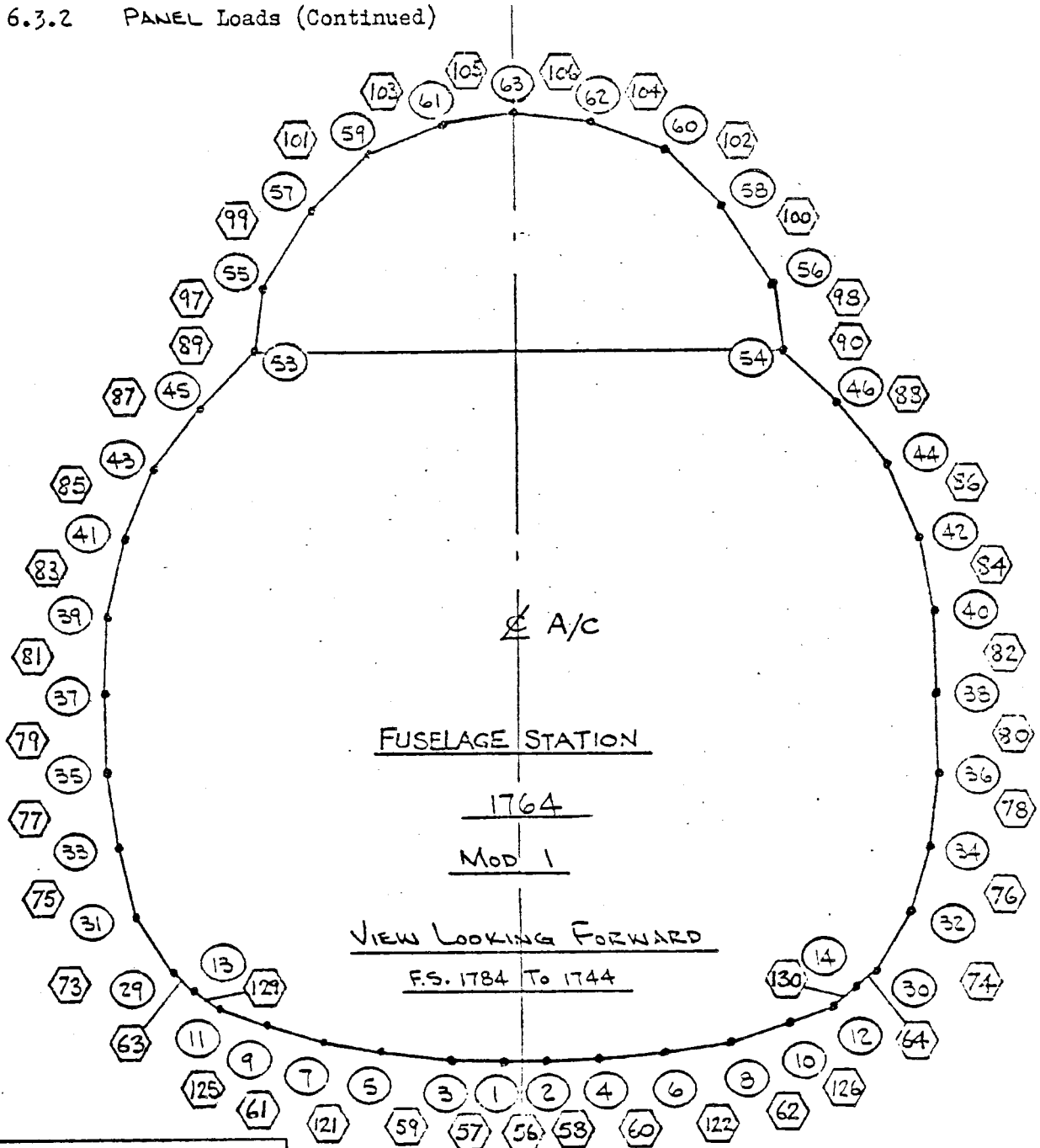
1	166	-	.312181+05	171	-	.226744+05	172	-	.230432+05	175	-	.124701+05
	176	-	.137884+05	179	-	.120155+05	180	-	.137702+05	183	-	.901682+04
	184	-	.120522+05	187	-	.368032+04	188	-	.666144+04	191		.233435+04
	192		.228888+02	193	-	.550777-C1	194	-	.933791-01	195		.412764-00
	196		.326825-C0	197	-	.167518+01	198	-	.137694+01	199	-	.238739+04
	200	-	.232228+04	201		.614495+C3	202		.597690+03	203	-	.364158+04
	204	-	.354217+04	205	-	.187751+C6	206	-	.181040+06			

LOADS ARE IN POUNDS

Prepared	NAME V.W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Vol. 7	PERM. 6.60
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved	L.G. Darby	16 Aug 68		Report No.	LGUS 46-2-2

REV. (A) M.B. 1 AUG. 69.

6.3.2 PANEL Loads (Continued)



KEY

—●— GRID POINT

○ FAMAS - MODEL
CONNECTING LINK

⬡ FAMAS PANEL

FIGURE No 6.3.2.1

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.61
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C=5A	
Approved	L G DARRY	1 JUN 69			LGIUS46=2=2	
					Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION SPMQ27) MOD 1

1	1	.435541+01	2	-.651600+01	3	.296246+03	4	.282084+03
	5	.566954+03	6	.551979+03	7	.646234+03	8	.626912+03
	9	.154319+03	10	.143657+03	11	.148806+03	12	.157442+03
	13	.309321+03	14	.318385+03	15	.687366+03	16	.695864+03
	17	-.101099+04	18	-.104169+04	19	.941778+03	20	.935353+03
	21	.105268+04	22	.104496+04	23	.119718+04	24	.118849+04
	25	.163585+04	26	.162255+04	27	.435618+03	28	.455833+03
	29	.715218+03	30	.744373+03	31	.892486+03	32	.932562+03
	33	.833529+03	34	.920024+03	35	.531058+03	36	.798302+03
	37	-.125643+03	38	-.523934+02	39	-.224824+03	40	-.525026+02
	41	-.336224+03	42	-.359731+02	43	.106356+04	44	.449018+03
	45	.468784+03	46	.558236+03	47	-.408218+03	48	.135614+03
	49	.184177+03	50	.833513+02	51	.107848+03	52	.602876+02
	53	.232324+02	54	.386967+02	55	.289073+01	56	-.764989+01
*	57	.242196+03	58	.227587+03	59	.470590+03	60	.455395+03
	61	.568099+03	62	.550418+03	63	.138477+04	64	.137868+04
	65	.149871+03	66	.155889+03	67	.291576+03	68	.298312+03
	69	.582928+03	70	.593716+03	71	-.637499+03	72	-.650228+03
	73	.111032+04	74	.110147+04	75	.137011+04	76	.136222+04
	77	.150584+04	78	.149876+04	79	.146370+04	80	.146329+04
	81	.103904+04	82	.105738+04	83	.877051+03	84	.919791+03
	85	.801324+03	86	.866020+03	87	.724475+03	88	.817760+03
	89	.681547+03	90	.731977+03	91	-.105522+03	92	-.142648+02
	93	-.111929+03	94	-.187974+02	95	.978792+01	96	-.236310+02
	97	.842237+03	98	.492589+03	99	.387712+03	100	.372682+03
	101	-.937909+02	102	.270978+03	103	.302650+02	104	.357500+02
	105	.710281+02	106	.174168+02	107	-.530320+03	108	-.539327+03
	109	-.711189+03	110	-.693444+03	111	.737174+03	112	.744109+03
	113	.587078+03	114	.564029+03	115	-.750433+03	116	-.759093+03
	117	-.363120+03	118	-.335290+03	119	-.462081+03	120	-.470209+03
	121	-.539351+03	122	-.522554+03	123	.639191+03	124	.654045+03
	125	.539446+03	126	.521732+03	127	-.713979+03	128	-.731439+03
	129	-.469428+03	130	-.452738+03				

* PANEL (57) REF. P. 6.60 ULTIMATE PANEL SHEAR FLOW = 242 LBS./IN.

NAME RF WILKINSON		DATE 1 JUN 69		LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I	6.62
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C=5A	
Approved	L G DARBY	1 JUN 69				Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION TPM096+P)MOD 1

1	1	.218109+01	2	-.726015+01	3	.360524+03	4	.344806+03
	5	.625903+03	6	.609103+03	7	.666992+03	8	.645016+03
	9	.210315+03	10	.196976+03	11	.196433+03	12	.201056+03
	13	.385390+03	14	.390961+03	15	.695903+03	16	.704587+03
	17	-.120904+04	18	-.124528+04	19	.103627+04	20	.103064+04
	21	.113973+04	22	.113216+04	23	.124381+04	24	.123528+04
	25	.151895+04	26	.150767+04	27	.353707+03	28	.379333+03
	29	.555621+03	30	.606637+03	31	.724959+03	32	.827929+03
	33	.530213+03	34	.737455+03	35	.205947+03	36	.593692+03
	37	-.969752+02	38	.346649+02	39	-.355067+03	40	-.111296+03
	41	-.241679+03	42	.112238+03	43	.131188+04	44	.276234+03
	45	.431795+03	46	.506449+03	47	-.797390+03	48	.134024+03
	49	.323331+03	50	.102510+03	51	.155544+03	52	.859941+02
	53	.511429+02	54	.610408+02	55	.912892-01	56	-.839445+01
	57	.249386+03	58	.233203+03	59	.547022+03	60	.530180+03
	61	.682855+03	62	.663116+03	63	.134450+04	64	.134117+04
	65	.185293+03	66	.185933+03	67	.361206+03	68	.363141+03
	69	.686209+03	70	.696148+03	71	-.398900+03	72	-.416248+03
	73	.984238+03	74	.976744+03	75	.114593+04	76	.114242+04
	77	.123752+04	78	.123785+04	79	.125591+04	80	.126797+04
	81	.100450+04	82	.103868+04	83	.782773+03	84	.843622+03
	85	.615158+03	86	.733690+03	87	.566138+03	88	.699549+03
	89	.573972+03	90	.611229+03	91	-.121941+03	92	.697820+01
	93	-.127197+03	94	-.169111+02	95	.863613+02	96	.212261+02
	97	.853774+03	98	.287018+03	99	.265563+03	100	.241464+03
	101	-.299369+03	102	.277361+03	103	.164639+02	104	.318014+01
	105	.982992+02	106	.515972+01	107	-.590506+03	108	-.597809+03
	109	-.742228+03	110	-.722186+03	111	.689945+03	112	.698450+03
	113	.640505+03	114	.613796+03	115	-.673207+03	116	-.684639+03
	117	-.447791+03	118	-.415030+03	119	-.550645+03	120	-.555870+03
	121	-.648885+03	122	-.630214+03	123	.759080+03	124	.774604+03
	125	.614810+03	126	.595080+03	127	-.831569+03	128	-.850931+03
	129	-.532912+03	130	-.514493+03				

LOADS ARE IN POUNDS / INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	NOL. I 6.63
Checked	A L PINCHOT	1 JUN 69	TITLE	Model C=5A	
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	LG IUS 46-2=2	
			AFT FUSELAGE	Report No.	

REV. (A) RFW 1 Aug. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION ALSIII) MOD 1

1	1	-.263102+02	2	-.866709+02	3	.369296+03	4	.200545+03
	5	.631501+03	6	.473988+03	7	.686383+03	8	.562234+03
	9	.179734+03	10	.895873+02	11	.177514+03	12	.128959+03
	13	.333429+03	14	.293749+03	15	.686143+03	16	.702312+03
	17	-.980949+03	18	-.106797+04	19	.954746+03	20	.894412+03
	21	.105088+04	22	.102343+04	23	.117713+04	24	.118568+04
	25	.158299+04	26	.166841+04	27	.433012+03	28	.437529+03
	29	.692676+03	30	.752001+03	31	.856587+03	32	.945448+03
	33	.778017+03	34	.958144+03	35	.478918+03	36	.851582+03
	37	-.321403+02	38	-.140938+03	39	-.144437+03	40	-.137084+03
	41	-.268680+03	42	-.121416+03	43	.106576+04	44	.449052+03
	45	.453077+03	46	.574515+03	47	-.421240+03	48	.159953+03
	49	.155868+03	50	.127820+03	51	.627421+02	52	.108061+03
	53	-.792558+02	54	-.736954+02	55	-.390182+02	56	-.851394+02
	57	.311116+03	58	.146292+03	59	.527420+03	60	.370748+03
	61	.605120+03	62	.465638+03	63	.136937+04	64	.132639+04
	65	.189785+03	66	.115494+03	67	.328062+03	68	.260453+03
	69	.597279+03	70	.581262+03	71	-.643855+03	72	-.675804+03
	73	.109325+04	74	.104425+04	75	.134201+04	76	.132324+04
	77	.146646+04	78	.147761+04	79	.141412+04	80	.146104+04
	81	.101070+04	82	.103886+04	83	.846418+03	84	.917033+03
	85	.759284+03	86	.882675+03	87	.673963+03	88	.855306+03
	89	.612278+03	90	.797510+03	91	.141027+00	92	-.121695+03
	93	-.775839+01	94	-.133538+03	95	.116235+03	96	-.152606+03
	97	.848139+03	98	.476465+03	99	.378580+03	100	.384492+03
	101	-.109278+03	102	.294331+03	103	.536171+00	104	.861181+02
	105	.274953+02	106	.659356+02	107	-.541878+03	108	-.530046+03
	109	-.763038+03	110	-.624936+03	111	.727574+03	112	.759000+03
	113	.622854+03	114	.493856+03	115	-.734005+03	116	-.779700+03
	117	-.399105+03	118	-.251013+03	119	-.487652+03	120	-.444411+03
	121	-.582004+03	122	-.438572+03	123	.646459+03	124	.651605+03
	125	.567972+03	126	.439619+03	127	-.716095+03	128	-.739751+03
	129	-.490724+03	130	-.371844+03				

LOADS ARE IN POUNDS / INCH

NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared				Page	No. I 6,64
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model C=5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	LGIUS46=2=2 Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION DMLOO1)MOD 1

1	1	-.404169+03	2	.319198+03	3	-.260592+03	4	.424268+03
	5	-.227262+03	6	.545980+03	7	-.213030+03	8	.566347+03
	9	.273368+03	10	-.248145+03	11	.419638+03	12	-.329631+03
	13	.405327+03	14	-.222077+03	15	.207912+03	16	.187186+03
	17	-.140790+03	18	-.447015+03	19	.139042+03	20	.343416+03
	21	-.588674+02	22	.614401+03	23	-.236603+03	24	.875448+03
	25	-.624669+03	26	.149171+04	27	-.486736+02	28	.267871+03
	29	-.288600+03	30	.644060+03	31	-.428350+03	32	.813695+03
	33	-.674052+03	34	.966406+03	35	-.543161+03	36	.947770+03
	37	.151694+03	38	-.222526+03	39	.176022+03	40	-.224257+03
	41	.277627+03	42	-.246578+03	43	.665939+02	44	.617965+03
	45	-.450942+03	46	.917114+03	47	-.120026+04	48	.457110+03
	49	-.216100+03	50	.572104+03	51	-.546858+03	52	.609443+03
	53	-.238754+03	54	-.204108+03	55	-.334861+03	56	.314970+03
	57	-.254781+03	58	.381869+03	59	-.201407+03	60	.448586+03
	61	-.147007+03	62	.432693+03	63	-.537779+02	64	.761519+03
	65	.357422+03	66	-.267775+03	67	.358885+03	68	-.186577+03
	69	.220324+03	70	.117093+03	71	-.320940+02	72	-.340589+03
	73	-.195999+02	74	.562642+03	75	-.147642+03	76	.834534+03
	77	-.288945+03	78	.104952+04	79	-.409100+03	80	.114844+04
	81	-.215014+03	82	.750538+03	83	-.268752+03	84	.745202+03
	85	-.427225+03	86	.807698+03	87	-.460280+03	88	.869225+03
	89	-.478052+03	90	.922593+03	91	.193755+03	92	-.205570+03
	93	.247532+03	94	-.225851+03	95	.329237+03	96	-.286940+03
	97	-.138136+03	98	.651649+03	99	-.555428+03	100	.715474+03
	101	-.856685+03	102	.590936+03	103	-.502263+03	104	.601197+03
	105	-.508933+03	106	.584900+03	107	-.306741+03	108	-.203252+01
	109	.238692+03	110	-.636997+03	111	.148098+03	112	.271313+03
	113	-.199134+03	114	.510395+03	115	-.110090+03	116	-.314053+03
	117	.137686+03	118	-.307919+03	119	-.293356+03	120	.241579+02
	121	.157567+03	122	-.431509+03	123	.167620+03	124	.203217+03
	125	-.126172+03	126	.388779+03	127	-.109907+03	128	-.307615+03
	129	.731940+02	130	-.289007+03				

LOADS ARE IN POUNDS / INCH

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.		PERM.	
Prepared	M BURTON	NOV 70		A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	VOL. I 6.64.1		
Checked	A L PINCHOT	NOV 70	TITLE	INTERNAL LOADS ANALYSIS			C=5A		
Approved	H H WADE	NOV 70	AFT FUSELAGE			Model			Report No.
						LGIUS46=2=2			

REV. (3) ADP 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION DMLOOR) MOD 1

1	1	.406061+03	2	-.322649+03	3	.428273+03	4	-.264105+03
	5	.550502+03	6	-.230739+03	7	.576063+03	8	-.213561+03
	9	-.229287+03	10	.285836+03	11	-.330852+03	12	.422225+03
	13	-.222155+03	14	.409499+03	15	.194541+03	16	.220060+03
	17	-.424780+03	18	-.134781+03	19	.349893+03	20	.141304+03
	21	.622454+03	22	-.549181+02	23	.887605+03	24	-.228204+03
	25	.152244+04	26	-.598187+03	27	.263922+03	28	-.391023+02
	29	.657058+03	30	-.254906+03	31	.888130+03	32	-.322888+03
	33	.108894+04	34	-.489966+03	35	.802331+03	36	-.538459+03
	37	-.232195+03	38	.188070+03	39	-.309556+03	40	.191054+03
	41	-.460279+03	42	.230523+03	43	.533212+03	44	-.406337+03
	45	.685314+03	46	-.657323+03	47	.874618+03	48	-.403678+03
	49	.328902+03	50	-.545687+03	51	.604420+03	52	-.579468+03
	53	.253753+03	54	.226863+03	55	.335828+03	56	-.318941+03
	57	.387661+03	58	-.256549+03	59	.456961+03	60	-.200903+03
	61	.444643+03	62	-.144214+03	63	.777052+03	64	-.412683+02
	65	-.268497+03	66	.358827+03	67	-.187278+03	68	.360806+03
	69	.115894+03	70	.224407+03	71	-.345116+03	72	-.441218+02
	73	.584327+03	74	-.173042+01	75	.860292+03	76	-.124417+03
	77	.108107+04	78	-.258438+03	79	.118227+04	80	-.371628+03
	81	.754240+03	82	-.198593+03	83	.716444+03	84	-.271332+03
	85	.832407+03	86	-.362820+03	87	.826489+03	88	-.451001+03
	89	.824361+03	90	-.556330+03	91	-.255800+03	92	.195730+03
	93	-.309234+03	94	.214239+03	95	-.315496+03	96	.276837+03
	97	.591324+03	98	-.421535+03	99	.728904+03	100	-.553880+03
	101	.740795+03	102	-.466531+03	103	.502260+03	104	-.608023+03
	105	.547432+03	106	-.582444+03	107	-.508214+01	108	-.314407+03
	109	-.643444+03	110	.241759+03	111	.282449+03	112	.163634+03
	113	.522948+03	114	-.198758+03	115	-.326351+03	116	-.128226+03
	117	-.325197+03	118	.134990+03	119	.246332+02	120	-.296508+03
	121	-.443159+03	122	.154716+03	123	.199273+03	124	.171329+03
	125	.402124+03	126	-.121899+03	127	-.300011+03	128	-.111518+03
	129	-.302559+03	130	.680831+02				

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.65
Checked	A L PINCHOT	1 JUN 69	TITLE	Model C=5A	
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	LGIUS46=2=2	
			AFT FUSELAGE	Report No.	

REV. (A) RFW 1 AUG. 69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS (CONDITION TPM096) MOD 1

1	1	.296010+01	2	-.583971+01	3	.300464+03	4	.287679+03
	5	.579787+03	6	.566125+03	7	.651498+03	8	.633768+03
	9	.385616+02	10	.298685+02	11	.162214+03	12	.168223+03
	13	.333902+03	14	.340738+03	15	.729138+03	16	.737567+03
	17	-.994359+03	18	-.102238+04	19	.816953+03	20	.810922+03
	21	.966444+03	22	.960088+03	23	.113006+04	24	.112400+04
	25	.157627+04	26	.156962+04	27	.354448+03	28	.379606+03
	29	.623109+03	30	.661674+03	31	.776547+03	32	.834532+03
	33	.695769+03	34	.807846+03	35	.429808+03	36	.688465+03
	37	-.140891+03	38	-.585633+02	39	-.230592+03	40	-.554517+02
	41	-.311951+03	42	-.264921+02	43	.103995+04	44	.348441+03
	45	.392350+03	46	.420560+03	47	-.619086+03	48	.774231+02
	49	.197001+03	50	.315406+02	51	.970215+02	52	.486033+02
	53	.270159+02	54	.398631+02	55	.132521+01	56	-.673134+01
	57	.236665+03	58	.223893+03	59	.454936+03	60	.441667+03
	61	.527170+03	62	.511615+03	63	.124538+04	64	.124172+04
	65	.161911+03	66	.164896+03	67	.311848+03	68	.315673+03
	69	.603637+03	70	.612640+03	71	-.668962+03	72	-.682471+03
	73	.969251+03	74	.963893+03	75	.122837+04	76	.122535+04
	77	.137265+04	78	.137216+04	79	.134461+04	80	.135238+04
	81	.926202+03	82	.949518+03	83	.761257+03	84	.808599+03
	85	.685140+03	86	.754017+03	87	.616557+03	88	.705637+03
	89	.584805+03	90	.614526+03	91	-.106949+03	92	-.162748+02
	93	-.105323+03	94	-.202128+02	95	.243482+02	96	-.202046+02
	97	.775509+03	98	.376472+03	99	.275866+03	100	.253294+03
	101	-.237517+03	102	.198051+03	103	-.118307+02	104	-.293077+02
	105	.645917+02	106	-.986728+00	107	-.566355+03	108	-.574090+03
	109	-.725491+03	110	-.709266+03	111	.779228+03	112	.787145+03
	113	.581125+03	114	.559926+03	115	-.788883+03	116	-.799357+03
	117	-.336536+03	118	-.310793+03	119	-.485156+03	120	-.491001+03
	121	-.505571+03	122	-.490797+03	123	.655900+03	124	.669068+03
	125	.486858+03	126	.471316+03	127	-.730785+03	128	-.746421+03
	129	-.399048+03	130	-.384489+03				

LOADS ARE IN POUNDS / INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Page	Vol. I 666
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Model C=5A LGIUS46=2=2 Report No.	

REV. (A) RFW 1 AUG. 69

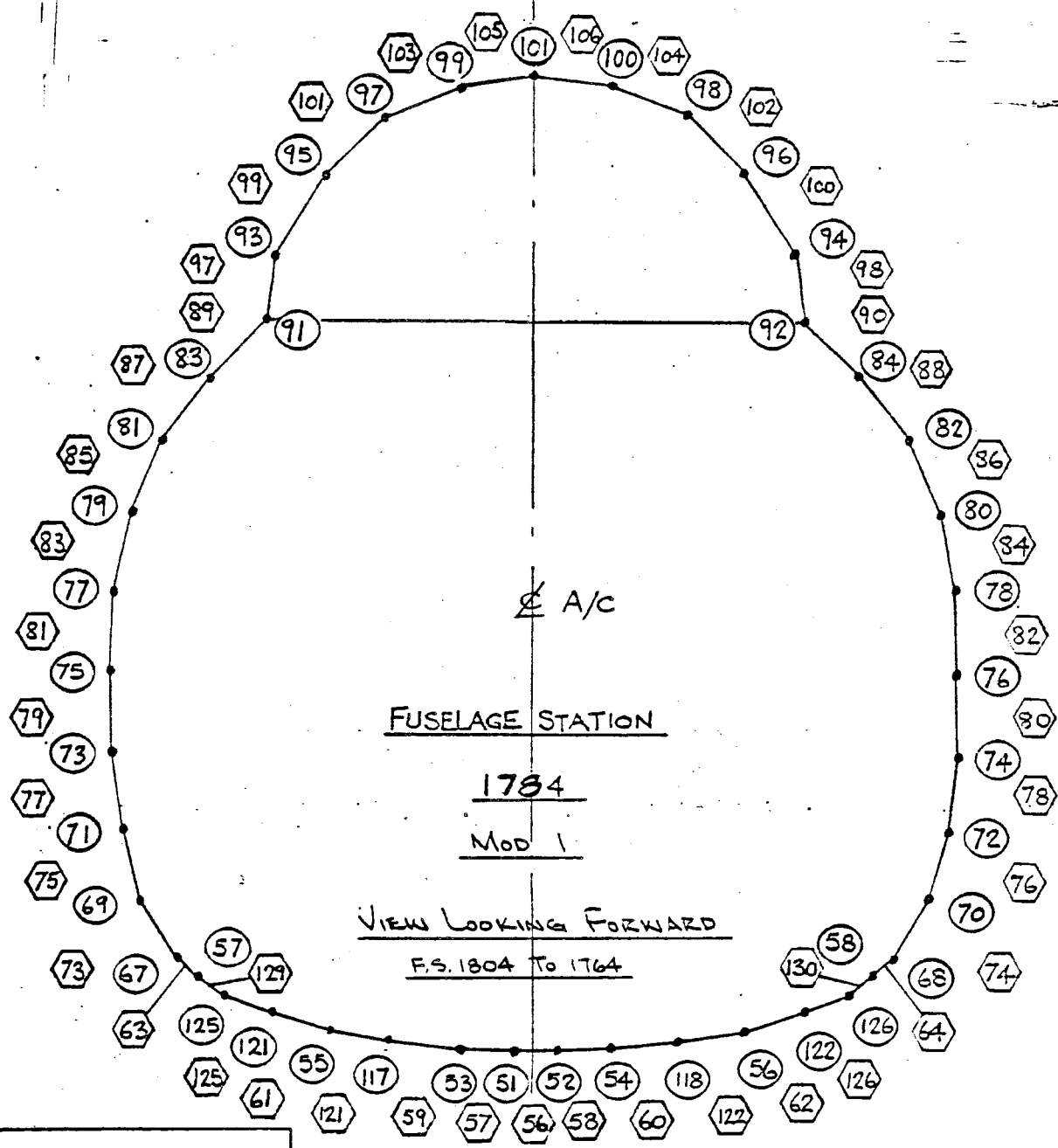
6.0 SHELL STRUCTURE F.S. 1744 TO 2101
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION 1PC242)MOD 1

1	1	-.380003+02	2	.759643+02	3	-.788127+02	4	.724654+02
	5	-.948569+02	6	.546949+02	7	-.484600+02	8	.749784+02
	9	.471052+03	10	.459514+03	11	-.271412+01	12	-.767580+02
	13	-.408817+02	14	-.110946+03	15	-.111776+03	16	-.168563+03
	17	-.162312+03	18	-.108556+03	19	.588721+03	20	.597596+03
	21	.421839+03	22	.447799+03	23	.348281+03	24	.377601+03
	25	.314199+03	26	.333844+03	27	.340923+03	28	.367258+03
	29	.391828+03	30	.407408+03	31	.484552+03	32	.496147+03
	33	.544588+03	34	.539523+03	35	.394891+03	36	.526401+03
	37	.118831+02	38	.528984+02	39	-.136582+02	40	.488931+02
	41	-.113786+03	42	.186608+02	43	.147061+03	44	.486600+03
	45	.283209+03	46	.695742+03	47	.688081+03	48	.283503+03
	49	-.680344+02	50	.257506+03	51	-.195456+01	52	.103028+03
	53	.292372+02	54	.354846+02	55	-.268138+02	56	.678048+02
	57	-.387482+02	58	.939279+02	59	.941981+01	60	.136616+03
	61	.116632+03	62	.226059+03	63	.604838+03	64	.639194+03
	65	-.712289+01	66	-.595091+02	67	-.335632+02	68	-.839680+02
	69	-.312064+02	70	-.761685+02	71	.488205+02	72	.892966+02
	73	.589069+03	74	.614040+03	75	.614631+03	76	.639384+03
	77	.584277+03	78	.609383+03	79	.518067+03	80	.540439+03
	81	.479663+03	82	.521325+03	83	.480383+03	84	.525281+03
	85	.472823+03	86	.533316+03	87	.437657+03	88	.532429+03
	89	.389661+03	90	.536845+03	91	-.311509+02	92	.460910+02
	93	-.524534+02	94	.548329+02	95	-.682449+02	96	.528440+02
	97	.277098+03	98	.554227+03	99	.399803+03	100	.569258+03
	101	.471476+03	102	.372438+03	103	.111478+03	104	.305829+03
	105	-.243101+02	106	.129599+03	107	.865757+02	108	.148140+03
	109	.901951+02	110	-.513059+02	111	-.113347+03	112	-.168939+03
	113	-.525021+01	114	.114894+03	115	.972849+02	116	.153779+03
	117	-.709288+02	118	-.184424+03	119	.424099+02	120	.895945+02
	121	-.846166+02	122	-.197433+03	123	-.102131+02	124	-.539436+02
	125	.165979+03	126	.264438+03	127	.104652+01	128	.409000+02
	129	-.245229+03	130	-.326184+03				

LOADS ARE IN POUNDS/INCH

REV. (A) M.B. 1 AUG 69.

6.3.2 PANEL Loads (Continued)



KEY

 GRID POINT

 FAMAS STRINGER

 FAMAS PANEL

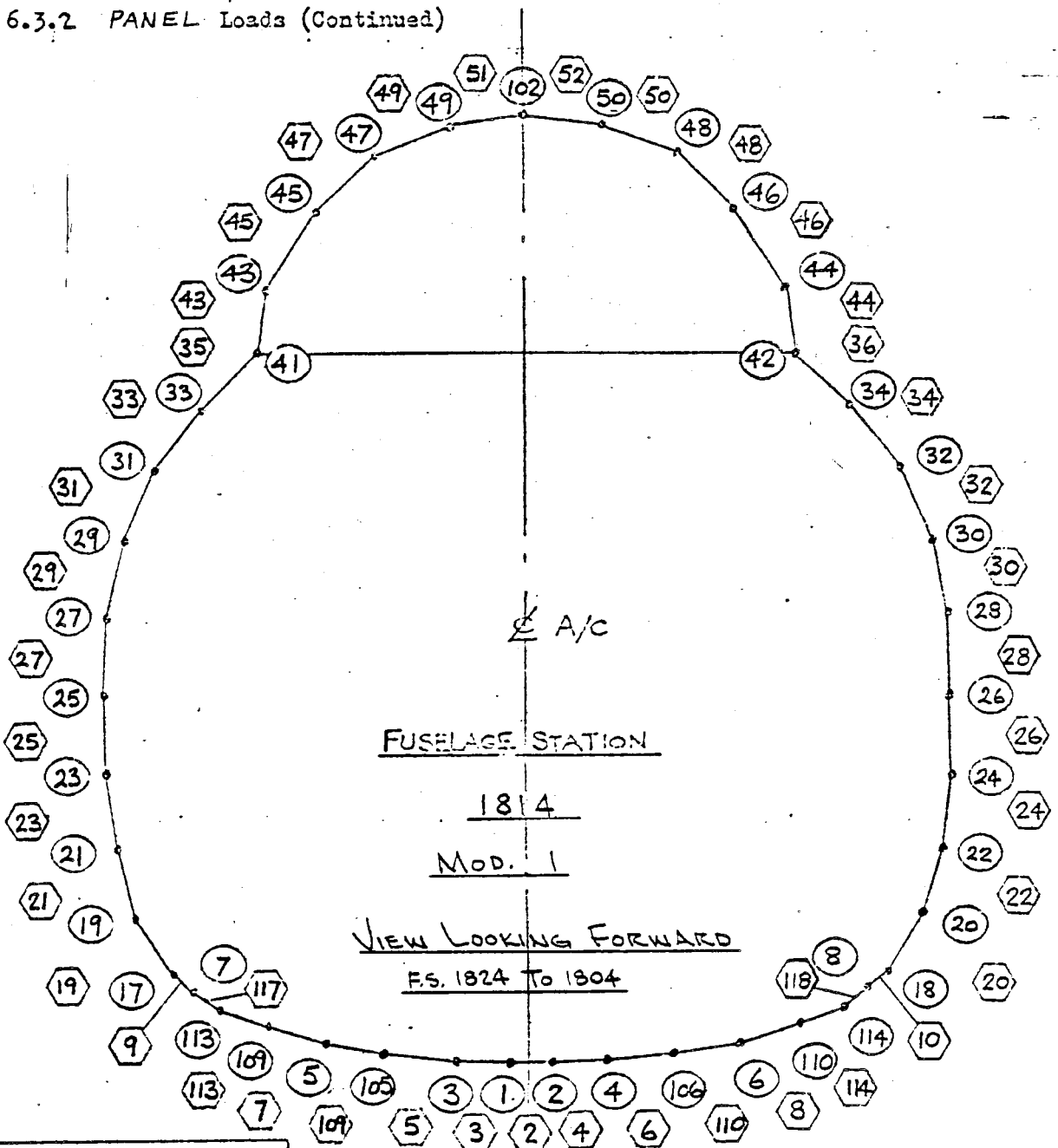
FIGURE No 6.3.2.2

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 5.2

Prepared	NAME V.W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Page	Vol. I 6.68
Approved	L.G. Darby	16 Aug 68	Model C-5A LGJUS 45-2-2 Report No.		

REV (A) M.B. 1 AUG 69

6.3.2 PANEL Loads (Continued)



KEY

—●— GRID POINT

○ FAMAS STRINGER

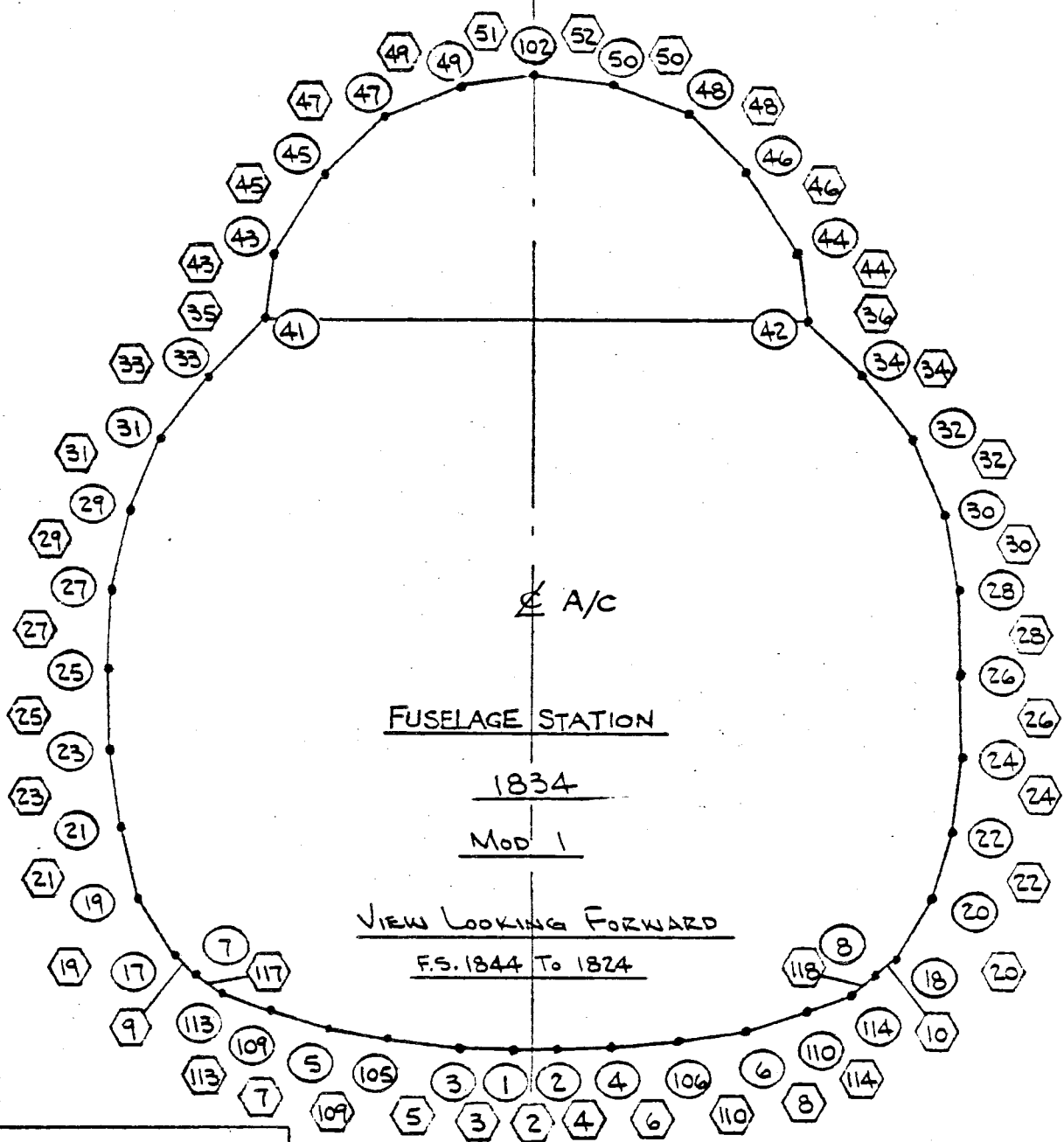
⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF B
VOLUME IV SECTION 52.

FIGURE No 6.3.2.3

REV. (A) M.F.D. 1 AUG. 69.

6.3.2 PANEL Loads (Continued)



KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FIGURE No 6.3.2.4

FOR AREAS AND THICKNESSES SEE REF 8.
VOLUME IV SECTION 5.2

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Page Vol. I	6.70
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Model C-5A	LG10S46-2-
					Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON.1
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS <CONDITION 5PM027 I MOD 1

1	1	.435541+01	2	-.651600+01	3	.296246+03	4	.282084+03
*	5	.566954+03	6	.551979+03	7	.646234+03	8	.626912+03
	9	.154319+03	10	.143657+03	11	.148806+03	12	.157442+03
	13	.309321+03	14	.318385+03	15	.687366+03	16	.695864+03
	17	-.101099+04	18	-.104169+04	19	.941778+03	20	.935353+03
	21	.105268+04	22	.104496+04	23	.119718+04	24	.118849+04
	25	.163585+04	26	.162255+04	27	.435618+03	28	.455833+03
	29	.715218+03	30	.744373+03	31	.392486+03	32	.932562+03
	33	.833529+03	34	.920024+03	35	.531058+03	36	.798302+03
	37	-.125643+03	38	-.523934+02	39	-.224824+03	40	-.525026+02
	41	-.336224+03	42	-.359731+02	43	.106356+04	44	.449018+03
	45	.468784+03	46	.558236+03	47	-.408218+03	48	.135614+03
	49	.184177+03	50	.833513+02	51	.107848+03	52	.602876+02
	53	.232324+02	54	.386967+02	55	.289073+01	56	-.764989+01
	57	.242196+03	58	.227587+03	59	.470590+03	60	.455395+03
	61	.568099+03	62	.550418+03	63	.138477+04	64	.137868+04
	65	.149871+03	66	.155889+03	67	.291576+03	68	.298312+03
	69	.582928+03	70	.593716+03	71	-.637499+03	72	-.650228+03
	73	.111032+04	74	.110147+04	75	.137011+04	76	.136222+04
	77	.150584+04	78	.149876+04	79	.146370+04	80	.146329+04
	81	.103904+04	82	.105738+04	83	.877051+03	84	.919791+03
	85	.801324+03	86	.866020+03	87	.724475+03	88	.817760+03
	89	.681547+03	90	.731977+03	91	-.105522+03	92	-.142648+02
	93	-.111929+03	94	-.187974+02	95	.978792+01	96	-.236310+02
	97	.842237+03	98	.492589+03	99	.387712+03	100	.372682+03
	101	-.937909+02	102	.270978+03	103	.302650+02	104	.357500+02
	105	.710281+02	106	.174168+02	107	-.530320+03	108	-.539327+03
	109	-.711189+03	110	-.693444+03	111	.737174+03	112	.744109+03
	113	.587078+03	114	.564029+03	115	-.750433+03	116	-.759093+03
	117	-.363120+03	118	-.335290+03	119	-.462081+03	120	-.470209+03
	121	-.539351+03	122	-.522554+03	123	.639191+03	124	.654045+03
	125	.539446+03	126	.521732+03	127	-.713979+03	128	-.731439+03
	129	-.469428+03	130	-.452738+03				

* PANEL (5) REF. P. 6.68 ULTIMATE PANEL SHEAR FLOW = 567 LBS./IN.

NAME RF WILKINSON		DATE 1 JUN 69		LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. No. I	PERM. 6.71
Prepared					Page		
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS			Model C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE			Report No. LG1US46-2-	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON-1
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS <CONDITION TPM096+P I MOD 1

1	1	.218109+01	2	-.726015+01	3	.360524+03	4	.344806+03
	5	.625903+03	6	.609103+03	7	.666992+03	8	.645016+03
	9	.210315+03	10	.196976+03	11	.196433+03	12	.201056+03
	13	.385390+03	14	.390961+03	15	.695903+03	16	.704587+03
	17	-.120904+04	18	-.124528+04	19	.103627+04	20	.103064+04
	21	.113973+04	22	.113216+04	23	.124381+04	24	.123528+04
	25	.151895+04	26	.150767+04	27	.353707+03	28	.379333+03
	29	.555621+03	30	.606637+03	31	.724959+03	32	.827929+03
	33	.530213+03	34	.737455+03	35	.205947+03	36	.593692+03
	37	-.969752+02	38	.346649+02	39	-.355067+03	40	-.111296+03
	41	-.241679+03	42	.112238+03	43	.131188+04	44	.276234+03
	45	.431795+03	46	.506449+03	47	-.797390+03	48	.134024+03
	49	.323331+03	50	.102510+03	51	.155544+03	52	.859941+02
	53	.511429+02	54	.610408+02	55	.912892-01	56	-.839445+01
	57	.249386+03	58	.233203+03	59	.547022+03	60	.530180+03
	61	.682955+03	62	.663116+03	63	.134450+04	64	.134117+04
	65	.185293+03	66	.185933+03	67	.361206+03	68	.363141+03
	69	.686209+03	70	.696148+03	71	-.398900+03	72	-.416248+03
	73	.984238+03	74	.976744+03	75	.114593+04	76	.114242+04
	77	.123752+04	78	.123735+04	79	.125591+04	80	.126797+04
	81	.100450+04	82	.103868+04	83	.782773+03	84	.843622+03
	85	.615158+03	86	.733690+03	87	.566138+03	88	.699549+03
	89	.573972+03	90	.611229+03	91	-.121941+03	92	.697820+01
	93	-.127197+03	94	-.169111+02	95	.863613+02	96	.212261+02
	97	.853774+03	98	.287018+03	99	.265563+03	100	.241464+03
	101	-.299369+03	102	.277361+03	103	.164639+02	104	.318014+01
	105	.982992+02	106	.515972+01	107	-.590506+03	108	-.597809+03
	109	-.742228+03	110	-.722186+03	111	.589945+03	112	.698450+03
	113	.640505+03	114	.613796+03	115	-.673207+03	116	-.684639+03
	117	-.447791+03	118	-.415030+03	119	-.550645+03	120	-.555370+03
	121	-.648885+03	122	-.630214+03	123	.759080+03	124	.774604+03
	125	.614810+03	126	.595080+03	127	-.831569+03	128	-.850931+03
	129	-.532912+03	130	-.514493+03				

LOADS ARE IN POUNDS / INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. JUL 16, 72	PERM. 6, 72
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1US46-2-

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON.1

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS <CONDITION ALSIII I MOD 1

1	1	-.263102+02	2	-.866709+02	3	.369296+03	4	.200545+03
	5	.631501+03	6	.473988+03	7	.686383+03	8	.562234+03
	9	.179734+03	10	.895873+02	11	.177514+03	12	.128959+03
	13	.333429+03	14	.293749+03	15	.686143+03	16	.702312+03
	17	-.980949+03	18	-.106797+04	19	.954746+03	20	.894412+03
	21	.105088+04	22	.102343+04	23	.117713+04	24	.118568+04
	25	.158299+04	26	.166841+04	27	.433012+03	28	.437529+03
	29	.692676+03	30	.752001+03	31	.956587+03	32	.945448+03
	33	.778017+03	34	.958144+03	35	.478918+03	36	.851582+03
	37	-.321403+02	38	-.140938+03	39	-.144437+03	40	-.137084+03
	41	-.268680+03	42	-.121416+03	43	.106576+04	44	.449052+03
	45	.453077+03	46	.574515+03	47	-.421240+03	48	.159953+03
	49	.155868+03	50	.127820+03	51	.627421+02	52	.108061+03
	53	-.792558+02	54	-.736954+02	55	-.390182+02	56	-.851394+02
	57	.311116+03	58	.146292+03	59	.527420+03	60	.370748+03
	61	.605120+03	62	.465638+03	63	.136937+04	64	.132639+04
	65	.189785+03	66	.115494+03	67	.328062+03	68	.260453+03
	69	.597279+03	70	.581262+03	71	-.643855+03	72	-.675804+03
	73	.109325+04	74	.104425+04	75	.134201+04	76	.132324+04
	77	.146646+04	78	.147751+04	79	.141412+04	80	.146104+04
	81	.101070+04	82	.103886+04	83	.846418+03	84	.917033+03
	85	.759284+03	86	.882675+03	87	.673963+03	88	.855306+03
	89	.612278+03	90	.797510+03	91	.141027-00	92	-.121695+03
	93	-.775839+01	94	-.133538+03	95	.116235+03	96	-.152606+03
	97	.848139+03	98	.476465+03	99	.378580+03	100	.384492+03
	101	-.109278+03	102	.294331+03	103	.536171-00	104	.861181+02
	105	.274953+02	106	.659356+02	107	-.541878+03	108	-.530046+03
	109	-.763038+03	110	-.624936+03	111	.727574+03	112	.759000+03
	113	.622854+03	114	.493856+03	115	-.734005+03	116	-.779700+03
	117	-.399105+03	118	-.251013+03	119	-.487652+03	120	-.444411+03
	121	-.582004+03	122	-.438572+03	123	.646459+03	124	.651605+03
	125	.567972+03	126	.439619+03	127	-.716095+03	128	-.739751+03
	129	-.490724+03	130	-.371844+03				

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. No. I	PERM. 6.73
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1US46-2-	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON. I
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS <CONDITION DMLOOI I MOD 1

1	1	- .404169+03	2	.319198+03	3	- .260592+03	4	.424268+03
	5	- .227262+03	6	.545980+03	7	- .213030+03	8	.566347+03
	9	.273368+03	10	- .248145+03	11	.419638+03	12	- .329631+03
	13	.405327+03	14	- .222077+03	15	.207912+03	16	.187186+03
	17	- .140790+03	18	- .447015+03	19	.139042+03	20	.343416+03
	21	- .588674+02	22	.614401+03	23	- .236603+03	24	.875448+03
	25	- .624669+03	26	.149171+04	27	- .486736+02	28	.267871+03
	29	- .288600+03	30	.644060+03	31	- .428350+03	32	.813695+03
	33	- .674052+03	34	.966406+03	35	- .543161+03	36	.947770+03
	37	.151694+03	38	- .222526+03	39	.176022+03	40	- .224257+03
	41	.277627+03	42	- .246578+03	43	.665939+02	44	.617965+03
	45	- .450942+03	46	.917114+03	47	- .120026+04	48	.457110+03
	49	- .216100+03	50	.572104+03	51	- .546858+03	52	.609443+03
	53	- .238754+03	54	- .204108+03	55	- .334861+03	56	.314970+03
	57	- .254781+03	58	.381869+03	59	- .201407+03	60	.448586+03
	61	- .147007+03	62	.432693+03	63	- .537779+02	64	.761519+03
	65	.357422+03	66	- .267775+03	67	.358885+03	68	- .186577+03
	69	.220324+03	70	.117093+03	71	- .320940+02	72	- .340589+03
	73	- .195999+02	74	.562642+03	75	- .147642+03	76	.834534+03
	77	- .288945+03	78	.104952+04	79	- .409100+03	80	.114844+04
	81	- .215014+03	82	.750538+03	83	- .268752+03	84	.745202+03
	85	- .427225+03	86	.807698+03	87	- .460280+03	88	.869225+03
	89	- .478052+03	90	.922593+03	91	.193755+03	92	- .205570+03
	93	.247532+03	94	- .225851+03	95	.329237+03	96	- .286940+03
	97	- .138136+03	98	.651649+03	99	- .555428+03	100	.715474+03
	101	- .856685+03	102	.590936+03	103	- .502263+03	104	.601197+03
	105	- .508933+03	106	.584900+03	107	- .306741+03	108	- .203252+01
	109	.238692+03	110	- .636997+03	111	.148098+03	112	.271313+03
	113	- .199134+03	114	.510395+03	115	- .110090+03	116	- .314053+03
	117	.137696+03	118	- .307919+03	119	- .293356+03	120	.241579+02
	121	.157567+03	122	- .431509+03	123	.167620+03	124	.203217+03
	125	- .126172+03	126	.388779+03	127	- .109907+03	128	- .307615+03
	129	.731940+02	130	- .289007+03				

LOADS ARE IN POUNDS / INCH

	NAME	DATE	LOCKHEED-GEORGIA COMPANY			TEMP.	PERM.
Prepared	M. Burton	Nov 70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	673.1
Checked	A. D. Pinchot	Nov 70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A		
Approved	H. H. Wade	Nov 70			LG1US46-2-2 Report No.		

REV (B) M.B. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (continued)

6.3 SKINS AND STRINGERS

6.3.2 Skin Loads (Condition DML00R) Mod.1

1	.406061+03	2	-.322649+03	3	.428273+03	4	-.264105+03
5	.550502+03	6	-.230739+03	7	.576063+03	8	-.213561+03
9	-.2229287+03	10	.285836+03	11	-.330852+03	12	.422225+03
13	-.222155+03	14	.409499+03	15	.194541+03	16	.220060+03
17	-.424780+03	18	-.134781+03	19	.349893+03	20	.141304+03
21	.622454+03	22	-.549181+02	23	.887605+03	24	-.228204+03
25	.152244+04	26	-.598187+03	27	.263922+03	28	-.391023+02
29	.657058+03	30	-.254906+03	31	.888130+03	32	-.322888+03
33	.108894+04	34	-.489966+03	35	.802331+03	36	-.538459+03
37	-.232195+03	38	.188070+03	39	-.309556+03	40	.191054+03
41	-.460279+03	42	.230523+03	43	.533212+03	44	-.406337+03
45	.685314+03	46	-.657323+03	47	.874618+03	48	-.403678+03
49	.328902+03	50	-.545687+03	51	.604420+03	52	-.579468+03
53	.253753+03	54	.226863+03	55	.335828+03	56	-.318941+03
57	.387661+03	58	-.256549+03	59	.456961+03	60	-.200903+03
61	.444643+03	62	-.144214+03	63	.777052+03	64	-.412683+02
65	-.268497+03	66	.358827+03	67	-.187278+03	68	.360806+03
69	.115894+03	70	.224407+03	71	-.345116+03	72	-.441218+02
73	.584327+03	74	-.173042+01	75	.860292+03	76	-.124417+03
77	.108107+04	78	-.258438+03	79	.118227+04	80	-.371628+03
81	.754240+03	82	-.198593+03	83	.716444+03	84	-.271332+03
85	.832407+03	86	-.362820+03	87	.826489+03	88	-.451001+03
89	.824361+03	90	-.556330+03	91	-.255800+03	92	.195730+03
93	-.309234+03	94	.214239+03	95	-.315496+03	96	.276837+03
97	.591324+03	98	-.421535+03	99	.728904+03	100	-.553880+03
101	.740795+03	102	-.466531+03	103	.502260+03	104	-.608023+03
105	.547432+03	106	-.582444+03	107	-.508214+01	108	-.314407+03
109	-.643444+03	110	.241759+03	111	.282449+03	112	.163634+03
113	.522948+03	114	-.198758+03	115	-.326351+03	116	-.128226+03
117	-.325197+03	118	.134990+03	119	.246332+02	120	-.296508+03
121	-.443159+03	122	.154716+03	123	.199273+03	124	.171329+03
125	.402124+03	126	-.121899+03	127	-.300011+03	128	-.111518+03
129	-.302559+03	130	.680831+02				

LOADS ARE IN POUNDS/INCH

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. JUL 1	PERM. 6.74
Prepared			TITLE INTERNAL LOADS ANALYSIS		Model C-5A		
Checked	A L PINCHOT	1 JUN 69			LG1US46-2-		
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No.		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 <CON.1

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS <CONDITION TPM096 I MOD 1

1	1	.296010+01	2	-.583971+01	3	.300464+03	4	.287679+03
	5	.579787+03	6	.566125+03	7	.651498+03	8	.633768+03
	9	.385616+02	10	.298685+02	11	.162214+03	12	.168223+03
	13	.333902+03	14	.340738+03	15	.729138+03	16	.737567+03
	17	-.994359+03	18	-.102238+04	19	.816953+03	20	.810922+03
	21	.966444+03	22	.960088+03	23	.113006+04	24	.112400+04
	25	.157627+04	26	.156962+04	27	.354448+03	28	.379606+03
	29	.623109+03	30	.661674+03	31	.776547+03	32	.834532+03
	33	.695769+03	34	.807846+03	35	.429808+03	36	.688465+03
	37	-.140891+03	38	-.585633+02	39	-.230592+03	40	-.554517+02
	41	-.311951+03	42	-.264921+02	43	.103995+04	44	.348441+03
	45	.392350+03	46	.420560+03	47	-.619086+03	48	.774231+02
	49	.197001+03	50	.315406+02	51	.970215+02	52	.486033+02
	53	.270159+02	54	.398631+02	55	.132521+01	56	-.673134+01
	57	.236665+03	58	.223893+03	59	.454936+03	60	.441667+03
	61	.527170+03	62	.511615+03	63	.124538+04	64	.124172+04
	65	.161911+03	66	.164896+03	67	.311848+03	68	.315673+03
	69	.603637+03	70	.612640+03	71	-.668962+03	72	-.682471+03
	73	.969251+03	74	.963893+03	75	.122837+04	76	.122535+04
	77	.137265+04	78	.137216+04	79	.134461+04	80	.135238+04
	81	.926202+03	82	.949518+03	83	.761257+03	84	.808599+03
	85	.685140+03	86	.754017+03	87	.616557+03	88	.705637+03
	89	.584805+03	90	.614526+03	91	-.106949+03	92	-.162748+02
	93	-.105323+03	94	-.202128+02	95	.243482+02	96	-.202046+02
	97	.775509+03	98	.376472+03	99	.275866+03	100	.253294+03
	101	-.237517+03	102	.198051+03	103	-.119307+02	104	-.293077+02
	105	.645917+02	106	-.986728-00	107	-.566355+03	108	-.574090+03
	109	-.725491+03	110	-.709266+03	111	.779228+03	112	.787145+03
	113	.581125+03	114	.559926+03	115	-.788883+03	116	-.799357+03
	117	-.336536+03	118	-.310793+03	119	-.485156+03	120	-.491001+03
	121	-.505571+03	122	-.490797+03	123	.655900+03	124	.669068+03
	125	.486858+03	126	.471316+03	127	-.730785+03	128	-.746421+03
	129	-.399048+03	130	-.384489+03				

LOADS ARE IN POUNDS / INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 6.75
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON. I)

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS (CONDITION 1PC242

I MOD I

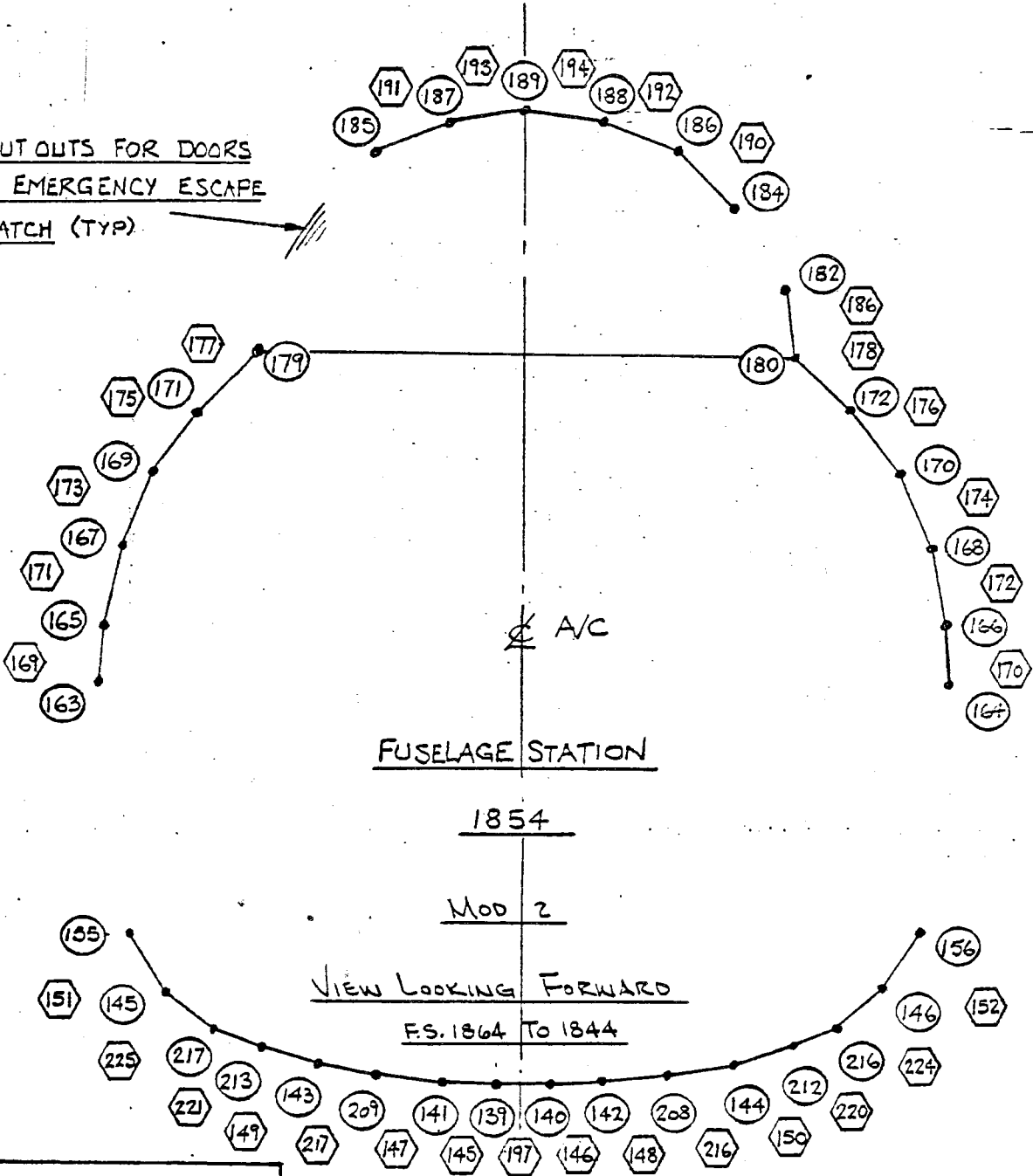
1	1	-.380003+02	2	.759643+02	3	-.788127+02	4	.724654+02
	5	-.948569+02	6	.546949+02	7	-.484600+02	8	.749784+02
	9	.471052+03	10	.459514+03	11	-.271412+01	12	-.767580+02
	13	-.408817+02	14	-.110946+03	15	-.111776+03	16	-.168563+03
	17	-.162312+03	18	-.108556+03	19	.588721+03	20	.597596+03
	21	.421839+03	22	.447799+03	23	.348291+03	24	.377601+03
	25	.314199+03	26	.333844+03	27	.340923+03	28	.367258+03
	29	.391823+03	30	.407408+03	31	.484552+03	32	.496147+03
	33	.544588+03	34	.539523+03	35	.394891+03	36	.526401+03
	37	.118831+02	38	.528984+02	39	-.136582+02	40	.488931+02
	41	-.113786+03	42	.186608+02	43	.147061+03	44	.486600+03
	45	.283209+03	46	.695742+03	47	.688031+03	48	.283503+03
	49	-.680344+02	50	.257506+03	51	-.195456+01	52	.103028+03
	53	.292372+02	54	.354846+02	55	-.268138+02	56	.678048+02
	57	-.387482+02	58	.939279+02	59	.941981+01	60	.136616+03
	61	.116632+03	62	.226059+03	63	.604838+03	64	.639194+03
	65	-.712289+01	66	-.595091+02	67	-.335632+02	68	-.839680+02
	69	-.312064+02	70	-.761685+02	71	.488205+02	72	.892966+02
	73	.589069+03	74	.614040+03	75	.614631+03	76	.639384+03
	77	.584277+03	78	.609383+03	79	.518067+03	80	.540439+03
	81	.479663+03	82	.521325+03	83	.480383+03	84	.525281+03
	85	.472823+03	86	.533316+03	87	.437657+03	88	.532429+03
	89	.389661+03	90	.536845+03	91	-.311509+02	92	.460910+02
	93	-.524534+02	94	.548329+02	95	-.682449+02	96	.528440+02
	97	.2177098+03	98	.554227+03	99	.399803+03	100	.569258+03
	101	.471476+03	102	.372439+03	103	.111478+03	104	.305829+03
	105	-.243101+02	106	.129599+03	107	.365757+02	108	.148140+03
	109	.901951+02	110	-.513059+02	111	-.113347+03	112	-.168939+03
	113	-.525021+01	114	.114894+03	115	.972849+02	116	.153779+03
	117	-.709288+02	118	-.184424+03	119	.424099+02	120	.895945+02
	121	-.846166+02	122	-.197433+03	123	-.102131+02	124	-.539436+02
	125	.165979+03	126	.264438+03	127	.104652+01	128	.409000+02
	129	-.245229+03	130	-.326184+03				

LOADS ARE IN POUNDS / INCH

6.3.2 PANEL Loads (Continued)

REV A MB. (AUG 69)

CUT OUTS FOR DOORS
& EMERGENCY ESCAPE
HATCH (TYP)



KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FIGURE No 6.3.2.5

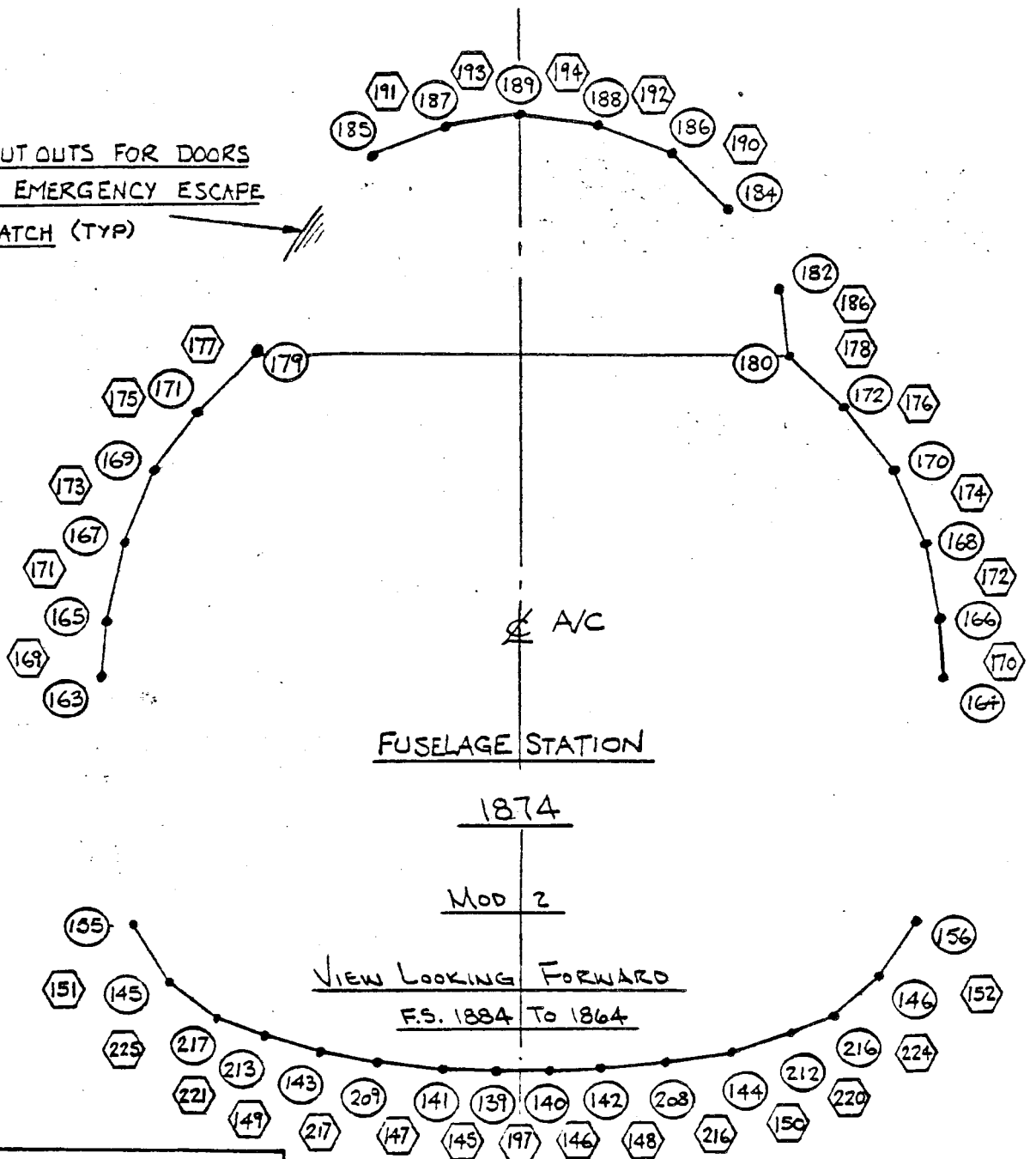
FOR AREAS AND THICKNESSES SEE REF8.
VOLUME IV SECTION 5.2

Prepared	V.W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Vol. I	PERM. 6.77
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Page	C-5A
Approved	L.G. Darby	16 Aug 68		Model	IGIUS 46-2-2 Report No.

6.3.2 PANEL Loads (Continued)

REV (A) M.B. 1 AUG 69.

CUT OUTS FOR DOORS
& EMERGENCY ESCAPE
HATCH (TYP)



KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

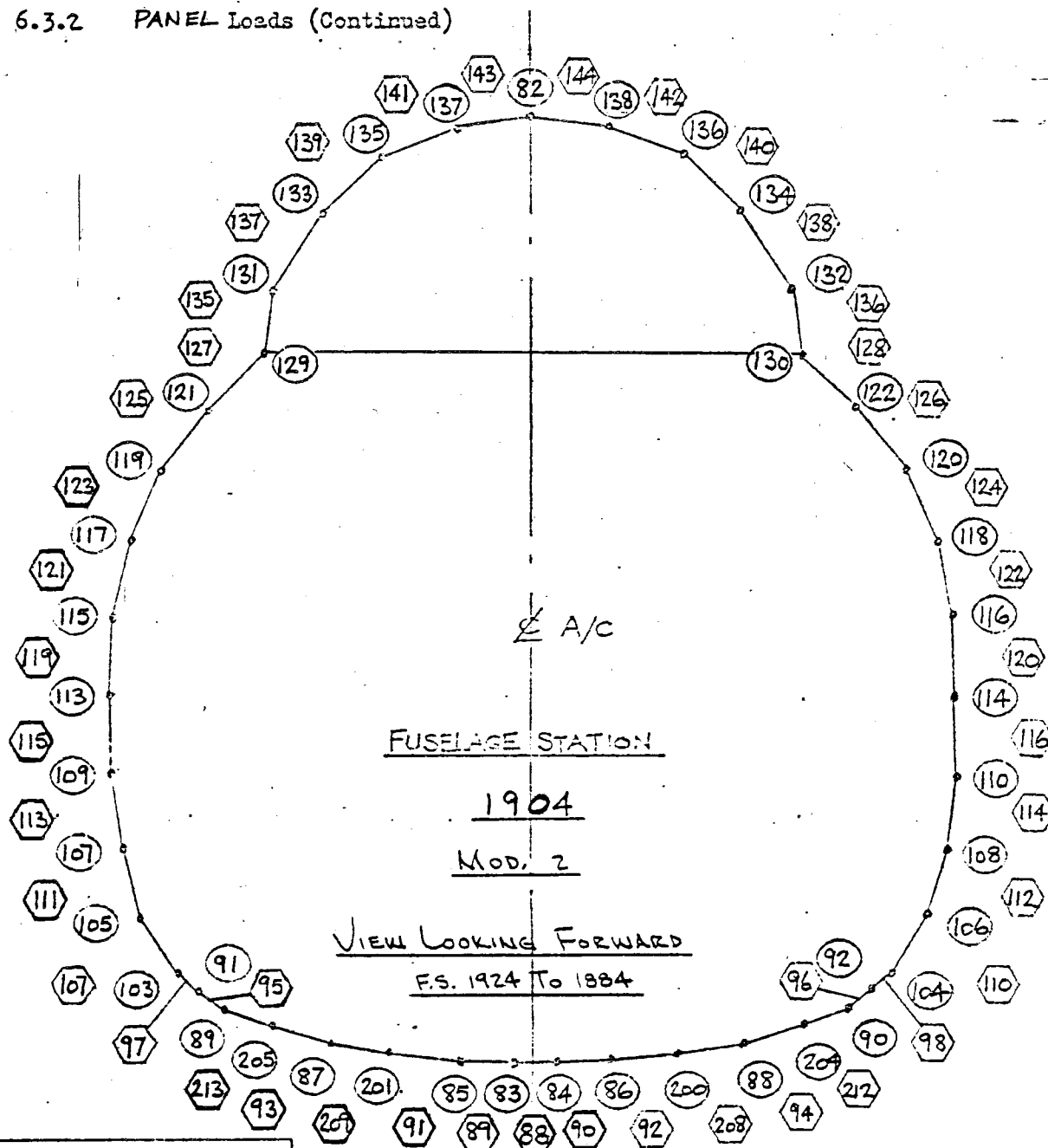
FIGURE No 6.3.2.6

FOR AREAS AND THICKNESSES SEE REF 3
VOLUME IV SECTION 5.2

Prepared	NAME V.W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	REV. A Vol. I 678
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved	L.C. Darby	16 Aug 68		IGLUS	45-2-2
				Report No.	

REV A MB. 1 AUG 69.

6.3.2 PANEL Loads (Continued)



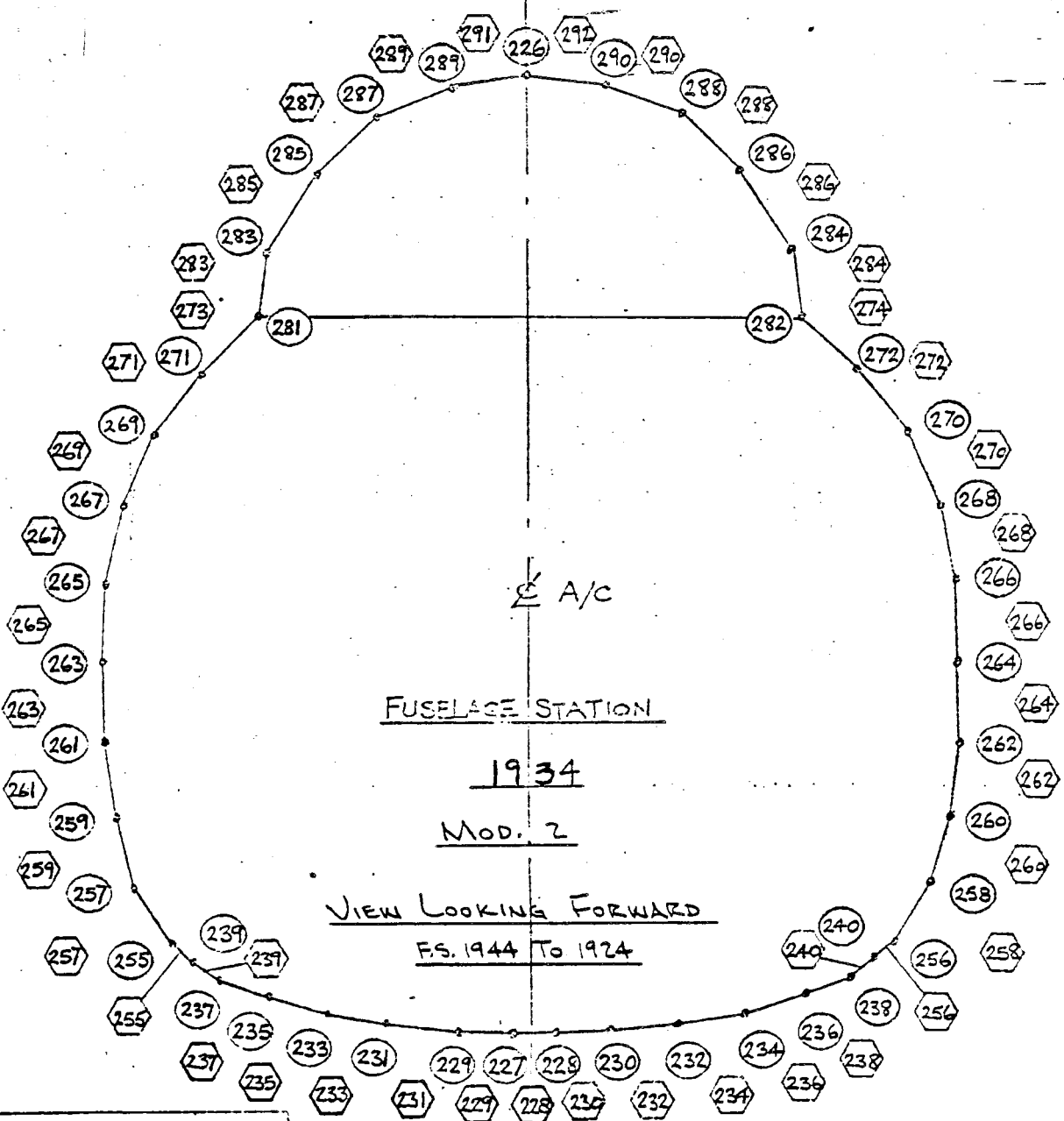
KEY	
	GRID POINT
	FAMAS STRINGER
	FAMAS PANEL

FIGURE No 6.3.2.7

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

REV (A) MB 1 AUG 69

6.3.2 PANEL Loads (Continued)



KEY

—●— GRID POINT

○ FAMAS STRINGER

⬡ FAMAS PANEL

FIGURE No 6.3.2.8

FOR AREAS AND THICKNESSES SEE REF. VOLUME IV SECTION 5.2.

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	FF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.31
Checked	A L FINCHET	1JUN69	TITLE		Model C-5A	
			INTERNAL LCADS ANALYSIS		LG1LS46-2-2	
Approved	L C CAFEY	1JUN69	AFT FUSELAGE		Report No.	

REV (A) RFW 8-1-69

5.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.2 SKIN LCADS CONDITION TPM096+P & MOD 2

1	* 1	- .225912+03	2	- .229245+03	3	- .354179+03	4	- .356033+03
	5	.307080+03	6	.293387+03	7	.879190+03	8	.854442+03
	9	.365902+04	10	.365807+04	11	.120535+03	12	.122712+03
	13	.241359+03	14	.246360+03	15	.969694+03	16	.998191+03
	17	.194115+04	18	.194956+04	19	.669113+04	20	.682678+04
	21	.372065+03	22	.324279+03	27	.730725+03	28	.724558+03
	31	.811179+03	32	.801120+03	35	.633113+03	36	.613654+03
	43	.642367+03	44	.610944+03	47	.721551+03	48	.714967+03
	51	.620804+03	52	.528856+03	55	.653763+03	56	-.183553+03
	59	.557987+03	60	.545448+03	63	.223259+03	64	.448224+03
	68	.393031+02	66	.307918+02	67	.232843+03	68	.251699+03
	69	.495561+03	70	.527919+03	71	.439531+03	72	.506290+03
	73	.255616+02	74	.436314+03	75	.526522+03	76	.437966+03
	77	.674563+03	78	.332710+03	79	-.847434+02	80	-.993109+02
	81	-.255495+03	82	-.525319+02	83	-.213086+02	84	.863572-00
	85	-.243133+01	86	-.276217+02	87	.123205+01	88	-.297850+01
	89	.208934+03	90	.201795+03	91	.399403+03	92	.393311+03
	93	.628314+03	94	.625064+03	95	.641158+03	96	.642147+03
	97	.165166+04	98	.166165+04	99	.140064+03	100	.142465+03
	101	.259093+03	102	.262529+03	103	.893371+03	104	.896372+03
	105	.100864+04	106	.101085+04	107	-.925522+03	108	-.924400+03
	109	.167451+04	110	.169445+04	111	.102975+04	112	.104387+04
	113	.675758+03	114	.680852+03	115	.501824+03	116	.472442+03
	119	.489659+03	120	.361339+03	121	.610965+03	122	.501939+03
	123	.789333+03	124	.829884+03	125	.881800+03	126	.631816+03
	127	.727614+03	128	.656764+03	129	.148141+03	130	.672168+02
	131	-.682098+02	132	-.125261+03	133	-.365222+02	134	-.372845+02
	135	-.491560+03	136	.408508+03	137	.193271+03	138	.443906+03
	139	.151359+04	140	.158285+03	141	-.375618+03	142	-.119541+02
	143	-.433689+02	144	.430321+02	145	.330182+03	146	.320766+03
	147	.725106+03	148	.715701+03	149	.158742+04	150	.158141+04
	151	.119393+04	152	.120414+04	153	.158334+03	154	.161102+03
	155	.310925+03	156	.313918+03	157	.712189+03	158	.719039+03
	159	.771826+03	160	.776711+03	169	.156397+04	170	.152867+04
	171	.142412+04	172	.137490+04	173	.593273+03	174	.864894+03
	175	.854955+03	176	.666457+03	177	.100714+04	178	.524049+03
	179	.158222+02	180	-.730947+01	181	.561199+02	182	-.226261+02
	183	.222863+03	184	.238444+02	186	.542211+03	190	.455801+03

* PANEL (1) REF. P. 6.80 ULTIMATE PANEL SHEAR FLOW = -226. LBS./IN.

NAME FF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.32
Prepared				Page	
Checked A L FINCHET	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved L C CAFEY	1 JUN 69			LG1LS46-2-2	
				Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CCN.6
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS CONDITION TFM096+P & MOD 2

1	191	.351183+03	192	.167813+03	193	.108299+03	194	.740647+02
	195	-.103450+01	196	.106798+01	197	-.458254+01	198	-.529231+03
	199	-.523959+03	200	-.548888+02	201	-.629092+02	202	-.134231+04
	203	-.133395+04	204	-.554633+03	205	-.573735+03	206	-.543118+03
	207	-.539285+03	208	-.635651+03	209	-.644399+03	210	-.100960+04
	211	-.100661+04	212	-.702477+03	213	-.703495+03	214	-.550348+03
	215	-.545623+03	216	-.122555+04	217	-.123724+04	218	-.720879+03
	219	-.711705+03	220	-.169836+04	221	-.170073+04	222	.625232+03
	223	.616813+03	224	.157028+04	225	.156879+04	226	.291556+02
	227	-.110361+01	228	.309401+01	229	-.163632+03	230	-.160796+03
	231	-.231819+03	232	-.230352+03	233	.307535+03	234	.296461+03
	235	-.546881+03	236	-.532537+03	237	.741440+03	238	.726202+03
	239	-.103383+04	240	-.101910+04	241	.757633+02	242	.715977+02
	243	.780137+01	244	.487374+01	245	.667831+03	246	.677732+03
	247	-.739532+03	248	-.741575+03	249	.104146+04	250	.104695+04
	251	-.133104+04	252	-.133360+04	253	.436698+03	254	.464955+03
	255	-.299446+04	256	-.299684+04	257	-.705851+03	258	-.706450+03
	259	-.738235+03	260	-.721580+03	261	-.847609+03	262	-.832874+03
	263	-.521421+03	264	-.521607+03	265	-.525956+03	266	-.557571+03
	267	-.595018+03	268	-.409516+03	269	-.653882+03	270	-.704518+03
	271	-.591934+03	272	-.512461+03	273	-.462743+03	274	-.581720+03
	275	.536373+02	276	.613229+02	277	.483789+03	278	.447128+03
	279	-.804401+02	280	-.853952+02	281	.438494+03	282	.320202+03
	283	.113692+03	284	-.443785+03	285	-.410916+03	286	-.400010+03
	287	-.725084+03	288	-.270441+02	289	-.124242+03	290	-.174574+03
	291	.835749+02	292	-.613815+02				

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUNE 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.33
					Page	
Checked	A L FINCHET	1 JUNE 69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	L C LAHEY	1 JUNE 69		LC1US46-2-2		
			AFT FUSELAGE	Report No.		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.8

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS CONDITION DMLCC1 8 MOD 2

1	1	-.85394E+02	2	.621611+02	3	-.122174+03	4	.180330+03
	5	-.413227+03	6	.603833+03	7	-.778856+03	8	.136421+04
	9	-.115137+04	10	.353289+04	11	.366103+03	12	-.389192+03
	13	.405721+03	14	-.413555+03	15	.151747+03	16	.152489+03
	17	-.385698+03	18	.138339+04	19	-.339422+04	20	.727297+04
	21	-.876091+03	22	.901213+03	27	-.356220+03	28	.738924+03
	31	-.293887+03	32	.617214+03	35	-.169637+03	36	.361956+03
	43	-.121524+03	44	.317917+03	47	-.204865+03	48	.518878+03
	51	-.196500+03	52	.824340+03	55	-.313947+03	59	-.454652+03
	60	.150088+04	63	-.617164+03	64	.106939+04	65	.173464+03
	66	-.189490+03	67	.280469+03	68	-.273043+03	69	.372246+03
	70	-.308720+03	71	.201972+03	72	-.272969+03	73	-.858065+03
	74	.669058+03	75	-.525410+03	76	.612222+03	77	-.185623+03
	78	.547308+03	79	-.350223+03	80	.477725+03	81	-.578580+03
	82	.543634+03	83	-.417457+03	84	-.392039+03	85	.476736+02
	86	-.336718+03	87	-.475221+03	88	.149951+03	89	-.131764+03
	90	.237166+03	91	-.162173+03	92	.365567+03	93	-.216069+03
	94	.484019+03	95	-.269640+03	96	.523393+03	97	-.698483+03
	98	.164931+04	99	.469543+03	100	-.420204+03	101	.436286+03
	102	-.322039+03	103	.126571+03	104	.299143+03	105	-.111335+02
	106	.536104+03	107	.292226+02	108	-.662914+03	109	-.520997+03
	110	.149607+04	111	-.338951+03	112	.955564+03	113	-.207740+03
	114	.613211+03	115	-.303592+01	116	.231573+03	119	-.401414+03
	120	.578435+03	121	-.433704+03	122	.713150+03	123	-.559267+03
	124	.119821+04	125	-.470176+03	126	.853010+03	127	-.250287+03
	128	.888250+03	129	.200385+03	130	-.245550+03	131	.256990+03
	132	-.242626+03	133	.426801+03	134	-.272701+03	135	-.103658+04
	136	.506410+03	137	-.397788+03	138	.651116+03	139	.656270+02
	140	.445680+03	141	-.512899+03	142	.547981+03	143	-.619992+03
	144	.559637+03	145	-.222563+03	146	.396102+03	147	-.207297+03
	148	.575619+03	149	-.312595+03	150	.111279+04	151	-.140992+03
	152	.821163+03	153	.478016+03	154	-.412751+03	155	.447305+03
	156	-.315190+03	157	.203663+03	158	.149270+03	159	-.322705+03
	160	.607671+03	169	-.563610+03	170	.140729+04	171	-.784283+03
	172	.151294+04	173	-.693913+03	174	.106248+04	175	-.736576+03
	176	.847444+03	177	-.111289+04	178	.692820+03	179	.163492+03
	180	-.251492+03	181	.527885+02	182	-.233513+03	183	.875946+02
	184	-.260441+03	186	.927415+03	190	.903531+03	191	-.100233+04

LOADS ARE IN POUNDS / INCH

	NAME FF WILKINSON	DATE 1JUN66	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 6.84
Checked	A L FINCHET	1JUN66	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L C CAFEY	1JUN66			LG10S46-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CON.8
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS CONDITION CMLCC1 & MOD 2

1	192	.580761403	193	-.496528+03	194	.483578+03	195	-.271346+03
	196	-.477412403	197	.267203+03	198	.213551+03	199	-.264207+03
	200	-.408041+03	201	.291676+03	202	-.656649+03	203	.726224+02
	204	-.554767+03	205	.613506+03	206	.135613+02	207	-.253329+03
	208	-.474772+03	209	.195091+03	210	-.462746+03	211	-.352206+02
	212	-.544103+03	213	.250637+03	214	.638564+02	215	-.317545+03
	216	-.851013+03	217	.268378+03	218	-.204571+03	219	-.156632+03
	220	-.117563+04	221	.297952+03	222	.175484+03	223	.135042+03
	224	.148806+04	225	-.407356+03	226	.355322+03	227	.434311+03
	228	-.563209+02	229	.625058+02	230	-.112272+03	231	.111080+03
	232	-.123290+03	233	-.260470+03	234	.338713+03	235	.367309+03
	236	-.539512+03	237	-.574367+03	238	.931437+03	239	.690120+03
	240	-.120381+04	241	-.429603+03	242	.401408+03	243	-.384542+03
	244	.324413+03	245	.356725+03	246	-.151934+02	247	-.389861+02
	248	-.417721+03	249	-.110801+03	250	.756601+03	251	.182668+03
	252	-.102852+04	253	.477525+03	254	-.362321+03	255	.103444+04
	256	-.284540+04	257	.433827+03	258	-.558415+03	259	.330615+03
	260	-.783330+03	261	.261335+03	262	-.657222+03	263	.767935+02
	264	-.270614+03	265	.271251+03	266	-.585459+03	267	.274464+03
	268	-.408316+03	269	.413575+03	270	-.531506+03	271	.527832+03
	272	-.108842+04	273	.555766+03	274	-.101830+04	275	-.231684+03
	276	.237403+03	277	-.818883+02	278	.190026+03	279	-.579473+03
	280	.412813+03	281	-.132485+03	282	.260727+03	283	.814139+03
	284	-.550702+03	285	.443032+03	286	-.637807+03	287	.129999+03
	288	-.542452+03	289	.385050+03	290	-.457850+03	291	.594080+03
	292	-.555610+03						

LOADS ARE IN POUNDS/INCH

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	6.84.1
Checked	A L PINCHOT	NOV70	TITLE		INTERNAL LOADS ANALYSIS		C-5A
Approved	H H WADE	NOV70			AFT FUSELAGE		Model
							LGIUS46-2-2
							Report No.

REV ⑤ M.B. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS (CONDITION DMLOOR) MOD 2

1	1	.615648+02	2	-.888509+02	3	.177755+03	4	-.127814+03
	5	.606999+03	6	-.416089+03	7	.137170+04	8	-.783137+03
	9	.352277+04	10	-.115294+04	11	-.390615+03	12	.387509+03
	13	-.414327+03	14	.408398+03	15	.149074+03	16	.152167+03
	17	.136851+04	18	-.398152+03	19	.723819+04	20	-.344702+04
	21	.911371+03	22	-.887205+03	27	.773110+03	28	-.322697+03
	31	.654682+03	32	-.260718+03	35	.399758+03	36	-.140499+03
	43	.352092+03	44	-.101271+03	47	.517015+03	48	-.214292+03
	51	.528634+03	52	-.345889+03	55	.753193+03	59	.932666+03
	60	-.771195+03	63	.971495+03	64	-.626472+03	65	-.202546+03
	66	.134283+03	67	-.331638+03	68	.241398+03	69	-.376187+03
	70	.328971+03	71	-.259197+03	72	.249616+03	73	.882030+03
	74	-.493905+03	75	.645113+03	76	-.518841+03	77	.446049+03
	78	-.510743+03	79	.377542+03	80	-.446761+03	81	.513357+03
	82	-.544203+03	83	.412729+03	84	.393428+03	85	-.489566+02
	86	.328473+03	87	.476327+03	88	-.151463+03	89	.234557+03
	90	-.138335+03	91	.359872+03	92	-.171457+03	93	.474095+03
	94	-.225804+03	95	.509814+03	96	-.280086+03	97	.166570+04
	98	-.673339+03	99	-.420440+03	100	.471371+03	101	-.321380+03
	102	.439752+03	103	.303145+03	104	.135098+03	105	.546832+03
	106	.390666+01	107	-.673711+03	108	.217453+02	109	.151027+04
	110	-.502317+03	111	.979271+03	112	-.315550+03	113	.647894+03
	114	-.180278+03	115	.295680+03	116	.386941+02	119	.671994+03
	120	-.355485+03	121	.788773+03	122	-.403637+03	123	.101693+04
	124	-.689675+03	125	.944292+03	126	-.491647+03	127	.645859+03
	128	-.501226+03	129	-.260487+03	130	.166310+03	131	-.398199+03
	132	.932674+02	133	-.599457+03	134	.117478+03	135	.896992+03
	136	-.341381+03	137	.508280+03	138	-.606143+03	139	.519276+03
	140	-.382940+03	141	.380286+03	142	-.532337+03	143	.587113+03
	144	-.556314+03	145	.396302+03	146	-.229033+03	147	.569190+03
	148	-.218444+03	149	.108944+04	150	-.339879+03	151	.806985+03
	152	-.153199+03	153	-.412088+03	154	.481566+03	155	-.312915+03
	156	.452553+03	157	.154534+03	158	.213769+03	159	.750460+03
	160	-.379635+03	169	.151854+04	170	-.470348+03	171	.167461+04
	172	-.644643+03	173	.129235+04	174	-.521578+03	175	.122079+04
	176	-.451797+03	177	.167784+04	178	-.560357+03	179	-.206192+03
	180	.200076+03	181	-.326352+02	182	.190516+03	183	-.262847+02
	184	.214955+03	186	-.658489+03	190	-.716904+03	191	.117503+04

LOADS ARE IN POUNDS/INCH

NAME		DATE		LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	M BURTON	NOV70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.84.2	
Checked	A L PINCHOT	NOV70	TITLE		INTERNAL LOADS ANALYSIS		
Approved	H H WADE	NOV70			Model		
					AFT FUSELAGE		
					Report No.		
					LGIUS46-2-2		

REV (B) M.D. 1-8-71

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 (CON.)
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS (CONDITION DMLOOR) MOD 2

1 192	-.494229+03	193	.540285+03	194	-.445501+03	195	.272261+03
196	.478658+03	197	-.269661+03	198	-.264614+03	199	.215941+03
200	.298879+03	201	-.402954+03	202	.775045+02	203	-.646599+03
204	.618293+03	205	-.997937+03	206	-.258569+03	207	.113763+02
208	.210222+03	209	-.464868+03	210	-.463570+02	211	-.469596+03
212	.263462+03	213	-.529086+03	214	-.325483+03	215	.597988+02
216	.288185+03	217	-.875000+03	218	-.168981+03	219	-.211091+03
220	.329092+03	221	-.114741+04	222	.150526+03	223	.185827+03
224	-.444489+03	225	.145222+04	226	-.389406+03	227	-.435900+03
228	.582929+02	229	-.112061+03	230	.647583+02	231	-.120033+03
232	.117372+03	233	.340066+03	234	-.264252+03	235	-.538230+03
236	.374658+03	237	.937945+03	238	-.574634+03	239	-.120763+04
240	.692513+03	241	.402213+03	242	-.432405+03	243	.325529+03
244	-.386396+03	245	-.169056+02	246	.361538+03	247	-.417640+03
248	-.422164+02	249	.798890+03	250	-.106270+03	251	-.103000+04
252	.181753+03	253	-.360890+03	254	.494719+03	255	-.285493+04
256	.102568+04	257	-.984074+03	258	.407274+03	259	-.782151+03
260	.303906+03	261	-.689779+03	262	.232578+03	263	-.282511+03
264	.658486+02	265	-.555553+03	266	.291983+03	267	-.592258+03
268	.194955+03	269	-.825200+03	270	.491127+03	271	-.939273+03
272	.686537+03	273	-.933874+03	274	.625567+03	275	.308069+03
276	-.150135+03	277	.277186+03	278	-.853236+01	279	.610167+03
280	-.387073+03	281	.323136+03	282	-.140831+03	283	-.818873+03
284	.378660+03	285	-.608780+03	286	.476076+03	287	-.451237+03
288	.503061+03	289	-.392644+03	290	.465210+03	291	-.536727+03
292	.554694+03						

LOADS ARE IN POUNDS/INCH

NAME F F WILKINSON		DATE 1JUN 65	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. VOL. I	PERM. 6.85
Prepared					Page	
Checked	A L FINCHET	1JUN 65	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L C CAFEY	1JUN 65			LGILS46-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CCA.8
 6.2 SKINS AND STRINGERS
 6.3.2 SKIN LOADS CONDITION TPNCS6 & MOD 2

1	1	-.433037+02	2	-.478832+02	3	.108838+03	4	.104249+03
	5	.384694+03	6	.375631+03	7	.112786+04	8	.111167+04
	9	.432458+04	10	.433829+04	11	-.802297+01	12	-.365678+01
	13	-.154413+02	14	-.998019+01	15	.561878+03	16	.567632+03
	17	.182972+04	18	.184035+04	19	.735937+04	20	.733019+04
	21	.179741+03	22	.146018+03	27	.720714+03	28	.720089+03
	31	.614015+03	32	.608028+03	35	.367896+03	36	.354838+03
	43	.370984+03	44	.348856+03	47	.515822+03	48	.506482+03
	51	.553787+03	52	.796036+03	55	.744953+03	59	.812260+03
	60	.123661+04	63	.591722+03	64	.745156+03	65	-.564253+02
	66	-.520965+02	67	-.955541+02	68	-.642479+02	69	-.756406+01
	70	.325731+02	71	-.106296+03	72	-.457707+02	73	-.983788-00
	74	.269009+03	75	.185561+03	76	.139063+03	77	.451382+03
	78	.447634+02	79	.448508+02	80	.477254+02	81	-.117486+03
	82	-.364039-00	83	-.101455+02	84	.205750+01	85	-.210180+01
	86	-.146997+02	87	.160211+01	88	-.235833+01	89	.189147+03
	90	.182665+03	91	.363261+03	92	.358159+03	93	.479334+03
	94	.480288+03	95	.459953+03	96	.465673+03	97	.179204+04
	98	.180183+04	99	.901816+02	100	.931874+02	101	.210724+03
	102	.215125+03	103	.791784+03	104	.799411+03	105	.980318+03
	106	.987625+03	107	-.113827+04	108	-.113139+04	109	.176544+04
	110	.177273+04	111	.112396+04	112	.112363+04	113	.753064+03
	114	.741329+03	115	.460282+03	116	.423990+03	119	.463412+03
	120	.382198+03	121	.602957+03	122	.525550+03	123	.773376+03
	124	.856938+03	125	.799125+03	126	.602714+03	127	.675442+03
	128	.643208+03	129	-.104516+03	130	-.138584+03	131	-.240957+03
	132	-.261385+03	133	-.288042+03	134	-.269237+03	135	-.290630+03
	136	.265587+03	137	.166920+03	138	.355844+03	139	.102919+04
	140	.954977+02	141	-.241706+03	142	.155893+02	143	-.641454+02
	144	.383174+01	145	.313248+03	146	.305384+03	147	.650555+03
	148	.642565+03	149	.138102+04	150	.137447+04	151	.115880+04
	152	.116143+04	153	.121989+03	154	.126614+03	155	.248569+03
	156	.283405+03	157	.660668+03	158	.668972+03	159	.731645+03
	160	.729999+03	169	.183019+04	170	.160446+04	171	.152048+04
	172	.148888+04	173	.100840+04	174	.917857+03	175	.807147+03
	176	.664619+03	177	.933202+03	178	.593676+03	179	-.759731+02
	180	-.885072+02	181	.289191+02	182	-.733068+02	183	.997889+02
	184	-.780009+02	186	.434829+03	190	.295645+03	191	.266058+03

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	FF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.36
Checked	A L FINCHET	1JUN69	TITLE	Model C-5A	
Approved	L C CAFEY	1JUN69	INTERNAL LOADS ANALYSIS	LC1LS46-2-2	
			AFT FUSELAGE	Report No.	

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TO 2101 CCN.8
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS CONDITION 1PNC56 8 MOD 2

1	192	.133954+03	193	.726667+02	194	.551583+02	195	.231896+01
	196	.154279+01	197	-.413844+01	198	-.949515+02	199	-.917730+02
	200	-.220688+03	201	-.227654+03	202	-.107877+04	203	-.107164+04
	204	-.730560+03	205	-.743673+03	206	-.450132+03	207	-.445125+03
	208	-.487510+03	209	-.469051+03	210	-.936003+03	211	-.928755+03
	212	-.526561+03	213	-.522032+03	214	-.482391+03	215	-.475892+03
	216	-.107735+04	217	-.106418+04	218	-.685076+03	219	-.679032+03
	220	-.149224+04	221	-.149770+04	222	.606130+03	223	.597179+03
	224	.183993+04	225	.184307+04	226	.184584+02	227	-.253612+01
	228	.323593+01	229	-.932454+02	230	-.502429+02	231	-.236997+02
	232	-.189945+02	233	.167681+03	234	.155464+03	235	-.341059+03
	236	-.331436+03	237	.707022+03	238	.696431+03	239	-.989633+03
	240	-.580045+03	241	-.503663+02	242	-.559801+02	243	-.108761+03
	244	-.112616+03	245	.628268+03	246	.636586+03	247	-.845651+03
	248	-.850617+03	249	.127350+04	250	.127656+04	251	-.156248+04
	252	-.156090+04	253	.128106+03	254	.153308+03	255	-.330914+04
	256	-.331346+04	257	-.576916+03	258	-.581206+03	259	-.781835+03
	260	-.779302+03	261	-.725087+03	262	-.720262+03	263	-.320548+03
	264	-.320236+03	265	-.472910+03	266	-.494974+03	267	-.530652+03
	268	-.351385+03	269	-.693565+03	270	-.740417+03	271	-.693967+03
	272	-.672740+03	273	-.564133+03	274	-.656830+03	275	.131619+03
	276	.153564+03	277	.342684+03	278	.321383+03	279	.403581+02
	280	.387259+02	281	.332221+03	282	.242679+03	283	.324981+02
	284	-.345103+03	285	-.266218+03	286	-.257946+03	287	-.559897+03
	288	-.491888+02	289	-.232069+01	290	-.501271+02	291	.104742+03
	292	-.856661+01						

LOADS ARE IN POUNDS / INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	FF WILKINSON	1 JUN 65	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 6.87
Checked	A L FINCHOT	1 JUN 65	TITLE	Model	C-5A
Approved	L C CAFEY	1 JUN 65	INTERNAL LOADS ANALYSIS	LG10S46-2-2	
			AFT FUSELAGE	Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCR.8

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS CONDITION SPE201+P & MOD 2

1	1	-.136210+03	2	-.245578+03	3	-.242026+03	4	-.384540+03
	5	.565671+03	6	.304911+03	7	.124679+04	8	.866832+03
	9	.375266+04	10	.375565+04	11	.572411+02	12	.163885+03
	13	.155016+03	14	.278256+03	15	.875708+03	16	.963301+03
	17	.188578+04	18	.182090+04	19	.758184+04	20	.656379+04
	21	.101403+04	22	.325283+03	23	.557178+03	24	.609731+03
	25	.146435+03	26	.666407+03	27	.481428+03	28	.478263+03
	29	.489457+03	30	.466063+03	31	.559754+03	32	.537201+03
	33	.467257+03	34	.637748+03	35	.466328+03	36	-.183553+03
	37	.337873+03	38	.547666+03	39	.564048+01	40	.142018+03
	41	.470841+02	42	.670457+01	43	.247802+03	44	.208542+03
	45	.501183+03	46	.475752+03	47	.431550+03	48	.448831+03
	49	-.156318+03	50	.146317+03	51	.378263+03	52	.174901+03
	53	.768205+03	54	.121504+03	55	-.131751+03	56	-.237202+03
	57	-.254617+03	58	-.168832+03	59	-.865494+02	60	.519808+02
	61	-.491467+02	62	-.733566+02	63	.513726+02	64	-.699472+02
	65	.275384+03	66	.125372+03	67	.477100+03	68	.321200+03
	69	.734106+03	70	.578156+03	71	.803271+03	72	.638248+03
	73	.164854+04	74	.162255+04	75	.818553+02	76	.182305+03
	77	.194575+03	78	.251828+03	79	.828048+03	80	.695201+03
	81	.517552+03	82	.550676+03	83	-.654189+03	84	-.783680+03
	85	.138549+04	86	.141968+04	87	.765218+03	88	.799596+03
	89	.439412+03	90	.439119+03	91	.235535+03	92	.166950+03
	93	.353647+03	94	.241156+03	95	.443563+03	96	.309382+03
	97	.546453+03	98	.480972+03	99	.629220+03	100	.340375+03
	101	.550610+03	102	.356071+03	103	.156042+03	104	.842915+02
	105	.102471+02	106	-.101651+03	107	.513710+02	108	-.143537+02
	109	-.630325+03	110	.187532+03	111	.365572+02	112	.809302+02
	111	.135761+04	112	.286367+02	113	-.406522+03	114	-.178275+03
	113	-.494563+02	114	-.356536+02	115	.361517+03	116	.214820+03
	115	.706284+03	116	.555081+03	117	.140188+04	118	.124593+04
	117	.865035+03	118	.872852+03	119	.114059+03	120	.194909+03
	119	.261380+03	120	.342168+03	121	.660896+03	122	.726908+03
	121	.535466+03	122	.454440+03	123	.101764+04	124	.946618+03
	123	.511252+03	124	.838004+03	125	.613539+03	126	.484895+03
	125	.521419+03	126	.353288+03	127	.574793+03	128	.220661+03
	127	.447665+02	128	-.215370+02	129	.677402+02	130	-.418624+02
	129	.173751+03	130	-.514516+01	131	.133638+03	132	.110846+03
	131							

LOADS ARE IN POUNDS/INCH

	NAME FF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Vol. I	PERM. 6.88
Prepared				Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A	
Approved	L C CAFEY	1 JUN 69	AFT FUSELAGE	LG10546-2-2	Report No.

REV (A) RFW 8-1-69

6.0 SPELL STRUCTURE F.S. 1744 TC 2101 CCN.8
6.3 SKINS AND STRINGERS
6.3.2 SKIN LOADS CONCITION SPE201+P 6 MCD 2

1	191	.146343+03	192	-.207981+02	193	.105886+03	194	-.231694+02
	195	-.370475+02	196	.411847+02	197	-.730434+02	198	-.535597+03
	199	-.436941+03	200	-.561336+02	201	-.268396+03	202	-.125832+04
	203	-.122991+04	204	-.573717+03	205	-.900629+03	206	-.553117+03
	207	-.461840+03	208	-.567094+03	209	-.726759+03	210	-.999290+03
	211	-.939771+03	212	-.668780+03	213	-.828295+03	214	-.566598+03
	215	-.498752+03	216	-.978313+03	217	-.112749+04	218	-.735577+03
	219	-.663054+03	220	-.129296+04	221	-.142388+04	222	.665162+03
	223	.590818+03	224	.149482+04	225	.162035+04	226	.861815+02
	227	-.514981+02	228	.565597+02	229	-.229688+03	230	-.107566+03
	231	-.391674+03	232	-.219276+03	233	.521167+03	234	.327184+03
	235	-.802298+03	236	-.976307+03	237	.100093+04	238	.787679+03
	239	-.134388+04	240	-.106898+04	241	.139997+03	242	.331228+02
	243	.712563+02	244	-.210159+02	245	.569765+03	246	.682795+03
	247	-.677811+03	248	-.726077+03	249	.982492+03	250	.101750+04
	251	-.124008+04	252	-.128525+04	253	-.661409+02	254	.359304+03
	255	-.285625+04	256	-.282505+04	257	-.593486+03	258	-.621215+03
	259	-.566917+03	260	-.562980+03	261	-.639948+03	262	-.618534+03
	263	-.360363+03	264	-.339461+03	265	-.382296+03	266	-.389069+03
	267	-.443975+03	268	-.268420+03	269	-.462853+03	270	-.430433+03
	271	-.389319+03	272	-.204498+03	273	-.259868+03	274	-.270091+03
	275	-.639220+01	276	.774488+02	277	.396911+03	278	.415500+03
	279	-.166179+03	280	-.687683+02	281	.361355+03	282	.284712+03
	283	.287599+03	284	-.198560+03	285	-.246718+03	286	-.125980+03
	287	-.609705+03	288	.184430+03	289	-.807288+02	290	-.298260+02
	291	.797809+02	292	.148498+02				

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
FF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Vol. I	6.89
Prepared		TITLE		Page	
Checked	A L FINCHET	INTERNAL LOADS ANALYSIS		Model	C-5A
Approved	L C CAFEY	AFT FUSELAGE		Report No.	LG1US46-2-2

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCM.6

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS CONDITION 1PC242 6 MOD 2

1	1	-.161967+03	2	-.401193+02	3	-.289506+03	4	-.129582+03
	5	-.769081+03	6	-.489244+03	7	-.139141+04	8	-.985662+03
	9	-.217812+04	10	-.216752+04	11	-.947369+02	12	-.244243+02
	13	-.147600+03	14	-.175631+02	15	-.258051+01	16	-.827385+02
	17	-.660030+03	18	-.554665+03	19	-.553409+04	20	-.395502+04
	21	-.130914+04	22	-.583217+03	27	-.385748+02	28	-.112768+03
	31	-.481499+01	32	-.413637+02	35	-.554494+02	36	-.661174+02
	43	-.120886+03	44	-.105155+03	47	-.118311+03	48	-.142939+03
	51	-.103033+03	52	-.252690+03	55	-.116856+03	59	-.175740+03
	60	-.388336+03	63	-.258884+03	64	-.344603+03	65	-.373099+02
	66	-.949012+02	67	-.543471+02	68	-.139861+03	69	-.316569+02
	70	-.107447+03	71	-.117085+03	72	-.149863+03	73	-.503467+03
	74	-.457814+03	75	-.307397+03	76	-.453574+03	77	-.743913+02
	78	-.414035+03	79	-.119045+03	80	-.258414+03	81	-.929556+02
	82	-.126132+03	83	-.541139+02	84	-.578729+02	85	-.547270+02
	86	-.588488+02	87	-.569879+02	88	-.769868+02	89	-.191160+03
	90	-.305690+02	91	-.328603+03	92	-.154611+03	93	-.487123+03
	94	-.313090+03	95	-.621969+03	96	-.436702+03	97	-.881974+03
	98	-.849649+03	99	-.340824+02	100	-.773449+02	101	-.830336+01
	102	-.115288+03	103	-.275290+02	104	-.348527+03	105	-.308700+03
	106	-.394007+03	107	-.390038+02	108	-.912343+02	109	-.180062+03
	110	-.199592+03	111	-.830970+02	112	-.461481+02	113	-.209187+03
	114	-.210913+03	115	-.430711+03	116	-.469877+03	119	-.616183+02
	120	-.825289+02	121	-.143106+03	122	-.177871+03	123	-.258602+03
	124	-.390962+03	125	-.275868+03	126	-.372289+03	127	-.128599+03
	128	-.391389+03	129	-.455277+02	130	-.268414+02	131	-.131265+03
	132	-.133800+03	133	-.219465+03	134	-.866152+01	135	-.637246+03
	136	-.347464+03	137	-.349538+03	138	-.616901+03	139	-.648226+02
	140	-.319296+03	141	-.209362+03	142	-.333062+03	143	-.102580+03
	144	-.137175+03	145	-.149618+03	146	-.143585+02	147	-.194964+03
	148	-.290936+02	149	-.142534+03	150	-.520543+02	151	-.247231+03
	152	-.267038+03	153	-.193537+01	154	-.957515+02	155	-.542664+02
	156	-.146694+03	157	-.226704+03	158	-.302689+03	159	-.250074+03
	160	-.404341+03	169	-.618838+03	170	-.626578+03	171	-.615156+03
	172	-.193314+03	173	-.536179+03	174	-.472949+03	175	-.529451+03
	176	-.424875+03	177	-.768823+03	178	-.460528+03	179	-.560373+00
	180	-.307916+02	181	-.927939+02	182	-.335539+02	183	-.111275+03
	184	-.557971+02	186	-.707668+03	190	-.631180+03	191	-.517165+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME FF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 6.90
Checked	A L FINCHET	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A		
Approved	L C CAFEY	1 JUN 69			LG 10546-2-2 Report No.		

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CCN.6
 6.3 SKINS AND STRINGERS
 6.3.2 SKIN LOADS CONCITION IPC242 & MOD 2

1	192	.354346+03	193	.255026+02	194	.165548+03	195	.353371+02
	196	-.496692+02	197	.806596+02	198	-.315752+02	199	-.135806+03
	200	.331558+03	201	.567586+03	202	.238422+03	203	.224404+03
	204	.755078+03	205	.110860+04	206	.204341+03	207	.126394+03
	208	.241955+03	209	.419767+03	210	.402280+03	211	.329808+03
	212	.363612+03	213	.562494+03	214	.233146+03	215	.153614+03
	216	.455122+01	217	.184137+03	218	.324369+03	219	.245419+03
	220	-.185636+03	221	-.102100+02	222	-.345125+03	223	-.260624+03
	224	.183333+03	225	.367031+01	226	-.800424+02	227	.578145+02
	228	-.617256+02	229	.151506+03	230	.154996+02	231	.272715+03
	232	.120201+03	233	-.552511+03	234	-.384585+03	235	.781674+03
	236	.541819+03	237	-.111510+04	238	-.622681+03	239	.125564+04
	240	.549546+03	241	-.580345+02	242	.537352+02	243	-.435872+02
	244	.565039+02	245	-.134706+03	246	-.250319+03	247	.305061+03
	248	.345452+03	249	-.517272+03	250	-.544587+03	251	.593707+03
	252	.630762+03	253	.587584+03	254	.533527+03	255	.142884+04
	256	.137888+04	257	.310293+03	258	.348295+03	259	.441085+02
	260	.634814+02	261	-.575062+02	262	-.505072+02	263	-.190349+03
	264	-.206282+03	265	-.569340+02	266	-.124120+03	267	-.113061+03
	268	-.125456+03	269	-.154618+03	270	-.262620+03	271	-.186070+03
	272	-.361452+03	273	-.245504+03	274	-.380599+03	275	.742499+02
	276	-.216514+02	277	.428532+02	278	-.424572+02	279	.228140+03
	280	.453854+02	281	.474062+02	282	.258060+02	283	-.484835+03
	284	-.353517+03	285	-.318493+03	286	-.463576+03	287	-.555297+02
	288	-.422363+03	289	-.131213+03	290	-.268305+03	291	-.904717+02
	292	-.121882+03						

LOADS ARE IN POUNDS / INCH

Prepared	NAME V W BOYD	DATE 9 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I 6.91	PERM.
Checked	J HALLIDAY	9 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	LC DARBY	9 SEPT 68		Report No.	LGIDS 46-2-2	

REV (A) MB 1 AUG 69

6.3.2 - PANEL LOADS (CONTINUED)

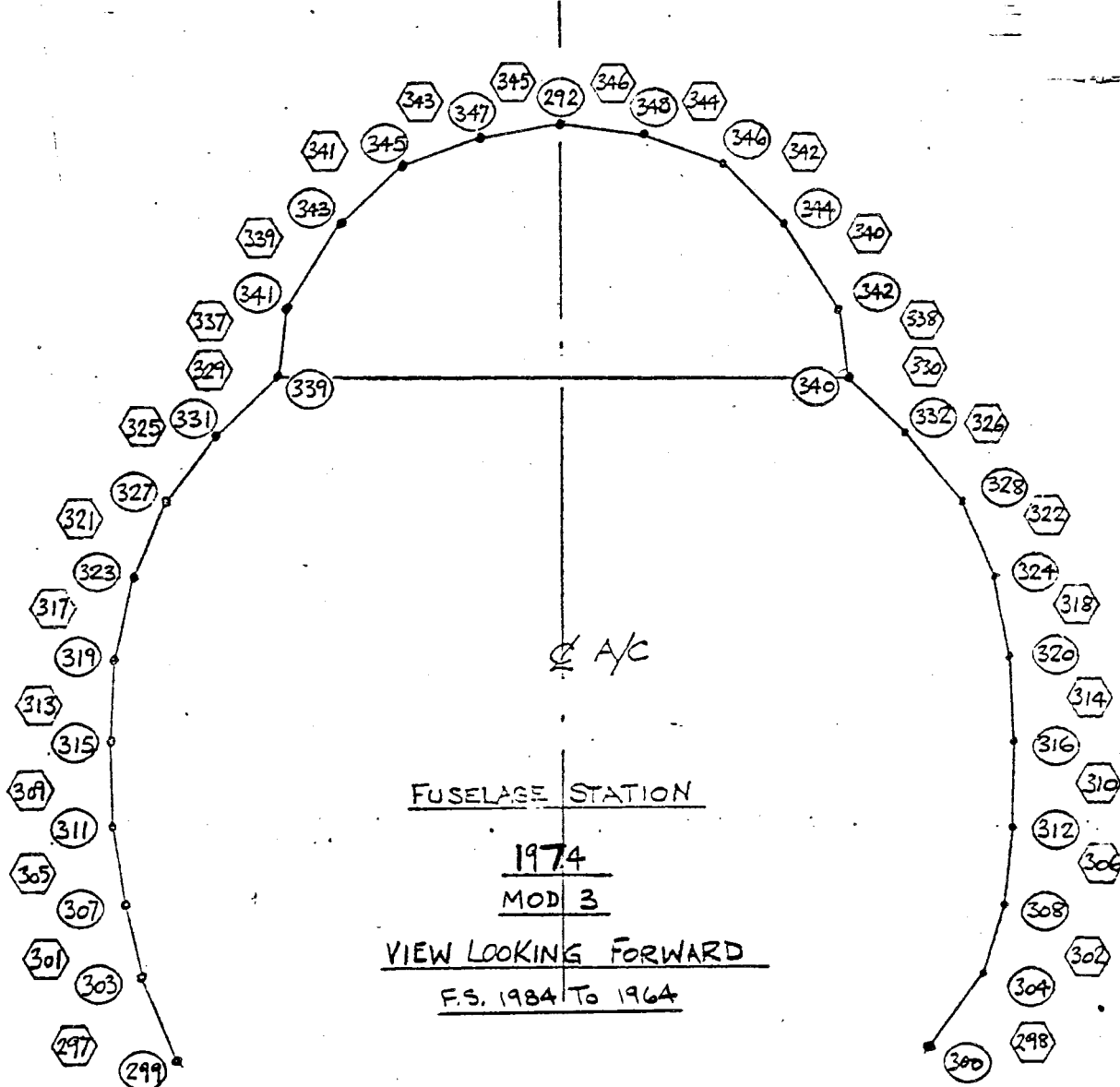


FIGURE No. 6.3.2.10

KEY

- GRID POINT
- FAMAS STRINGER
- FAMAS PANEL

FOR AREAS AND THICKNESS SEE REF 8
VOLUME IV SECTION 5.2

Prepared	NAME K.W. BOYD	DATE 10 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. VOL. I	PERM. 6.92
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved	L. G. DARBY	10 SEPT 68		Report No.	LGUS 46-2-2

REV (A) MB 1 AUG 69.

6.3.2 PANEL LOADS (CONTINUED)

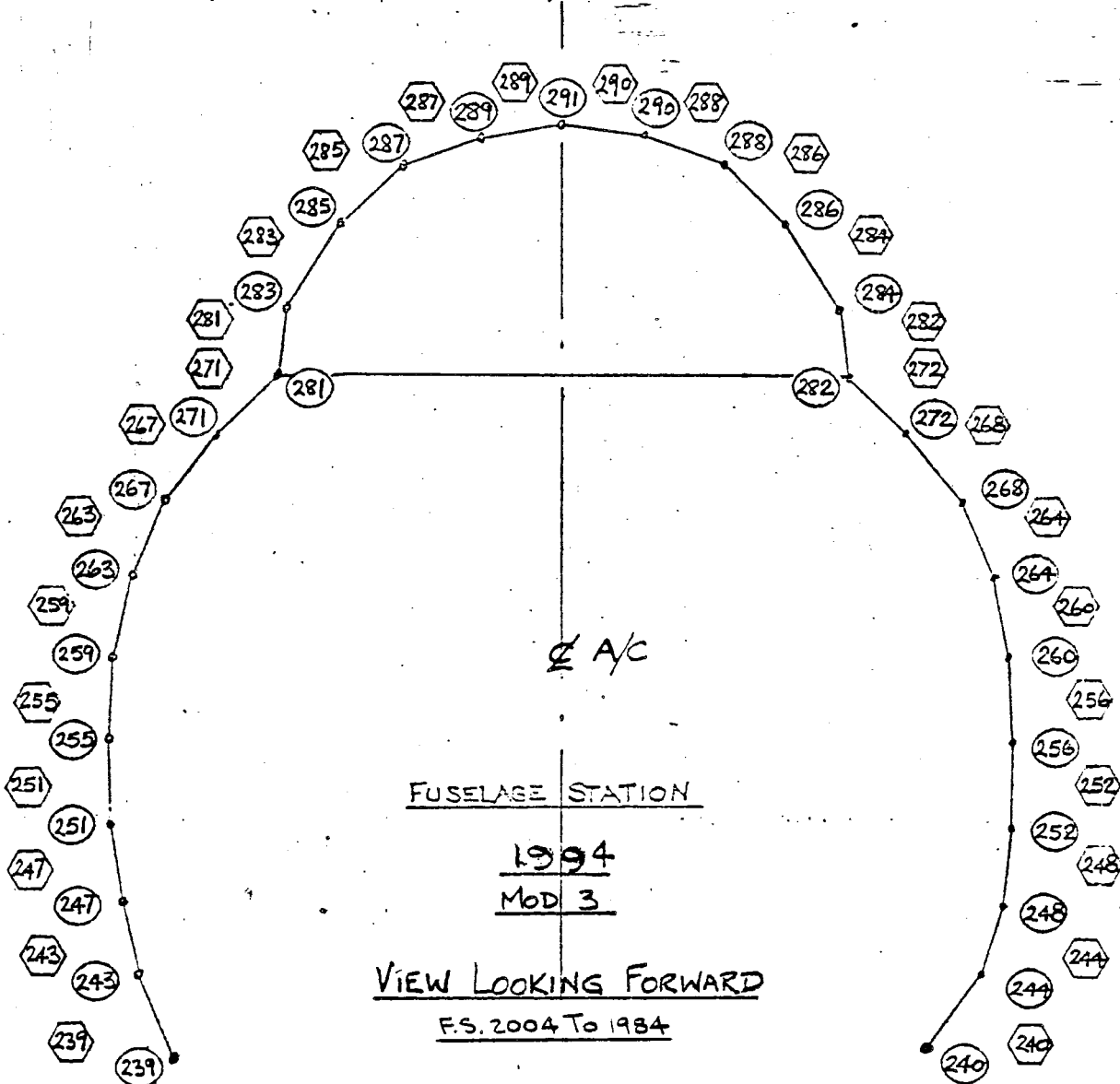


FIGURE No. 6.3.2.11

KEY

—●— GRID POINT

○ FAMAS STRINGER

⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 3
VOLUME IV SECTION 5.2.

PREPARED	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
	J. W. BOYD	16 AUG. 68			VOL. I	⚠
CHECKED	J. HALLIDAY	16 AUG. 68	TITLE INTERNAL LOADS REPORT AFT FUSELAGE	MODEL C-5A		
	APPROVED	L. G. DARBY		16 AUG. 68	LGIUS46-2-2 REPORT NO.	

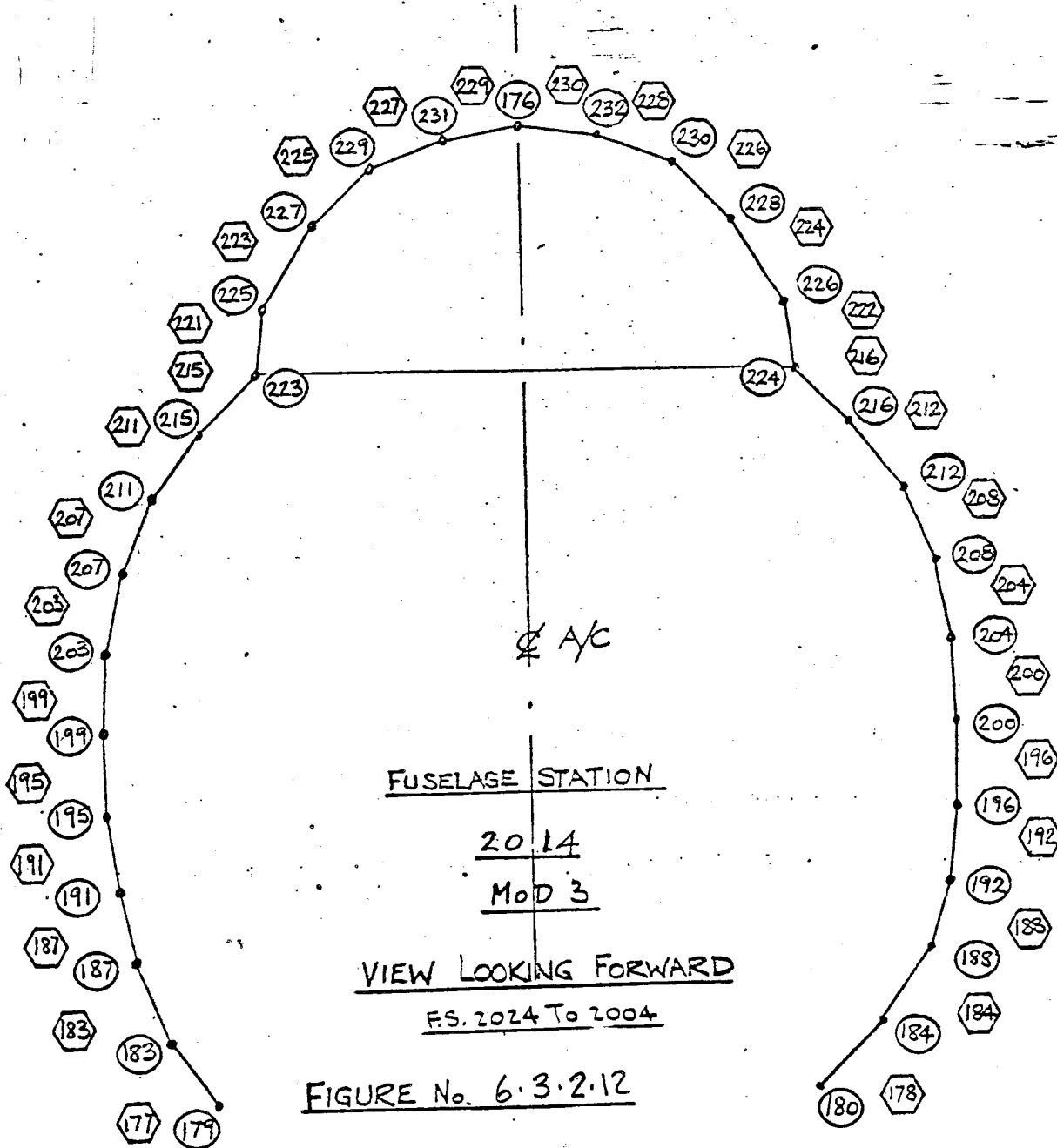
REV. (B) LDP 1-8-71

⚠ PAGES 6.92.1, 6.92.2 & 6.92.3

"DELETED"

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP.	PERM.
	Y. W. BOYD	10 SEPT 68		Vol. I	6.93	
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A		
	L. G. DARBY	10 SEPT 68		LGUS 46-2-2 Report No.		
Approved				REV (A) MB. 1 AUG 69		

6.3.2 PANEL LOADS (CONTINUED)



KEY

— GRID POINT

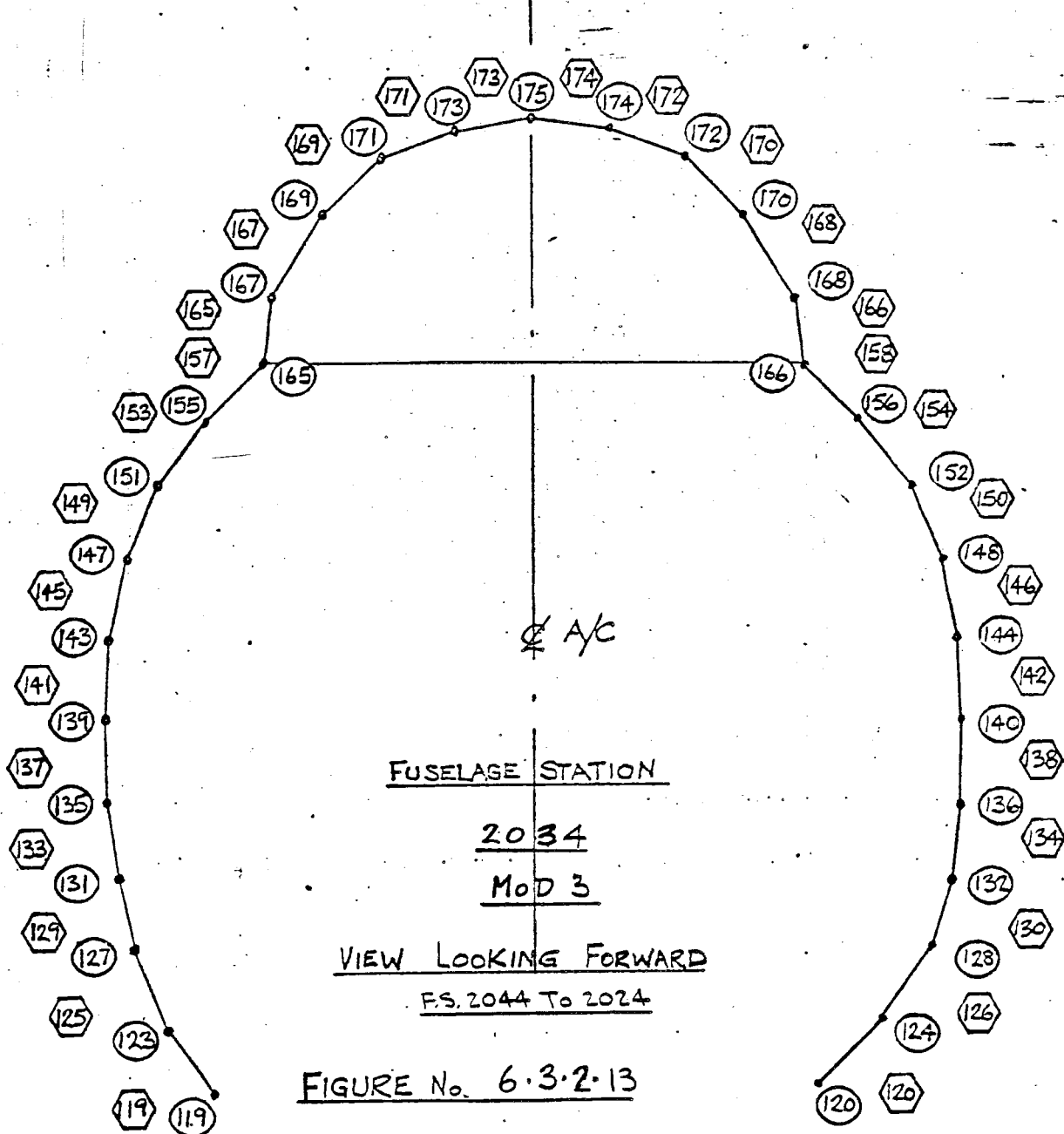
○ FAMAS STRINGER

⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF B
VOLUME IV SECTION 5.2

REV (A) MB. 1 AUG 69

6.3.2 PANEL LOADS (CONTINUED)



- KEY
- GRID POINT
 - FAMAS STRINGER
 - FAMAS PANEL

FOR AREAS AND THICKNESS SEE REF 8.
VOLUME IV SECTION 5.2.

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	V. W. BOYD	10 SEPT 68	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	VOL. I 6,95
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
	L. G. DARBY	10 SEPT 68			LG/US 46-2-2 Report No.	

6.3.2 PANEL LOADS (CONTINUED)

REV(A) MB 1 AUG 69

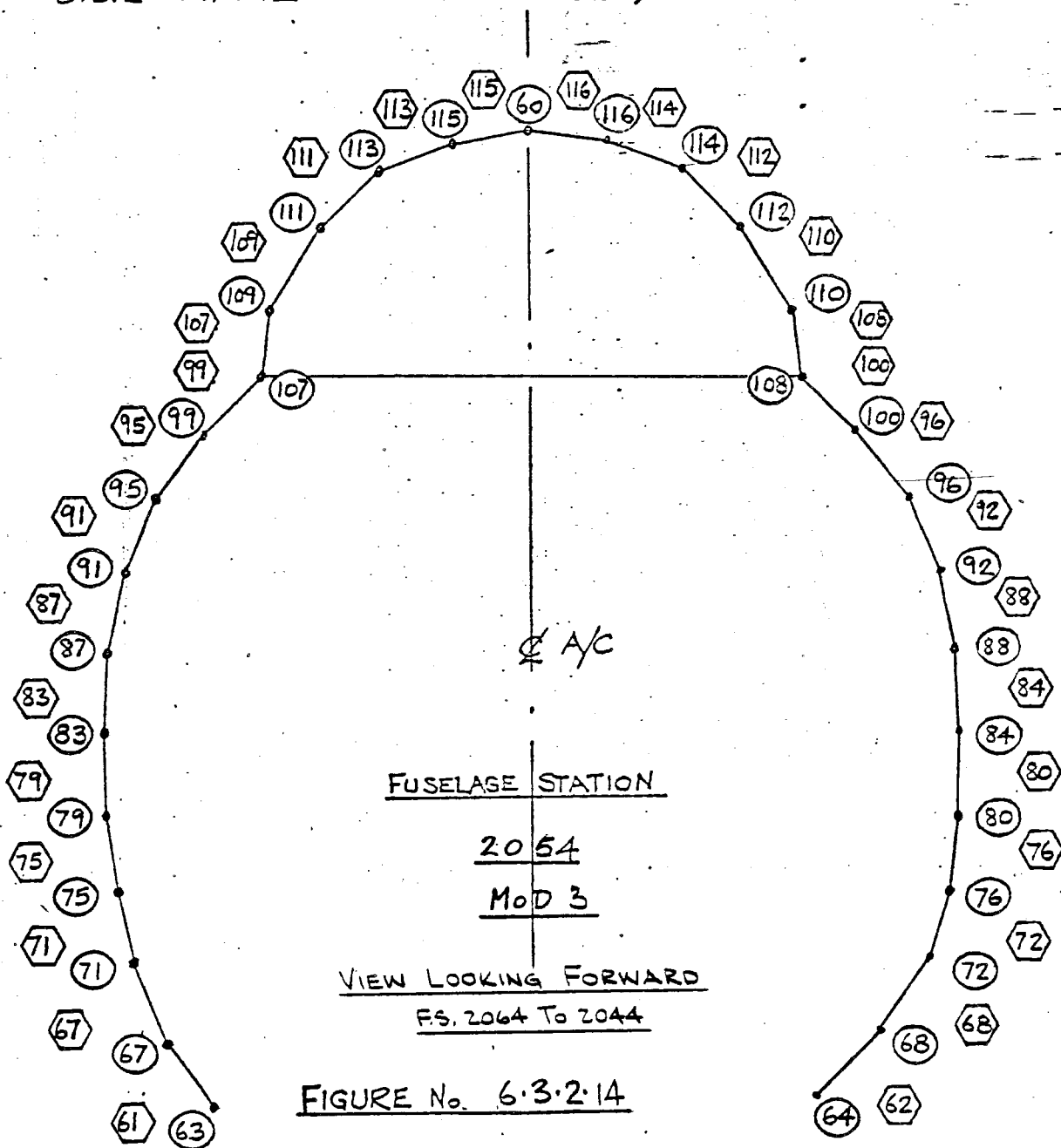


FIGURE No. 6.3.2.14

KEY

GRID POINT

FAMAS STRINGER

FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 3
VOLUME IV SECTION 5.2

Prepared	NAME V.W. BOYD	DATE 10 SEPT 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 6.96
Checked	J. HALLIDAY	10 SEPT 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. DARBY	10 SEPT 68		Report No.	LG105 46-2-2	

REV (A) MB. 1 AUG 69

6.3.2. PANEL LOADS (CON'T)

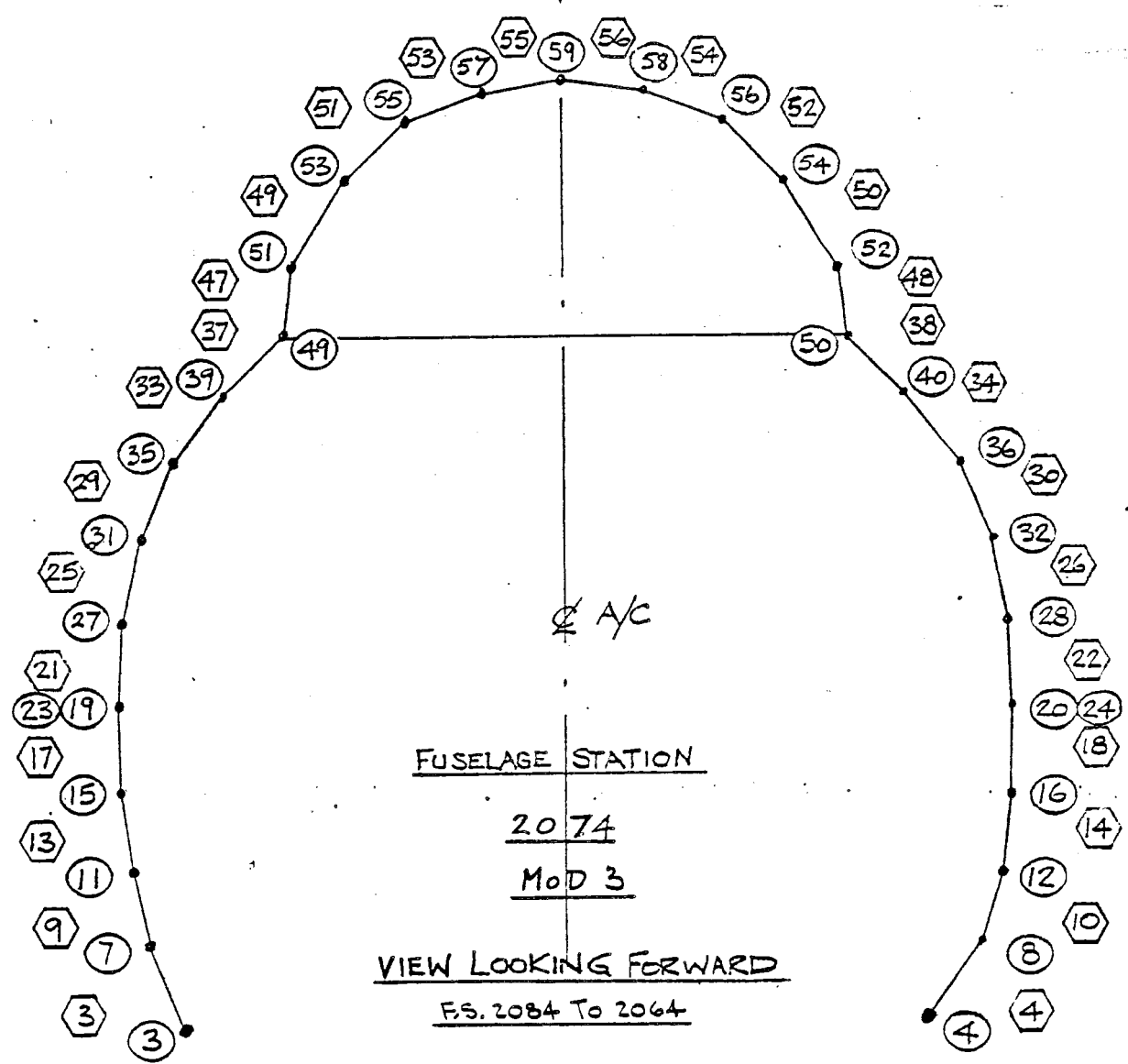


FIGURE No. 6.3.2.15

KEY

 GRID POINT

 FAMAS STRINGER

 FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	FF WILKINSON	1JUN69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 6.97
Checked	A L FINCHET	1JUN69	TITLE		Model C-5A	
					LG1LS46-2-2	
Approved	L C CAFEY	1JUN69	AFT FUSELAGE		Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TO 2101 CON.6

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS CONDITION ENLCC1 E MCD 3

1	1	-.107108+03	2	.259163+03	3	-.638798+02	4	.163857+03
	5	-.739576+01	6	-.425355+02	7	-.165337+02	8	.168801+03
*	9	-.123318+03	10	.245807+03	11	.301585+02	12	.420708+02
	13	-.141616+03	14	.314741+03	15	.423810+02	16	-.114233+02
	17	-.625690+02	18	.179653+03	19	.430745+02	20	-.237136-00
	21	-.302620+03	22	.531123+03	23	.387449+02	24	-.856567+01
	25	-.256933+03	26	.459152+03	27	.359679+02	28	.340247+01
	29	-.371577+03	30	.516565+03	31	.348275+02	32	-.504621+01
	33	-.459661+03	34	.578228+03	35	.786544+02	36	-.286862+02
	37	-.683702+03	38	.744071+03	39	.186563+02	40	-.110708+03
	41	.115957+01	42	-.200661+03	43	-.736103+01	44	-.191592+03
	45	.314665+02	46	-.251132+03	47	-.661230+03	48	.483123+03
	49	-.652440+03	50	.503183+03	51	-.593362+03	52	.521746+03
	53	-.540012+03	54	.536767+03	55	-.827639+03	56	.536004+03
	57	-.417603+03	58	-.680327+02	59	.104920+03	60	-.702903+02
	62	.129913+03	63	-.526244+02	64	.116073+02	65	-.172894+03
	68	.334952+03	69	-.141663+03	70	.314229+03	71	-.141291+03
	74	.308753+03	75	-.241467+03	76	.489173+03	77	-.234356+03
	84	.432415+03	85	-.296296+03	86	.491034+03	87	-.357661+03
	92	.507175+03	93	-.465416+03	94	.560058+03	95	-.676901+03
	100	.750140+03	101	.342316+02	102	-.185864+03	103	-.321846+02
	104	-.151882+03	105	.720246+02	106	-.294838+03	107	-.598774+03
	108	.419832+03	109	-.595907+03	110	.470585+03	111	-.566216+03
	112	.515055+03	113	-.543719+03	114	.554070+03	115	-.533943+03
	116	.544377+03	117	-.345287+02	118	.802348+02	119	-.407761+02
	120	.219185+02	121	-.124948+03	122	.848792+01	123	-.179070+03
	126	.375104+03	127	-.937562+02	128	.280603+03	129	-.483853+02
	134	.234811+03	135	-.110208+03	136	.338453+03	137	-.178146+03
	142	.391701+03	143	-.224862+03	144	.452777+03	145	-.309760+03
	150	.539725+03	151	-.413357+03	152	.629373+03	153	-.704109+03
	156	.648181+03	157	.237390+03	158	-.377465+03	159	.134041+03
	162	-.355185+03	163	.159125+03	164	-.375727+03	165	-.637285+03
	166	.500737+03	167	-.566043+03	168	.465842+03	169	-.552028+03
	170	.532548+03	171	-.522370+03	172	.551545+03	173	-.535349+03
	174	.547019+03	175	-.524810+02	176	-.657068+02	177	-.116503+03
	180	-.484205+02	181	-.214027+03	182	.427477+03	183	-.112293+03
	188	.265450+03	189	-.108205+03	190	.269474+03	191	-.129115+03
	196	.342738+03	197	-.173717+03	198	.425877+03	199	-.228661+03

* PANEL (9) REF. P. 6.96 ULTIMATE PANEL SHEAR FLOW = -123. LBS./IN.

	NAME FF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 6.98
Prepared					Page	
Checked	A L FINCHET	1JUN69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L C CAFEY	1JUN69	AFT FUSELAGE		LG1LS46-2-2	
					Report No.	

REV (A) RFW 8-1-69

6.0 SHELL STRUCTURE F.S. 1744 TC 2101 CCN.6

6.3 SKINS AND STRINGERS

6.3.2 SKIN LOADS CONDITION CML001 & MOD 3

1 204	.461508+03	207	-.323771+03	208	.585204+03	211	-.451621+03
212	.630842+03	215	-.661862+03	216	.632274+03	217	.210801+03
218	-.376309+03	219	.186640+03	220	-.410029+03	221	-.562047+03
222	.454558+03	223	-.575359+03	224	.510713+03	225	-.487009+03
226	.512357+03	227	-.501706+03	228	.534257+03	229	-.567523+03
230	.552628+03	235	-.220472+03	236	.114963+03	239	-.161476+03
240	.127284+03	243	-.884251+02	244	.763391+02	247	-.629872+02
248	.141347+03	251	-.792003+02	252	.242741+03	255	-.174547+03
256	.347827+03	259	-.187800+03	260	.397824+03	263	-.309123+03
264	.722459+03	267	-.474617+03	268	.753625+03	271	-.702606+03
272	.931727+03	273	.299830+03	274	-.357116+03	275	.231781+03
276	-.384675+03	277	.275126+03	278	-.435947+03	279	.211124+03
280	-.407824+03	281	-.640674+03	282	.580229+03	283	-.510308+03
284	.509044+03	285	-.408712+03	286	.462948+03	287	-.491015+03
288	.531628+03	289	-.575888+03	290	.573275+03	293	.329006+02
294	-.696636+03	297	.388889+01	298	-.184206+03	301	-.812626+02
302	.130564+03	305	-.497562+02	306	.155662+03	309	-.129498+03
310	.333355+03	313	-.229073+03	314	.441926+03	317	-.224287+03
318	.330540+03	321	-.388402+03	322	.891087+03	325	-.564578+03
326	.711770+03	329	-.522226+03	330	.878902+03	331	.314727+03
332	-.388951+03	333	.262454+03	334	-.374292+03	335	.223918+03
336	-.407258+03	337	-.576831+03	338	.529117+03	339	-.536081+03
340	.542445+03	341	-.295547+03	342	.506602+03	343	-.488721+03
344	.554565+03	345	-.586454+03	346	.571613+03	347	.735403+02
348	-.131903+03	349	.138492+02	350	.846339+02	351	.276929+02
352	.274931+02	353	.575495+02	354	-.624024+02	355	-.596339+02
356	.306880+02	357	.455633+02	358	-.131873+03		

LOADS ARE IN POUNDS / INCH

NAME M BURTON		DATE NOV70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. No. I	PERM. 698.1
Prepared	A L PINCHOT	NOV70	TITLE INTERNAL LOADS ANALYSIS			C-5A	
Checked					Model		
Approved	H H WADE	NOV70	AFT FUSELAGE			LGIUS46-2-2	
					Report No.		

REV (B) M.B. 1-8-71

6.0 SHELL STRUCTURE P.S. 1744 TO 2101 (CON.)
6.3 SKINS AND STRINGERS
6.3.2 SKIN LOADS (CONDITION DMLOOR) MOD 3

1	1	.259870+03	2	-.107084+03	3	.166351+03	4	-.651322+02
	5	-.335180+02	6	-.672717+01	7	.166674+03	8	-.195820+02
	9	.244921+03	10	-.135346+03	11	.402744+02	12	.281139+02
	13	.312116+03	14	-.153481+03	15	-.128309+02	16	.407158+02
	17	.178700+03	18	-.742958+02	19	-.277832+01	20	.406168+02
	21	.523356+03	22	-.309575+03	23	-.107052+02	24	.367443+02
	25	.450645+03	26	-.302743+03	27	.517522+00	28	.334235+02
	29	.505321+03	30	-.372575+03	31	-.795462+01	32	.321063+02
	33	.571283+03	34	-.451665+03	35	.345747+02	36	.737029+02
	37	.748096+03	38	-.662725+03	39	-.939132+02	40	.367590+02
	41	-.163794+03	42	.492366+02	43	-.156774+03	44	.280834+02
	45	-.205947+03	46	.779043+02	47	.536301+03	48	-.593827+03
	49	.543485+03	50	-.619528+03	51	.528395+03	52	-.609827+03
	53	.505839+03	54	-.584829+03	55	.510907+03	56	-.552230+03
	57	.416340+03	59	.112068+03	60	-.661826+02	61	.129931+03
	62	-.702870+02	63	.174588+02	64	-.550258+02	67	.337234+03
	68	-.182357+03	71	.318884+03	72	-.150974+03	75	.314394+03
	76	-.150956+03	79	.489581+03	80	-.250656+03	83	.428699+03
	84	-.240724+03	87	.478887+03	88	-.301676+03	91	.489041+03
	92	-.361277+03	95	.549890+03	96	-.459414+03	99	.750081+03
	100	-.656728+03	101	-.161710+03	102	.654389+02	103	-.150245+03
	104	.108092+02	105	-.248465+03	106	.121405+03	107	.479267+03
	108	-.520310+03	109	.516277+03	110	-.558350+03	111	.524194+03
	112	-.592068+03	113	.515125+03	114	-.597489+03	115	.515169+03
	116	-.559095+03	117	.711706+02	118	-.418659+02	119	.315131+02
	120	-.425259+02	121	.243316+02	122	-.111826+03	125	.384953+03
	126	-.186852+03	129	.290123+03	130	-.105807+03	133	.240324+03
	134	-.578447+02	137	.338796+03	138	-.118037+03	141	.388030+03
	142	-.185358+03	145	.438999+03	146	-.232454+03	149	.506931+03
	150	-.320388+03	153	.607594+03	154	-.413046+03	157	.839760+03
	158	-.686864+03	159	-.362269+03	160	.254349+03	161	-.315553+03
	162	.176265+03	163	-.324005+03	164	.216947+03	165	.567838+03
	166	-.541723+03	167	.518534+03	168	-.521208+03	169	.538874+03
	170	-.593951+03	171	.499624+03	172	-.590661+03	173	.511795+03
	174	-.559593+03	175	-.412403+02	176	-.449338+02	179	-.798119+02
	180	-.114887+03	183	.443767+03	184	-.218706+03	187	.277583+03
	188	-.116930+03	191	.276753+03	192	-.114956+03	195	.351297+03
	196	-.134518+03	199	.431227+03	200	-.176372+03	203	.454666+03

LOADS ARE IN POUNDS / INCH