

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.29
Checked	A L PINCHOT	1 JUN 69	TITLE	Model	C-54
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	Report No.	LG1J346-2-2
			AFT FUSELAGE		

REV (A) RFW 8-1-69

7.0 SPELL STRUCTURE F.S. 2101 TO 2539 CON. 2

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 6

1	1	.236419+05	2	.241889+05	3	.171774+05	4	.177154+05
	5	.173731+05	6	.182340+05	9	.938069+05	10	.978736+05
	11	.370315+05	12	.394803+05	13	.240966+05	14	.251430+05
*	15	.161560+04	16	.878511+03	17	-.223699+05	18	-.253269+05
	19	-.515917+05	20	-.563091+05	21	-.850701+05	22	-.947168+05
	23	-.282995+05	24	-.294746+05	25	-.588747+05	26	-.603641+05
	27	.101835+05	28	.106503+05	29	.175453+05	30	.180076+05
	31	.228881+05	32	.233076+05	33	.161143+05	34	.164654+05
	35	.166218+05	36	.167649+05	37	.422385+05	38	.450149+05
	39	.909538+05	40	.943671+05	41	.316396+05	42	.332654+05
	43	.211291+05	44	.219239+05	45	-.333925+04	46	-.401873+04
	47	-.280725+05	48	-.305086+05	49	-.563147+05	50	-.601653+05
	51	-.972197+05	52	-.101676+06	53	-.791262+05	54	-.790288+05
	55	.302928+05	56	.316330+05	57	.400012+05	58	.415289+05
	59	.218696+05	60	.222007+05	61	.150720+05	62	.153170+05
	63	.158212+05	64	.158338+05	69	.377663+05	70	.387311+05
	71	.718015+05	72	.746462+05	73	.271277+05	74	.279801+05
	75	.188445+05	76	.191551+05	77	-.607352+04	78	-.674144+04
	79	-.319369+05	80	-.336414+05	81	-.597849+05	82	-.625530+05
	83	-.102157+06	84	-.105382+06	85	-.138239+06	86	-.125425+06
	87	.442829+05	88	.471519+05	89	.468803+05	90	.437239+05
	91	.614276+04	92	.644067+04	93	.671610+04	94	.679092+04
	95	.205225+05	96	.207885+05	97	.139819+05	98	.141378+05
	99	.150032+05	100	.148390+05	105	.306404+05	106	.302573+05
	107	.576531+05	108	.538527+05	109	.246713+05	110	.248775+05
	111	.166686+05	112	.165828+05	113	-.684870+04	114	-.721111+04
	115	-.347580+05	116	-.355407+05	117	-.637284+05	118	-.654669+05
	119	-.104711+06	120	-.105845+06	121	.571494+05	122	.630433+05
	123	.527485+05	124	.536736+05	125	.717099+04	126	.738133+04
	127	.102591+05	128	.101091+05	129	.374941+04	130	.324797+04
	131	.363044+04	132	.345578+04	133	.186872+05	134	.189141+05
	135	.125981+05	136	.126827+05	137	.136998+05	138	.133719+05
	139	.207931+05	140	.196867+05	141	.463122+05	142	.454675+05
	143	.234964+05	144	.233140+05	145	.152930+05	146	.142527+05
	147	-.722551+04	148	-.697825+04	149	-.365759+05	150	-.365875+05
	151	-.661974+05	152	-.670192+05	153	-.105513+06	154	-.106846+06
	155	-.119656+06	156	-.115611+06	157	.773902+05	158	.870229+05
	159	.562712+05	160	.550697+05	161	.361003+04	162	.396512+04

* STRINGER (15) REF. P. 7.27 ULTIMATE TENSION LOAD = 1616 LB.

	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 7.29
Prepared			TITLE INTERNAL LOADS ANALYSIS			Model C-5A	
Checked	A L PINCHOT	1 JUN 69				LG1JS46-2-2	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE			Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 6

1	163	.463039+04	164	.441859+04	165	.132066+04	166	.876366+03
	167	.382814+04	168	.838112+04	169	.164603+05	170	.156829+05
	171	.109840+05	172	.110369+05	173	.120798+05	174	.116309+05
	177	.126771+05	178	.115620+05	179	.162054+05	180	.145369+05
	181	.316002+05	182	.301994+05	183	.209438+05	184	.205018+05
	185	.144488+05	186	.138260+05	187	-.826921+04	188	-.727934+04
	189	-.377249+05	190	-.363883+05	191	-.676460+05	192	-.675600+05
	193	-.105281+06	194	-.105998+06	195	.429473+05	196	.463301+05

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. 7.30	PERM. 7.30
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model	C-5A
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No.	LG10545-2-2
REV (A) RFW 8-1-69						

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.6

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION SPE201 & MOD 6

1	1	-.363268+05	2	-.369599+05	3	-.262830+05	4	-.263777+05
	5	-.271535+05	6	-.277942+05	9	-.144270+06	10	-.149633+06
	11	-.575166+05	12	-.604349+05	13	-.371559+05	14	-.387135+05
	15	-.213856+04	16	-.194955+04	17	.353711+05	18	.375479+05
	19	.820779+05	20	.859913+05	21	.143414+06	22	.143060+06
	23	.457613+05	24	.464306+05	25	.969567+05	26	.981253+05
	27	-.155193+05	28	-.161367+05	29	-.264549+05	30	-.272204+05
	31	-.350228+05	32	-.355395+05	33	-.245759+05	34	-.251857+05
	35	-.251588+05	36	-.255685+05	37	-.648661+05	38	-.681264+05
	39	-.138963+06	40	-.143742+06	41	-.487724+05	42	-.509345+05
	43	-.323479+05	44	-.335681+05	45	.564922+04	46	.594814+04
	47	.442658+05	48	.462073+05	49	.890501+05	50	.923106+05
	51	.154961+06	52	.158639+06	53	.126984+06	54	.126554+06
	55	-.456454+05	56	-.474566+05	57	-.600301+05	58	-.622367+05
	59	-.332401+05	60	-.336664+05	61	-.228233+05	62	-.232182+05
	63	-.237624+05	64	-.240290+05	69	-.568131+05	70	-.583999+05
	71	-.109460+06	72	-.113331+06	73	-.414515+05	74	-.427078+05
	75	-.284915+05	76	-.291954+05	77	.997904+04	78	.103303+05
	79	.499371+05	80	.513211+05	81	.935876+05	82	.959824+05
	83	.160821+06	84	.163525+06	85	.218622+06	86	.197614+06
	87	-.662241+05	88	-.697871+05	89	-.701473+05	90	-.727932+05
	91	-.911142+04	92	-.950965+04	93	-.100282+05	94	-.102263+05
	95	-.310132+05	96	-.313661+05	97	-.210441+05	98	-.213394+05
	99	-.223546+05	100	-.224380+05	105	-.454061+05	106	-.455412+05
	107	-.874701+05	108	-.895156+05	109	-.372661+05	110	-.373651+05
	111	-.248058+05	112	-.250852+05	113	.113011+05	114	.113930+05
	115	.539027+05	116	.546156+05	117	.987977+05	118	.100361+06
	119	.163019+06	120	.164868+06	121	-.851586+05	122	-.919061+05
	123	-.783268+05	124	-.802117+05	125	-.106845+05	126	-.114946+05
	127	-.151690+05	128	-.152272+05	129	-.543491+04	130	-.503999+04
	131	-.529354+04	132	-.518492+04	133	-.250039+05	134	-.283040+05
	135	-.187805+05	136	-.139871+05	137	-.201910+05	138	-.200900+05
	139	-.303661+05	140	-.297250+05	141	-.692510+05	142	-.691693+05
	143	-.349578+05	144	-.352110+05	145	-.223932+05	146	-.224003+05
	147	.113288+05	148	.113652+05	149	.559827+05	150	.559574+05
	151	.101396+06	152	.102210+06	153	.162449+06	154	.163679+06
	155	.184890+06	156	.176828+06	157	-.115344+06	158	-.125095+06
	159	-.825624+05	160	-.825341+05	161	-.533279+04	162	-.573772+04

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 7.31
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-5A	
Approved	L G DARBY	1 JUN 69			Report No.	LG1J545-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. &

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION SPE201 & MOD 6

1	163	-.665879+04	164	-.651641+04	165	-.186076+04	166	-.147486+04
	167	-.128426+05	168	-.125807+05	169	-.244157+05	170	-.246921+05
	171	-.161850+05	172	-.163354+05	173	-.175558+05	174	-.173121+05
	177	-.181830+05	178	-.173976+05	179	-.233574+05	180	-.220564+05
	181	-.463823+05	182	-.455102+05	183	-.307131+05	184	-.306471+05
	185	-.207474+05	186	-.204623+05	187	.131354+05	188	.120037+05
	189	.568821+05	190	.561250+05	191	.102319+06	192	.102464+06
	193	.160396+06	194	.161149+06	195	-.664953+05	196	-.703051+05

LOADS ARE IN POUNDS

NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared				Page	VOL. I 732
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A
Approved	L G CARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1J346-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.2

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION DML001 & MOD 6

1	1	-.382664+04	2	-.129347+05	3	.575441+03	4	-.123354+05
	5	.403028+04	6	-.167780+05	9	.290851+05	10	-.983298+05
	11	.129531+05	12	-.405185+05	13	.164477+05	14	-.335925+05
	15	.220921+05	16	-.218257+05	17	.349362+05	18	-.158930+05
	19	.468382+05	20	-.520199+04	21	.610829+05	22	.291644+04
	23	.163557+05	24	.602331+04	25	.299738+05	26	.174080+05
	27	.364071+04	28	-.110343+05	29	.687810+04	30	-.194866+05
	31	-.315747+04	32	-.128863+05	33	.117512+04	34	-.125290+05
	35	.445308+04	36	-.161266+05	37	.119994+05	38	-.430121+05
	39	.321595+05	40	-.986170+05	41	.132703+05	42	-.364686+05
	43	.166053+05	44	-.214832+05	45	.233465+05	46	-.192612+05
	47	.329875+05	48	-.989538+04	49	.437737+05	50	.897369+03
	51	.584984+05	52	.176676+05	53	.297255+05	54	.316323+05
	55	.115680+05	56	-.333052+05	57	.154958+05	58	-.441832+05
	59	-.256698+04	60	-.125486+05	61	.157309+04	62	-.120286+05
	63	.483118+04	64	-.157465+05	69	.154593+05	70	-.419701+05
	71	.268687+05	72	-.791596+05	73	.138482+05	74	-.333961+05
	75	.172839+05	76	-.302497+05	77	.255329+05	78	-.193327+05
	79	.311831+05	80	-.566015+04	81	.402834+05	82	.618819+04
	83	.544614+05	84	.241292+05	85	.528743+05	86	.475762+05
	87	.170967+05	88	-.435695+05	89	.184184+05	90	-.520415+05
	91	.279398+04	92	-.700028+04	93	.301018+04	94	-.769325+04
	95	-.194295+04	96	-.120152+05	97	.200695+04	98	-.115321+05
	99	.540102+04	100	-.155246+05	105	.153041+05	106	-.360340+05
	107	.246109+05	108	-.660491+05	109	.154552+05	110	-.323101+05
	111	.178512+05	112	-.290246+05	113	.230870+05	114	-.213467+05
	115	.295558+05	116	-.289633+04	117	.375475+05	118	.110071+05
	119	.500655+05	120	.291695+05	121	.197390+05	122	-.507610+05
	123	.225095+05	124	-.539135+05	125	.263875+04	126	-.770892+04
	127	.494200+04	128	-.121213+05	129	.252765+04	130	-.500693+04
	131	.219458+04	132	-.464687+04	133	-.128176+04	134	-.112509+05
	135	.235484+04	136	-.108236+05	137	.575735+04	138	-.148444+05
	139	.124528+05	140	-.261362+05	141	.247245+05	142	-.570385+05
	143	.172137+05	144	-.333747+05	145	.189770+05	146	-.289330+05
	147	.302354+05	148	-.233278+05	149	.290155+05	150	-.112036+04
	151	.353185+05	152	.144105+05	153	.465849+05	154	.324581+05
	155	.447425+05	156	.427223+05	157	.239900+05	158	-.800721+05
	159	.275567+05	160	-.661653+05	161	.132592+04	162	-.384529+04

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. VOL. I	PERM. 733
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L G DARBY	1 JUN 69		Report No.	LG1JS46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. &
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION DML001 & MOD 6

1 163 .296524+04 164 -.537127+04 165 .116067+04 166 -.196666+04
 167 .518733+04 168 -.110767+05 169 -.484835+03 170 -.102323+05
 171 .273565+04 172 -.998311+04 173 .609353+04 174 -.138224+05
 177 .851501+04 178 -.168222+05 179 .117012+05 180 -.220627+05
 181 .212022+05 182 -.423442+05 183 .177747+05 184 -.313367+05
 185 .209502+05 186 -.300020+05 187 .313629+05 188 -.242020+05
 189 .283131+05 190 -.421317+03 191 .334155+05 192 .163502+05
 193 .434840+05 194 .342742+05 195 .925550+04 196 -.413963+05

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	VOL. I 734
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		LG1J545-2-2 Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION 1004001.18 & MOD 6

1	1	.162664+05	2	.162770+05	3	.113500+05	4	.113535+05
	5	.108906+05	6	.108901+05	9	.520069+05	10	.519436+05
	11	.225455+05	12	.225365+05	13	.154877+05	14	.155263+05
	15	.369513+04	16	.376304+04	17	-.121531+05	18	-.121282+05
	19	-.407670+05	20	-.408161+05	21	-.877539+05	22	-.878161+05
	23	-.294822+05	24	-.294484+05	25	-.702177+05	26	-.702032+05
	27	.518041+04	28	.516248+04	29	.733233+04	30	.728578+04
	31	.154034+05	32	.154184+05	33	.103253+05	34	.103356+05
	35	.975683+04	36	.976497+04	37	.241830+05	38	.241676+05
	39	.468174+05	40	.467619+05	41	.179493+05	42	.179646+05
	43	.126683+05	44	.127308+05	45	.125102+03	46	.209038+03
	47	-.166155+05	48	-.165891+05	49	-.435406+05	50	-.436199+05
	51	-.906552+05	52	-.907352+05	53	-.824353+05	54	-.823744+05
	55	.140424+05	56	.139772+05	57	.159159+05	58	.157856+05
	59	.144454+05	60	.144642+05	61	.939650+04	62	.941381+04
	63	.895859+04	64	.897699+04	69	.195781+05	70	.195816+05
	71	.352926+05	72	.352976+05	73	.143431+05	74	.144059+05
	75	.102648+05	76	.103569+05	77	-.210009+04	78	-.202813+04
	79	-.194323+05	80	-.194313+05	81	-.444750+05	82	-.445388+05
	83	-.892264+05	84	-.893264+05	85	-.132810+06	86	-.111752+06
	87	.195885+05	88	.194141+05	89	.166009+05	90	.164330+05
	91	.250518+04	92	.247986+04	93	.184732+04	94	.183701+04
	95	.133707+05	96	.133939+05	97	.847935+04	98	.850468+04
	99	.813565+04	100	.816879+04	105	.144100+05	106	.144416+05
	107	.275478+05	108	.276732+05	109	.122261+05	110	.123453+05
	111	.807993+04	112	.819654+04	113	-.355412+04	114	-.347263+04
	115	-.212868+05	116	-.213246+05	117	-.454090+05	118	-.455606+05
	119	-.854773+05	120	-.856032+05	121	.244453+05	122	.241016+05
	123	.188725+05	124	.187549+05	125	.355080+04	126	.342901+04
	127	.345402+04	128	.345093+04	129	.192247+04	130	.193957+04
	131	.252623+04	132	.255795+04	133	.130298+05	134	.130601+05
	135	.839284+04	136	.842832+04	137	.814370+04	138	.819694+04
	139	.978948+04	140	.987333+04	141	.223465+05	142	.225119+05
	143	.110232+05	144	.111791+05	145	.631986+04	146	.643693+04
	147	-.593334+04	148	-.587640+04	149	-.239058+05	150	-.239926+05
	151	-.474684+05	152	-.476638+05	153	-.839223+05	154	-.840826+05
	155	-.101114+06	156	-.908020+05	157	.285509+05	158	.277308+05
	159	.175201+05	160	.175227+05	161	.178570+04	162	.175981+04

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.35
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1J545-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8

7.3 SKINS AND STRINGERS

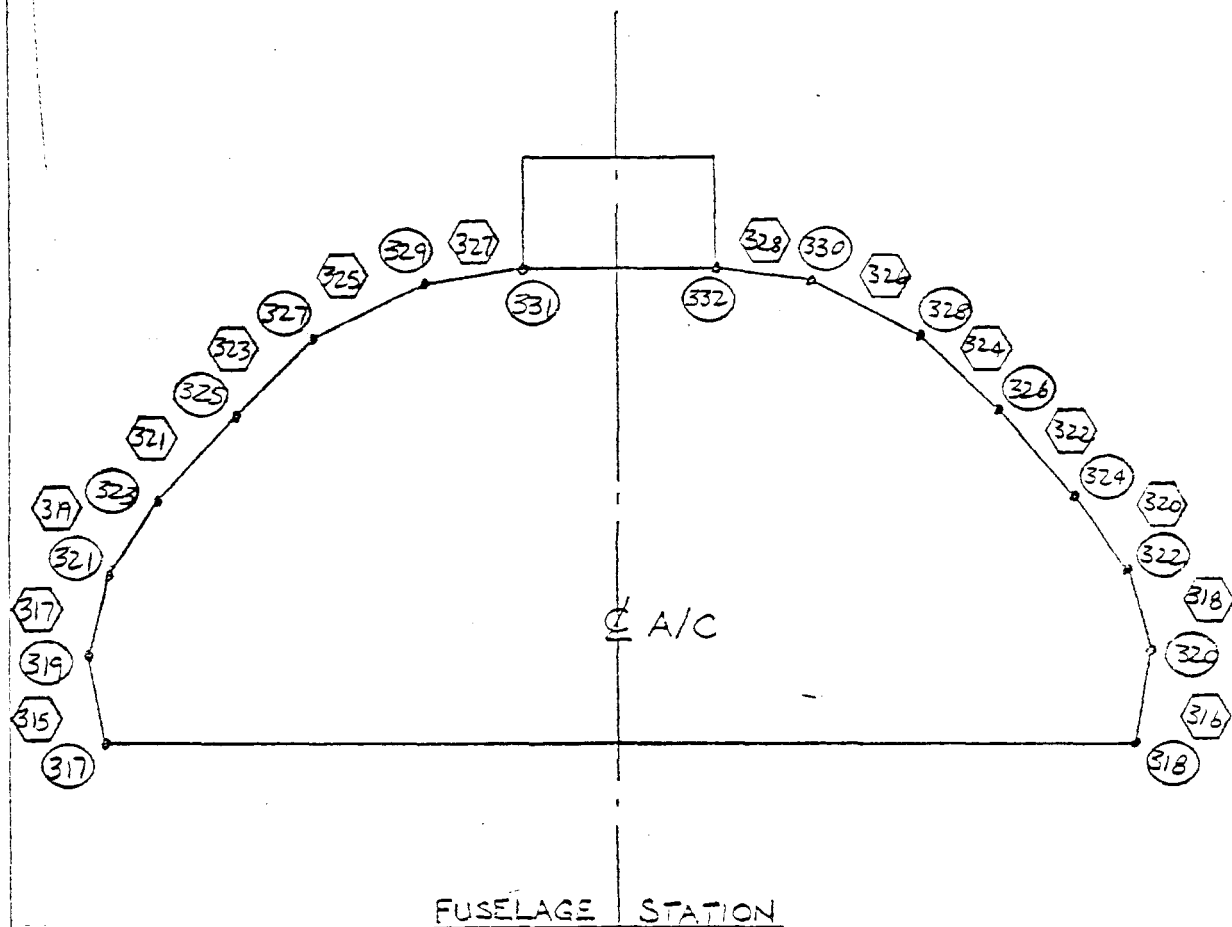
7.3.1 STRINGER LOADS CONDITION 1004001.13 & MOD 6

1	163	.345940+04	164	.352102+04	165	.610015+03	166	.626391+03
	167	.312539+04	168	.316636+04	169	.116952+05	170	.117353+05
	171	.740302+04	172	.745055+04	173	.725409+04	174	.733141+04
	177	.212144+04	178	.217615+04	179	.782661+04	180	.799073+04
	181	.170095+05	182	.173174+05	183	.895539+04	184	.912455+04
	185	.544438+04	186	.552001+04	187	-.680152+04	188	-.677729+04
	189	-.236488+05	190	-.237863+05	191	-.458472+05	192	-.460937+05
	193	-.784981+05	194	-.787043+05	195	.260344+05	196	.260044+05

LOADS ARE IN POUNDS

REV. (A) M.B. 1 AUG. 69

7.3.1 Stringer Loads (Continued)



FUSELAGE STATION

2461

MOD. 7

VIEW LOOKING FORWARD FS 2472 TO FS 2450

FIGURE No 7.3.1.3

KEY

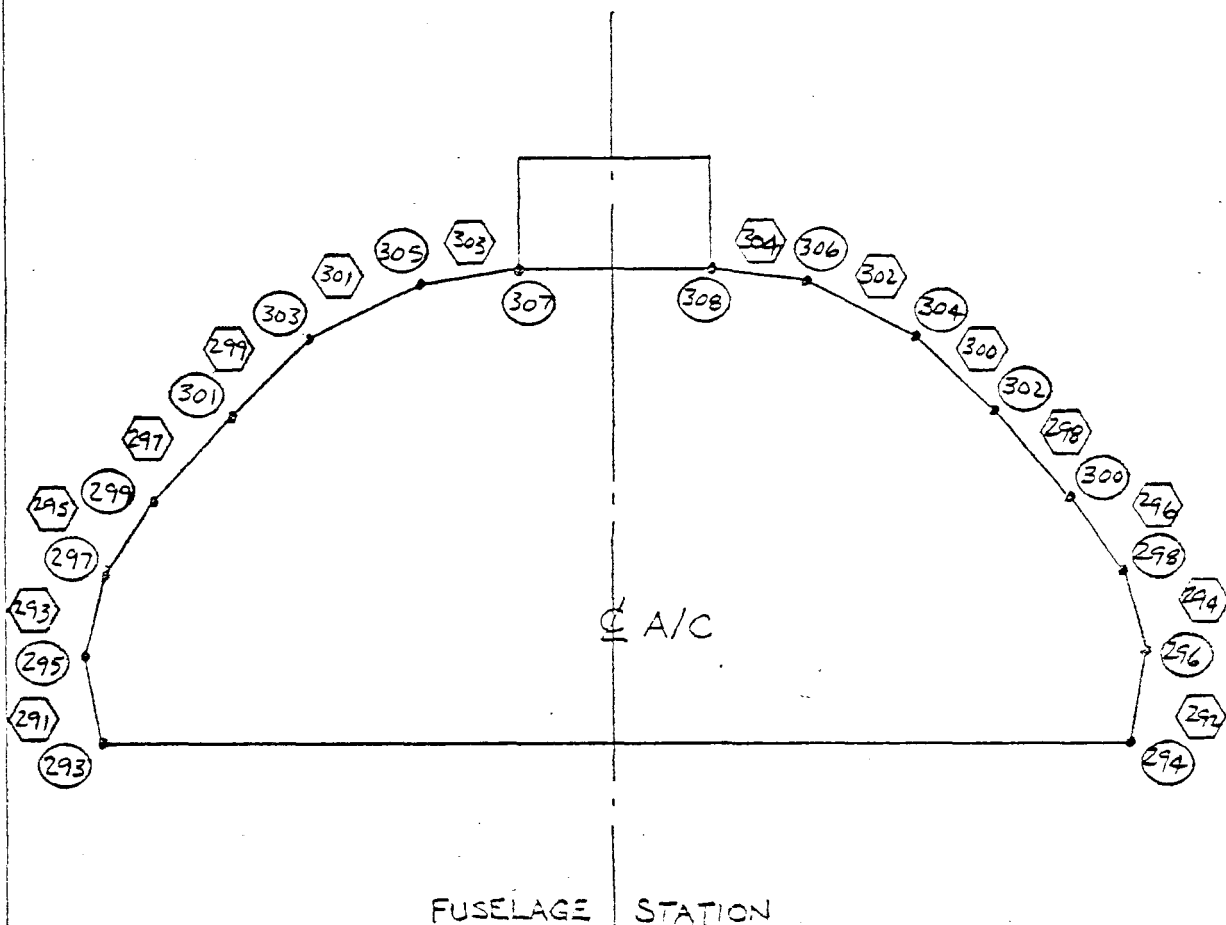
- GRID POINT
- FAMAS STRINGER
- ⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF B. VOLUME IV SECTION 5.2

Prepared	NAME L. H. HYDE	DATE 9-9-68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. 1	PERM. 7.37
Checked	J. HALLIDAY	9-9-68	TITLE AFT FUSELAGE	Model	C-5A	
Approved	L. G. DARBY	9-9-68	INTERNAL LOADS ANALYSIS	Report No.	LGIVS 46-2-2	

REV. (A) M.B. 1 AUG. 69

7.3.1 Stringer Loads (Continued)



FUSELAGE STATION

2483.00

MOD. 7

VIEW LOOKING FORWARD FS 2494 TO FS 2470

FIGURE No 7.3.1.14

KEY

—●— GRID POINT

○ FAMAS STRINGER

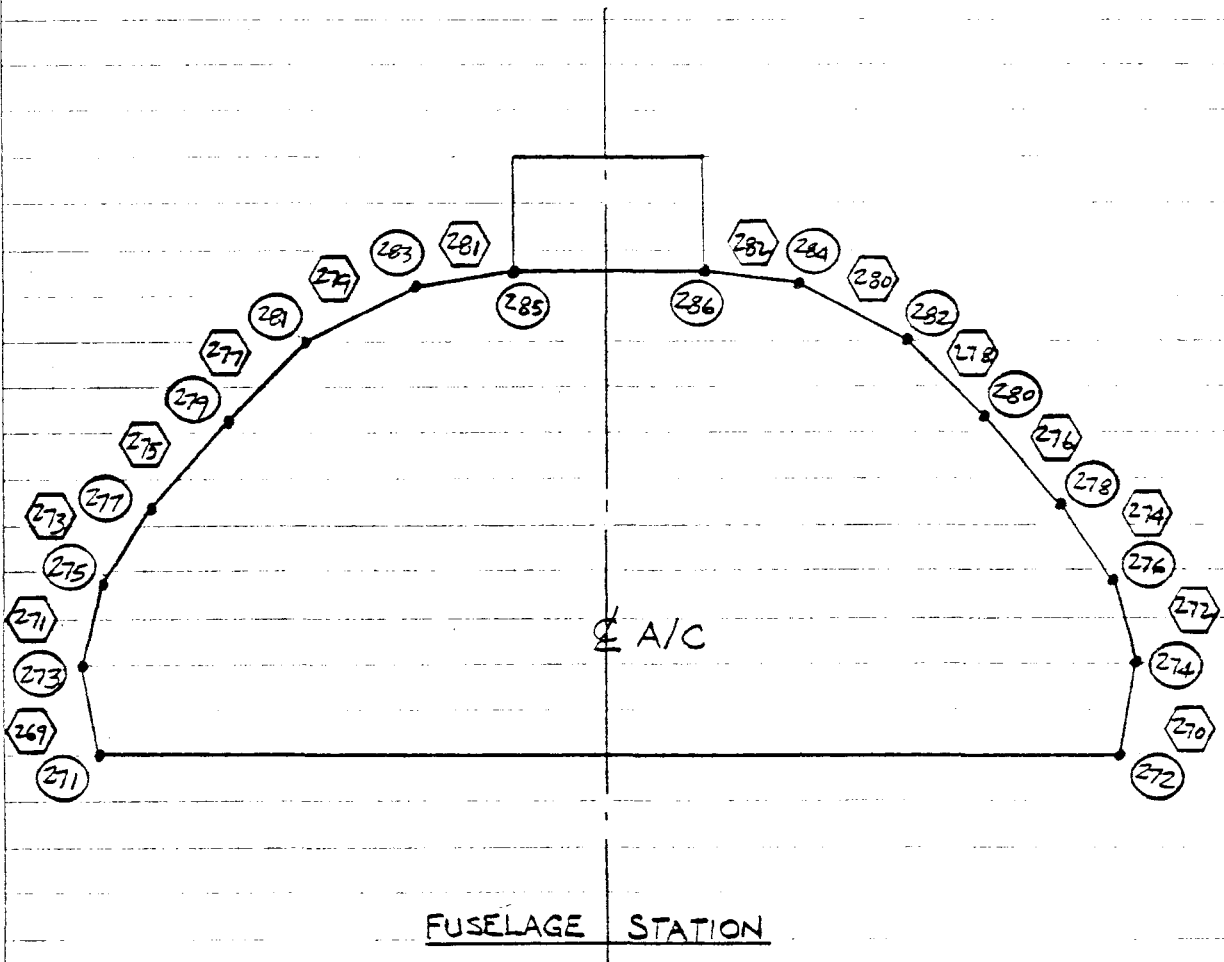
⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF B. VOLUME IV SECTION 5.2

Prepared	V. W. Boyd	DATE	9 Sep 68	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
				A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.33
Checked	J. Halliday	9 Sep 68	TITLE	INTERNAL LOADS ANALYSIS	Model	C-5A
				AFT FUSELAGE	LGJUS 46-2-2	
Approved	L.G. Darby	9 Sep 68			Report No.	

REV. (A) M.B. 1 AUG 69

7.3.1 Stringer Loads (Continued)



FUSELAGE STATION

2505

MOD. 7
VIEW LOOKING FORWARD FS.2516 TO FS.2494

FIGURE No 7.3.1.15

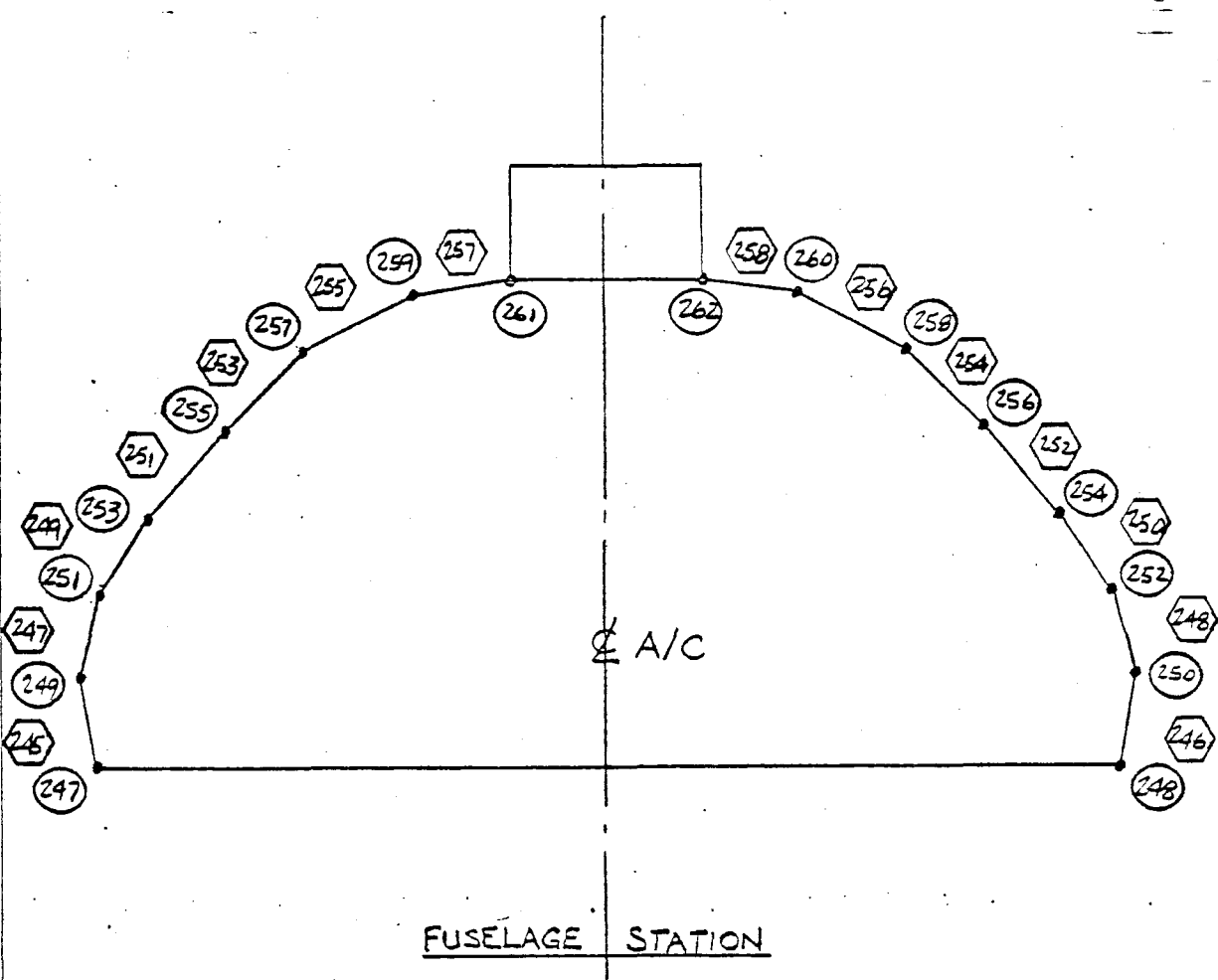
KEY

- GRID POINT
- FAMAS STRINGER
- ⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF'S
VOLUME IV SECTION 5.2

REV. (A) M.B. 1 AUG 67

7.3.1 STRINGER LOADS (CONTINUED)



FUSELAGE STATION

2527
MOD 7
VIEW LOOKING FORWARD FS. 2538 TO FS 2516
FIGURE No 7.3.1.16

KEY

 GRID POINT
  FAMAS STRINGER
  FAMAS PANEL

FOR AREAS AND THICKNESSES
 SEE REF 8. VOLUME IV SECTION 5.2

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	No. I 7.40
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		LG1JS46-2-2 Report No.	

REV ④ RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 2

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 7

1	.145272+04	2	.155331+04	3	.612059+03	4	.725875+03
5	.740102+01	6	.224630+03	7	-.127498+04	8	.211326+03
9	.799857+03	10	.126716+04	11	.119424+04	12	.122524+04
13	.245475+03	14	.248032+03	15	.140754+03	16	-.315776+03
17	.922722+03	18	.146864+03	19	-.706927+02	20	-.104197+03
21	.852813+03	22	.107561+04	23	.342970+05	24	.241553+03
25	.484784+04	26	.531564+04	27	.205695+04	28	.271238+04
29	.259730+03	30	.107506+04	31	-.180827+04	32	.235223+04
33	.310565+04	34	.431804+04	35	.439291+04	36	.458097+04
37	.186802+04	38	.126884+04	39	.159881+04	40	-.211195+03
41	.200593+04	42	-.119205+03	43	-.828367+03	44	-.711916+03
45	-.953635+03	46	-.121045+03	47	.356234+05	48	.275247+05
49	.870057+04	50	.968363+04	51	.413802+04	52	.555377+04
53	.130655+04	54	.291367+04	55	.128235+04	56	.731247+04
57	.654540+04	58	.991475+04	59	.895402+04	60	.941978+04
61	.494569+04	62	.329455+04	63	.363503+04	64	-.158935+02
65	.103622+04	66	-.238717+04	67	-.307157+04	68	-.267950+04
69	-.419088+04	70	-.259286+04	71	.254061+05	72	.235053+05
73	.120799+05	74	.134105+05	75	.648843+04	76	.837367+04
77	.315908+04	78	.535675+04	79	.671636+04	80	.142946+05
81	.103150+05	82	.150857+05	83	.134067+05	84	.143145+05
85	.793300+04	86	.564055+04	87	.462360+04	88	-.630901+03
89	-.131630+04	90	-.622834+04	91	-.590031+04	92	-.555976+04
93	-.592084+04	94	-.387989+04	95	.143692+05	96	.150563+05
97	.147458+05	98	.152283+05	99	.853833+04	100	.106398+05
101	.518866+04	102	.776655+04	103	.131141+05	104	.225049+05
105	.141576+05	106	.196656+05	107	.172653+05	108	.185235+05
109	.105669+05	110	.778217+04	111	.505004+04	112	-.114055+04
113	-.423605+04	114	-.102753+05	115	-.843071+04	116	-.874933+04
117	-.885757+04	118	-.672491+04	119	.810032+04	120	.712616+04
121	.171089+05	122	.186533+05	123	.105373+05	124	.128739+05
125	.726991+04	126	.102070+05	127	.207939+05	128	.311369+05
129	.131291+05	130	.238350+05	131	.211207+05	132	.225475+05
133	.130530+05	134	.101046+05	135	.540492+04	136	-.134357+04
137	-.762865+04	138	-.145030+05	139	-.118512+05	140	-.132303+05
141	-.115789+05	142	-.970335+04	143	.362143+03	144	-.137384+04
145	.193651+05	146	.209746+05	147	.125123+05	148	.143510+05
149	.935775+04	150	.122723+05	151	.288700+05	152	.392027+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7-1
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1JS45-2-2	

REV (A) RFW 8-1-69

7.0 SPELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION 1PC242 & MOD 7

1 153	.218748+05	154	.274703+05	155	.248419+05	156	.264411+05
157	.154869+05	158	.128127+05	159	.591690+04	160	-.115793+04
161	-.114627+05	162	-.139510+05	163	-.165520+05	164	-.185304+05
165	-.146358+05	166	-.133483+05	167	-.966867+04	168	-.118683+05
169	.213556+05	170	.230062+05	171	.147054+05	172	.167381+05
173	.117324+05	174	.143548+05	175	.368519+05	176	.482146+05
177	.253667+05	178	.305953+05	179	.279686+05	180	.295569+05
181	.182298+05	182	.157222+05	183	.638424+04	184	-.420856+03
185	-.157512+05	186	-.232426+05	187	-.219316+05	188	-.247045+05
189	-.191011+05	190	-.135375+05	191	-.170610+05	192	-.207287+05
193	.236120+05	194	.249354+05	195	.163680+05	196	.181625+05
197	.137849+05	198	.160079+05	199	.444907+05	200	.555872+05
201	.282308+05	202	.329661+05	203	.314907+05	204	.331227+05
205	.207270+05	206	.134435+05	207	.628383+04	208	.229470+03
209	-.196999+05	210	-.267690+05	211	-.275832+05	212	-.316824+05
213	-.217541+05	214	-.222257+05	215	-.282172+05	216	-.329636+05
217	.252219+05	218	.264007+05	219	.175570+05	220	.191123+05
221	.151052+05	222	.166772+05	223	.523869+05	224	.626556+05
225	.307228+05	226	.352265+05	227	.347012+05	228	.363207+05
229	.221047+05	230	.203791+05	231	.538161+04	232	.518869+02
233	-.232745+05	234	-.298774+05	235	-.339663+05	236	-.390457+05
237	-.158816+05	238	-.187103+05	239	-.558653+05	240	-.634415+05
241	.256150+05	242	.267152+05	243	.185630+05	244	.199695+05
245	.166297+05	246	.178922+05	247	.624112+05	248	.718215+05
* 249	.341670+05	250	.382690+05	251	.344648+05	252	.359127+05
253	.210852+05	254	.199639+05	255	.312937+04	256	-.160033+04
257	-.272618+05	258	-.336872+05	259	-.435701+05	260	-.502843+05
261	-.161088+05	262	-.206829+05	263	-.938382+05	264	-.109528+06
265	.251501+05	266	.261466+05	267	.191502+05	268	.204093+05
269	.198061+05	270	.208744+05	271	.734777+05	272	.821975+05
273	.374070+05	274	.409854+05	275	.323327+05	276	.334510+05
277	.178694+05	278	.169703+05	279	-.111365+04	280	-.573409+04
281	-.321496+05	282	-.385195+05	283	-.559323+05	284	-.640445+05
285	-.282231+05	286	-.330332+05	287	.244585+05	288	.252998+05
289	.182903+05	290	.192805+05	291	.153030+05	292	.203046+05
293	.612755+05	294	.691119+05	295	.396964+05	296	.429734+05
297	.304209+05	298	.314232+05	299	.132681+05	300	.122930+05
301	-.715294+04	302	-.110179+05	303	-.377591+05	304	-.438304+05

* STRINGER (249) REF. P. 7.39 ULTIMATE TENSION LOAD = 34,167, LB.

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.42
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	LG1J946-2-2		
				Report No.		

REV ① RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
7.3 SKINS AND STRINGERS
7.3.1 STRINGER LOADS CONDITION 1PC242 6 MOD 7

1	305	-.674988+05	306	-.751819+05	307	-.372065+05	308	-.419775+05
	309	-.741669+05	310	-.790316+05	311	.242808+05	312	.249981+05
	313	.177851+05	314	.185912+05	315	.179593+05	316	.188345+05
	317	.784577+05	318	.846498+05	319	.405756+05	320	.435766+05
	321	.278005+05	322	.289345+05	323	.753084+04	324	.671643+04
	325	-.149026+05	326	-.182396+05	327	-.446624+05	328	-.500234+05
	329	-.789144+05	330	-.856390+05	331	-.384911+05	332	-.421240+05
	333	-.772756+05	334	-.853297+05	335	-.626517+05	336	-.643903+05
	337	.824284+04	338	.712838+04	339	.246310+04	340	.233204+04
	341	.527914+04	342	.472983+04	343	.338947+04	344	.118020+04
	345	.646356+03	346	.248098+03	347	-.139338+04	348	-.154038+04
	349	-.415279+04	350	-.439645+04	351	-.776429+04	352	-.818991+04
	353	-.120054+05	354	-.126212+05	355	-.140989+05	356	-.147068+05
	357	-.171305+05	358	-.136910+05	359	-.160640+05	360	-.167753+05
	361	.213699+04	362	.184961+04	363	.450889+04	364	.537893+04
	365	.313750+05	366	.313301+05	367	.359659+05	368	.374670+05

LOADS ARE IN POUNDS

NAME FF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 7.43
Prepared			TITLE INTERNAL LOADS ANALYSIS			Model C-54	
Checked	A L PINCHOT	1 JUN 69				LG1US46-2-2	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No.		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.6
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION SPE201 & MOD 7

1	1	-.245035+04	2	-.253903+04	3	-.107839+04	4	-.117386+04
	5	-.118946+03	6	-.305312+03	7	.146131+04	8	.193588+03
	9	-.165166+04	10	-.205163+04	11	-.209429+04	12	-.212934+04
	13	-.331671+03	14	-.398965+03	15	.832060+02	16	.436682+03
	17	-.107279+04	18	-.470631+03	19	.157709+03	20	.158566+03
	21	-.153633+04	22	-.174254+04	23	-.533344+05	24	-.443418+05
	25	-.819310+04	26	-.860148+04	27	-.369876+04	28	-.425812+04
	29	-.809098+03	30	-.150832+04	31	.110484+04	32	-.244931+04
	33	-.624246+04	34	-.771700+04	35	-.763755+04	36	-.733962+04
	37	-.274920+04	38	-.230457+04	39	-.148716+04	40	-.749427+02
	41	-.200760+04	42	-.363051+03	43	.141724+04	44	.124344+04
	45	.120495+04	46	.454808+03	47	-.561994+05	48	-.498238+05
	49	-.147016+05	50	-.155573+05	51	-.745534+04	52	-.859003+04
	53	-.291514+04	54	-.429491+04	55	-.525025+04	56	-.108453+05
	57	-.128297+05	58	-.137479+05	59	-.154179+05	60	-.159128+05
	61	-.733225+04	62	-.616426+04	63	-.380198+04	64	-.930607+03
	65	.479367+03	66	.313523+04	67	.513131+04	68	.468047+04
	69	.632142+04	70	.487244+04	71	-.414657+05	72	-.406273+05
	73	-.203112+05	74	-.214720+05	75	-.115064+05	76	-.131443+05
	77	-.627631+04	78	-.817222+04	79	-.151937+05	80	-.223118+05
	81	-.197293+05	82	-.238780+05	83	-.229906+05	84	-.239212+05
	85	-.119337+05	86	-.101926+05	87	-.457336+04	88	-.423789+03
	89	.526520+04	90	.910346+04	91	.997429+04	92	.951616+04
	93	.905973+04	94	.713880+04	95	-.239271+05	96	-.252132+05
	97	-.246391+05	98	-.259375+05	99	-.149218+05	100	-.157548+05
	101	-.979299+04	102	-.120284+05	103	-.263254+05	104	-.344294+05
	105	-.263771+05	106	-.311980+05	107	-.294872+05	108	-.307679+05
	109	-.160493+05	110	-.139293+05	111	-.479497+04	112	.850454+02
	113	.107326+05	114	.134582+05	115	.145652+05	116	.145009+05
	117	.138936+05	118	.119364+05	119	-.125499+05	120	-.126627+05
	121	-.284104+05	122	-.297692+05	123	-.182002+05	124	-.202105+05
	125	-.133266+05	126	-.158839+05	127	-.394093+05	128	-.484314+05
	129	-.329483+05	130	-.379777+05	131	-.358703+05	132	-.373337+05
	133	-.200978+05	134	-.178806+05	135	-.512032+04	136	.179477+03
	137	.167968+05	138	.221729+05	139	.207280+05	140	.214306+05
	141	.185227+05	142	.167530+05	143	.860025+03	144	.141979+04
	145	-.319497+05	146	-.332856+05	147	-.214799+05	148	-.234507+05
	149	-.166777+05	150	-.192322+05	151	-.528025+05	152	-.624485+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. No. I	PERM. 7.44
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG10546-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION SPE201 & MOD 7

1	153	-.389271+05	154	-.439031+05	155	-.419839+05	156	-.436388+05
	157	-.242730+05	158	-.223231+05	159	-.587491+04	160	-.333303+03
	161	.233966+05	162	.292529+05	163	.289617+05	164	.302879+05
	165	.238177+05	166	.225128+05	167	.173274+05	168	.188140+05
	169	-.353619+05	170	-.366333+05	171	-.247029+05	172	-.265063+05
	173	-.203381+05	174	-.226605+05	175	-.656777+05	176	-.756978+05
	177	-.443027+05	178	-.490080+05	179	-.470647+05	180	-.438265+05
	181	-.289169+05	182	-.271248+05	183	-.688306+04	184	-.161021+04
	185	.303675+05	186	.352011+05	187	.381272+05	188	.401732+05
	189	.314317+05	190	.307219+05	191	.310661+05	192	.332706+05
	193	-.385009+05	194	-.396889+05	195	-.271608+05	196	-.287663+05
	197	-.233301+05	198	-.253275+05	199	-.775876+05	200	-.875736+05
	201	-.485085+05	202	-.528332+05	203	-.526735+05	204	-.544135+05
	205	-.331863+05	206	-.315929+05	207	-.724142+04	208	-.260306+04
	209	.363950+05	210	.418598+05	211	.480946+05	212	.511460+05
	213	.362830+05	214	.364207+05	215	.506170+05	216	.535988+05
	217	-.408543+05	218	-.419643+05	219	-.288303+05	220	-.302335+05
	221	-.248640+05	222	-.263991+05	223	-.395363+05	224	-.988360+05
	225	-.519801+05	226	-.561617+05	227	-.575222+05	228	-.532822+05
	229	-.357002+05	230	-.345783+05	231	-.634639+04	232	-.230629+04
	233	.415480+05	234	.466347+05	235	.587114+05	236	.625680+05
	237	.274203+05	238	.295582+05	239	.995733+05	240	.104955+06
	241	-.413655+05	242	-.423709+05	243	-.302490+05	244	-.315301+05
	245	-.271186+05	246	-.283109+05	247	-.104464+06	248	-.113179+06
	249	-.566269+05	250	-.605442+05	251	-.562318+05	252	-.578675+05
	253	-.340013+05	254	-.333763+05	255	-.343893+04	256	.165967+03
	257	.470953+05	258	.521420+05	259	.748404+05	260	.801108+05
	261	.283955+05	262	.319399+05	263	.168779+06	264	.181211+06
	265	-.404585+05	266	-.413747+05	267	-.310287+05	268	-.321825+05
	269	-.318957+05	270	-.329428+05	271	-.120448+06	272	-.128765+06
	273	-.608150+05	274	-.643622+05	275	-.517488+05	276	-.531112+05
	277	-.284593+05	278	-.279864+05	279	.266629+04	280	.595938+04
	281	.540008+05	282	.591806+05	283	.951749+05	284	.101749+06
	285	.488194+05	286	.524374+05	287	-.391544+05	288	-.399375+05
	289	-.295110+05	290	-.304115+05	291	-.309990+05	292	-.319601+05
	293	-.997174+05	294	-.107280+06	295	-.636326+05	296	-.668668+05
	297	-.480179+05	298	-.492904+05	299	-.207955+05	300	-.201959+05
	301	.119572+05	302	.150791+05	303	.621786+05	304	.672671+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. 17.45	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No. LG1JS45-2-2		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION SPE201 & MOD 7

1 305	.112924+06	306	.119266+06	307	.640988+05	308	.675837+05
309	.127822+06	310	.131697+06	311	-.374991+05	312	-.382793+05
313	-.273579+05	314	-.282899+05	315	-.275169+05	316	-.285233+05
317	-.122970+06	318	-.129825+06	319	-.634997+05	320	-.670759+05
321	-.430469+05	322	-.447921+05	323	-.115801+05	324	-.114031+05
325	.235553+05	326	.260267+05	327	.715043+05	328	.753593+05
329	.128527+06	330	.134024+06	331	.643101+05	332	.669646+05
333	.136229+06	334	.143266+06	335	.104146+06	336	.105817+06
337	-.133935+05	338	-.124364+05	339	-.408929+04	340	-.396599+04
341	-.857432+04	342	-.820112+04	343	-.459937+04	344	-.370557+04
345	-.818215+03	346	-.575781+03	347	.252526+04	348	.257844+04
349	.726732+04	350	.738565+04	351	.135114+05	352	.137657+05
353	.208272+05	354	.212258+05	355	.242597+05	356	.246493+05
357	.301986+05	358	.313979+05	359	.279373+05	360	.285755+05
361	-.347408+04	362	-.321902+04	363	-.670884+04	364	-.940413+04
365	-.488807+05	366	-.491501+05	367	-.551632+05	368	-.572184+05

LOADS ARE IN POUNDS

Prepared	NAME R F WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Page	VOL. I 7.46
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Model	C-54
				Report No.	LG10545-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION DML001 & MOD 7

1	1	-.661999+03	2	-.487132+03	3	-.419261+03	4	-.921522+02
	5	-.534861+03	6	.447758+03	7	-.336069+04	8	.377460+04
	9	-.136921+04	10	.523030+03	11	.995752+01	12	-.986349+03
	13	.704142+03	14	-.833719+03	15	.305335+04	16	-.291444+04
	17	.496084+04	18	-.526140+04	19	.222480+04	20	-.210750+04
	21	.207514+04	22	-.279559+04	23	.363135+05	24	-.585304+05
	25	-.256214+04	26	-.133303+04	27	-.206757+04	28	.228557+03
	29	-.207750+04	30	.154839+04	31	-.999734+04	32	.973370+04
	33	-.463688+04	34	.141376+04	35	.490706+03	36	-.408631+04
	37	.459322+04	38	-.576144+04	39	.112814+05	40	-.115561+05
	41	.141920+05	42	-.145300+05	43	.580833+04	44	-.501477+04
	45	.620479+04	46	-.569202+04	47	.504113+05	48	-.740106+05
	49	-.490390+04	50	-.212567+04	51	-.445225+04	52	.706288+03
	53	-.429910+04	54	.253328+04	55	-.167638+05	56	.130797+05
	57	-.855346+04	58	.193125+04	59	.163397+04	60	-.892120+04
	61	.108910+05	62	-.139714+05	63	.221336+05	64	-.230265+05
	65	.241142+05	66	-.228469+05	67	.965955+04	68	-.699301+04
	69	.894275+04	70	-.603005+04	71	.457551+05	72	-.633165+05
	73	-.654806+04	74	-.316993+04	75	-.592563+04	76	.197150+03
	77	-.604282+04	78	.269077+04	79	-.222104+05	80	.135369+05
	81	-.117880+05	82	.165443+04	83	.239962+04	84	-.133089+05
	85	.159149+05	86	-.209857+05	87	.320130+05	88	-.328450+05
	89	.348309+05	90	-.308343+05	91	.142762+05	92	-.911528+04
	93	.913615+04	94	-.481346+04	95	.403484+05	96	-.439625+05
	97	-.739795+04	98	-.437293+04	99	-.660292+04	100	-.766061+03
	101	-.714303+04	102	.207061+04	103	-.257455+05	104	.115380+05
	105	-.133087+05	106	-.835937+02	107	.296902+04	108	-.159891+05
	109	.197383+05	110	-.255934+05	111	.392164+05	112	-.398651+05
	113	.435634+05	114	-.355932+05	115	.190495+05	116	-.113993+05
	117	.792201+04	118	-.119844+04	119	.446725+05	120	-.484663+05
	121	-.788130+04	122	-.556493+04	123	-.714119+04	124	-.180324+04
	125	-.806657+04	126	.126456+04	127	-.284952+05	128	.804023+04
	129	-.132824+05	130	-.324496+04	131	.359250+04	132	-.206358+05
	133	.219917+05	134	-.306891+05	135	.448749+05	136	-.454612+05
	137	.509441+05	138	-.407740+05	139	.240607+05	140	-.131239+05
	141	.783662+04	142	.124923+04	143	.439753+05	144	-.412618+05
	145	-.806137+04	146	-.713218+04	147	-.693357+04	148	-.354130+04
	149	-.779940+04	150	-.579686+03	151	-.295236+05	152	.263406+04

LOADS ARE IN POUNDS

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	VOL. I 7.47
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-FA	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		L 610546-2-2 Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION DML001 & MOD 7

1	153	-.122410+05	154	-.708236+04	155	.395206+04	156	-.233958+05
	157	.222662+05	158	-.329583+05	159	.490393+05	160	-.498587+05
	161	.563885+05	162	-.433150+05	163	.288857+05	164	-.137032+05
	165	.873775+04	166	.306855+04	167	.405836+05	168	-.298338+05
	169	-.817028+04	170	-.860419+04	171	-.628811+04	172	-.566177+04
	173	-.673073+04	174	-.332319+04	175	-.286427+05	176	-.440018+04
	177	-.102953+05	178	-.115010+05	179	.440580+04	180	-.267508+05
	181	.223773+05	182	-.352455+05	183	.504677+05	184	-.516582+05
	185	.601713+05	186	-.431472+05	187	.337224+05	188	-.133641+05
	189	.100946+05	190	.554714+04	191	.429844+05	192	-.256108+05
	193	-.803220+04	194	-.101738+05	195	-.538462+04	196	-.766951+04
	197	-.518382+04	198	-.620819+04	199	-.252264+05	200	-.134665+05
	201	-.769339+04	202	-.160167+05	203	.506475+04	204	-.300384+05
	205	.218728+05	206	-.367679+05	207	.494791+05	208	-.503676+05
	209	.609280+05	210	-.410019+05	211	.397154+05	212	-.147502+05
	213	.124851+05	214	.578832+04	215	.466800+05	216	-.109779+05
	217	-.768231+04	218	-.115558+05	219	-.439348+04	220	-.938966+04
	221	-.315713+04	222	-.885607+04	223	-.191984+05	224	-.250666+05
	225	-.436049+04	226	-.204562+05	227	.630506+04	228	-.335459+05
	229	.201679+05	230	-.363007+05	231	.477114+05	232	-.436765+05
	233	.597489+05	234	-.373625+05	235	.449722+05	236	-.145459+05
	237	.169057+05	238	-.225184+04	239	.650508+05	240	-.145342+05
	241	-.714243+04	242	-.123102+05	243	-.350274+04	244	-.109059+05
	245	-.139889+04	246	-.116013+05	247	-.111982+05	248	-.401110+05
	249	-.357575+03	250	-.271155+05	251	.955560+04	252	-.261106+05
	253	.188631+05	254	-.341850+05	255	.443713+05	256	-.433832+05
	257	.574130+05	258	-.322530+05	259	.539698+05	260	-.155320+05
	261	.236842+05	262	-.829895+04	263	.113184+06	264	-.280654+05
	265	-.634361+04	266	-.125716+05	267	-.247677+04	268	-.122029+05
	269	.591298+03	270	-.157577+05	271	-.172406+04	272	-.570941+05
	273	.473502+04	274	-.340953+05	275	.133313+05	276	-.376150+05
	277	.191395+05	278	-.317016+05	279	.405368+05	280	-.370375+05
	281	.547681+05	282	-.261616+05	283	.628264+05	284	-.143423+05
	285	.314749+05	286	-.652203+04	287	-.551469+04	288	-.125625+05
	289	-.152132+04	290	-.123163+05	291	.158166+04	292	-.162532+05
	293	.791071+03	294	-.497353+05	295	.861703+04	296	-.392131+05
	297	.155215+05	298	-.379305+05	299	.201345+05	300	-.238917+05
	301	.382026+05	302	-.301980+05	303	.521010+05	304	-.195925+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.48
Checked	A. L. PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L. G. DARBY	1 JUN 69		AFT FUSELAGE	Report No.	LG10646-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LOADS CONDITION DML001 & MOD 7

1 305	.637502+05	306	-.688455+04	307	.361229+05	308	-.401153+04
309	.527074+05	310	.103178+05	311	-.464934+04	312	-.127683+05
313	-.501204+03	314	-.123727+05	315	.242962+04	316	-.155350+05
317	.147167+05	318	-.744416+05	319	.116406+05	320	-.422462+05
321	.165911+05	322	-.367178+05	323	.211840+05	324	-.255159+05
325	.365304+05	325	-.230463+05	327	.492603+05	328	-.124112+05
329	.627038+05	330	.146210+04	331	.316950+05	332	.287587+03
333	.746327+05	334	-.667525+04	335	.351083+05	336	.159337+05
337	.701169+03	338	-.653895+04	339	-.706487+03	340	-.111089+04
341	.448696+04	342	-.809178+04	343	.951691+04	344	-.109404+05
345	.396568+04	346	-.389346+04	347	.405257+04	348	-.235451+04
349	.561145+04	350	-.156188+04	351	.759150+04	352	-.477425+03
353	.960495+04	354	.104245+04	355	.985577+04	356	.234754+04
357	.146064+05	358	.465929+03	359	.977746+04	360	.395562+04
361	.341315+03	362	-.185120+04	363	-.214524+04	364	-.139255+04
365	.101116+05	366	-.331038+05	367	.932821+04	368	-.357715+05

LOADS ARE IN POUNDS

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.&

7.3 SKINS AND STRINGERS

7.3.1 STRINGER LCADS CONDITION 1004001.18 & MOD 7

1	.142984+04	2	.142917+04	3	.676521+03	4	.676318+03
5	.134120+03	6	.134030+03	7	-.527008+03	8	-.529832+03
9	.136554+04	10	.136452+04	11	.151886+04	12	.151892+04
13	.201550+03	14	.201327+03	15	-.367647+03	16	-.366213+03
17	.218878+03	18	.219288+03	19	-.133989+03	20	-.133796+03
21	.977254+03	22	.977802+03	23	.309134+05	24	.309089+05
25	.476597+04	26	.476360+04	27	.236887+04	28	.236757+04
29	.754202+03	30	.752776+03	31	.574127+03	32	.567012+03
33	.511691+04	34	.511233+04	35	.554069+04	36	.554040+04
37	.149785+04	38	.149913+04	39	-.293296+03	40	-.289072+03
41	-.276796+03	42	-.274079+03	43	-.109153+04	44	-.109048+04
45	-.522255+03	46	-.522953+03	47	.333989+05	48	.333774+05
49	.837487+04	50	.837025+04	51	.470144+04	52	.469903+04
53	.229613+04	54	.229138+04	55	.557584+04	56	.556552+04
57	.102963+05	58	.102870+05	59	.110343+05	60	.110309+05
61	.411540+04	62	.412161+04	63	-.248859+02	64	-.169130+02
65	-.319735+04	66	-.318973+04	67	-.380038+04	68	-.379797+04
69	-.355701+04	70	-.355781+04	71	.254935+05	72	.254912+05
73	.112492+05	74	.112426+05	75	.726812+04	76	.706335+04
77	.457838+04	78	.457316+04	79	.126101+05	80	.125958+05
81	.153893+05	82	.153743+05	83	.163122+05	84	.163063+05
85	.687584+04	86	.688659+04	87	-.443083+03	88	-.430337+03
89	-.755378+04	90	-.754152+04	91	-.740152+04	92	-.739690+04
93	-.550099+04	94	-.550110+04	95	.144064+05	96	.144017+05
97	.132722+05	98	.132641+05	99	.888953+04	100	.888138+04
101	.677757+04	102	.677235+04	103	.199639+05	104	.199385+05
105	.198559+05	106	.198438+05	107	.207316+05	108	.207262+05
109	.941516+04	110	.941378+04	111	-.739373+03	112	-.722759+03
113	-.120309+05	114	-.120144+05	115	-.112242+05	116	-.112261+05
117	-.870962+04	118	-.871145+04	119	.558760+04	120	.558885+04
121	.149124+05	122	.149031+05	123	.105006+05	124	.104904+05
125	.380843+04	126	.380034+04	127	.281772+05	128	.281409+05
129	.238669+05	130	.238523+05	131	.248237+05	132	.248164+05
133	.121983+05	134	.122010+05	135	-.699152+03	136	-.679607+03
137	-.166934+05	138	-.166714+05	139	-.162727+05	140	-.162713+05
141	-.119133+05	142	-.119161+05	143	-.424538+04	144	-.424052+04
145	.163717+05	146	.163618+05	147	.120259+05	148	.120119+05
149	.105664+05	150	.105552+05	151	.358777+05	152	.358493+05

LOADS ARE IN POUNDS

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.51
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-54	
Approved	L G DAREY	1 JUN 69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.1 STRINGER LOADS CONDITION 1004001.18 & MOD 7

1 305	-.756246+05	306	-.756457+05	307	-.471682+05	308	-.471500+05
309	-.100661+06	310	-.100638+06	311	.170842+05	312	.170925+05
313	.122209+05	314	.122156+05	315	.116409+05	316	.115353+05
317	.503361+05	318	.502928+05	319	.266461+05	320	.266298+05
321	.187402+05	322	.187614+05	323	.319729+04	324	.324797+04
325	-.599227+04	326	-.595732+04	327	-.359238+05	328	-.359496+05
329	-.821694+05	330	-.822104+05	331	-.465025+05	332	-.465748+05
333	-.111564+06	334	-.111524+06	335	-.778313+05	336	-.778147+05
337	.812103+04	338	.812071+04	339	.253731+04	340	.253746+04
341	.514034+04	342	.513870+04	343	.205399+04	344	.204879+04
345	-.177876+03	346	-.179223+03	347	-.269136+04	348	-.269159+04
349	-.640979+04	350	-.640993+04	351	-.113596+05	352	-.113597+05
353	-.170673+05	354	-.170673+05	355	-.190270+05	356	-.190263+05
357	-.253586+05	358	-.253544+05	359	-.231169+05	360	-.231210+05
361	.211149+04	362	.211154+04	363	.567566+04	364	.567795+04
365	.189180+05	366	.189969+05	367	.217842+05	368	.217592+05

LOADS ARE IN POUNDS

Prepared	NAME V. W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.52
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT. FUSELAGE	Model	C-5A	
Approved	L.G. Darby	16 Aug 68		IGIUS 46-2-2 Report No.		

REV. (A) MB. AUG 69

7.3.2. Panel Loads (Continued)

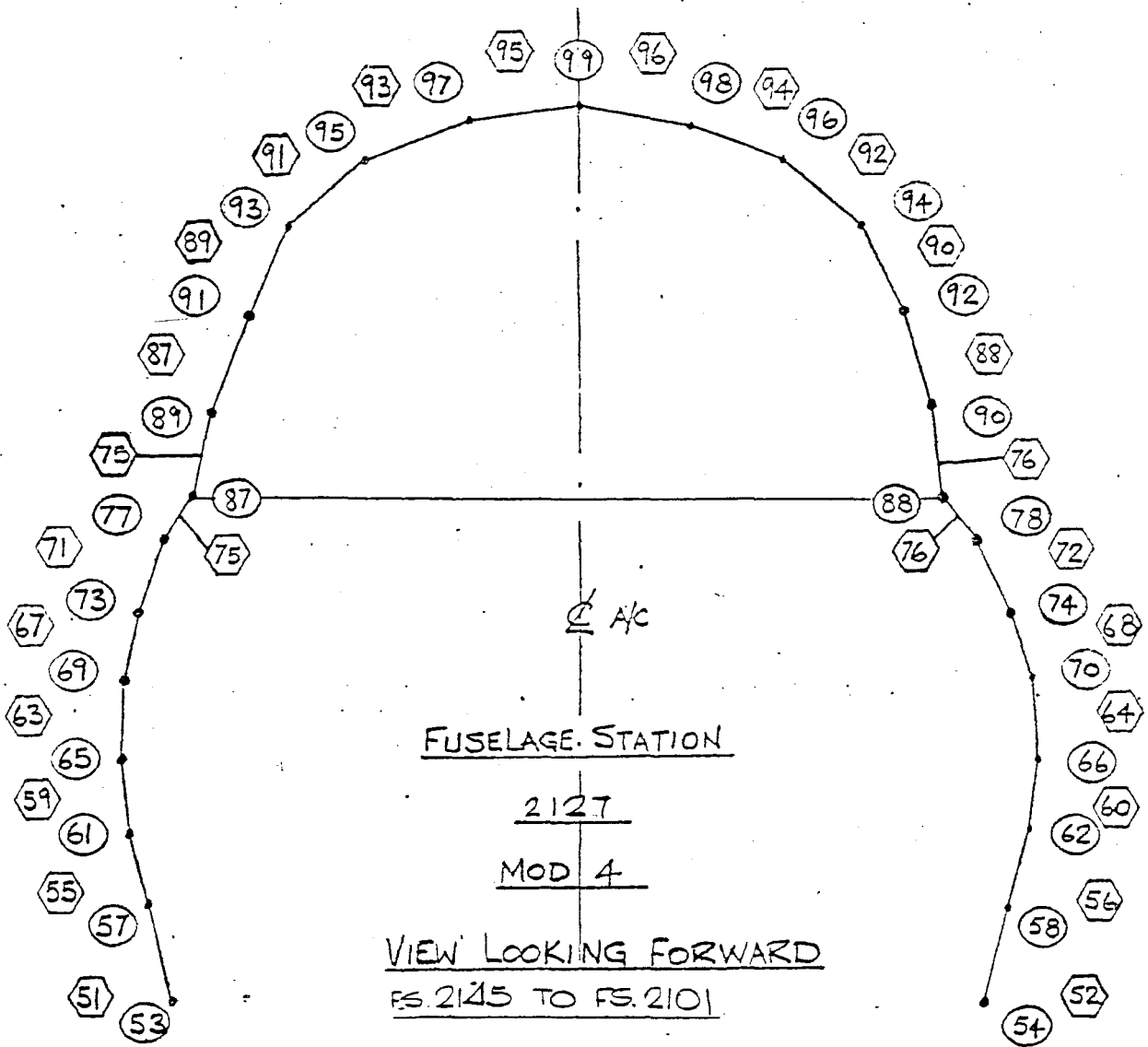


FIGURE No 7.3.2.1

KEY

— GRID POINT

○ FAMAS STRINGER

⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 3.2.

Prepared	NAME V. W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. No. I	PERIL 7.53
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. Darby	16 Aug 68		Report No.	LGUS 46-2-2	

REV. (A) MO. 1 AUG. 69

7.3.2 Panel Loads (Continued)

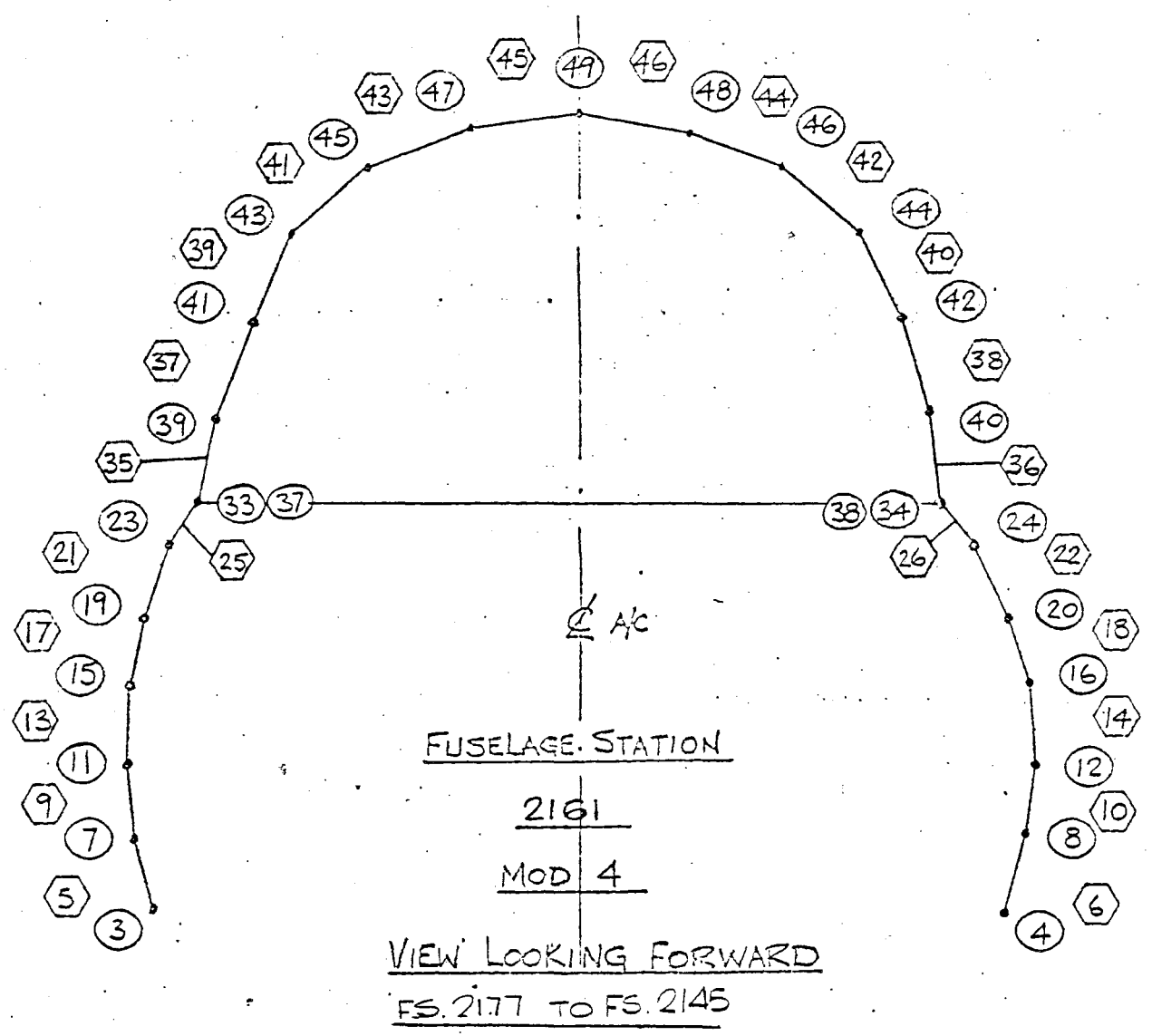


FIGURE No 7.3.2.2

KEY

 GRID POINT

 FAMAS STRINGER

 FAMAS PANEL

FOR ACES AND THICKNESS SEE REF 8
VOLUME IV SECTION 5.2.

NAME RF WILKINSON		DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. JUL I	PERM. 7.54
Prepared					Page	
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS		Model	C-5A
Approved	L G DARRY	1JUN69	AFT FUSELAGE		Report No.	LG1US46-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 & CON. I

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS & CONDITION IPC242 I MOD 4

1	1	.931842+02	2	.217076+03	5	-.272977+03	6	-.116019+03
* 9	-.224929+03	10	-.772279+02	13	-.191531+03	14	-.735947+02	
17	-.476751+03	18	-.323523+03	21	-.295393+03	22	-.225416+03	
25	-.339835+03	26	-.254934+03	27	-.131362+02	28	.122232+03	
29	.216693+02	30	.143824+03	31	.475317+02	32	.166744+03	
33	.786900+02	34	.177583+03	35	.156365+02	36	.198626+03	
37	.460550+02	38	.181182+03	39	.120546+03	40	.230728+03	
41	.967350+02	42	.222422+03	43	.407825+02	44	.194372+03	
45	-.405226+02	46	.119252+03	47	-.239803+03	48	-.136521+03	
49	.288794+03	50	.240379+03	51	-.289278+02	52	.151517+03	
53	.496536+02	54	.223967+02	55	-.725110+02	56	.982727+02	
57	-.394137+02	58	-.616889+02	59	-.183171+03	60	-.944739+01	
61	-.684455+02	62	-.859030+02	63	-.168546+03	64	-.366357+02	
65	-.518979+02	66	-.637287+02	67	-.203044+03	68	-.964747+02	
69	-.472101+02	70	-.651208+02	71	-.878287+02	72	-.336858+01	
73	-.422931+02	74	-.641542+02	75	-.124380+03	76	-.771247+02	
77	-.900828+02	78	.713115+02	79	.559108+01	80	.126599+03	
81	.492925+02	82	.175557+03	83	.112045+03	84	.229530+03	
85	.143956+03	86	.261693+03	87	.158131+03	88	.326171+03	
89	.170422+03	90	.297945+03	91	.128473+03	92	.244459+03	
93	.346377+02	94	.166868+03	95	-.652616+02	96	.105014+03	
97	.172023+03	98	.159271+03	99	.502791+03	100	.456058+03	
101	.178954+02	102	.127975+03	103	-.816515+02	104	-.103319+03	
105	.171052+02	106	.166894+03	107	-.749570+02	108	-.948436+02	
109	-.855141+02	110	.767945+02	111	-.668333+02	112	-.873402+02	
113	-.164088+03	114	.553827+01	115	-.392354+02	116	-.658366+02	
117	-.247383+03	118	-.112974+03	119	-.369999+02	120	-.723781+02	
121	-.153507+03	122	-.420835+02	123	-.914049+01	124	-.449567+02	
125	-.995340+02	126	-.373563+02	127	-.193646+02	128	-.637718+02	
129	-.308521+02	130	.519205+02	131	.254222+02	132	.254222+02	
133	-.728244+02	134	.206492+03	135	.485633+02	136	.206052+03	
137	.734869+02	138	.250949+03	139	.132680+03	140	.262524+03	
141	.220705+03	142	.322779+03	143	.216774+03	144	.350494+03	
145	.151963+03	146	.308394+03	147	.911971+02	148	.272628+03	
149	-.515141+01	150	.162082+03	151	.814227+02	153	.124813+03	
154	.205432+03	155	-.340907+03	156	-.310835+03	159	-.197083+03	
160	-.439245+02	163	-.193781+03	164	-.284068+02	167	-.880501+02	
168	.747712+02	171	-.331673+03	172	-.173553+03	175	-.225936+03	

* PANEL (9) REF. P. 7.53 ULTIMATE PANEL SHEAR FLOW = -225. LBS./IN.

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.55
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	L G DAPRY	1JUN69	AFT FUSELAGE	Report No. LG1US46-2-2		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 & CON. I

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS & CONDITION IPC242 I MOD 4

1	176	-	.995216+02	179	-	.160577+03	180	-	.390317+02	183	-	.482041+02
	184		.490784+02	185		.227304+03	186		.185352+03	187		.405765+01
	188		.374305+01	189		.170408-00	190		.123392+00	191	-	.394086-00
	192	-	.280449-00	193		.171962+03	194		.159210+03	195		.171698+03
	196		.158965+03	197	-	.178920+02	198	-	.127973+03			

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I 7.56	PERM.
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DAPPY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 *CON.1

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS *CONDITION SPE201 I MOD 4

1	1	-.145042+03	2	-.268643+03	5	.417092+03	6	.276036+03
	9	.341271+03	10	.209355+03	13	.289548+03	14	.186074+03
	17	.667151+03	18	.543304+03	21	.394636+03	22	.341426+03
	25	.416649+03	26	.353122+03	27	-.358824+02	28	-.154140+03
	29	-.975263+02	30	-.205890+03	31	-.147702+03	32	-.254293+03
	33	-.201066+03	34	-.291291+03	35	-.152807+03	36	-.297162+03
	37	-.173472+03	38	-.294156+03	39	-.305513+03	40	-.403968+03
	41	-.283357+03	42	-.397625+03	43	-.179435+03	44	-.319315+03
	45	-.110531+02	46	-.155251+03	47	.336885+03	48	.240621+03
	49	-.349327+03	50	-.308089+03	51	.140519+03	52	-.319284+02
	53	-.206694+02	54	.491412+01	55	.176986+03	56	.175036+02
	57	.107596+03	58	.129956+03	59	.305967+03	60	.149716+03
	61	.131776+03	62	.149640+03	63	.267895+03	64	.153995+03
	65	.113326+03	66	.125135+03	67	.282786+03	68	.196828+03
	69	.117143+03	70	.133615+03	71	.126594+03	72	.591948+02
	73	.112028+03	74	.130856+03	75	.152847+03	76	.122957+03
	77	.738997+02	78	-.637993+02	79	-.542709+02	80	-.160032+03
	81	-.133652+03	82	-.245615+03	83	-.227785+03	84	-.333131+03
	85	-.225753+03	86	-.335006+03	87	-.292211+03	88	-.441336+03
	89	-.325012+03	90	-.443277+03	91	-.274374+03	92	-.384614+03
	93	-.168677+03	94	-.288245+03	95	-.622924-01	96	-.152269+03
	97	-.283052+03	98	-.275312+03	99	-.739087+03	100	-.708543+03
	101	.386204+02	102	-.724945+02	103	.695851+02	104	.919445+02
	105	-.646152+02	106	-.211904+03	107	.105207+03	108	.125697+03
	109	.982632+02	110	-.552305+02	111	.112339+03	112	.133227+03
	113	.217131+03	114	.624729+02	115	.951792+02	116	.121001+03
	117	.306032+03	118	.192418+03	119	.843317+02	120	.118514+03
	121	.203508+03	122	.114026+03	123	.774003+02	124	.113332+03
	125	.135174+03	126	.932876+02	127	.339803+02	128	.787118+02
	129	.410497+02	130	-.206711+02	131	-.151096+02	132	-.151096+02
	133	.187677+01	134	-.236176+03	135	-.169695+03	136	-.307049+03
	137	-.251509+03	138	-.414477+03	139	-.322135+03	140	-.438409+03
	141	-.356038+03	142	-.439687+03	143	-.286553+03	144	-.417026+03
	145	-.204737+03	146	-.361668+03	147	-.789558+02	148	-.247955+03
	149	.265201+02	150	-.112988+03	151	-.702447+02	153	.737496+01
	154	-.750073+02	155	.385644+03	156	.363247+03	159	.215764+03
	160	.704974+02	163	.197102+03	164	.416072+02	167	.691902+02
	168	-.867222+02	171	.431607+03	172	.296803+03	175	.270864+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.57
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DABBY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 (CON.1

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS (CONDITION SPE201 1 MOD 4

1	176	.165026+03	179	.210540+03	180	.111287+03	183	.128401+03
	184	.589884+02	185	-.304029+03	186	-.267368+03	187	-.611948+01
	188	-.590068+01	189	-.240738-00	190	-.199217-00	191	.551075-00
	192	.450222-00	193	-.282943+03	194	-.275205+03	195	-.282508+03
	196	-.274782+03	197	-.386278+02	198	.724877+02		

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. No. 1	PERM. 7.53
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARRY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 (CON. 1)

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS (CONDITION DML001 1 MOD 4

1	1	-.402230+02	2	-.523206+02	5	-.161726+03	6	.386008+03
	9	-.129566+03	10	.322220+03	13	-.112803+03	14	.279952+03
	17	-.391239+03	18	.742372+03	21	-.350440+03	22	.556920+03
	25	-.509853+03	26	.721325+03	27	-.346145+01	28	-.516621+02
	29	-.617969+02	30	-.281724+02	31	-.819107+02	32	-.319787+02
	33	-.116341+03	34	-.210749+02	35	-.619441+03	36	.510414+03
	37	-.497234+03	38	.384919+03	39	-.567799+03	40	.378676+03
	41	-.687740+03	42	.500921+03	43	-.788336+03	44	.653927+03
	45	-.802253+03	46	.758239+03	47	-.121289+03	48	.280751+03
	49	.166699+03	50	-.312545+03	51	-.100321+03	52	.177263+03
	53	.668846+02	54	-.600229+02	55	-.128637+03	56	.221714+03
	57	.339449+02	58	.360560+02	59	-.233518+03	60	.385395+03
	61	.221669+02	62	.550664+02	63	-.272912+03	64	.404488+03
	65	.231047+02	66	.456024+02	67	-.420898+03	68	.556999+03
	69	.294778+02	70	.435466+02	71	-.368142+03	72	.431469+03
	73	-.451965+01	74	.721804+02	75	-.554642+03	76	.635122+03
	77	.987107+02	78	-.105094+03	79	-.129087+02	80	-.476757+02
	81	-.473620+02	82	-.656189+02	83	-.613926+02	84	-.100195+03
	85	-.677215+02	86	-.821913+02	87	-.566071+03	88	.377566+03
	89	-.597963+03	90	.387135+03	91	-.632158+03	92	.462124+03
	93	-.769708+03	94	.647035+03	95	-.858127+03	96	.812763+03
	97	.634344+02	98	-.210096+03	99	.215644+03	100	-.573207+03
	101	-.760518+02	102	.108726+03	103	.769387+01	104	.256251+02
	105	-.641285+02	106	.811021+01	107	.516605+01	108	.554127+02
	109	-.125840+03	110	.150483+03	111	.627504+01	112	.618847+02
	113	-.196794+03	114	.282441+03	115	.177553+02	116	.481926+02
	117	-.347418+03	118	.474820+03	119	.192866+02	120	.409821+02
	121	-.353638+03	122	.447093+03	123	.791742+01	124	.589663+02
	125	-.381866+03	126	.452729+03	127	-.140382+02	128	.496049+02
	129	-.580498+03	130	.604318+03	131	.772909+01	132	.772906+01
	133	.331295+03	134	-.414386+03	135	.882121+02	136	-.245552+03
	137	-.668484+02	138	-.156750+03	139	-.902119+02	140	-.150737+03
	141	-.687769+03	142	.466851+03	143	-.692765+03	144	.530234+03
	145	-.702689+03	146	.597550+03	147	-.579101+03	148	.566912+03
	149	-.501041+03	150	.525357+03	151	-.611388+02	153	-.360137+02
	154	.850831+02	155	-.133751+03	156	.308771+03	159	-.165076+03
	160	.234904+03	163	-.165748+03	164	.221761+03	167	-.553529+02
	168	.601791+02	171	-.375693+03	172	.561413+03	175	-.343585+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 759
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 & CON. I

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS & CONDITION DM L001 I MOD 4

1	176	.457252+03	179	-.322856+03	180	.420156+03	183	-.615680+03
	184	.696707+03	185	.732653+02	186	-.217315+03	187	.161224+01
	188	-.457866+01	189	-.222296-01	190	-.903198-01	191	.635488-01
	192	.192062-00	193	.634100+02	194	-.210015+03	195	.633123+02
	196	-.209691+03	197	.760527+02	198	-.108731+03		

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.60
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1JUN69	AFT FUSELAGE	Report No.	LG1US46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 (CON.)
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS (CONDITION 1004001.18 1 MOD 4

1	1	.130435+03	2	.121314+03	5	.328225+03	6	.328251+03
	9	.347165+03	10	.348967+03	13	.304764+03	14	.307503+03
	17	.391709+03	18	.395169+03	21	.244757+03	22	.245343+03
	25	.310865+03	26	.309111+03	27	.232696+02	28	.311790+02
	29	.474314+02	30	.473826+02	31	.773139+02	32	.761998+02
	33	.120078+03	34	.117920+03	35	.531607+03	36	.524256+03
	37	.432150+03	38	.420952+03	39	.416399+03	40	.408939+03
	41	.379357+03	42	.375839+03	43	.300963+03	44	.302459+03
	45	.112883+03	46	.120102+03	47	.155568+03	48	.151451+03
	49	.731393+02	50	.778550+02	51	.313829+03	52	.308005+03
	53	.623641+02	54	.673483+02	55	.290288+03	56	.286211+03
	57	.734914+02	58	.791056+02	59	.288903+03	60	.286337+03
	61	.621498+02	62	.670764+02	63	.241215+03	64	.239900+03
	65	.561805+02	66	.592525+02	67	.256264+03	68	.254745+03
	69	.536937+02	70	.552146+02	71	.263859+03	72	.261478+03
	73	.486547+02	74	.482777+02	75	.318231+03	76	.313798+03
	77	-.180956+02	78	-.885708+01	79	.308677+02	80	.328198+02
	81	.440904+02	82	.438894+02	83	.630361+02	84	.624317+02
	85	.130123+03	86	.125367+03	87	.481676+03	88	.467909+03
	89	.447830+03	90	.435224+03	91	.377562+03	92	.371349+03
	93	.274500+03	94	.278622+03	95	.107271+03	96	.116455+03
	97	-.242189+02	98	-.254351+02	99	-.153634+02	100	-.187002+02
	101	.227449+03	102	.216234+03	103	.314425+02	104	.356887+02
	105	.180932+03	106	.173125+03	107	.595143+02	108	.646273+02
	109	.190951+03	110	.185752+03	111	.675631+02	112	.725491+02
	113	.209245+03	114	.206718+03	115	.820759+02	116	.871422+02
	117	.184442+03	118	.184722+03	119	.784076+02	120	.843959+02
	121	.238972+03	122	.240713+03	123	.102576+03	124	.110478+03
	125	.246651+03	126	.249875+03	127	.974530+02	128	.107576+03
	129	.365974+03	130	.352330+03	131	.352325+02	132	.352325+02
	133	-.915515+01	134	.483193+01	135	.305414+01	136	.388158+01
	137	-.333587+01	138	-.962320+01	139	.162242+03	140	.158771+03
	141	.492120+03	142	.480406+03	143	.477167+03	144	.459361+03
	145	.409267+03	146	.396489+03	147	.274423+03	148	.273980+03
	149	.837301+02	150	.977703+02	151	.301739+01	153	.260883+03
	154	.250456+03	155	-.103730+02	156	-.935080+01	159	.147820+03
	160	.140325+03	163	.149203+03	164	.143760+03	167	.157358+03
	168	.163627+03	171	.190733+03	172	.191770+03	175	.191414+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1JUN69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. 1	PERM. 761
Checked	A L PINCHOT	1JUN69	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	L G DARBY	1JUN69	AFT FUSELAGE	Report No. LG1US46-2-2		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 *CON. 1

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS *CONDITION 1004001-18 1 MOD 4

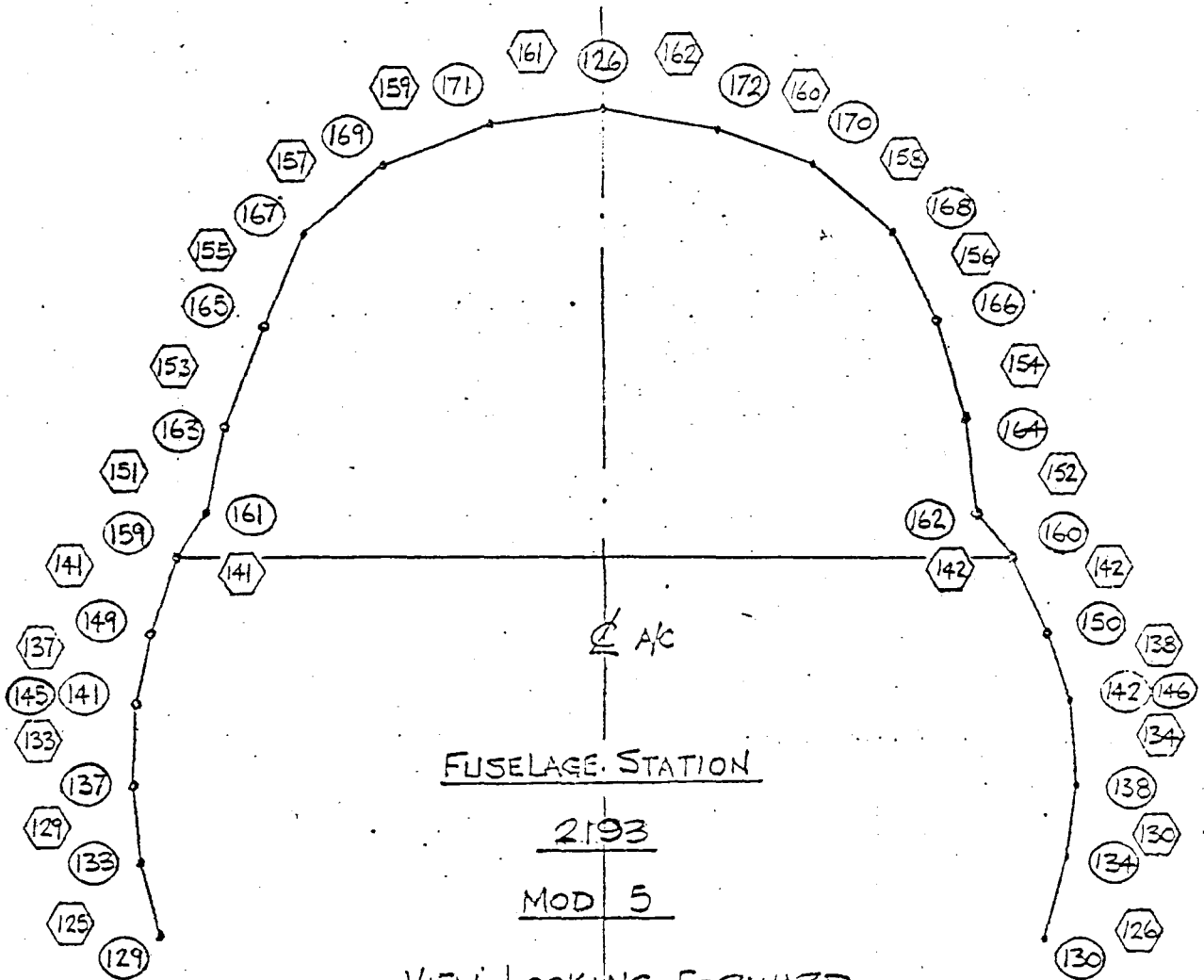
1	176	.193133+03	179	.224271+03	180	.226826+03	183	.399018+03
	184	.398198+03	185	-.175434+02	186	-.178408+02	187	-.106093+00
	188	-.124187+00	189	-.589678-02	190	-.443341-02	191	.110588-01
	192	.728959-02	193	-.242095+02	194	-.254253+02	195	-.241723+02
	196	-.253861+02	197	-.227450+03	198	-.216284+03		

LOADS ARE IN POUNDS/INCH

Prepared	NAME V. W. Boyd	DATE 16 Aug 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERIAL 7.62
Checked	J. Halliday	16 Aug 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. Darby	16 Aug 68		Report No.	LG1US 46-2-2	

REV. (A) M5 1 AUG 69

7.3.2 Panel Loads (Continued)



VIEW LOOKING FORWARD

FS. 2209 TO FS. 2177

FIGURE No 7.3.2.3

KEY

— GRID POINT

○ FAMAS STRINGER

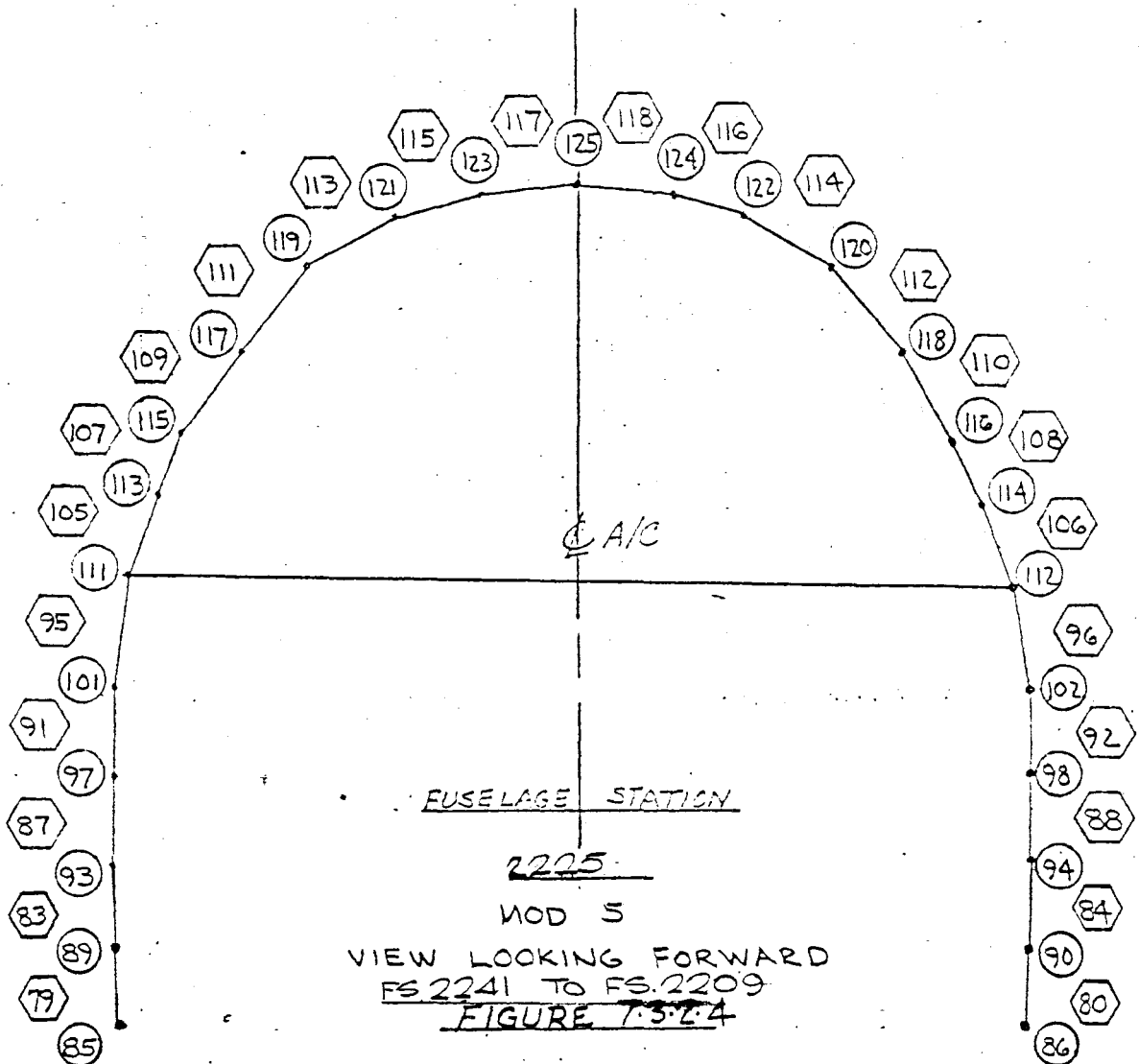
⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2

Prepared	NAME J. H. RAY	DATE 10 SEP 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. VOL. 1	PERM. 7.63
Checked	V. BOYD	10 SEP 68	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A
Approved	L. G. DARBY	10 SEP 68	AFT FUSELAGE	Report No.	LG10546-2-2

REV. (A) M. B. 1 AUG. 69

7.3.2 PANEL LOADS (CONTINUED)



KEY

-  GRID POINT
-  FAMAS STRINGER
-  FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 8. VOLUME IV SECTION 5.2

Prepared	NAME J. H. RAY	DATE 10 SEP 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.64
Checked	V. BOYD	10 SEP 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L. G. DARBY	10 SEP 68		Report No.	LGUS 46-2-2	

REV. (A) M.O. 1 AUG 69

7.3.2. PANEL LOADS (CONTINUED)

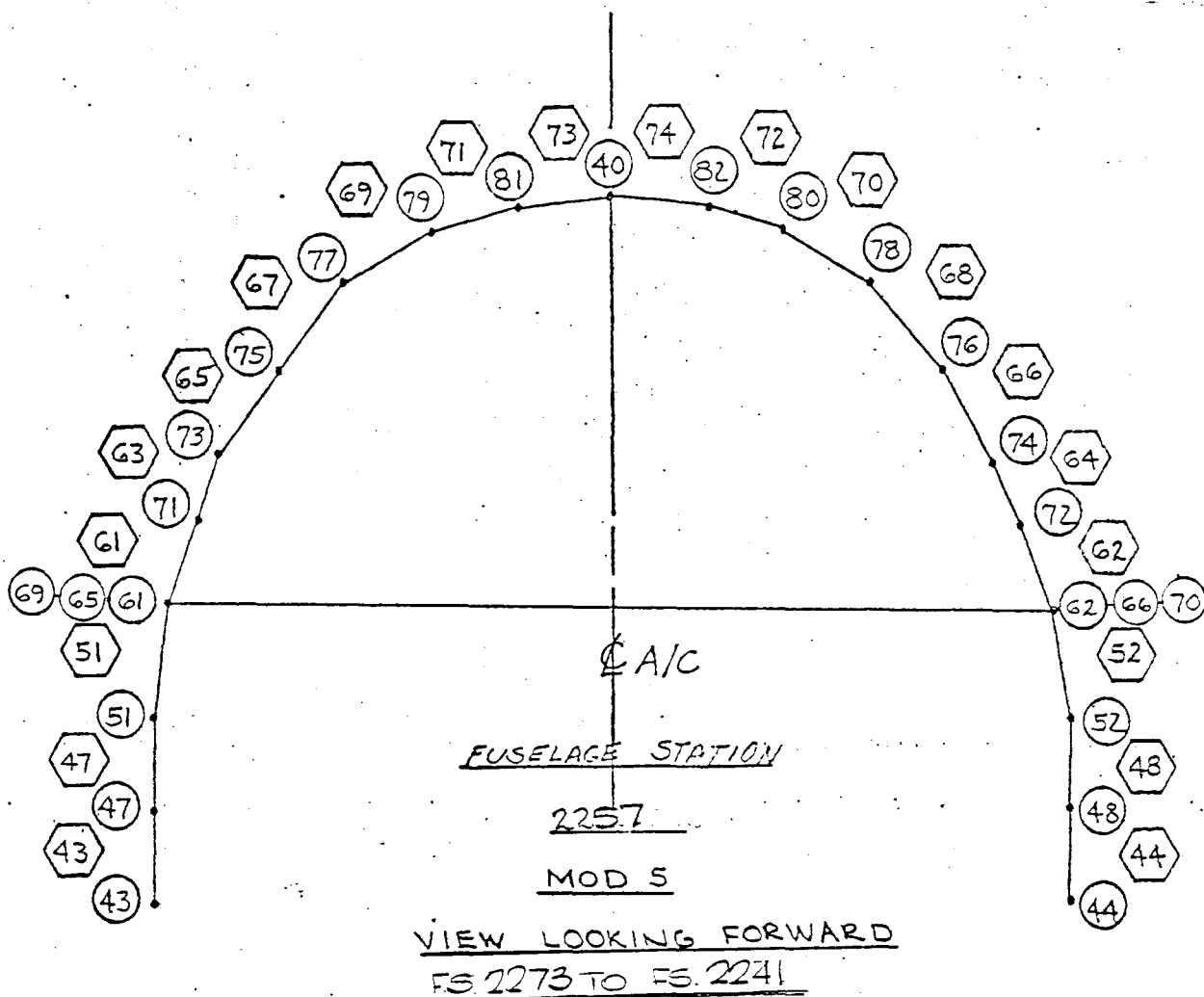


FIGURE 7.3.2.5

KEY

-  GRID POINT
-  FAMAS STRINGER
-  FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 5.2

Prepared	NAME J. H. RAY	DATE 10 SEP 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.65
Checked	V. BOYD	10 SEP 68	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L. G. DARBY	10 SEP 68	AFT FUSELAGE	Report No.	LG10546-2-2	

REV. (A) M.B. 1 AUG 69.

7.3.2 PANEL LOADS (CONTINUED)

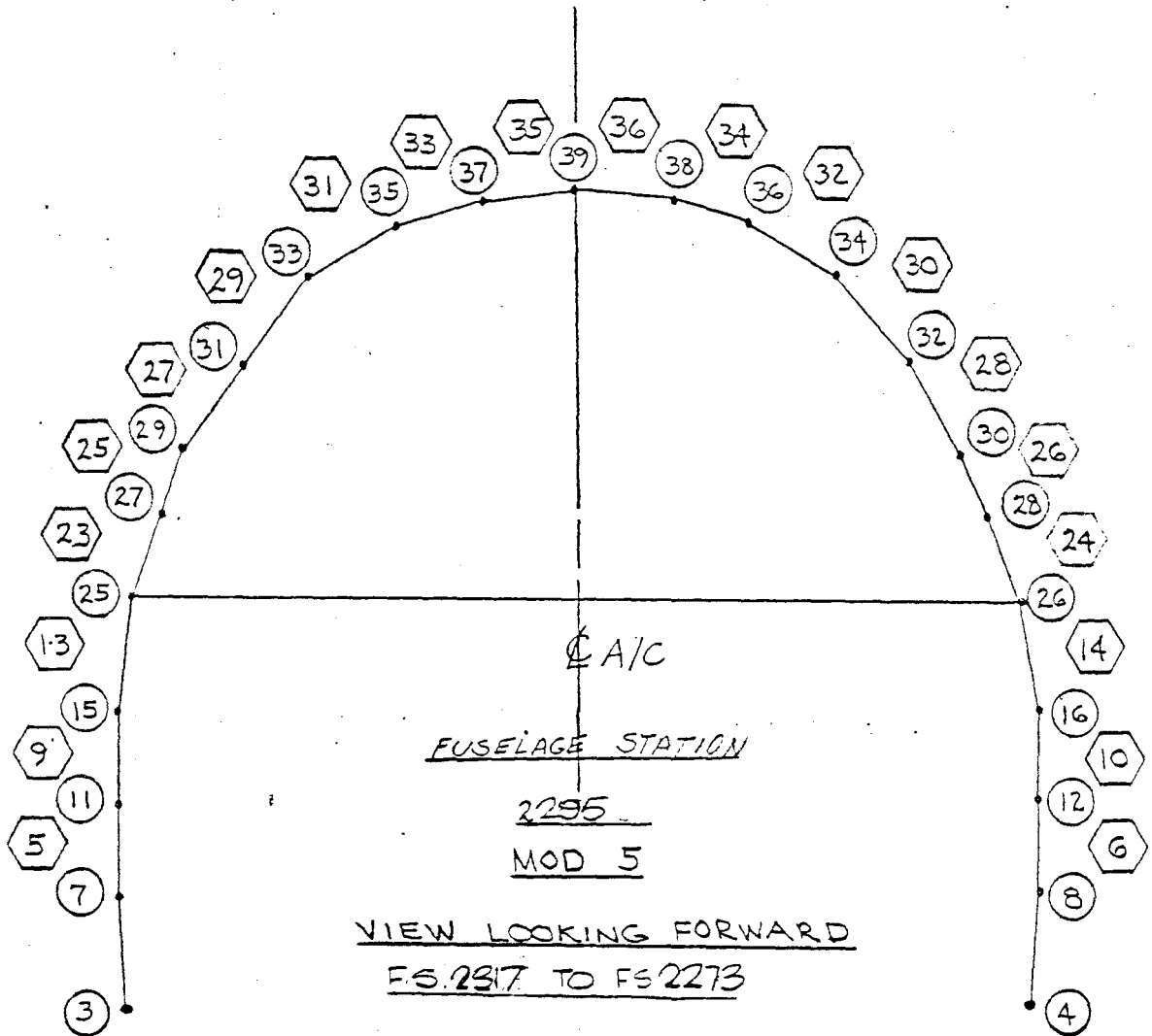


FIGURE 7.3.2.6

KEY

-  GRID POINT
-  FAMAS STRINGER
-  FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 5.2

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	Vol. I 7.66
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		LG1JS45-2-2	
					Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. &
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1PC242 & MOD 5

1	1	-.211756+03	2	-.199357+03	5	-.349603+03	6	-.325205+03
	* 9	-.493561+03	10	-.460802+03	13	-.637148+03	14	-.595752+03
	15	.177912+02	16	.191511+03	17	.845407+02	18	.253898+03
	19	.145058+03	20	.302889+03	21	.234732+03	22	.359945+03
	23	-.270928+03	24	-.851843+02	25	-.170109+03	26	.332365+02
	27	-.805936+02	28	.137207+03	29	-.950554+01	30	.191847+03
	31	.190168+02	32	.198834+03	33	.492585+02	34	.199679+03
	35	-.168772+02	36	.114399+03	37	.888090+02	38	.772327+02
	39	-.352213+03	40	-.183460+03	43	-.356761+03	44	-.252267+03
	47	-.471946+03	48	-.378416+03	51	-.594530+03	52	-.478656+03
	53	-.382816+02	54	.107985+03	55	-.965631+01	56	.126575+03
	57	.329812+02	58	.155407+03	59	.117368+03	60	.243287+03
	61	-.391108+03	62	-.222755+03	63	-.259714+03	64	-.704512+02
	65	-.953718+02	66	.291085+02	67	.345832+02	68	.205998+03
	69	.123876+03	70	.290771+03	71	.103812+03	72	.261026+03
	73	-.279872+02	74	.111401+03	75	-.171490+03	76	-.222575+02
	79	-.295939+03	80	-.172790+03	83	-.332957+03	84	-.217940+03
	87	-.222610+03	88	-.134991+03	91	-.587577+03	92	-.469643+03
	95	-.622835+03	96	-.541866+03	97	-.336808+02	98	.110870+03
	99	-.561695+01	100	.129182+03	101	.354997+02	102	.165559+03
	103	.104195+03	104	.202312+03	105	-.227227+03	106	-.144582+02
	107	-.992895+02	108	.241520+02	109	-.883188+02	110	.569136+02
	111	.573913+02	112	.194395+03	113	.875563+02	114	.257514+03
	115	.812751+02	116	.241972+03	117	-.205559+02	118	.122062+03
	119	.781275+02	120	.797694+02	121	-.331799+03	122	-.172213+03
	125	-.348391+03	126	-.188638+03	129	-.319394+03	130	-.156030+03
	133	-.298738+03	134	-.143485+03	137	-.462777+03	138	-.348088+03
	141	-.417770+03	142	-.349453+03	143	-.256847+02	144	.115014+03
	145	.744004+01	146	.135878+03	147	.532759+02	148	.167984+03
	149	.113153+03	150	.203638+03	151	-.130327+03	152	.313505+02
	153	.205872+02	154	.121467+03	155	.813439+02	156	.219737+03
	157	.119069+03	158	.259502+03	159	.645381+02	160	.212775+03
	161	-.400945+02	162	.114490+03				

* PANEL (9) REF. P. 7.65 ULTIMATE PANEL SHEAR FLOW = -494. LBS./IN.

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
					Page	Vol. I 7.67
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L G DARBY	1 JUN 69			Report No. LG1J546-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
7.3 SKINS AND STRINGERS
7.3.2 SKIN LOADS CONDITION SPE201 & MOD 5

1	1	.314693+03	2	.309908+03	5	.512477+03	6	.500360+03
	9	.706880+03	10	.689839+03	13	.893523+03	14	.872846+03
	15	-.888791+02	16	-.243825+03	17	-.193465+03	18	-.345695+03
	19	-.286887+03	20	-.430155+03	21	-.432205+03	22	-.555161+03
	23	.238254+03	24	.805913+02	25	.530427+02	26	-.123104+03
	27	-.942259+02	28	-.285058+03	29	-.199727+03	30	-.377933+03
	31	-.205611+03	32	-.355914+03	33	-.190519+03	34	-.326373+03
	35	-.361512+02	36	-.155876+03	37	-.787167+02	38	-.677763+02
	39	.545189+03	40	.397040+03	43	.500082+03	44	.419071+03
	47	.650284+03	48	.573804+03	51	.810720+03	52	.715914+03
	53	-.213024+01	54	-.131616+03	55	-.500169+02	56	-.172107+03
	57	-.113495+03	58	-.225633+03	59	-.256955+03	60	-.374626+03
	61	.415826+03	62	.274681+03	63	.208513+03	64	.431261+02
	65	-.696425+02	66	-.175510+03	67	-.232270+03	68	-.383894+03
	69	-.329430+03	70	-.480673+03	71	-.265016+03	72	-.408177+03
	73	-.198326+02	74	-.145872+03	75	.255945+03	76	.116626+03
	79	.426983+03	80	.322758+03	83	.473283+03	84	.378850+03
	87	.315521+03	88	.245684+03	91	.800595+03	92	.711420+03
	95	.832448+03	96	.777526+03	97	-.954156+01	98	-.137035+03
	99	-.591031+02	100	-.179328+03	101	-.133500+03	102	-.248613+03
	103	-.219765+03	104	-.310398+03	105	.146387+03	106	-.314404+02
	107	-.736105+01	108	-.110054+03	109	-.429516+02	110	-.164813+03
	111	-.265014+03	112	-.335837+03	113	-.287070+03	114	-.441380+03
	115	-.230198+03	116	-.376636+03	117	-.305829+02	118	-.159823+03
	119	-.683088+02	120	-.693547+02	121	.481446+03	122	.358020+03
	125	.493571+03	126	.360293+03	129	.450386+03	130	.324913+03
	133	.414645+03	134	.298239+03	137	.609821+03	138	.526679+03
	141	.523876+03	142	.481952+03	143	-.206675+02	144	-.145050+03
	145	-.777737+02	146	-.191484+03	147	-.146100+03	148	-.243056+03
	149	-.233403+03	150	-.314855+03	151	.395327+02	152	-.943276+02
	153	-.184941+03	154	-.265369+03	155	-.264157+03	156	-.338818+03
	157	-.304398+03	158	-.433848+03	159	-.209053+03	160	-.343821+03
	161	-.154116+02	162	-.154884+03				

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. VOL. I	PERM. 7.63
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG10546-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS CONDITION DML001 & MOD 5

1	1	-.589845+02	2	.220569+03	5	-.157589+03	6	.415352+03
	9	-.275080+03	10	.619332+03	13	-.392603+03	14	.319724+03
	15	-.160631+03	16	.770800+02	17	-.176222+03	18	.405963+02
	19	-.187207+03	20	.555117+01	21	-.173723+03	22	-.844472+02
	23	-.591512+03	24	.643123+03	25	-.664278+03	26	.613806+03
	27	-.753157+03	28	.630028+03	29	-.804737+03	30	.633781+03
	31	-.812342+03	32	.654107+03	33	-.759501+03	34	.625784+03
	35	-.658618+03	36	.610482+03	37	.122091+03	38	.712384+01
	39	-.187945+03	40	.463710+03	43	-.199253+03	44	.444944+03
	47	-.285594+03	48	.601149+03	51	-.412619+03	52	.803725+03
	53	-.420986+02	54	.111930+01	55	-.701977+02	56	.295261+01
	57	-.824796+02	58	-.158936+02	59	-.108226+03	60	-.695429+02
	61	-.592966+03	62	.752637+03	63	-.631173+03	64	.639075+03
	65	-.680872+03	66	.584294+03	67	-.725571+03	68	.559971+03
	69	-.733520+03	70	.523947+03	71	-.749301+03	72	.576190+03
	73	-.710094+03	74	.667399+03	75	-.141015+03	76	.258712+03
	79	-.177672+03	80	.398754+03	83	-.192545+03	84	.438714+03
	87	-.111454+03	88	.285464+03	91	-.465465+03	92	.861778+03
	95	-.647999+03	96	.104937+04	97	-.320950+02	98	-.120865+02
	99	-.669658+02	100	-.530826+01	101	-.827151+02	102	-.285366+02
	103	-.652016+02	104	-.800166+02	105	-.681061+03	106	.705003+03
	107	-.451867+03	108	.411624+03	109	-.628060+03	110	.557800+03
	111	-.659679+03	112	.476419+03	113	-.764477+03	114	.569831+03
	115	-.764996+03	116	.608713+03	117	-.732210+03	118	.637273+03
	119	-.105722+02	120	-.335243+02	121	-.238327+03	122	.453186+03
	125	-.235623+03	126	.485848+03	129	-.210280+03	130	.449179+03
	133	-.192148+03	134	.420259+03	137	-.446251+03	138	.736551+03
	141	-.560693+03	142	.819715+03	143	-.189157+02	144	-.293907+02
	145	-.613904+02	146	-.180970+02	147	-.685483+02	148	-.404563+02
	149	-.269347+02	150	-.120626+03	151	-.573171+03	152	.563630+03
	153	-.481228+03	154	.359295+03	155	-.627304+03	156	.453256+03
	157	-.698218+03	158	.502041+03	159	-.764341+03	160	.617259+03
	161	-.774858+03	162	.728497+03				

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 7.69
Prepared		TITLE			
Checked	A L PINCHOT	1 JUN 69	INTERNAL LOADS ANALYSIS		Model C-5A
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		Report No. LG1JS46-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1004001.18 & MOD 5

1	1	.927727+02	2	.974136+02	5	.157989+03	6	.152650+03
	9	.123858+03	10	.126952+03	13	.144545+03	14	.147854+03
	15	.595510+02	16	.585575+02	17	.960026+02	18	.944435+02
	19	.121865+03	20	.119572+03	21	.151915+03	22	.149855+03
	23	.375931+03	24	.374780+03	25	.434846+03	26	.431624+03
	27	.506428+03	28	.502025+03	29	.519522+03	30	.514550+03
	31	.494547+03	32	.490972+03	33	.381112+03	34	.380908+03
	35	.137672+03	36	.141355+03	37	.177680-00	38	.111438+01
	39	.269105+03	40	.267022+03	43	.208594+03	44	.209764+03
	47	.204242+03	48	.205515+03	51	.230248+03	52	.230175+03
	53	.193719+02	54	.203104+02	55	.307357+02	56	.303551+02
	57	.456831+02	58	.440876+02	59	.798159+02	60	.770638+02
	61	.328902+03	62	.326743+03	63	.434889+03	64	.430536+03
	65	.381157+03	66	.377764+03	67	.458770+03	68	.492527+03
	69	.454366+03	70	.450334+03	71	.347826+03	72	.348117+03
	73	.113024+03	74	.117568+03	75	.203577+03	76	.199150+03
	79	.271774+03	80	.271845+03	83	.274002+03	84	.275306+03
	87	.254201+03	88	.256481+03	91	.271470+03	92	.271227+03
	95	.195694+03	96	.192077+03	97	.218081+02	98	.231454+02
	99	.342749+02	100	.340431+02	101	.546660+02	102	.534293+02
	103	.805589+02	104	.788625+02	105	.461893+03	106	.455262+03
	107	.338618+03	108	.337218+03	109	.451289+03	110	.445689+03
	111	.426624+03	112	.421752+03	113	.445945+03	114	.441301+03
	115	.341148+03	116	.341191+03	117	.112465+03	118	.117763+03
	119	.152539+01	120	.217464+01	121	.114008+03	122	.108801+03
	125	.275604+03	126	.274166+03	129	.321230+03	130	.321716+03
	133	.373830+03	134	.375796+03	137	.299468+03	138	.299774+03
	141	.261576+03	142	.259817+03	143	.259411+02	144	.278850+02
	145	.434383+02	146	.434284+02	147	.712546+02	148	.702555+02
	149	.830907+02	150	.820504+02	151	.450269+03	152	.444974+03
	153	.399141+03	154	.395208+03	155	.433212+03	156	.424474+03
	157	.392321+03	158	.387979+03	159	.303835+03	160	.305414+03
	161	.117878+03	162	.124220+03				

LOADS ARE IN POUNDS/INCH

REV. (A) M.S. 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)

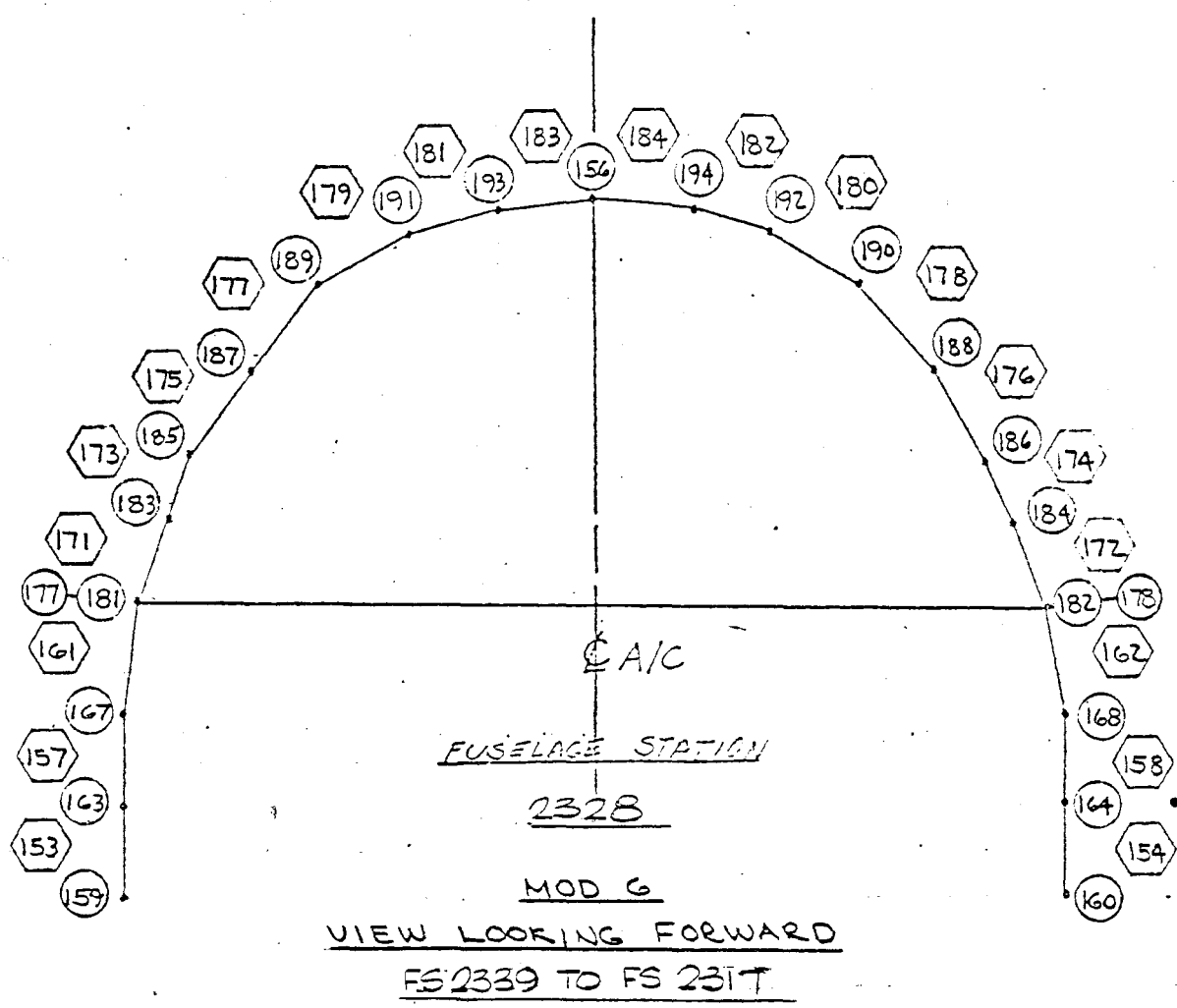


FIGURE 7.3.2.7

KEY	
	GRID POINT
	FAMAS STRINGER
	FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF. 8. VOLUME IV SECTION 5.2

Prepared	NAME L.H. HYDE	DATE 9 Sept 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I 7.71
Checked	J. HALLIDAY	9 Sept 68	TITLE AFT FUSELAGE	Model	C-5A
Approved	L.G. DARBY	9 Sept 68	INTERNAL LOADS ANALYSIS	Report No.	LGIUS-46-2-2

REV. (A) MB. 1 AUG. 69

7.3-2 PANEL LOADS (CONTINUED)

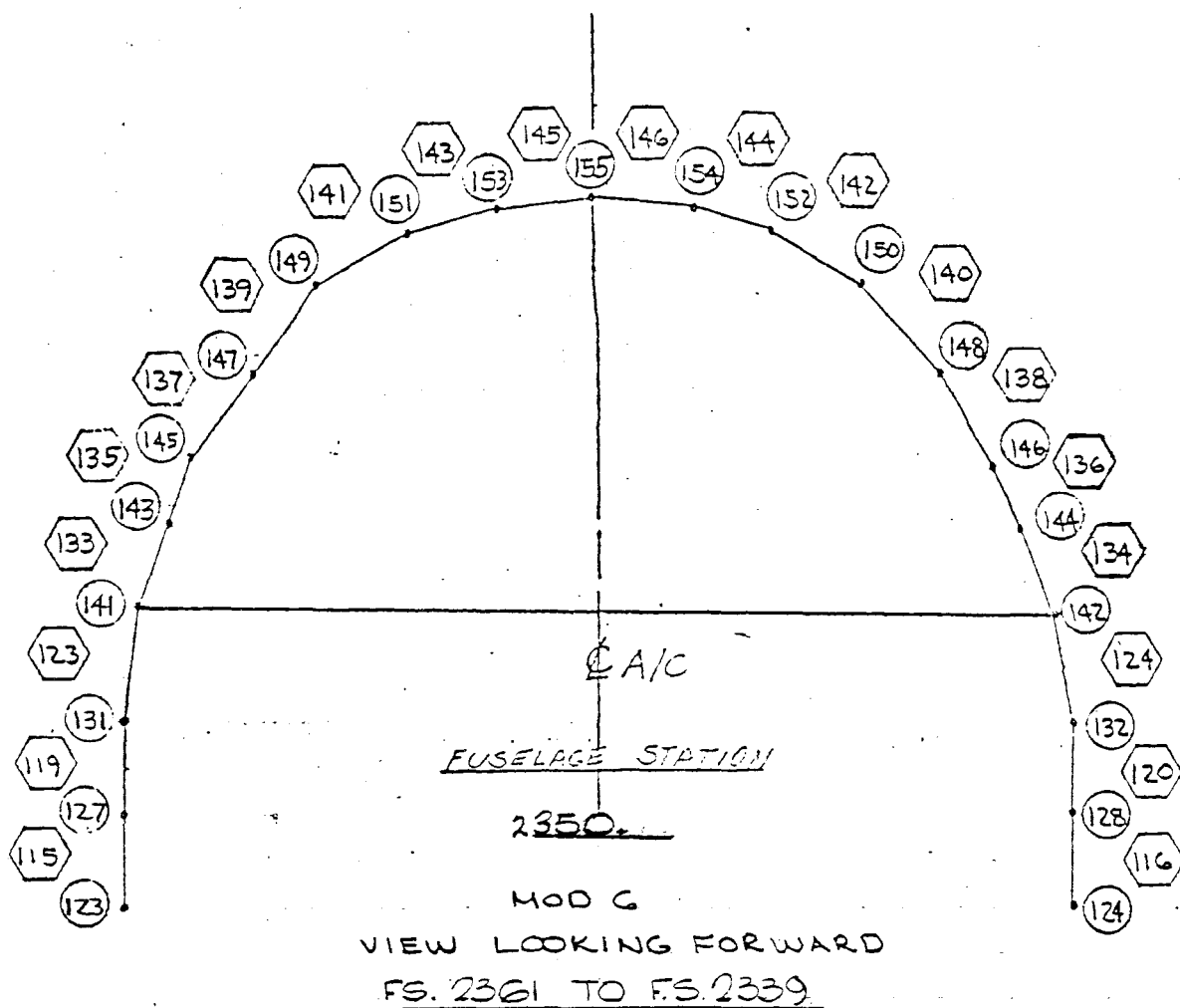


FIGURE 7.3.1.8

KEY



GRID POINT



FAMAS STRINGER



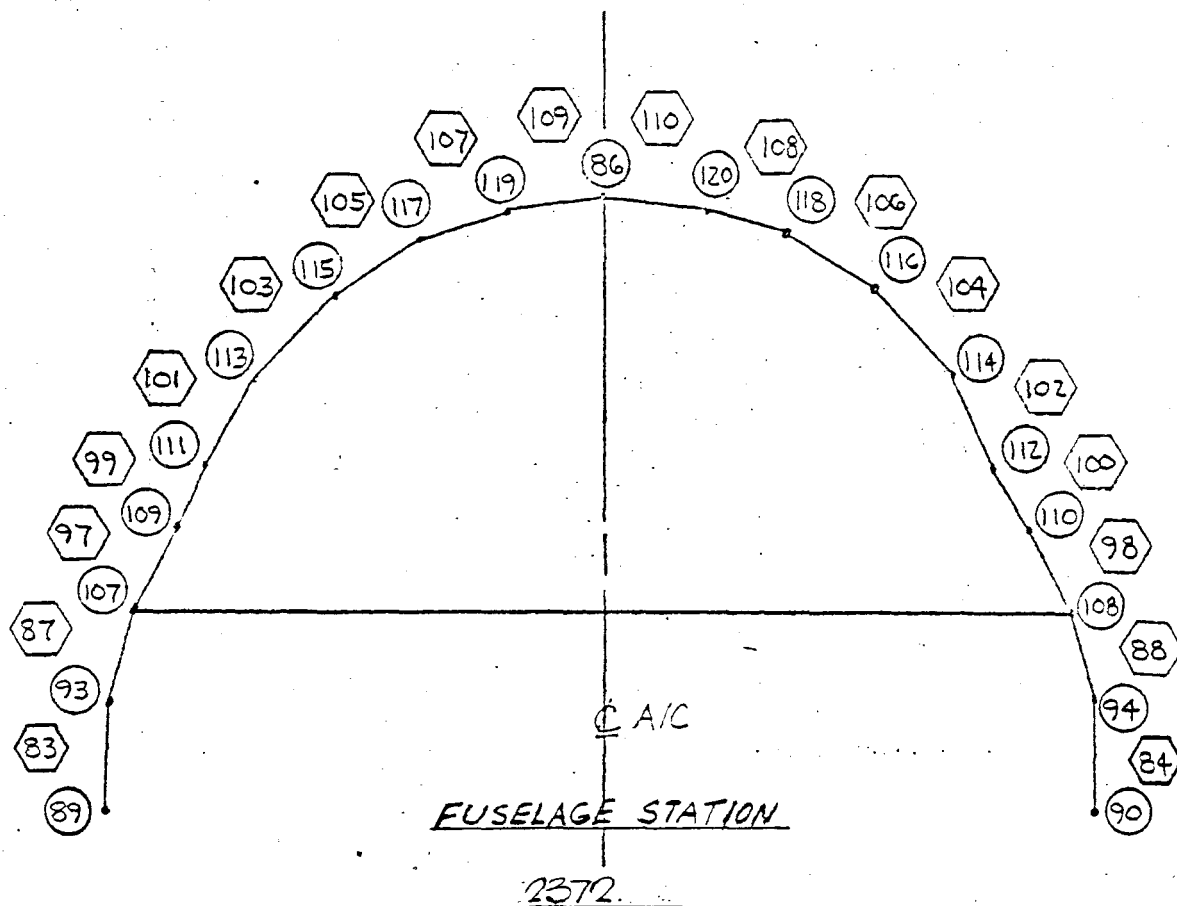
FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 3. VOLUME IV SECTION 5.2

Prepared	NAME L. H. HYDE	DATE 9 Sept 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	J. HALLIDAY	9 Sept 68	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	Page	Vol. I 7.12
Approved	L. G. DARBY	9 Sept 68		Model	C-5A
				Report No.	LGIUS-46-2-2

REV (A) M.B. 1 AUG. 69

7.3.2 PANEL LOADS (CONTINUED)



KEY



GRID POINT



FAMAS STRINGER

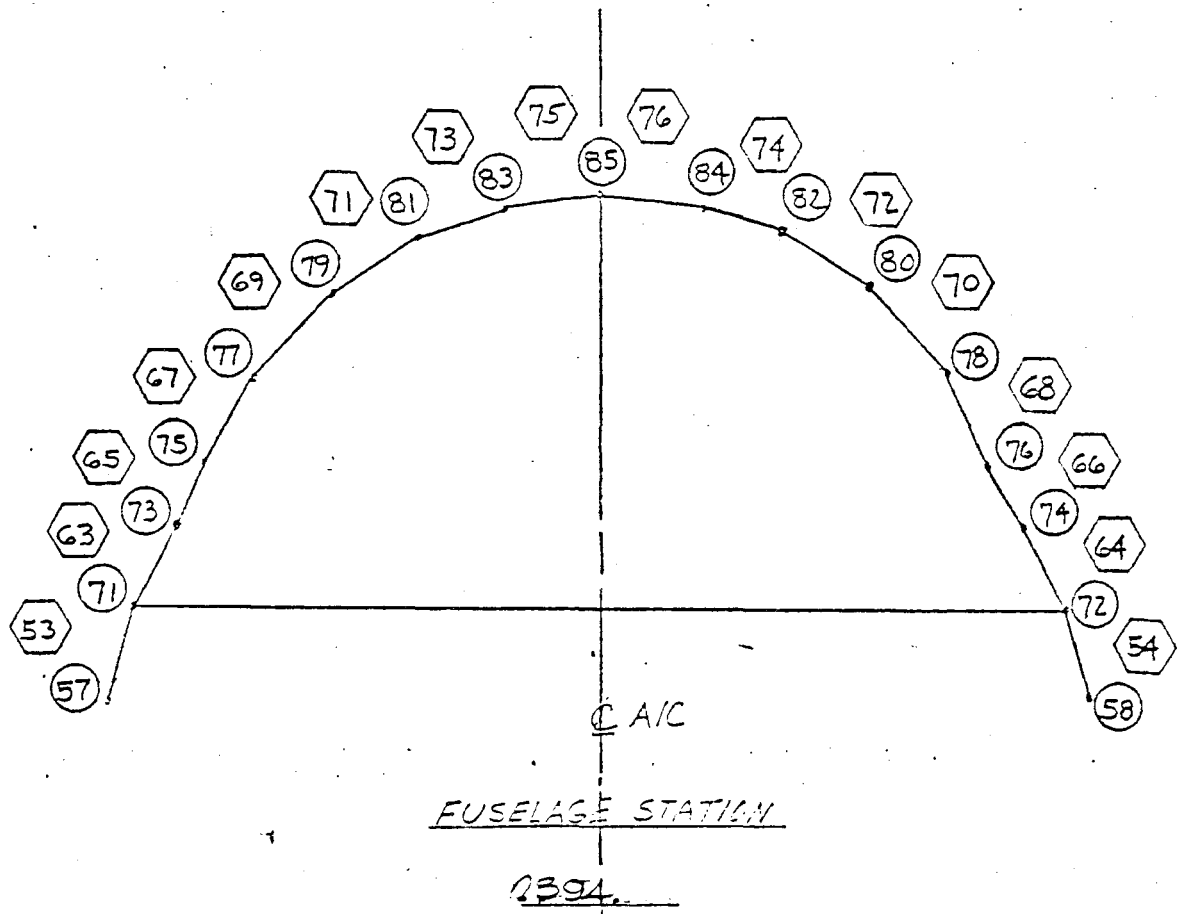


FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 8. VOLUME IV SECTION 5.2

REV. (A) MB 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)



MOD 6
VIEW LOOKING FORWARD
FS.2405 TO FS.2383.
FIGURE 7.3.2.10

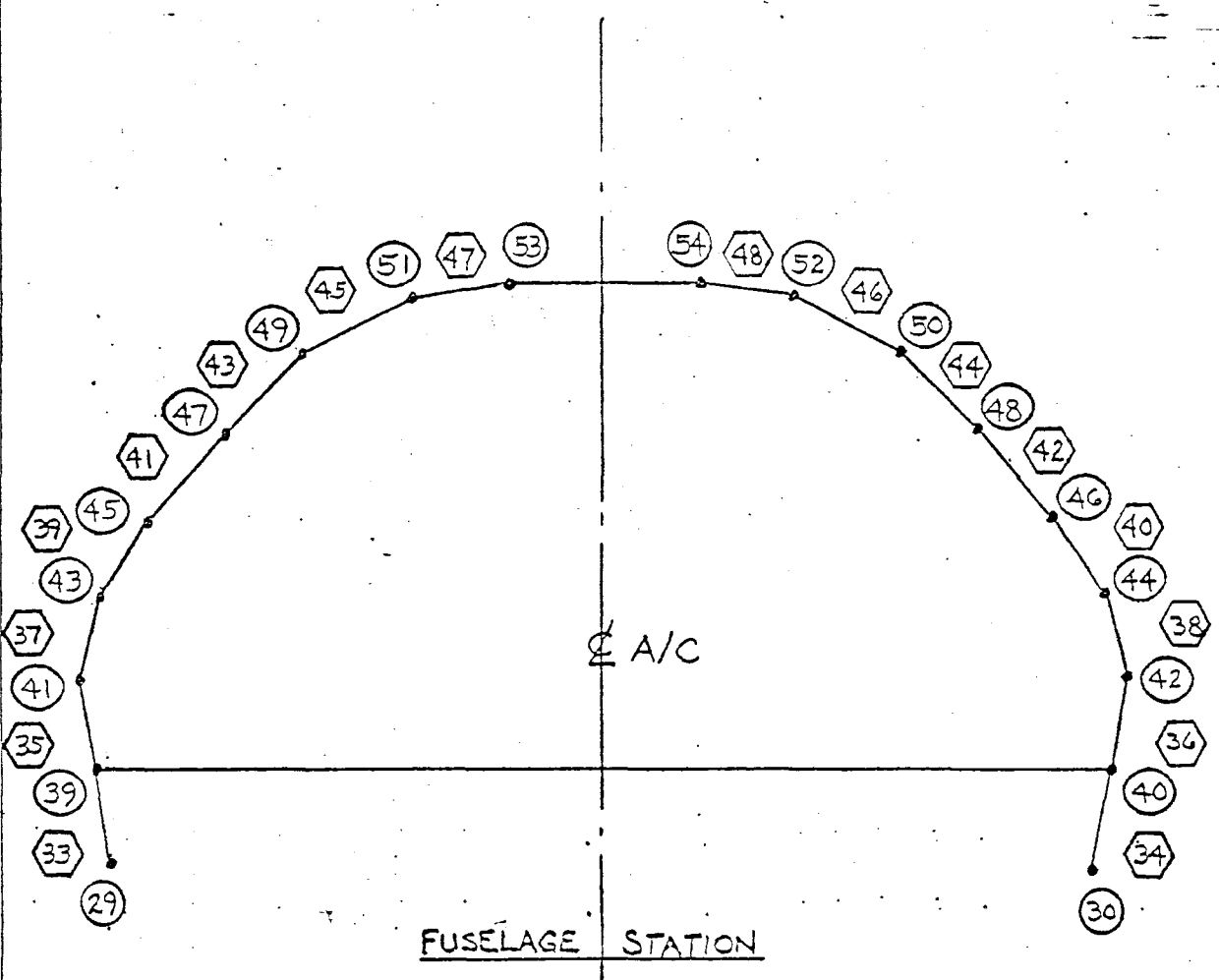
KEY

GRID POINT
 FAMAS STRINGER
 FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 8, VOLUME IV SECTION 5.2

REV. (A) M.B. 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)



2416

VIEW LOOKING FORWARD FS. 2427. TO FS. 2405
MOD 6

FIGURE No 7.3.2.11

KEY

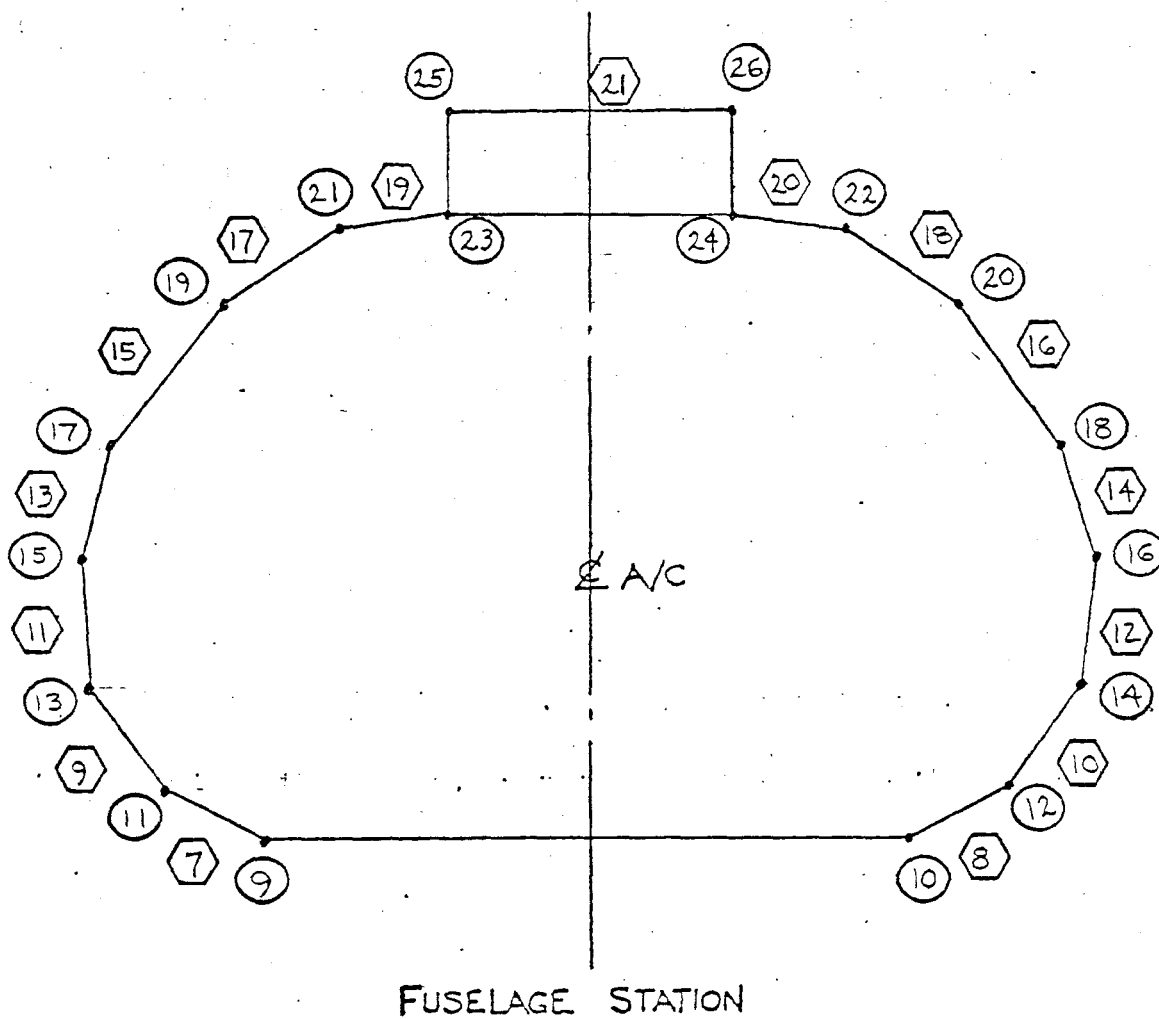
GRID POINT
 FAMAS PANEL
 FAMAS STRINGER

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 5.2

Prepared	NAME V. W. Boyd	DATE 9 Sep 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.75
Checked	J. Halliday	9 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. Darby	9 Sep 68		Report No.	IGIUS 46-2-2	

REV (A) M.B. 1 AUG. 69

7.3.2 PANEL LOADS (CONTINUED)



2439

VIEW LOOKING FORWARD FS 2450 TO FS 2427

MOD 6

FIGURE 7.3.2.12

KEY

-  GRID POINT
-  FAMAS STRINGER
-  FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 8, VOLUME IV SECTION 5.2

Prepared	RF NAME WILKINSON	DATE 1 JUNE 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	A L PINCHOT	1 JUNE 69	TITLE INTERNAL LOADS ANALYSIS		Page	VOL. I 1.76
Approved	L G DARBY	1 JUNE 69	AFT FUSELAGE		Model	C-54
					Report No.	LG11746-2-2

REV (1) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1PC242 8 MOD 6

1	1	-.528609+02	2	.126061+03	3	-.129183+02	4	.157723+03
	5	-.235124+03	6	-.104217+03	7	-.118572+04	8	-.986635+03
*	9	-.919611+03	10	-.684146+03	11	-.746370+03	12	-.510143+03
	13	-.463489+03	14	-.235842+03	15	-.139331+03	16	.720129+02
	17	.190231+03	18	.373063+03	19	.664411+03	20	.794567+03
	21	-.408650+01	22	.647872+01	23	.759904+02	24	.744847+02
	25	-.349249+02	26	.113567+03	27	.156751+02	28	.174283+03
	29	.565494+03	30	.656037+03	31	-.164914+03	32	.157598+02
	33	-.181320+04	34	-.180174+04	35	-.792910+03	36	-.610230+03
	37	-.582821+03	38	-.363101+03	39	-.494039+03	40	-.252633+03
	41	-.328944+03	42	-.844262+02	43	-.132954+03	44	.815353+02
	45	-.254410+02	46	.140304+03	47	.249679+03	48	.351049+03
	49	-.852332+03	50	-.591850+03	51	-.187033+03	52	-.282149+03
	53	-.107063+04	54	-.106215+04	55	-.238514+02	56	.171450+03
	57	.194249+02	58	.179081+03	59	.395389+02	60	.205875+03
	61	-.779938+02	62	.474001+02	63	-.633925+03	64	-.439238+03
	65	-.443356+03	66	-.216550+03	67	-.228920+03	68	-.303139+02
	69	-.247826+03	70	-.329318+01	71	-.942303+02	72	.114633+03
	73	.113457+03	74	.274175+03	75	.286722+03	76	.391015+03
	77	.759282+02	78	.735221+02	79	-.398940+03	80	-.410401+03
	81	-.162023+02	82	-.104368+03	83	-.569465+03	84	-.555378+03
	85	-.542080+03	86	-.355742+03	87	-.114662+04	88	-.113943+04
	89	-.912946+01	90	.150552+03	91	.459415+02	92	.209303+02
	93	.995397+02	94	.271305+03	95	.124688+03	96	.250853+03
	97	-.274072+03	98	-.540346+02	99	-.247592+03	100	-.153509+01
	101	-.167136+03	102	.927365+02	103	-.178130+03	104	.582357+02
	105	-.755848+02	106	.132346+03	107	.749753+02	108	.203371+03
	109	.135710+03	110	.247146+03	111	-.265089+03	112	-.314207+03
	113	-.434314+03	114	-.547829+03	115	-.612765+03	116	-.607947+03
	119	-.337489+02	120	-.305923+03	121	-.332138+03	122	-.411079+03
	123	-.650849+03	124	-.605090+03	125	.137430+02	126	.184325+03
	127	.857729+02	128	.259133+03	129	.151949+03	130	.331780+03
	131	.141217+03	132	.239464+03	133	-.239563+03	134	.182715+01
	135	-.152396+03	136	.882183+02	137	-.119891+03	138	.139940+03
	139	-.772113+02	140	.150892+03	141	-.184483+02	142	.162311+03
	143	.505039+02	144	.199956+03	145	.580982+02	146	.177011+03
	147	.847395+02	148	.872273+02	149	-.562069+03	150	-.617376+03
	151	-.523143+03	152	-.633610+03	153	-.456173+03	154	-.421631+03

* PANEL (9) REF. P. 7.75 ULTIMATE PANEL SHEAR FLOW = -920. LBS./IN.

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.77
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG10S46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1PC242 & MOD 6

1	155	-.212204+03	156	-.292087+03	157	-.283332+03	158	-.229207+03
	159	-.943064+02	160	-.213853+03	161	-.418710+03	162	-.355061+03
	163	.183118+02	164	.192054+03	165	.948324+02	166	.268662+03
	167	.177744+03	168	.356058+03	169	.179309+03	170	.291581+03
	171	-.279812+03	172	-.557381+02	173	-.128192+03	174	.110225+03
	175	-.961051+02	176	.158769+03	177	-.464541+02	178	.170837+03
	179	-.859501+01	180	.174272+03	181	.453561+02	182	.193457+03
	183	.801569+01	184	.130929+03	185	.427513+02	186	.228887+03
	187	.299224+03	188	.248066+03	189	.674693+02	190	.230261+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Page	Vol. I 7.79
Approved	L G DARBY	1 JUN 69			Model	C-FA

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.&

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS CONDITION SPE201 & MOD 6

1	155	.322066+03	156	.401286+03	157	.384498+03	158	.341871+03
	159	.155952+03	160	.259622+03	161	.580364+03	162	.533842+03
	163	-.899921+02	164	-.244497+03	165	-.210115+03	166	-.366000+03
	167	-.341066+03	168	-.503162+03	169	-.338132+03	170	-.440336+03
	171	.232771+03	172	.401296+02	173	-.140179+02	174	-.222332+03
	175	-.854084+02	176	-.310380+03	177	-.143815+03	178	-.335799+03
	179	-.171853+03	180	-.333281+03	181	-.204035+03	182	-.335685+03
	183	-.828276+02	184	-.195954+03	185	-.150122+03	186	-.312609+03
	187	-.417316+03	188	-.378905+03	189	-.171518+03	190	-.314721+03

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	No. I 7.30
Checked	A L PINCHOT	1 JUN 69	TITLE	Model	C-5A
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	Report No.	LG1JS46-2-2
			AFT FUSELAGE		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION DMLC01 & MCD 6

1	1	-.287598+03	2	.261542+03	3	-.326587+03	4	.263024+03
	5	-.422054+03	6	.520358+03	7	-.110359+04	8	.183011+04
	9	-.112024+04	10	.162489+04	11	-.109007+04	12	.145279+04
	13	-.111764+04	14	.127609+04	15	-.104585+04	16	.966317+03
	17	-.956617+03	18	.651171+03	19	-.884041+03	20	.261652+03
	21	-.308354+02	22	.440375+02	23	.253665+03	24	.239704+03
	25	-.264800+03	26	.228735+03	27	-.278577+03	28	.200916+03
	29	.498770+02	30	-.494022+03	31	-.515543+03	32	.522531+03
	33	-.845963+03	34	.211481+04	35	-.953040+03	36	.137729+04
	37	-.970382+03	38	.122528+04	39	-.101670+04	40	.119488+04
	41	-.110345+04	42	.115292+04	43	-.100279+04	44	.923688+03
	45	-.816137+03	46	.690659+03	47	-.661088+03	48	.382386+03
	49	-.403499+03	50	.828693+03	51	.391206+02	52	.127512+03
	53	-.553597+03	54	.127848+04	55	-.249238+03	56	.204151+03
	57	-.270078+03	58	.188125+03	59	-.302424+03	60	.197159+03
	61	-.385566+03	62	.354397+03	63	-.877834+03	64	.117502+04
	65	-.913091+03	66	.105676+04	67	-.928945+03	68	.980039+03
	69	-.104081+04	70	.103014+04	71	-.981764+03	72	.878594+03
	73	-.857493+03	74	.634705+03	75	-.652849+03	76	.359673+03
	77	.222371+03	78	.205105+03	79	-.228482+03	80	.473803+03
	81	.990984+02	82	-.660563+02	83	-.327565+03	84	.703889+03
	85	-.138970+03	86	.534122+03	87	-.585135+03	88	.135590+04
	89	-.233658+03	90	.175080+03	91	-.251049+03	92	.148183+03
	93	-.269189+03	94	.119969+03	95	-.273061+03	96	.963093+02
	97	-.718838+03	98	.752388+03	99	-.829616+03	100	.823358+03
	101	-.865786+03	102	.792154+03	103	-.585946+03	104	.925108+03
	105	-.936233+03	106	.827303+03	107	-.816059+03	108	.635527+03
	109	-.626988+03	110	.454195+03	111	-.337388+02	112	.250138+03
	113	-.734931+02	114	.388516+03	115	-.300638+03	116	.715265+03
	119	-.214830+03	120	.427698+03	121	-.541876+02	122	.293960+03
	123	-.410332+03	124	.826473+03	125	-.196245+03	126	.117353+03
	127	-.213870+03	128	.772143+02	129	-.231422+03	130	.398598+02
	131	-.168142+03	132	-.309339+01	133	-.733926+03	134	.723868+03
	135	-.784045+03	136	.706842+03	137	-.348405+03	138	.732605+03
	139	-.917247+03	140	.785733+03	141	-.872512+03	142	.732863+03
	143	-.778549+03	144	.629598+03	145	-.637802+03	146	.523691+03
	147	.161324+03	148	.141984+03	149	-.145094+03	150	.573866+03
	151	-.125838+03	152	.499597+03	153	-.305089+03	154	.630758+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Vol. I	PERM. 7.31
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-54
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LGJS46-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. &

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS CONDITION DML001 & MOD 6

1	155	-.116345+02	156	.168396+03	157	-.219928+03	158	.332897+03
	159	.919811+02	160	-.717175+01	161	-.283984+03	162	.545354+03
	163	-.178378+03	164	.946571+02	165	-.194425+03	166	.505489+02
	167	-.206984+03	168	-.369633+01	169	-.163839+03	170	-.412817+02
	171	-.712164+03	172	.738201+03	173	-.713662+03	174	.612594+03
	175	-.828620+03	176	.691619+03	177	-.363600+03	178	.717772+03
	179	-.847236+03	180	.705921+03	181	-.771625+03	182	.631767+03
	183	-.635159+03	184	.565355+03	185	-.362934+03	186	.247630+03
	187	.255566+03	188	-.438226+03	189	-.279058+03	190	.153877+03

LOADS ARE IN POUNDS/INCH

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Prepared					Page	VOL. I 7.32
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L G DAREY	1 JUN 69	AFT FUSELAGE		LG11345-2-2 Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1004001.18 & MOD 6

1	1	.392601+02	2	.395314+02	3	.903312+02	4	-.902950+02
	5	.210526+02	6	.207675+02	7	-.198232+03	8	-.197087+03
	9	.384369+02	10	.339682+02	11	.180843+03	12	.130507+03
	13	.381049+03	14	.378921+03	15	.642153+03	16	.641032+03
	17	.388564+03	18	.888644+03	19	.115355+04	20	.115452+04
	21	-.145099-00	22	-.349193-00	23	.242985-00	24	.233313-00
	25	.436607+02	26	.439395+02	27	.905504+02	28	.905212+02
	29	.368043+03	30	.366945+03	31	.694984+02	32	.687184+02
	33	-.687728+03	34	-.683312+03	35	.619770+02	36	.634932+02
	37	.238249+03	38	.238204+03	39	.349497+03	40	.348175+03
	41	.471455+03	42	.469473+03	43	.609130+03	44	.607902+03
	45	.612259+03	46	.612545+03	47	.609467+03	48	.610472+03
	49	-.135904+03	50	-.133774+03	51	-.177736+03	52	-.177381+03
	53	-.243743+03	54	-.238407+03	55	.477841+02	56	.479375+02
	57	.900973+02	58	.899248+02	59	.124708+03	60	.124037+03
	61	.623326+02	62	.617523+02	63	.168285+03	64	.169560+03
	65	.316725+03	66	.315485+03	67	.422943+03	68	.420320+03
	69	.501398+03	70	.498706+03	71	.616467+03	72	.615298+03
	73	.696251+03	74	.696682+03	75	.569374+03	76	.570885+03
	77	.149417-00	78	.731068-01	79	-.288955+02	80	-.221187+02
	81	-.152824+03	82	-.152543+03	83	.267591+02	84	.335535+02
	85	-.362554+03	86	-.360107+03	87	-.127302+03	88	-.112622+03
	89	.532615+02	90	.531532+02	91	.979818+02	92	.974746+02
	93	.141910+03	94	.140570+03	95	.131081+03	96	.123907+03
	97	.349852+03	98	.348816+03	99	.352354+03	100	.338908+03
	101	.484366+03	102	.480095+03	103	.537113+03	104	.533852+03
	105	.588477+03	106	.587184+03	107	.591548+03	108	.592073+03
	109	.373804+03	110	.380596+03	111	.156658+03	112	.167916+03
	113	-.252421+03	114	-.247386+03	115	.442656+02	116	.511936+02
	119	.188342+03	120	.193766+03	121	-.187789+03	122	-.184153+03
	123	.772917+02	124	.799557+02	125	.609426+02	126	.602413+02
	127	.107313+03	128	.106092+03	129	.145916+03	130	.143708+03
	131	.105732+03	132	.104712+03	133	.386105+03	134	.382658+03
	135	.467690+03	136	.463358+03	137	.510221+03	138	.504859+03
	139	.562024+03	140	.557868+03	141	.560413+03	142	.558720+03
	143	.493623+03	144	.494081+03	145	.262648+03	146	.264946+03
	147	-.153417-00	148	-.215824-00	149	-.186121+02	150	-.863564+01
	151	-.252932+03	152	-.246214+03	153	.218319+03	154	.222219+03

LOADS ARE IN POUNDS / INCH

	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 7.83
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-54	
Approved	L G DARBY	1 JUN 69			AFT FUSELAGE	

REV (A) REF 3-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1004001.18 & MOD 6

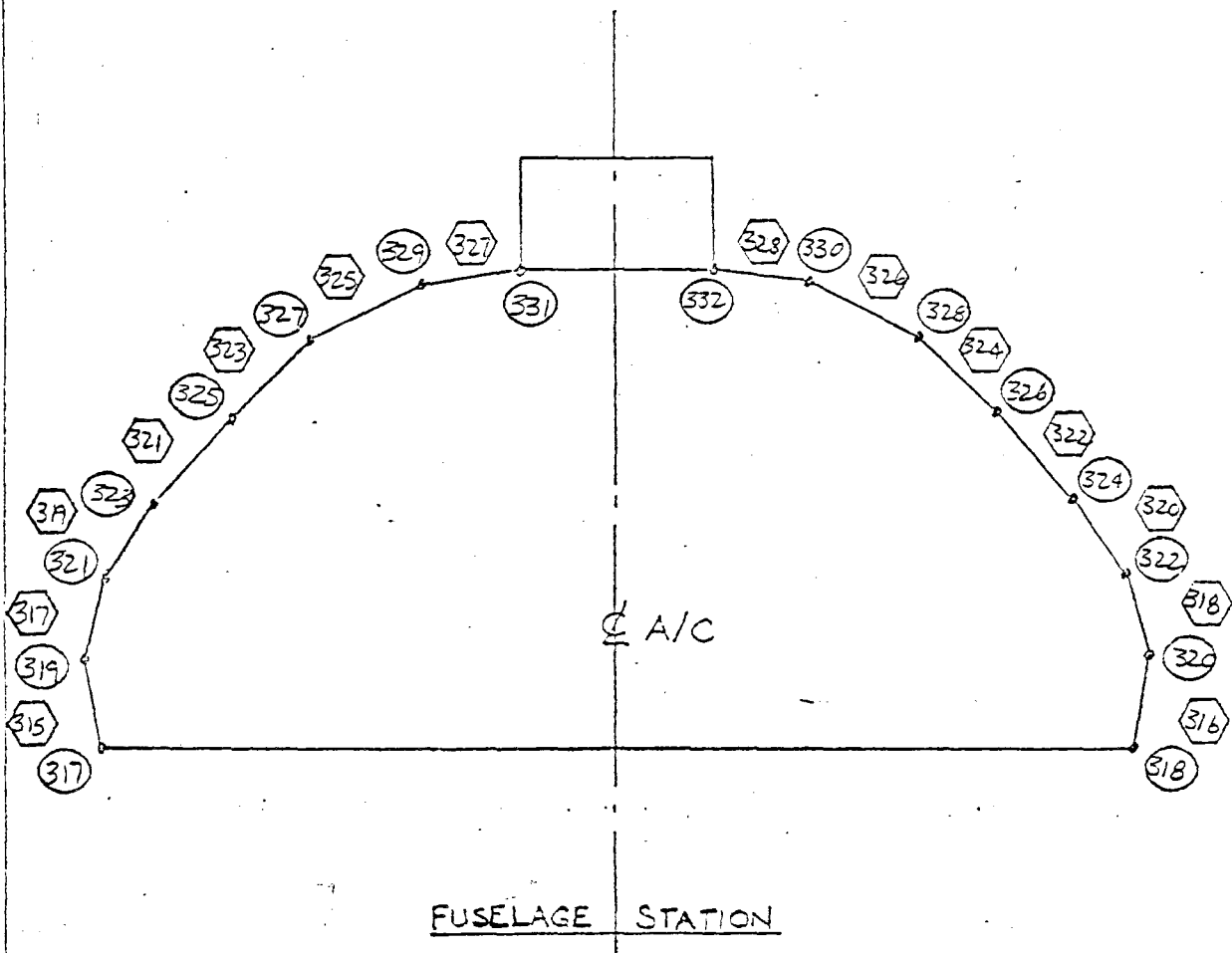
1	155	-.991268+02	156	-.946324+02	157	.754270+02	158	.752663+02
	159	-.446695+02	160	-.387192+02	161	.136101+03	162	.137393+03
	163	.623089+02	164	.613960+02	165	.107615+03	166	.106131+03
	167	.151618+03	168	.148942+03	169	.138900+03	170	.137361+03
	171	.369854+03	172	.366614+03	173	.476573+03	174	.473006+03
	175	.515573+03	176	.509544+03	177	.545362+03	178	.541086+03
	179	.528015+03	180	.525871+03	181	.452265+03	182	.452613+03
	183	.197955+03	184	.200620+03	185	.155293+03	186	.154901+03
	187	.631913+02	188	.627762+02	189	.133377+03	190	.132923+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME L. W. NYDE	DATE 9-9-68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	J. HALLIDAY	9-9-68	TITLE AFT FUSELAGE	Page	Vol. I 7.34
Approved	L. G. DARBY	9-9-68	INTERNAL LOAD ANALYSIS	Model	C-5A
				LGIUS	Report No. 46-2-2

REV. (A) MB. 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)



2461.

MOD. 7

VIEW LOOKING FORWARD FS 2472 TO FS 2450

FIGURE No 7.3.2.13

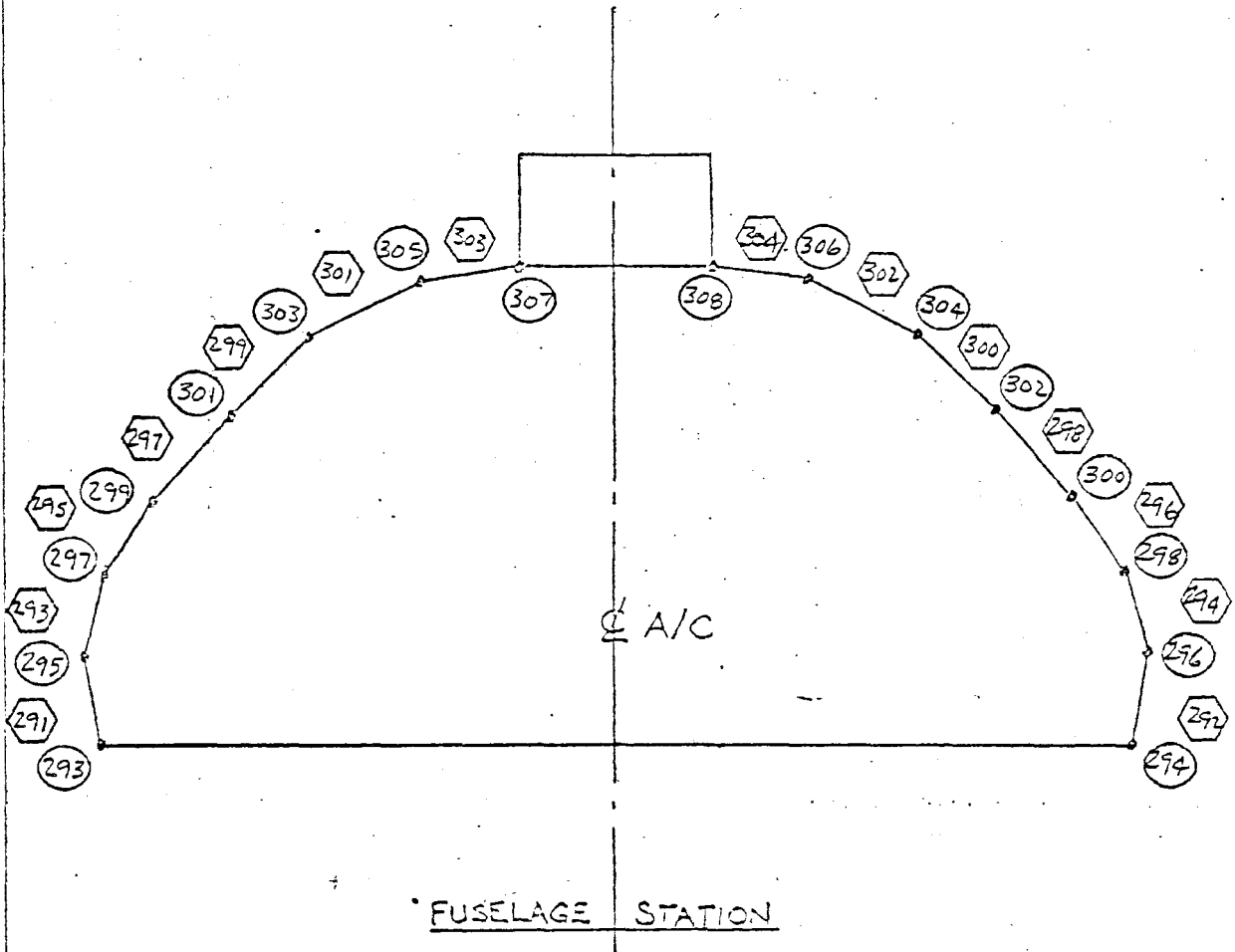
KEY

- GRID POINT
- FAMAS STRINGER
- ⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF. 3.
VOLUME IV SECTION 5.2

REV. (A) MB. 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)



2483..

MOD. 7

VIEW LOOKING FORWARD FS 2494 TO FS 2472.

FIGURE No 7.3.2.14

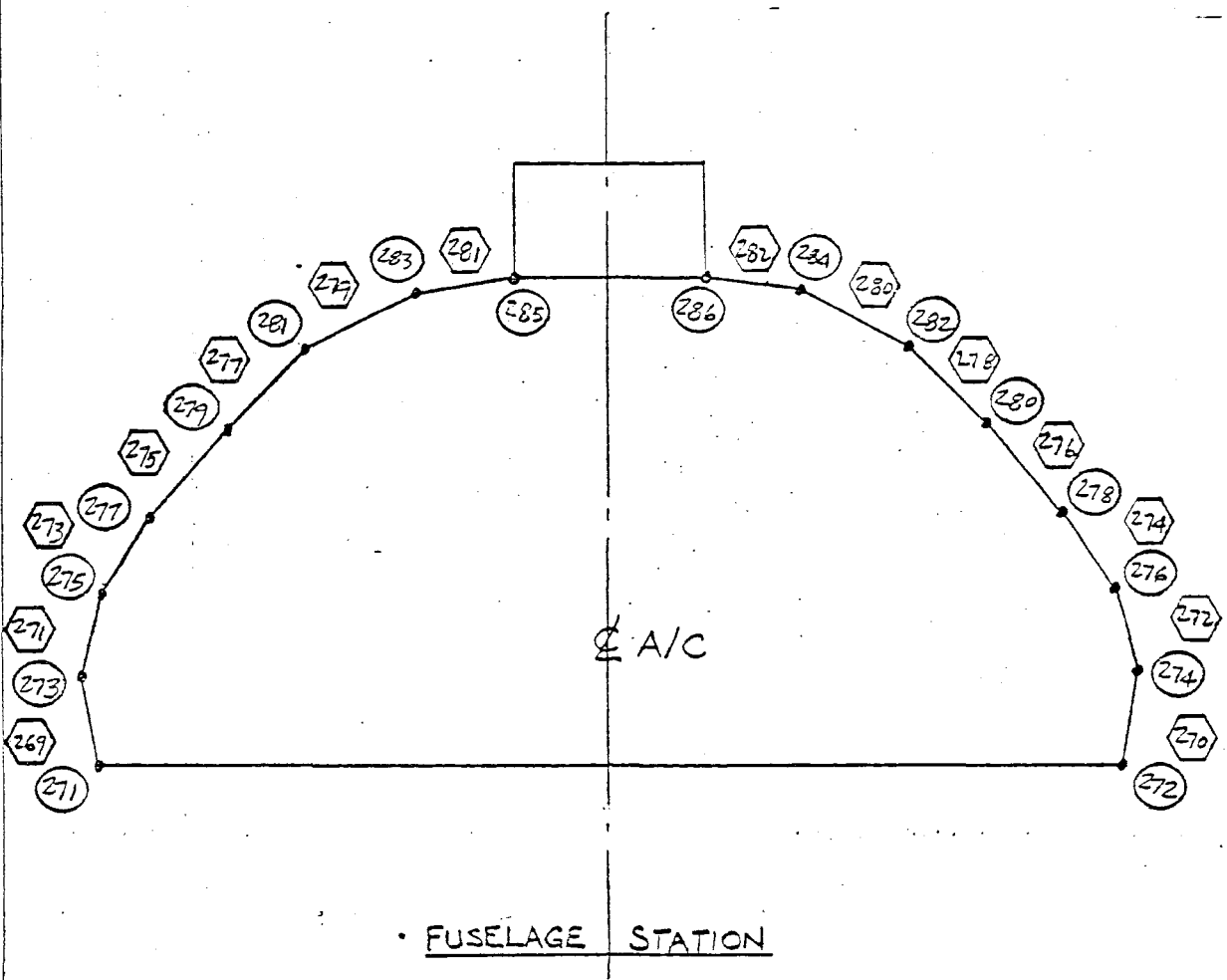
KEY

- GRID POINT
- FAMAS STRINGER
- ⬡ FAMAS PANEL

FOR AREAS AND THICKNESSES SEE REF 8
VOLUME IV SECTION 5.2.

REV. (A) M.B 1 AUG 69

7.3.2 PANEL LOADS (CONTINUED)



2505.0
MOD. 7
VIEW LOOKING FORWARD FS 2516 TO FS 6494

FIGURE No 7.3.2-15

KEY

GRID POINT

FAMAS STRINGER

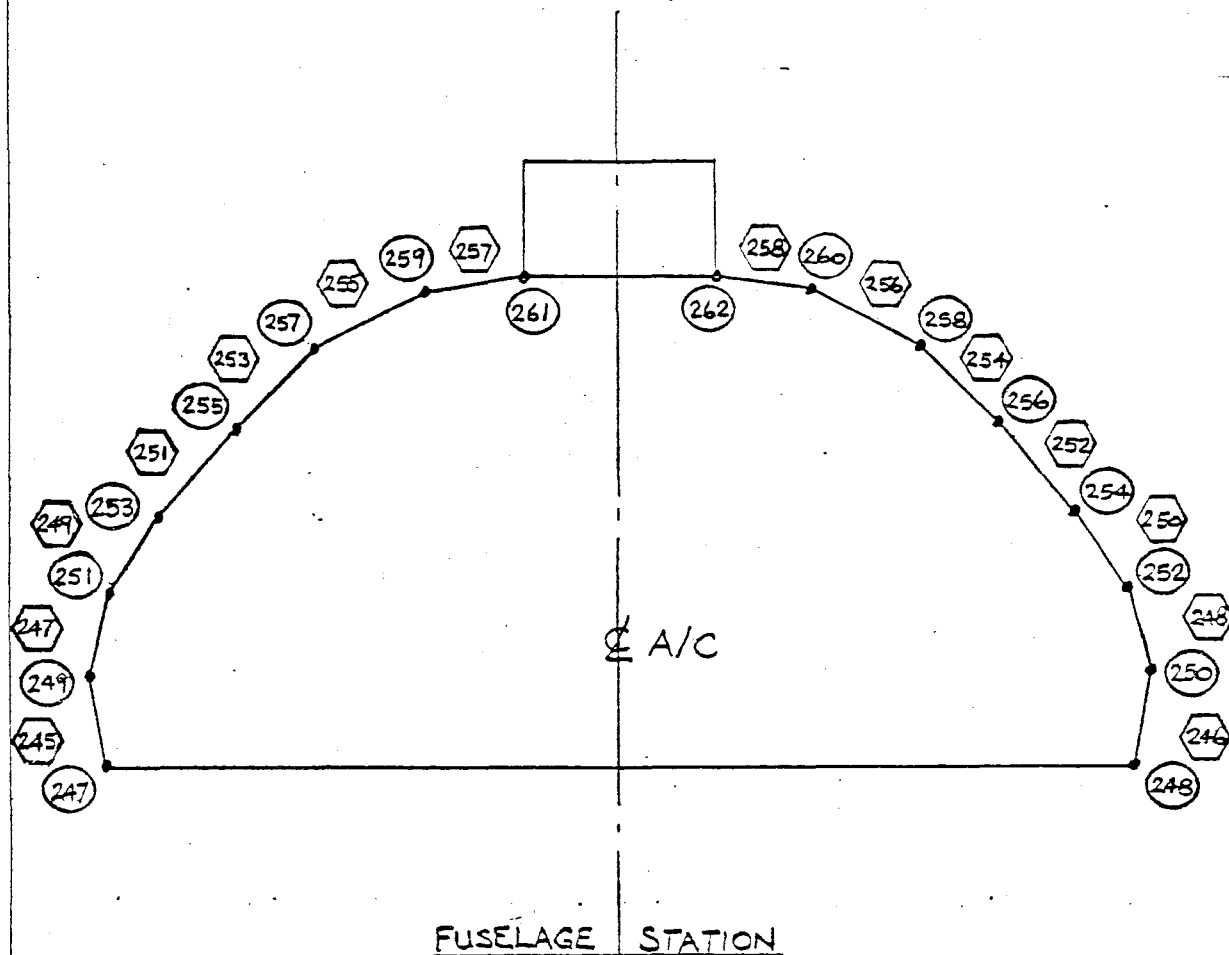
FAMAS PANEL

FOR AREAS AND THICKNESSES SEE
REF 8. VOLUME IV SECTION 5.2

Prepared	V. W. Boyd	NAME	DATE	9 Sep 68	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
					A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol I 7.37
Checked	J. Halliday	9 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model	C-5A
Approved	L.G. Darby	9 Sep 68				Report No.	LG1US 46-2-2

REV. (A) M.B. 1 AUG 1969

7.3.2 PANEL LOADS (CONTINUED)



FUSELAGE STATION

2527

MOD. 7

VIEW LOOKING FORWARD FS 2538 TO FS 2516

FIGURE No 7.3.2.16

KEY

-  GRID POINT
-  FAMAS STRINGER
-  FAMAS PANEL

FOR AREAS AND THICKNESSES
SEE REF 3 VOLUME IV SECTION 5.2

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 738
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model: C-54	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE		LG1JS46-2-2	
					Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1PC242 & MOD 7

1	1	-.300115+03	2	-.107458+03	3	-.375925+03	4	-.197740+03
	5	-.375427+03	6	-.221922+03	7	-.236232+03	8	-.243433+03
	9	-.337932+03	10	-.404612+03	11	-.500202+03	12	-.571527+03
	13	-.533650+03	14	-.605328+03	15	-.553012+03	16	-.551072+03
	17	-.681485+03	18	-.531541+03	19	-.671842+03	20	-.567235+03
	21	-.790339+03	22	-.716627+03	23	.196325+03	24	.103192+03
	25	-.314136+03	26	-.107086+03	27	-.400483+03	28	-.237966+03
	29	-.423754+03	30	-.295038+03	31	-.463905+03	32	-.463022+03
	33	-.608863+03	34	-.687508+03	35	-.806081+03	36	-.897050+03
	37	-.942235+03	38	-.973704+03	39	-.107277+04	40	-.101559+04
	41	-.108888+04	42	-.974712+03	43	-.102067+04	44	-.924653+03
	45	-.758214+03	46	-.700543+03	47	.139961+03	48	.115824+03
	49	-.256610+03	50	-.779136+02	51	-.356809+03	52	-.208519+03
	53	-.420279+03	54	-.305051+03	55	-.615132+03	56	-.589091+03
	57	-.772413+03	58	-.816904+03	59	-.982032+03	60	-.103898+04
	61	-.112238+04	62	-.114201+04	63	-.118199+04	64	-.112325+04
	65	-.103867+04	66	-.969855+03	67	-.960062+03	68	-.843905+03
	69	-.911332+03	70	-.831567+03	72	.996336+02	73	-.231315+03
	74	-.767812+02	75	-.342672+03	76	-.197142+03	77	-.443561+03
	78	-.315845+03	79	-.716666+03	80	-.654134+03	81	-.809761+03
	82	-.889521+03	83	-.108726+04	84	-.110598+04	85	-.121482+04
	86	-.121503+04	87	-.124194+04	88	-.117554+04	89	-.112037+04
	90	-.978758+03	91	-.993419+03	92	-.836564+03	93	-.879401+03
	94	-.729086+03	95	.744910+02	96	.821958+02	97	-.205447+03
	98	-.396452+02	99	-.303897+03	100	-.151303+03	101	-.406056+03
	102	-.276907+03	103	-.761890+03	104	-.696151+03	105	-.950183+03
	106	-.914656+03	107	-.115302+04	108	-.111454+04	109	-.128573+04
	110	-.121852+04	111	-.130154+04	112	-.120633+04	113	-.113262+04
	114	-.100103+04	115	-.100743+04	116	-.827743+03	117	-.823893+03
	118	-.652023+03	119	.571133+02	120	.362194+02	121	-.195961+03
	122	-.374750+02	123	-.297537+03	124	-.144445+03	125	-.399565+03
	126	-.258811+03	127	-.798469+03	128	-.684977+03	129	-.985072+03
	130	-.883662+03	131	-.117653+04	132	-.108499+04	133	-.129189+04
	134	-.121244+04	135	-.131192+04	136	-.120436+04	137	-.114277+04
	138	-.987926+03	139	-.926901+03	140	-.734546+03	141	-.834646+03
	142	-.613999+03	143	.541894+02	144	.795463+02	145	-.131489+03
	146	-.346859+02	147	-.287909+03	148	-.130534+03	149	-.393366+03
	150	-.222884+03	151	-.799128+03	152	-.659687+03	153	-.976016+03

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Page Vol. I 7.39	TEMP.	PERM.
	Checked A L PINCHOT	1 JUN 69			Model C-5A	
Approved	L G DARBY	1 JUN 69		Report No.		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
7.3 SKINS AND STRINGERS
7.3.2 SKIN LOADS CONDITION 1PC242 & MOD 7

1	154	-.827130+03	155	-.115660+04	156	-.101408+04	157	-.128464+04
	158	-.115687+04	159	-.131609+04	160	-.118274+04	161	-.110272+04
	162	-.954953+03	163	-.849582+03	164	-.659567+03	165	-.673553+03
	166	-.418453+03	167	.652448+02	168	.714074+02	169	-.179916+03
	170	-.365603+02	171	-.231770+03	172	-.128759+03	173	-.409862+03
	174	-.241845+03	175	-.793353+03	176	-.618214+03	177	-.957109+03
	178	-.767330+03	179	-.108964+04	180	-.902323+03	181	-.123616+04
	182	-.105056+04	183	-.125177+04	184	-.109866+04	185	-.103687+04
	186	-.897354+03	187	-.768224+03	188	-.582330+03	189	-.536998+03
	190	-.309352+03	191	.813798+02	192	.707488+02	193	-.172867+03
	194	-.214484+02	195	-.245132+03	196	-.830488+02	197	-.337579+03
	198	-.154702+03	199	-.725543+03	200	-.528157+03	201	-.841300+03
	202	-.621804+03	203	-.106150+04	204	-.833937+03	205	-.116541+04
	206	-.958497+03	207	-.114026+04	208	-.975228+03	209	-.960379+03
	210	-.823423+03	211	-.643274+03	212	-.429734+03	213	-.614519+03
	214	-.339065+03	215	.101452+03	216	.694352+02	217	-.137751+03
	218	.891140+01	219	-.193335+03	220	-.354205+02	221	-.238182+03
	222	-.456413+02	223	-.691288+03	224	-.425278+03	225	-.833051+03
	226	-.573416+03	227	-.945626+03	228	-.688369+03	229	-.983427+03
	230	-.763381+03	231	-.914402+03	232	-.726537+03	233	-.724012+03
	234	-.555510+03	235	-.385583+03	236	-.196112+03	237	.140444+03
	238	.712957+02	239	-.722331+02	240	.105919+03	241	-.120495+03
	242	.612676+02	243	-.234075+03	244	-.440720+02	245	-.707378+03
*	246	-.499212+03	247	-.896916+03	248	-.653096+03	249	-.792500+03
	250	-.527499+03	251	-.671213+03	252	-.428991+03	253	-.527397+03
	254	-.311731+03	255	-.328488+03	256	-.112230+03	257	.250673+03
	258	.596307+03	259	-.829259+03	260	-.117756+04	261	.873539+02
	262	.821302+02	263	-.554808+02	264	.114667+03	265	-.602630+02
	266	.119558+03	267	-.233025+03	268	-.430502+02	269	-.780113+03
	270	-.555008+03	271	-.885799+03	272	-.649935+03	273	-.802912+03
	274	-.558894+03	275	-.637445+03	276	-.390481+03	277	-.400625+03
	278	-.172607+03	279	-.159792+03	280	.628991+02	281	.372743+03
	282	.591812+03	283	.875461+03	284	.111399+04	285	-.436002+02
	286	.122786+03	287	.385460+02	288	.219484+03	289	.256517+03
	290	.433311+03	291	-.810321+03	292	-.589028+03	293	-.914215+03
	294	-.677872+03	295	-.829822+03	296	-.591388+03	297	-.583625+03
	298	-.341260+03	299	-.278203+03	300	-.466494+02	301	-.143144+02
	302	.193953+03	303	.494324+03	304	.668060+03	305	.791442+03

* PANEL (240) REF. P. 7.37 ULTIMATE PANEL SHEAR FLOW = -499, LBS./IN.

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. Vol. I	PERM. 7.90
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A	
Approved	L G CARBY	1 JUN 69	AFT FUSELAGE		LG10546-2-2 Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1PC242 & MOD 7

1	306	.945459+03	307	.791405+02	308	.773017+02	309	-.501541+02
	310	.112830+03	311	-.426486+02	312	.132783+03	313	-.978574+02
	314	.108420+03	315	-.971755+03	316	-.762097+03	317	-.929647+03
	318	-.699301+03	319	-.776805+03	320	-.538507+03	321	-.527535+03
	322	-.287881+03	323	-.176067+03	324	.461269+02	325	.125860+03
	326	.318955+03	327	.577982+03	328	.727207+03	329	-.104534+04
	330	-.672074+03	332	.248808+02	334	.196363+02	336	.381067+01
	337	.322347+03	338	.394738+03	340	.141099+01	341	.321402+02
	342	-.281108+03	343	-.436977+02	344	-.155646+03	345	-.216341+02
	346	-.277600+03	347	-.100024+03	348	-.101273+03	349	-.329878+02
	350	-.120341+03	351	-.604581+02	352	-.170520+03	353	-.969602+02
	354	-.235455+03	355	-.118584+03	356	-.291573+03	357	-.123613+03
	358	-.335337+03	359	.157333+03	360	-.113227+03	361	.219705+03
	362	.210654+03	363	.270320+03	364	.543018+03	365	.105532+03
	366	.373351+03	367	-.531370+03	368	-.441903+03		

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 7.91
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS		Model C-5A		
Approved	L G DARBY	1 JUN 69			AFT FUSELAGE Report No. LG1J346-2-2		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION SPE201 & MCD 7

1	1	.419527+03	2	.254863+03	3	.552183+03	4	.400135+03
	5	.562425+03	6	.431595+03	7	.401152+03	8	.407393+03
	9	.609880+03	10	.667373+03	11	.894566+03	12	.955858+03
	13	.946114+03	14	.101080+04	15	.934034+03	16	.943567+03
	17	.108343+04	18	.101514+04	19	.106195+04	20	.993550+03
	21	.127558+04	22	.123577+04	23	-.164003+03	24	-.883837+02
	25	.435329+03	26	.258040+03	27	.595194+03	28	.452853+03
	29	.651179+03	30	.541257+03	31	.782642+03	32	.786496+03
	33	.106633+04	34	.113480+04	35	.140616+04	36	.148753+04
	37	.160292+04	38	.163712+04	39	.176691+04	40	.173125+04
	41	.175347+04	42	.167406+04	43	.164394+04	44	.158171+04
	45	.122169+04	46	.119292+04	47	-.111875+03	48	-.100169+03
	49	.349217+03	50	.196001+03	51	.526734+03	52	.399942+03
	53	.647715+03	54	.549400+03	55	.102060+04	56	.998542+03
	57	.131521+04	58	.135582+04	59	.167001+04	60	.172383+04
	61	.188355+04	62	.190892+04	63	.193831+04	64	.190123+04
	65	.174421+04	66	.165980+04	67	.152843+04	68	.145278+04
	69	.145765+04	70	.141195+04	72	-.856008+02	73	.314666+03
	74	.181993+03	75	.501028+03	76	.376561+03	77	.577646+03
	78	.569228+03	79	.116364+04	80	.111246+04	81	.143872+04
	82	.148406+04	83	.131040+04	84	.183279+04	85	.201066+04
	86	.202017+04	87	.202211+04	88	.197931+04	89	.177955+04
	90	.167577+04	91	.155692+04	92	.144341+04	93	.176883+04
	94	.126149+04	95	-.530868+02	96	-.707531+02	97	.267860+03
	98	.124616+03	99	.430182+03	100	.299605+03	101	.607075+03
	102	.497158+03	103	.122460+04	104	.117144+04	105	.155834+04
	106	.152520+04	107	.187948+04	108	.185387+04	109	.208584+04
	110	.203826+04	111	.209700+04	112	.202867+04	113	.179530+04
	114	.169933+04	115	.156053+04	116	.142708+04	117	.126713+04
	118	.113631+04	119	-.338378+02	120	-.742204+02	121	.252702+03
	122	.116514+03	123	.417034+03	124	.235981+03	125	.587226+03
	126	.467544+03	127	.124039+04	128	.115593+04	129	.155171+04
	130	.148019+04	131	.187707+04	132	.180683+04	133	.207454+04
	134	.201632+04	135	.209628+04	136	.201656+04	137	.179340+04
	138	.167615+04	139	.141371+04	140	.126760+04	141	.124652+04
	142	.107920+04	143	-.372644+02	144	-.685362+02	145	.232919+03
	146	.106698+03	147	.397806+03	148	.263013+03	149	.561547+03
	150	.416343+03	151	.122933+04	152	.111485+04	153	.150756+04

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	VOL I 7.92
Checked	A L PINCHOT	1 JUN 69	TITLE	Model	C-5A
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	Report No.	LG1JS46-2-2
			AFT FUSELAGE		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 8

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS CONDITION SPE201 & MOD 7

1	154	.138706+04	155	.130203+04	156	.168888+04	157	.202229+04
	158	.192288+04	159	.207617+04	160	.197372+04	161	.172031+04
	162	.160712+04	163	.127849+04	164	.113204+04	165	.920275+03
	166	.748809+03	167	-.466206+02	168	-.615684+02	169	.231292+03
	170	.108011+03	171	.387363+03	172	.256239+03	173	.584057+03
	174	.441054+03	175	.117441+04	176	.103864+04	177	.143790+04
	178	.128329+04	179	.165250+04	180	.150135+04	181	.189640+04
	182	.174745+04	183	.194346+04	184	.182290+04	185	.160285+04
	186	.149488+04	187	.113888+04	188	.992881+03	189	.742380+03
	190	.562743+03	191	-.602663+02	192	-.609764+02	193	.211714+03
	194	.831879+02	195	.318198+03	196	.179051+03	197	.447781+03
	198	.291654+03	199	.104841+04	200	.885419+03	201	.121219+04
	202	.103241+04	203	.155969+04	204	.137454+04	205	.174360+04
	206	.157609+04	207	.173278+04	208	.160077+04	209	.147039+04
	210	.136222+04	211	.920097+03	212	.747540+03	213	.840144+03
	214	.617638+03	215	-.775445+02	216	-.599006+02	217	.159743+03
	218	.333828+02	219	.237478+03	220	.101549+03	221	.285365+03
	222	.120011+03	223	.965392+03	224	.743731+03	225	.116958+04
	226	.951860+03	227	.131651+04	228	.110680+04	229	.139114+04
	230	.121266+04	231	.130761+04	232	.115474+04	233	.103599+04
	234	.897710+03	235	.496885+03	236	.343053+03	237	-.111133+03
	238	-.615848+02	239	.441922+02	240	-.109871+03	241	.113863+03
	242	-.428582+02	243	.284910+03	244	.122445+03	245	.980466+03
	246	.809326+03	247	.123246+04	248	.107531+04	249	.100557+04
	250	.790955+03	251	.793632+03	252	.597232+03	253	.603276+03
	254	.424231+03	255	.331807+03	256	.143903+03	257	-.565582+03
	258	-.967448+03	259	.163105+04	260	.192629+04	261	-.755623+02
	262	-.709766+02	263	.202154+02	264	-.126843+03	265	.203890+02
	266	-.135479+03	267	.279389+03	268	.116181+03	269	.106743+04
	270	.883643+03	271	.119815+04	272	.100811+04	273	.102933+04
	274	.832463+03	275	.746715+03	276	.545227+03	277	.389301+03
	278	.198283+03	279	.392702+02	280	-.155032+03	281	-.791905+03
	282	-.939569+03	283	-.155037+04	284	-.186153+04	285	.104395+01
	286	-.142263+03	287	-.136304+03	288	-.293627+03	289	-.476537+03
	290	-.634264+03	291	.109841+04	292	.919753+03	293	.122810+04
	294	.103723+04	295	.106881+04	296	.876870+03	297	.665509+03
	298	.466790+03	299	.188706+03	300	-.523467+01	301	-.197155+03
	302	-.379776+03	303	-.963829+03	304	-.112256+04	305	-.147059+04

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.93
Checked	A L PINCHET	1 JUN 69	TITLE	Model	C-5A
Approved	L G DARBY	1 JUN 69	INTERNAL LOADS ANALYSIS	Report No.	LG1JS46-2-2
			AFT FUSELAGE		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. &
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION SPE201 & MOD 7

1 306	-.161300+04	307	-.632109+02	308	-.663388+02	309	.111215+02
310	-.128790+03	311	-.121051+02	312	-.162380+03	313	.542163+02
314	-.122610+03	315	.132723+04	316	.116567+04	317	.123487+04
318	.105287+04	319	.987562+03	320	.794343+03	321	.536618+03
322	.386666+03	323	.311101+02	324	-.159790+03	325	-.424842+03
326	-.595891+03	327	-.108083+04	328	-.121942+04	329	.153208+04
330	.122122+04	332	-.822537+01	334	-.315070+01	336	.670171+01
337	-.619372+03	338	-.696491+03	340	.656155+01	341	.758946+02
342	.341300+03	343	.122743+03	344	.213313+03	345	.157143+03
346	.347106+03	347	.171961+03	348	.167505+03	349	.101213+03
350	.159060+03	351	.158346+03	352	.233734+03	353	.233565+03
354	.332478+03	355	.285741+03	356	.414224+03	357	.309630+03
358	.471226+03	359	-.158284+03	360	.560264+02	361	-.361429+03
362	-.383885+03	363	-.568297+03	364	-.797713+03	365	-.297399+03
366	-.508821+03	367	.759896+03	368	.691836+03		

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. 7.94
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	L G DAFBY	1 JUN 69	AFT ELSE LAGE	Report No.	LG1JS46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION DML001 & MOD 7

1	1	-.366220+03	2	.523049+03	3	-.313373+03	4	.534665+03
	5	-.252618+03	6	.433804+03	7	.110753+03	8	.755790+02
	9	.286200+03	10	.665943+01	11	.284836+03	12	.143700+03
	13	.187411+03	14	.265979+03	15	-.237150+03	16	.671224+03
	17	-.927732+03	18	.140365+04	19	-.123736+04	20	.169696+04
	21	-.152513+04	22	.208463+04	23	.190093+03	24	.455567+03
	25	-.385373+03	26	.546416+03	27	-.264745+03	28	.503277+03
	29	-.165393+03	30	.442160+03	31	.186170+03	32	.177067+03
	33	.369077+03	34	.140520+03	35	.323120+03	36	.347468+03
	37	.928625+01	38	.741025+03	39	-.503411+03	40	.130835+04
	41	-.927580+03	42	.170654+04	43	-.106259+04	44	.173620+04
	45	-.126664+04	46	.179763+04	47	.406194+03	48	.509505+03
	49	-.334857+03	50	.461778+03	51	-.239051+03	52	.454571+03
	53	-.141420+03	54	.419702+03	55	.125589+03	56	.343553+03
	57	.289918+03	58	.330980+03	59	.234317+03	60	.554525+03
	61	-.226802+02	62	.902499+03	63	-.492936+03	64	.137330+04
	65	-.961284+03	66	.173567+04	67	-.116754+04	68	.183327+04
	69	-.122479+04	70	.185285+04	72	.426320+03	73	-.279147+03
	74	.394886+03	75	-.248301+03	76	.452757+03	77	-.106168+03
	78	.486477+03	79	.117048+02	80	.513800+03	81	.175166+03
	82	.558278+03	83	.126865+03	84	.722743+03	85	-.633299+02
	86	.100109+04	87	-.482495+03	88	.140770+04	89	-.938002+03
	90	.177516+04	91	-.119990+04	92	.186954+04	93	-.115189+04
	94	.172637+04	95	.557141+03	96	.333406+03	97	-.321649+03
	98	.413104+03	99	-.279787+03	100	.449832+03	101	-.213179+03
	102	.470503+03	103	-.254462+02	104	.585134+03	105	.505071+01
	106	.713606+03	107	-.410905+02	108	.913045+03	109	-.231760+03
	110	.119315+04	111	-.532014+03	112	.148785+04	113	-.898439+03
	114	.169626+04	115	-.116227+04	116	.182961+04	117	-.109073+04
	118	.161429+04	119	.508248+03	120	.353870+03	121	-.301123+03
	122	.387162+03	123	-.289749+03	124	.452832+03	125	-.263179+03
	126	.509590+03	127	-.174752+03	128	.735471+03	129	-.202430+03
	130	.914450+03	131	-.216511+03	132	.107840+04	133	-.252089+03
	134	.120813+04	135	-.513661+03	136	.147411+04	137	-.890763+03
	138	.168515+04	139	-.112707+04	140	.172437+04	141	-.119031+04
	142	.169653+04	143	.489926+03	144	.317427+03	145	-.271810+03
	146	.350907+03	147	-.303025+03	148	.457441+03	149	-.350717+03
	150	.579846+03	151	-.322061+03	152	.872091+03	153	-.390571+03

LOADS ARE IN POUNDS/INCH

NAME RF WILKINSON		DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. NOL. I	PERM. 7.95	
Prepared			TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE					
Checked	A L PINCHOT	1 JUN 69			Model	C-54		
Approved	L G DARBY	1 JUN 69			Report No.	LG1J846-2-2		

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION DML001 & MOD 7

1	154	.106985+04	155	-.412584+03	156	.123109+04	157	-.407076+03
	158	.133056+04	159	-.554106+03	160	.150645+04	161	-.738112+03
	162	.155176+04	163	-.103407+04	164	.157129+04	165	-.106102+04
	166	.141849+04	167	.560378+03	168	.273968+03	169	-.251315+03
	170	.340174+03	171	-.293481+03	172	.443584+03	173	-.347023+03
	174	.589083+03	175	-.457889+03	176	.978267+03	177	-.576158+03
	178	.121668+04	179	-.599612+03	180	.134130+04	181	-.618761+03
	182	.147341+04	183	-.602948+03	184	.148074+04	185	-.710040+03
	186	.142170+04	187	-.939554+03	188	.141573+04	189	-.104765+04
	190	.132276+04	191	.676026+03	192	.270170+03	193	-.263579+03
	194	.336414+03	195	-.322060+03	196	.436858+03	197	-.410101+03
	198	.581892+03	199	-.621315+03	200	.107765+04	201	-.755265+03
	202	.128640+04	203	-.798661+03	204	.149090+04	205	-.732828+03
	206	.151356+04	207	-.653560+03	208	.143096+04	209	-.625506+03
	210	.127864+04	211	-.101182+04	212	.139082+04	213	-.113808+04
	214	.145078+04	215	.838558+03	216	.247680+03	217	-.254918+03
	218	.298792+03	219	-.301377+03	220	.378879+03	221	-.419130+03
	222	.512669+03	223	-.805216+03	224	.121286+04	225	-.953130+03
	226	.146242+04	227	-.104474+04	228	.161795+04	229	-.935542+03
	230	.154743+04	231	-.834645+03	232	.140496+04	233	-.741923+03
	234	.118141+04	235	-.834051+03	236	.105875+04	237	.115366+04
	238	.240550+03	239	-.324434+03	240	.307074+03	241	-.364720+03
	242	.378580+03	243	-.423372+03	244	.516303+03	245	-.810812+03
	246	.123638+04	247	-.108177+04	248	.162268+04	249	-.130044+04
	250	.172761+04	251	-.127673+04	252	.160001+04	253	-.106404+04
	254	.129191+04	255	-.934716+03	256	.102266+04	257	-.162477+04
	258	.120585+04	259	.195981+04	260	-.109265+04	261	.289334+03
	262	.261184+03	263	-.295158+03	264	.268900+03	265	-.345414+03
	266	.316175+03	267	-.465138+03	268	.554882+03	269	-.905071+03
	270	.136558+04	271	-.108705+04	272	.160725+04	273	-.120545+04
	274	.164114+04	275	-.125473+04	276	.154914+04	277	-.112512+04
	278	.124490+04	279	-.101950+04	280	.968361+03	281	-.112761+04
	282	.673566+03	283	-.149046+04	284	.633476+03	285	-.289362+03
	286	.254011+03	287	-.323253+03	288	.218894+03	289	-.291770+03
	290	.267767+02	291	-.933626+03	292	.140408+04	293	-.109816+04
	294	.162580+04	295	-.117609+04	296	.162552+04	297	-.121673+04
	298	.146791+04	299	-.113832+04	300	.115943+04	301	-.100670+04
	302	.846542+03	303	-.985225+03	304	.466525+03	305	-.103951+04

LOADS ARE IN POUNDS/INCH

Prepared	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. VOL. I	PERM. 7.96
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS	Model	C-FA	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	LG1JS46-2-2	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON. 6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION DML001 3 MOD 7

1	306	.287002+03	307	.251590+03	308	.251367+03	309	-.290434+03
	310	.260622+03	311	-.346822+03	312	.302465+03	313	-.453084+03
	314	.431500+03	315	-.100198+04	316	.157868+04	317	-.110747+04
	318	.163453+04	319	-.112598+04	320	.153095+04	321	-.113015+04
	322	.138739+04	323	-.110968+04	324	.105443+04	325	-.038127+03
	326	.722422+03	327	-.919039+03	328	.352147+03	329	-.121050+04
	330	.183606+04	332	.273009+03	334	.216495+03	336	.519767+02
	337	-.471031+03	338	.147230+03	340	.176563+02	341	-.592502+02
	342	.156160+03	343	.193319+03	344	-.112662+03	345	.790260+03
	346	-.670741+03	347	.830228+02	348	-.328512+01	349	.455797+03
	350	-.390067+03	351	.551074+03	352	-.453813+03	353	.655467+03
	354	-.517016+03	355	.779327+03	356	-.610669+03	357	.936836+03
	358	-.751599+03	359	.107588+04	360	-.110686+04	361	.728794+01
	362	-.193801+03	363	.107409+03	364	-.417428+03	365	-.476824+03
	366	.295135+02	367	-.405994+03	368	.742373+03		

LOADS ARE IN POUNDS/INCH

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
RF WILKINSON	1 JUN 69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 7.97
Prepared		TITLE			
Checked	A L PINCHOT	1 JUN 69	INTERNAL LOADS ANALYSIS	Model C-5A	
Approved	L G DARBY	1 JUN 69	AFT FUSELAGE	Report No.	

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.6
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LOADS CONDITION 1004001.18 & MOD 7

1	1	-.194976+03	2	-.195211+03	3	-.280249+03	4	-.280459+03
	5	-.295484+03	6	-.295683+03	7	-.238502+03	8	-.238396+03
	9	-.413478+03	10	-.413242+03	11	-.621438+03	12	-.621210+03
	13	-.649324+03	14	-.649065+03	15	-.598204+03	16	-.598144+03
	17	-.628674+03	18	-.628670+03	19	-.610025+03	20	-.610042+03
	21	-.745545+03	22	-.745644+03	23	.116068+00	24	-.162905-00
	25	-.192258+03	26	-.192495+03	27	-.295333+03	28	-.295484+03
	29	-.341092+03	30	-.341130+03	31	-.471896+03	32	-.471751+03
	33	-.700215+03	34	-.699828+03	35	-.945517+03	36	-.945090+03
	37	-.105338+04	38	-.105312+04	39	-.109700+04	40	-.109688+04
	41	-.102630+04	42	-.102636+04	43	-.944742+03	44	-.944875+03
	45	-.700597+03	46	-.700553+03	47	.158146-00	48	-.171614-00
	49	-.140324+03	50	-.140469+03	51	-.247107+03	52	-.247231+03
	53	-.331801+03	54	-.331755+03	55	-.608058+03	56	-.607890+03
	57	-.835180+03	58	-.834836+03	59	-.108099+04	60	-.108043+04
	61	-.120657+04	62	-.120627+04	63	-.119235+04	64	-.119224+04
	65	-.100987+04	66	-.101001+04	67	-.853956+03	68	-.854160+03
	69	-.806839+03	70	-.807181+03	72	-.121568+00	73	-.118054+03
	74	-.118195+03	75	-.222864+03	76	-.222799+03	77	-.339409+03
	78	-.339500+03	79	-.669919+03	80	-.669772+03	81	-.898645+03
	82	-.898140+03	83	-.112428+04	84	-.112385+04	85	-.124721+04
	86	-.124691+04	87	-.122279+04	88	-.122272+04	89	-.100870+04
	90	-.100879+04	91	-.837915+03	92	-.838160+03	93	-.701002+03
	94	-.701151+03	95	.167508-00	96	-.109537+00	97	-.863868+02
	98	-.865000+02	99	-.168057+03	100	-.168085+03	101	-.273434+03
	102	-.273296+03	103	-.678196+03	104	-.677934+03	105	-.893223+03
	106	-.892800+03	107	-.110465+04	108	-.110416+04	109	-.123253+04
	110	-.123237+04	111	-.122757+04	112	-.122745+04	113	-.995723+03
	114	-.995869+03	115	-.785940+03	116	-.786253+03	117	-.603485+03
	118	-.603723+03	119	.170988-00	120	-.949248-01	121	-.775344+02
	122	-.776514+02	123	-.156278+03	124	-.156221+03	125	-.250349+03
	126	-.250179+03	127	-.644337+03	128	-.644090+03	129	-.830810+03
	130	-.830474+03	131	-.102755+04	132	-.102710+04	133	-.117680+04
	134	-.117652+04	135	-.118578+04	136	-.118565+04	137	-.952537+03
	138	-.952692+03	139	-.659212+03	140	-.659526+03	141	-.523105+03
	142	-.523404+03	143	.140605-00	144	-.806238-01	145	-.637806+02
	146	-.638911+02	147	-.143262+03	148	-.143198+03	149	-.224345+03
	150	-.224588+03	151	-.593315+03	152	-.593007+03	153	-.733163+03

LOADS ARE IN POUNDS/INCH

NAME

DATE

LOCKHEED-GEORGIA COMPANY

TEMP.

PERM.

Prepared

RF WILKINSON

1 JUN 69

A DIVISION OF LOCKHEED AIRCRAFT CORPORATION

Page

Vol. I 17.93

Checked

A L PINCHOT

1 JUN 69

TITLE

INTERNAL LOADS ANALYSIS

Model

C-54

LG1JS46-2-2

Approved

L G DARBY

1 JUN 69

AFT FUSELAGE

Report No.

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.6

7.3 SKINS AND STRINGERS

7.3.2 SKIN LOADS CONDITION 1004001.18 & MOD 7

1	154	-.732842+03	155	-.902312+03	156	-.902005+03	157	-.105894+04
	158	-.105866+04	159	-.110748+04	160	-.110734+04	161	-.876173+03
	162	-.876336+03	163	-.541082+03	164	-.541363+03	165	-.305756+03
	166	-.305953+03	167	.114037+00	168	-.747253-01	169	-.673272+02
	170	-.679251+02	171	-.130922+03	172	-.130793+03	173	-.219521+03
	174	-.219239+03	175	-.490024+03	176	-.489680+03	177	-.614720+03
	178	-.614597+03	179	-.736545+03	180	-.736253+03	181	-.893952+03
	182	-.893676+03	183	-.955819+03	184	-.955684+03	185	-.762124+03
	186	-.762273+03	187	-.424714+03	188	-.425045+03	189	-.155613+03
	190	-.155804+03	191	.740121-01	192	-.719773-01	193	-.461663+02
	194	-.463379+02	195	-.629792+02	196	-.628684+02	197	-.790601+02
	198	-.788048+02	199	-.346831+03	200	-.346483+03	201	-.383782+03
	202	-.383760+03	203	-.567353+03	204	-.567111+03	205	-.703773+03
	206	-.703598+03	207	-.754146+03	208	-.754074+03	209	-.670943+03
	210	-.671061+03	211	-.283639+03	212	-.283885+03	213	-.135571+03
	214	-.136716+03	215	.349046-02	216	-.297401-01	217	-.234879+02
	218	-.236216+02	219	-.313092+02	220	-.313292+02	221	-.875042+01
	222	-.830261+01	223	-.313113+03	224	-.312681+03	225	-.283918+03
	226	-.289925+03	227	-.248663+03	228	-.248554+03	229	-.293516+03
	230	-.293511+03	231	-.324080+03	232	-.324269+03	233	-.270476+03
	234	-.270546+03	235	-.190887+01	236	-.187132+01	237	-.791935-01
	238	.173248-01	239	.238600+02	240	.291276+02	241	.191514+02
	242	.189618+02	243	-.311197+02	244	-.311245+02	245	-.144290+03
	246	-.144052+03	247	-.134458+03	248	-.134437+03	249	.189572+03
	250	.189456+03	251	.366999+03	252	.366452+03	253	.307247+03
	254	.306611+03	255	.266749+03	256	.266953+03	257	.100437+04
	258	.100496+04	259	-.163410+04	260	-.163376+04	261	.176460-00
	262	.228538-00	263	.353569+02	264	.355494+02	265	.574900+02
	266	.574509+02	267	-.420379+01	268	-.442274+01	269	-.115984+03
	270	-.115873+03	271	-.554144+02	272	-.551752+02	273	.145947+03
	274	.145838+03	275	.333117+03	276	.332467+03	277	.440939+03
	278	.440272+03	279	.520070+03	280	.520154+03	281	.962438+03
	282	.963153+03	283	.157664+04	284	.157669+04	285	.437771+02
	286	.439237+02	287	.121580+03	288	.121634+03	289	.290665+03
	290	.290106+03	291	-.873526+02	292	-.873959+02	293	-.252663+02
	294	-.245433+02	295	.121651+03	296	.121450+03	297	.342240+03
	298	.341474+03	299	.560190+03	300	.559406+03	301	.634928+03
	302	.684959+03	303	.103488+04	304	.103566+04	305	.137382+04

LOADS ARE IN POUNDS/INCH

	NAME RF WILKINSON	DATE 1 JUN 69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. 7.99	PERM. 7.99
Prepared					Page	
Checked	A L PINCHOT	1 JUN 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-5A
Approved	L G DARBY	1 JUN 69			Report No.	LG1J346-2-2

REV (A) RFW 8-1-69

7.0 SHELL STRUCTURE F.S. 2101 TO 2538 CON.8
 7.3 SKINS AND STRINGERS
 7.3.2 SKIN LEADS CONDITION 1004001.18 & MOD 7

1	306	.137402+04	307	.214213-00	308	.171557-00	309	.412557+02
	310	.416073+02	311	.758968+02	312	.757722+02	313	.857827+02
	314	.853597+02	315	-.133120+03	316	-.132580+03	317	.309852-00
	318	.825064-00	319	.151368+03	320	.151192+03	321	.357970+03
	322	.357031+03	323	.650017+03	324	.649082+03	325	.838127+03
	326	.838096+03	327	.107579+04	328	.107679+04	329	-.637913+03
	330	-.587074+03	332	-.249695-00	334	-.375970-00	336	-.412906-00
	337	.682938+03	338	.683681+03	340	-.312038-00	341	-.130570+03
	342	-.130708+03	343	-.107900+03	344	-.108013+03	345	-.166042+03
	346	-.166552+03	347	-.116452+03	348	-.116516+03	349	-.100498+03
	350	-.100741+03	351	-.151168+03	352	-.151435+03	353	-.221307+03
	354	-.221539+03	355	-.274555+03	356	-.274695+03	357	-.297048+03
	358	-.297054+03	359	.108736+03	360	.108971+03	361	.319223+03
	362	.319754+03	363	.428588+03	364	.428724+03	365	.253505+03
	366	.253836+03	367	-.201132+03	368	-.200958+03		

LOADS ARE IN POUNDS/INCH

PREPARED	NAME L.D. PINCHOT	DATE 3-19-68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	FORM. 
CHECKED	J. HALLIDAY	3-19-68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L.G. DARBY	3-19-68		LGIUS46-2-2 REPORT NO.		

REV. (3) ADP 1-8-71

 PAGES 7.100 & 7.101 "NOT USED"

7.102 - 7.113 "DELETED"

7.114 - 7.139 "NOT USED"

PREPARED	NAME J.P. Kimmel	DATE 9 Sep 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	REF. 7.140
CHECKED	J. Halliday	9 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L.G. Darby	9 Sep 68		LG1US 46-2-2 REPORT NO.		

7.4 LONGERONS

7.4.1 W.L. 314 Longerons

The longeron from F.S. 2101 to F.S. 2450 is in two lengths of 7075-T6 extrusions which are spliced at F.S. 2273. The internal loads in this region are found in Sect. 7.3.1.

PREPARED	NAME A. D. Pinchot	DATE 1 Aug. 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. 701.1	FORM 
CHECKED	R.F. Wilkinson	1 Aug. 69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L. G. Darby	1 Aug. 69		LG10S46-2-2 REPORT NO.		

Rev. (A) ADP 1 Aug. 69

 Pages 7.141 and 7.142

PAGES NOT USED

	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
PREPARED	K. Watson	9 Sep 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	VOL. I	7.143	
CHECKED	F. Schreiber	9 Sep 68		MODEL	C-5A	
APPROVED	L. G. Darby	9 Sep 68		LG1US46-2-2	REPORT NO.	

REV ⑧ FES 1-8-71

7.5 FRAMES

The frames located at F.S. 2273.000 and F.S. 2383.472 typify the frames between F.S. 2124.572 and F.S. 2516.664. The loads for these two frames are detailed in this report.

The frame at F.S. 2383.472 is the most critical of the four frames located immediately above the longeron box beam. The primary function of the frame is to react vertical and horizontal kick loads induced by axial loads in the sloping longeron. Internal load distributions for the frame are obtained from FAMAS (Reference 8).

The most critical combination of aft cargo door loads is reacted by the frame at F.S. 2273.000. The frame reacts aft center cargo door radius arm loads and side brace loads. It also reacts aft side cargo door hinge loads. In addition to reacting cargo door loads, this frame also reacts the vertical kick load induced by the change in slope of the torque deck at F.S. 2273.000. Internal load distributions for the frame are obtained utilizing the 2-D Computer Aided Structural Design Computer Program (Reference 17).

Dorsal stability frames from F.S. 2427 through F.S. 2516 are not direct empennage load redistribution frames (these are from F.S. 2538 through F.S. 2735). They do, however, distribute loads from the dorsal area forward of the empennage down into the fuselage. The critical loads for these frames are presented in this report.

	NAME	DATE		TEMP.	PERM.
PREPARED	J.W. Morehead	9 May 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I 7.144
CHECKED	F.E. Schreiber	9 May 68	TITLE AFT FUSELAGE INTERNAL LOADS	MODEL	C-5A
APPROVED	L.G. Darby	9 May 68		LGIUS 46-2-2 REPORT NO. 1	

7.5.1 F.S. 2273.000 Frame

7.5.1.1 Discussion

The frame at F.S. 2273.000 is located at the intersection of the sloping torque deck forward of FS 2273.000 and the horizontal torque deck aft of F.S. 2273.000. The frame is constructed from the sections of two circular rings with the lower ring having the larger radius. Intersecting at WL 314, the sections of the two circular rings form an inverted U-shaped structure with the transverse beam located below the intersection (refer to sketches on the following pages). The lower ring is open at the location of the sloping longeron at WL 225.45.

The frame is manufactured from 7075-T6 aluminum. Caps of the frame are multi-element tee, angle, and J extrusions. The frame web is constructed from sheet metal plus stiffeners on the transverse beam, upper lobe, and on the side frames above WL 265.00. Below WL 265.00 to the sloping longeron at WL 225.45, the frame web is honeycomb.

The external loads on the frame originate from the torque deck, the aft cargo center door, and the aft cargo side doors, and are balanced by shear flow in the skin and along the torque deck. There is no positive internal pressure load as this frame is aft of the pressure bulkhead. A negative internal pressure does exist during high speed flight due to the aft pressure vent. This pressure is small, however, and subtracts from the negative external pressure which exists at F.S. 2273.000 (Ref. 10). The torque deck is loaded axially, similar to the skin and stringers, as the fuselage is subjected to a bending moment, and this axial load produces a vertical kick load at F.S. 2273.000. Air loads on the aft cargo side doors produce the hinge moments that load the side frames. This load is applied to the side frame at the sloping longeron. The aft cargo center door loads are from a jam condition of the door and also air loads. These loads are applied to the transverse beam.

All critical load conditions are defined by a sketch on the following pages. Loads and conditions are tabulated.

Internal bending moments, axial loads, and shears for this frame are obtained utilizing the 2-D computer aided structural design computer program (ref. 17).

PREPARED	NAME A.D. PINCHOT	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	PERM. 7.145
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APPROVED	L.G. DARBY	5-20-69		LGIUS46-2-2 REPORT NO.		

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	NAME	DATE	LOCKHEED-GEOORGIA COMPANY	TEMP.	PERM.
Prepared	K. R. Watson	5-20-69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.145.1
Checked	F.E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved	L. G. Darby	5-20-69		LG1US46-2-2	Report No.

REV. (A) KRW 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273

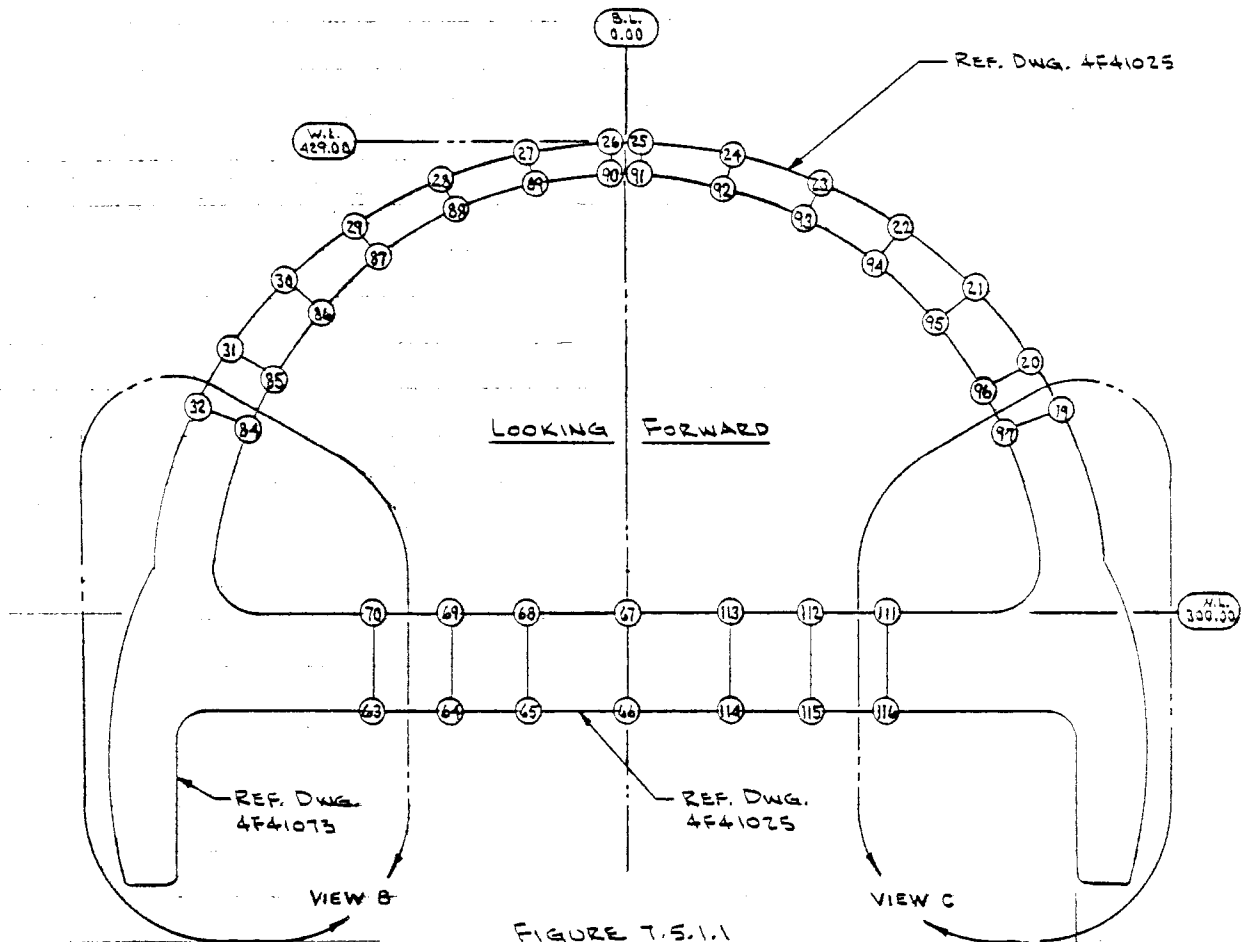


FIGURE 7.5.1.1
F.S. 2273 FRAME MODEL (REF. 17)

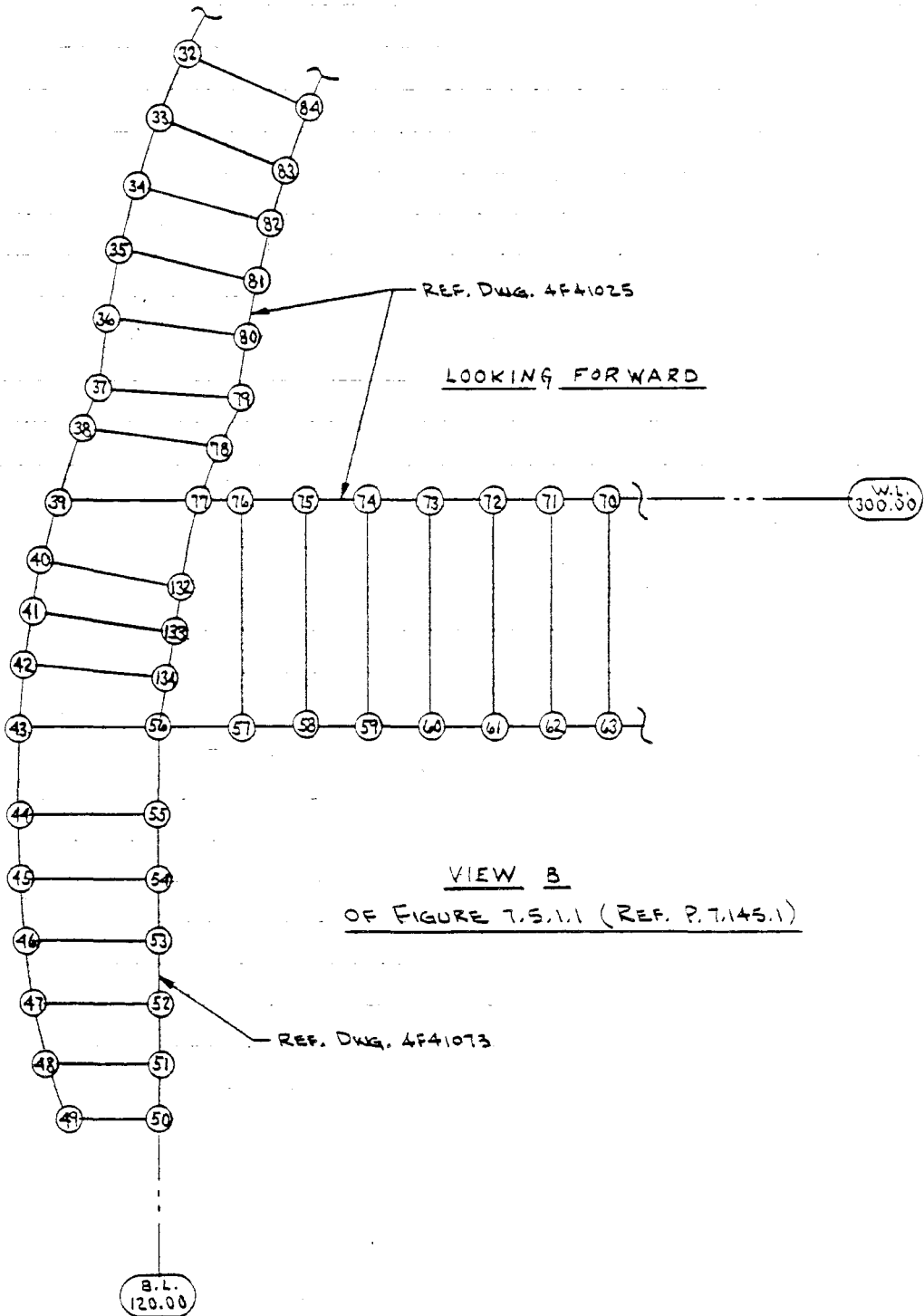
NOTES:

1. SEE P. 7.145.2 FOR VIEW B.
2. SEE P. 7.145.3 FOR VIEW C.
3. NUMBERS ENCIRCLED ARE GRIDPOINT NUMBERS.
4. LINES CONNECTING GRIDPOINT NUMBERS ○—○ ARE AXIAL ELEMENTS.

S.L.
120.00

REV. (A) KRW 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S.2273 (CONTINUED)



Prepared	A. D. Pinchot	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I 7.146.2	PERM.
Checked	F.E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L. G. Darby	5-20-69		Report No.	LG1US46-2-2	

REV. (A) A.D.P. 1 AUG, 69

7.5.1.2 FRAME GEOMETRY - F.S.2273 (CONTINUED)

TABLE 7.5.1.1 (CONT'D.)
FRAME MODEL GRID POINT LOCATIONS (SEE P.7.145.1-7.145.3)

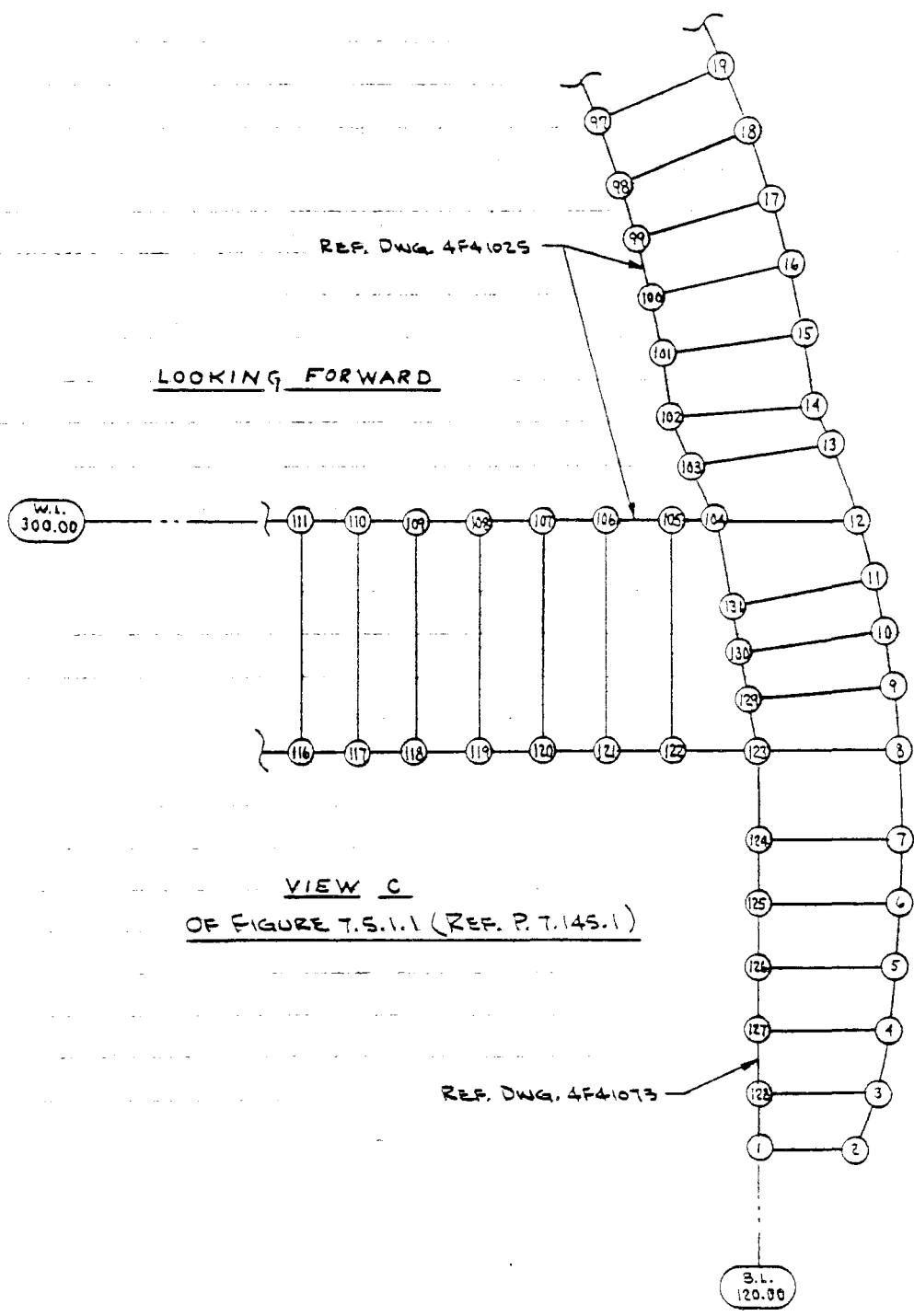
G.P.	B.L.	W.L.
82	-106.34	333.56
83	-104.17	340.48
84	-101.67	347.32
85	-95.75	360.74
86	-83.68	379.52
87	-68.05	396.11
88	-49.08	409.47
89	-27.01	416.91
90	-3.92	420.00
91	3.92	420.00
92	27.01	416.91
93	49.08	409.47
94	68.05	396.11
95	83.68	379.52
96	95.75	360.74
97	101.67	347.32
98	104.17	340.48
99	106.34	333.56

G.P.	B.L.	W.L.
100	108.13	326.56
101	109.54	319.62
102	111.30	310.00
103	114.23	303.65
104	115.05	300.62
105	109.66	300.62
106	101.37	300.62
107	94.29	300.62
108	87.15	300.62
109	80.00	300.62
110	73.40	300.62
111	66.73	300.62
112	46.73	300.62
113	26.73	300.62
114	26.73	273.33
115	46.73	273.33
116	66.73	273.33
117	73.40	273.33

G.P.	B.L.	W.L.
118	80.00	273.33
119	87.15	273.33
120	94.29	273.33
121	101.37	273.33
122	109.66	273.33
123	120.50	273.33
124	120.50	262.50
125	120.50	254.95
126	120.50	247.38
127	120.50	240.00
128	120.50	232.65
129	120.00	279.15
130	119.25	284.63
131	118.15	289.75
132	-118.15	289.75
133	-119.25	284.63
134	-120.00	279.15

REV. (A) KRW 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273 (CONTINUED)



PREPARED	NAME L.D. PINCHOT	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	FORM. 7.146
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APPROVED	L.G. DARBY	5-20-69	AFT FUSELAGE	LG10546-2-2 REPORT NO.		

REV. (3) LDP 1-8-71

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Prepared	NAME A. D. Pinchot	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. 1	PERM. 7.146.1
Checked	F. E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	L. G. Darby	5-20-69				LG:US46-2-2 Report No.	

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - FS. 2273 (CONTINUED)

TABLE 7.5.1.1
FRAME MODEL GRID POINT LOCATIONS (SEE P. 7.145.1 - 7.145.3)

G.P.	B.L.	W.L.	G.P.	B.L.	W.L.	G.P.	B.L.	W.L.
1	121.90	226.10	28	-52.73	417.52	55	-120.50	262.50
2	130.62	226.10	29	-74.42	405.01	56	-120.50	273.33
3	134.20	232.65	30	-93.26	388.51	57	-109.66	273.33
4	135.25	240.00	31	-108.50	368.69	58	-101.37	273.33
5	136.08	247.38	32	-116.38	354.01	59	-94.29	273.33
6	136.58	254.95	33	-119.58	346.32	60	-87.15	273.33
7	136.80	262.50	34	-122.28	338.44	61	-80.00	273.33
8	136.80	273.33	35	-124.45	330.41	62	-73.40	273.33
9	136.08	280.70	36	-126.07	322.43	63	-66.73	273.33
10	135.20	286.96	37	-127.21	314.00	64	-46.73	273.33
11	134.03	292.96	38	-129.16	309.21	65	-26.73	273.33
12	132.21	300.18	39	-132.21	300.18	66	0.0	273.33
13	129.16	309.21	40	-134.03	292.96	67	0.0	300.62
14	127.21	314.00	41	-135.20	286.96	68	-26.73	300.62
15	126.07	322.43	42	-136.08	280.70	69	-46.73	300.62
16	124.45	330.41	43	-136.80	273.33	70	-66.73	300.62
17	122.28	338.44	44	-136.80	262.50	71	-73.40	300.62
18	119.58	346.32	45	-136.58	254.95	72	-80.00	300.62
19	116.38	354.01	46	-136.08	247.38	73	-87.15	300.62
20	108.50	368.69	47	-135.25	240.00	74	-94.29	300.62
21	93.26	388.51	48	-134.20	232.65	75	-101.37	300.62
22	74.42	405.01	49	-130.62	226.10	76	-109.66	300.62
23	52.73	417.52	50	-121.90	226.10	77	-115.05	300.62
24	29.03	425.56	51	-120.50	232.65	78	-114.23	303.65
25	4.21	428.83	52	-120.50	240.00	79	-111.30	310.00
26	-4.21	428.83	53	-120.50	247.38	80	-109.54	319.62
27	-29.03	425.56	54	-120.50	254.95	81	-108.13	326.56

PREPARED	NAME A.D. PINCHOT	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	FORM. 7.147
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APPROVED	L.G. DAZBY	5-20-69		LG10546-2-2 REPORT NO.		

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Approved	L. G. Darby	5-20-69		Report No.	LG10546-2-2	

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273 (CONTINUED)

TABLE 7.5.1.2
AXIAL ELEMENT AREAS (SEE P.7.145.1 - 7.145.3)

AXIAL ELEM.	GRID PTS.	A ₁ IN. ²	A ₂ IN. ²	AXIAL ELEM.	GRID PTS.	A ₁ IN. ²	A ₂ IN. ²	AXIAL ELEM.	GRID PTS.	A ₁ IN. ²	A ₂ IN. ²
1	1-2	4.000	4.000	26	26-27	1.173	1.173	51	51-52	1.900	1.310
2	2-3	4.736	2.487	27	27-28	1.173	1.173	52	52-53	1.310	1.173
3	3-4	2.487	0.781	28	28-29	1.173	1.173	53	53-54	1.173	0.960
4	4-5	0.781	0.781	29	29-30	1.173	1.173	54	54-55	0.960	1.750
5	5-6	0.781	0.781	30	30-31	1.173	1.173	55	55-56	1.750	0.720
6	6-7	0.781	0.781	31	31-32	1.173	1.173	56	56-57	1.500	1.500
7	7-8	0.781	1.229	32	32-33	1.173	1.173	57	57-58	1.500	1.830
8	8-9	1.229	1.229	33	33-34	1.173	1.173	58	58-59	1.830	2.112
9	9-10	1.229	1.442	34	34-35	1.173	1.173	59	59-60	2.112	2.281
10	10-11	1.442	1.327	35	35-36	1.173	1.173	60	60-61	2.281	2.340
11	11-12	1.327	1.212	36	36-37	1.212	1.212	61	61-62	2.340	2.390
12	12-13	1.212	1.212	37	37-38	1.212	1.212	62	62-63	2.390	2.493
13	13-14	1.212	1.212	38	38-39	1.212	1.212	63	63-64	2.493	2.442
14	14-15	1.212	1.212	39	39-40	1.212	1.327	64	64-65	2.442	2.757
15	15-16	1.173	1.173	40	40-41	1.327	1.442	65	65-66	2.757	2.341
16	16-17	1.173	1.173	41	41-42	1.442	1.229	66	66-67	0.337	0.337
17	17-18	1.173	1.173	42	42-43	1.229	1.229	67	67-68	1.412	1.232
18	18-19	1.173	1.173	43	43-44	1.229	0.781	68	68-69	1.232	1.185
19	19-20	1.173	1.173	44	44-45	0.781	0.781	69	69-70	1.185	1.088
20	20-21	1.173	1.173	45	45-46	0.781	0.781	70	70-71	1.088	1.056
21	21-22	1.173	1.173	46	46-47	0.781	0.781	71	71-72	1.056	1.008
22	22-23	1.173	1.173	47	47-48	0.781	2.487	72	72-73	1.008	0.989
23	23-24	1.173	1.173	48	48-49	2.487	4.736	73	73-74	0.989	0.955
24	24-25	1.173	1.173	49	49-50	4.000	4.000	74	74-75	0.955	0.918
25	25-26	1.173	1.173	50	50-51	3.520	1.900	75	75-76	0.918	0.851

△ A₁ = AREA AT FIRST GRID POINT
 A₂ = AREA AT SECOND GRID POINT
 AREA TAPERS LINEARLY BETWEEN GRID POINTS.

Checked
F.E.Schreiber

5-20-69

TITLE
INTERNAL LOADS ANALYSIS
AFT FUSELAGE

Model
C-5A

Approved
L. G. Darby

5-20-69

Report No.
LG10S46-2-2

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273 (CONTINUED)

TABLE 7.5.1.2 (CONT'D.)
AXIAL ELEMENT AREAS (SEE P. 7.145.1 - 7.145.3)

AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²	AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²	AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²
76	76-77	0.851	0.851	101	101-102	0.913	0.975	126	126-127	1.175	1.310
77	77-78	1.112	1.050	102	102-103	0.985	1.050	127	127-128	1.310	1.900
78	78-79	1.050	0.985	103	103-104	1.030	1.112	128	1-128	3.520	1.900
79	79-80	0.975	0.913	104	104-105	0.851	0.851	129	3-128	1.023	1.023
80	80-81	0.913	0.852	105	105-106	0.851	0.913	130	4-127	0.363	0.363
81	81-82	0.852	0.790	106	106-107	0.913	0.955	131	5-126	0.374	0.374
82	82-83	0.790	0.729	107	107-108	0.955	0.989	132	6-125	0.378	0.378
83	83-84	0.729	0.670	108	108-109	0.989	1.008	133	7-124	0.460	0.460
84	84-85	0.670	0.493	109	109-110	1.008	1.056	134	8-123	1.500	1.500
85	85-86	0.493	0.423	110	110-111	1.056	1.088	135	9-129	0.223	0.223
86	86-87	0.423	0.423	111	111-112	1.088	1.185	136	10-130	0.223	0.223
87	87-88	0.422	0.423	112	112-113	1.185	1.282	137	11-131	0.223	0.223
88	88-89	0.413	0.495	113	113-114	0.252	0.252	138	12-104	0.351	0.351
89	89-90	0.488	0.488	114	114-115	2.757	2.442	139	13-103	0.211	0.211
90	90-91	0.488	0.488	115	115-116	2.442	2.493	140	14-102	0.682	0.682
91	91-92	0.488	0.488	116	116-117	2.493	2.390	141	15-101	0.158	0.158
92	92-93	0.495	0.413	117	117-118	2.390	2.340	142	16-100	0.158	0.158
93	93-94	0.423	0.422	118	118-119	2.340	2.231	143	17-99	0.158	0.158
94	94-95	0.423	0.423	119	119-120	2.231	2.112	144	18-98	0.158	0.158
95	95-96	0.423	0.493	120	120-121	2.112	1.830	145	19-97	0.158	0.158
96	96-97	0.493	0.670	121	121-122	1.830	1.500	146	20-96	0.158	0.158
97	97-98	0.670	0.729	122	122-123	1.500	1.500	147	21-95	0.158	0.158
98	98-99	0.729	0.790	123	123-124	0.720	1.750	148	22-94	0.158	0.158
99	99-100	0.790	0.852	124	124-125	1.750	0.960	149	23-93	0.151	0.151
100	100-101	0.852	0.913	125	125-126	0.960	1.175	150	24-92	0.117	0.117

△ A₁ = AREA AT FIRST GRID POINT
A₂ = AREA AT SECOND GRID POINT
AREA TAPERS LINEARLY BETWEEN GRID POINTS

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Checked	F.E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Page	Model C-5A
Approved	L. G. Darby	5-20-69		Report No.	IG1US-16-2-2

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - P.S. 2273 (CONTINUED)

TABLE 7.5.1.2 (CONT'D.)
AXIAL ELEMENT AREAS (SEE P.7.145.1 - 7.145.3)

AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²
151	25-91	0.109	0.109
152	26-90	0.109	0.109
153	27-89	0.117	0.117
154	28-88	0.151	0.151
155	29-87	0.158	0.158
156	30-86	0.158	0.158
157	31-85	0.158	0.158
158	32-84	0.158	0.158
159	33-83	0.158	0.158
160	34-82	0.158	0.158
161	35-81	0.158	0.158
162	36-80	0.158	0.158
163	37-79	0.682	0.682
164	38-78	0.211	0.211
165	39-77	0.851	0.851
166	40-76	0.223	0.223
167	41-75	0.223	0.223

AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²
168	42-74	0.223	0.223
169	43-73	1.500	1.500
170	44-72	0.460	0.460
171	45-71	0.378	0.378
172	46-70	0.374	0.374
173	47-69	0.368	0.368
174	48-68	1.023	1.023
175	49-67	0.417	0.417
176	50-66	0.401	0.401
177	51-65	0.401	0.401
178	52-64	0.401	0.401
179	53-63	0.423	0.423
180	54-62	0.276	0.276
181	55-61	0.276	0.276
182	56-60	0.276	0.276
183	57-59	0.252	0.252
184	58-58	0.276	0.276

AXIAL ELEM.	GRID PTS.	A ₁ IN ²	A ₂ IN ²
185	111-116	0.276	0.276
186	110-117	0.276	0.276
187	109-118	0.423	0.423
188	108-119	0.401	0.401
189	107-120	0.401	0.401
190	106-121	0.401	0.401
191	105-122	0.417	0.417
192	67-113	1.412	1.232
193	66-114	2.341	2.757
194	123-129	0.720	1.070
195	129-130	1.070	1.310
196	130-131	1.310	1.250
197	104-131	1.190	1.250
198	56-134	0.720	1.070
199	133-134	1.070	1.310
200	132-133	1.310	1.250
201	77-132	1.190	1.250

△ A₁ = AREA AT FIRST GRID POINT

A₂ = AREA AT SECOND GRID POINT

AREA TAPERS LINEARLY BETWEEN GRID POINTS.

PREPARED	NAME A.D. Pincot	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	PERM. 7.148
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APPROVED	L.G. DARBY	5-20-69		LG10546-2-2 REPORT NO.		

REV. (3) ADP 1-8-71

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Prepared	NAME A. D. Pinchot	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I 7.145.1
Checked	F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-51
Approved	L. G. Darby	5-20-69		Report No.	LG1US46-2-2

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S.2273 (CONTINUED)

TABLE 7.5.1.3
SHEAR PANEL THICKNESSES (SEE P.7.145.1 - 7.145.3)

PANEL NO.	THK.	CONNECTING GRID PTS.			
1	0.100	1	128	3	2
2	0.050	3	128	127	4
3	0.050	4	127	126	5
4	0.050	5	126	125	6
5	0.050	6	125	124	7
6	0.050	7	124	123	8
7	0.096	8	123	129	9
8	0.096	9	129	130	10
9	0.096	10	130	131	11
10	0.096	11	131	104	12
11	0.081	12	104	103	13
12	0.081	13	103	102	14
13	0.081	14	102	101	15
14	0.081	15	101	100	16
15	0.081	16	100	99	17
16	0.081	17	99	98	18
17	0.081	18	98	97	19
18	0.081	19	97	96	20
19	0.081	20	96	95	21
20	0.081	21	95	94	22
21	0.072	22	94	93	23
22	0.060	23	93	92	24
23	0.049	24	92	91	25
24	0.049	25	91	90	26
25	0.049	26	90	89	27

PANEL NO.	THK.	CONNECTING GRID PTS.			
26	0.060	27	39	88	28
27	0.072	28	88	87	29
28	0.081	29	87	86	30
29	0.081	30	86	85	31
30	0.081	31	85	84	32
31	0.081	32	84	83	33
32	0.081	33	83	82	34
33	0.081	34	82	81	35
34	0.081	35	81	80	36
35	0.081	36	80	79	37
36	0.081	37	79	78	38
37	0.081	38	78	77	39
38	0.096	39	77	132	40
39	0.096	40	132	133	41
40	0.096	41	133	134	42
41	0.096	42	134	56	43
42	0.050	43	56	55	44
43	0.050	44	55	54	45
44	0.050	45	54	53	46
45	0.050	46	53	52	47
46	0.050	47	52	51	48
47	0.100	48	51	50	49
48	0.096	56	77	76	57
49	0.096	57	76	75	58
50	0.096	58	75	74	59

	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
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Approved	L. C. Darby	5-20-69			IG1US46-2-2 Report No.	

REV. (A) A.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273 (CONTINUED)

TABLE 7.5.1.3 (CONT'D.)

SHEAR PANEL THICKNESSES (SEE P. 7.145.1 - 7.145.3)

PANEL NO.	THK.	CONNECTING GRID PTS.			
51	0.096	59	74	73	60
52	0.096	60	73	72	61
53	0.054	61	72	71	62
54	0.054	62	71	70	63
55	0.054	63	70	69	64
56	0.057	64	69	68	65
57	0.054	65	68	67	66
58	0.054	66	67	113	114
59	0.054	112	115	114	113

PANEL NO.	THK.	CONNECTING GRID PTS.			
60	0.054	111	116	115	112
61	0.054	110	117	116	111
62	0.054	109	118	117	110
63	0.096	108	119	118	109
64	0.096	107	120	119	108
65	0.096	106	121	120	107
66	0.096	105	122	121	106
67	0.096	104	123	122	105

Prepared	A. D. Pinchot	5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Val. I	PERM. 7.149
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REV. (A) L.D.P. 1 AUG. 69

7.5.1.2 FRAME GEOMETRY - F.S. 2273 (CONTINUED)

TABLE 7.5.1.4

SHELL SUPPORT SPECIFICATION (SEE P. 7.145.1 - 7.145.3)

LENGTH = 200.0 IN.

SHELL NO.	GRID PTS.		THK.
1	1	2	3.630
2	2	3	0.070
3	3	4	0.070
4	4	5	0.070
5	5	6	0.070
6	6	7	0.070
7	7	8	0.070
8	8	9	0.093
9	9	10	0.093
10	10	11	0.093
11	11	12	0.093
12	12	13	0.093
13	13	14	0.093
14	14	15	0.098
15	15	16	0.098
16	16	17	0.098
17	17	18	0.098
18	18	19	0.093
19	19	20	0.098
20	20	21	0.098
21	21	22	0.098
22	22	23	0.098
23	23	24	0.098
24	24	25	0.098

SHELL NO.	GRID PTS.		THK.
25	25	26	0.098
26	26	27	0.098
27	27	28	0.098
28	28	29	0.098
29	29	30	0.098
30	30	31	0.098
31	31	32	0.098
32	32	33	0.098
33	33	34	0.098
34	34	35	0.098
35	35	36	0.098
36	36	37	0.098
37	37	38	0.093
38	38	39	0.093
39	39	40	0.093
40	40	41	0.093
41	41	42	0.093
42	42	43	0.093
43	43	44	0.070
44	44	45	0.070
45	45	46	0.070
46	46	47	0.070
47	47	48	0.070
48	48	49	0.070

SHELL NO.	GRID PTS.		THK.
49	49	50	3.630
50	43	56	0.040
51	56	57	0.040
52	57	58	0.040
53	58	59	0.040
54	59	60	0.040
55	60	61	0.040
56	61	62	0.040
57	62	63	0.040
58	63	64	0.040
59	64	65	0.040
60	65	66	0.040
61	66	114	0.040
62	114	115	0.040
63	115	116	0.040
64	116	117	0.040
65	117	118	0.040
66	118	119	0.040
67	119	120	0.040
68	120	121	0.040
69	121	122	0.040
70	122	123	0.040
71	3	23	0.040

	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
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CHECKED	F.E. SCHREIBER	5-20-69	TITLE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. DARBY	5-20-69	LEFT FUSELAGE	LG10546-2-2	REPORT NO.	

REV. (B) ADP 1-8-71

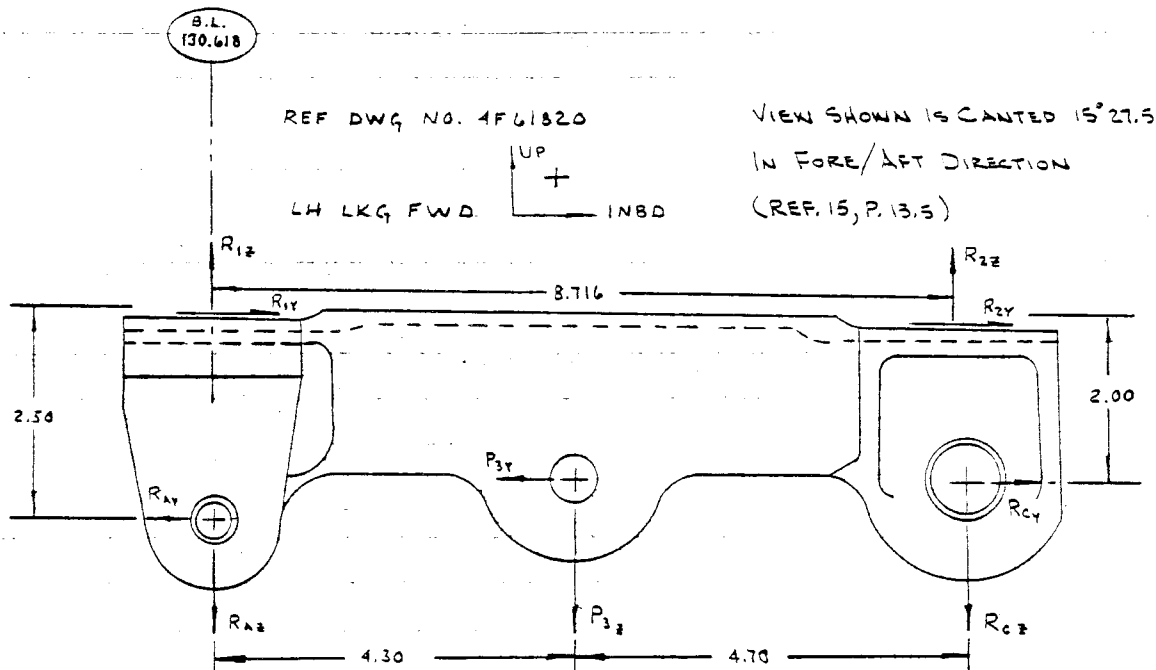
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Prepared	K. R. Watson	5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I	7.150.1
Checked	F. E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model	C-5A
Approved	L. G. Darby	5-20-69				IG1US46-2-2	Report No.

REV (A) KRW 1 AUG. 69

T.S.1.3 FRAME LOADS - F.S.2273

CALCULATION OF SIDE CARGO DOOR LOADS



$$R_{2z} = (-2.00R_{cy} + 9.00R_{cz} + 2.00P_{3y} + 2.50R_{ay} + 4.30P_{3z}) / 8.716 \quad \text{NOTES: } \triangleright \text{ LEFT HAND}$$

(REF. 15, P. 13.21) $\triangleright$ RIGHT HAND

$$R_{1z} = (2.00R_{cy} - 2.84R_{cz} - 2.00P_{3y} - 2.50R_{ay} + 8.716R_{az} + 4.416P_{3z}) / 8.716 \quad \triangleright \text{ REF. 15, P. 13.4}$$

(REF. 15, P. 13.21) $\triangleright$ REF. 15, P. 13.4

$$R_{1y} + R_{2y} = -R_{cy} + P_{3y} + R_{ay}; \text{ ASSUMPTION: } R_{1y} = R_{2y} = (-R_{cy} + P_{3y} + R_{ay}) / 2. \quad \triangleright \theta = 15^\circ 27.5';$$

REF. 15, P. 13.5

TABLE T.S.1.5 - SIDE CARGO DOOR LOADS - L.H. SIDE

LOAD CASE	R_{cy}	R_{cz}	P_{3y}	P_{3z}	R_{ay}	R_{az}	R_{2z}	R_{1z}	$R_{1y} \mp R_{2y}$	$\frac{R_{2z}}{\cos \theta}$	$\frac{R_{1z}}{\cos \theta}$
1	19162.	31829.	4111.	1233.	12678.	-20813.	33657.	-21408.	-1187.	34922.	-22213.
2	12937.	21488.	3140.	942.	8560.	-14051.	22860.	-14431.	-619.	23719.	-15025.
3	9945.	16518.	1950.	584.	6580.	-10801.	17397.	-11096.	-708.	18051.	-11513.
4	-12561.	-20864.	-2297.	-688.	-8311.	13643.	-21912.	14003.	977.	-22736.	14529.

TABLE T.S.1.6 - SIDE CARGO DOOR LOADS - R.H. SIDE

LOAD CASE	R_{cy}	R_{cz}	P_{3y}	P_{3z}	R_{ay}	R_{az}	R_{2z}	R_{1z}	$R_{1y} \mp R_{2y}$	$\frac{R_{2z}}{\cos \theta}$	$\frac{R_{1z}}{\cos \theta}$
1	12150.	20182.	3067.	920.	8039.	-13197.	21515.	-13610.	-522.	22324.	-14122.
2	18155.	30156.	4049.	1215.	12012.	-19719.	31947.	-20295.	-1047.	33147.	-21053.
3	1472.	2444.	749.	224.	974.	-1598.	2743.	-1678.	126.	2952.	-1742.
4	-12561.	-20864.	-2297.	-688.	-8311.	13643.	-21912.	14003.	977.	-22736.	14529.

REV. (A) A.D.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S.2273 (CONTINUED)

TABLE 7.5.1.7
CROSS BEAM & TORQUE DECK REACTIVE LOADS
FROM CENTER DOOR DRIVER ARM & SIDE BRACE

$\sin \theta = \sin 15^{\circ} 27' = .26640$
 $\cos \theta = \cos 15^{\circ} 27' = .96386$

$\sin \phi = \sin 13^{\circ} 31' 44'' = .23394$
 $\cos \phi = \cos 13^{\circ} 31' 44'' = .97225$

LOAD CASE	P.7.145.2 & P.7.145.3		REF. 15, P. 13.4		$P_x \cos \theta$	$P_z \sin \theta$	$\cos \phi$	R_T	$P_z \cos \theta$	$P_x \sin \theta$	$R_{T \sin \theta}$	R_B
	G.P.	P_x	P_z									
1	57	64	718		61.7	191.3	.97225	-133.	692.1	17.1	-31.1	678.
	61	-458	-7040		-441.5	-1875.5		1475.	-6785.6	-122.	345.1	-6563.
	118	36	-3182		34.7	-847.7		908.	-3067.	9.6	212.4	-2345.
2	57	-1080	-12,104		-1041.	-3224.5	.97225	2246.	-11,666.6	-287.7	525.4	-11,429.
	61	450	5188		433.7	1382.1		-976.	5000.1	119.9	-223.3	4392.
	118	-313	-5638		-301.7	-1502.		1235.	-5434.2	-83.4	288.9	-5229.
3	57	477	5347		459.8	1424.4	.97225	-992.	5153.8	127.1	-232.1	5049.
	61	-1022	-9444		-985.1	-2515.9		1575.	-9102.7	-272.3	368.5	-9007.
	118	-252	-2043		-242.9	-544.3		310.	-1969.2	-67.1	72.5	-1964.

FOR JAM CASES; $\theta = 10.53^{\circ}$, $P_x = -R_{Az}'$ & $P_z = R_{Ax}'$ (REF. 15, P. 13.40 & P. 13.42)

5	118	-5010.5	13,975.8	-4926.1	2554.1	.97225	-7694.	13,740.5	-915.7	-1799.9	11,025.
6	118	5010.5	-13,975.8	4926.1	-2554.1	.97225	7694.	-13,740.5	915.7	1799.9	-11,025.

1 APPLIED LOAD P_y (REF. 15, P. 13.4) IS REACTED Laterally BY CROSS BEAM & IS IN THE DIRECTION OF NORMAL FUSELAGE COORDINATES. THEREFORE, NO FURTHER CALCULATIONS ARE NECESSARY.

PREPARED	A.D. PINCHOT	5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	VOL. I	7,151
CHECKED	F.E. SCHREIBER	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L.G. DARBY	5-20-69		LG10546-2-2 REPORT NO.		

REV. (B) LDP 1-8-71

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REV (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED TO THE FRAME IN THE DIRECTION SHOWN.

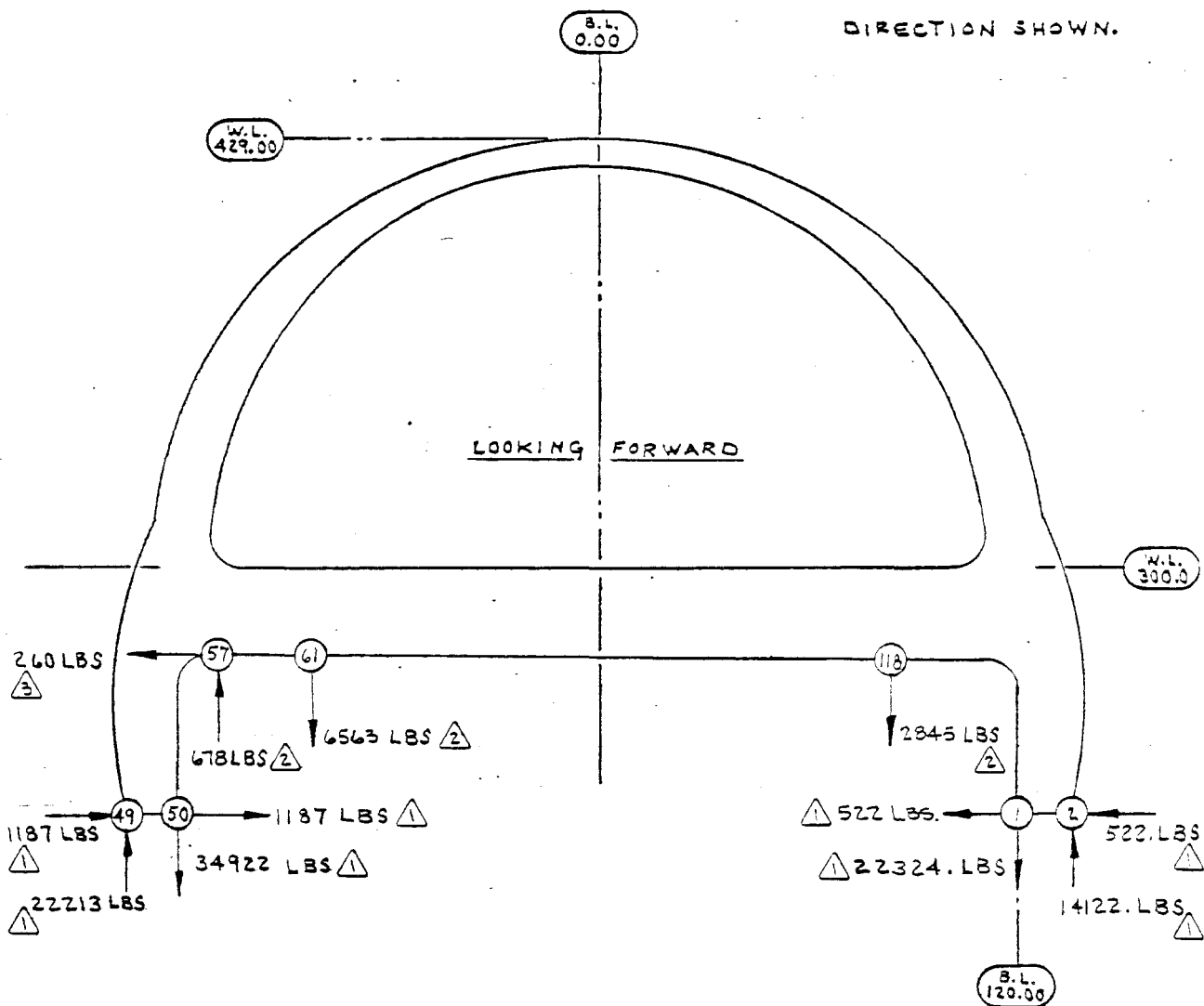


FIGURE 7.5.1.2
LOAD CASE NO. 1
POSITIVE YAW - CARGO DOOR LOADS

- ① REF. P. T. 150.1
- ② REF. P. T. 150.3
- ③ REF. 15, P. 13.4

Prepared	NAME K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. No L.T	PERM. 7.151.2
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Approved	L. G. Darby	5-20-69		LG1US46-2-2 Report No.	

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED TO THE FRAME IN THE DIRECTION SHOWN.

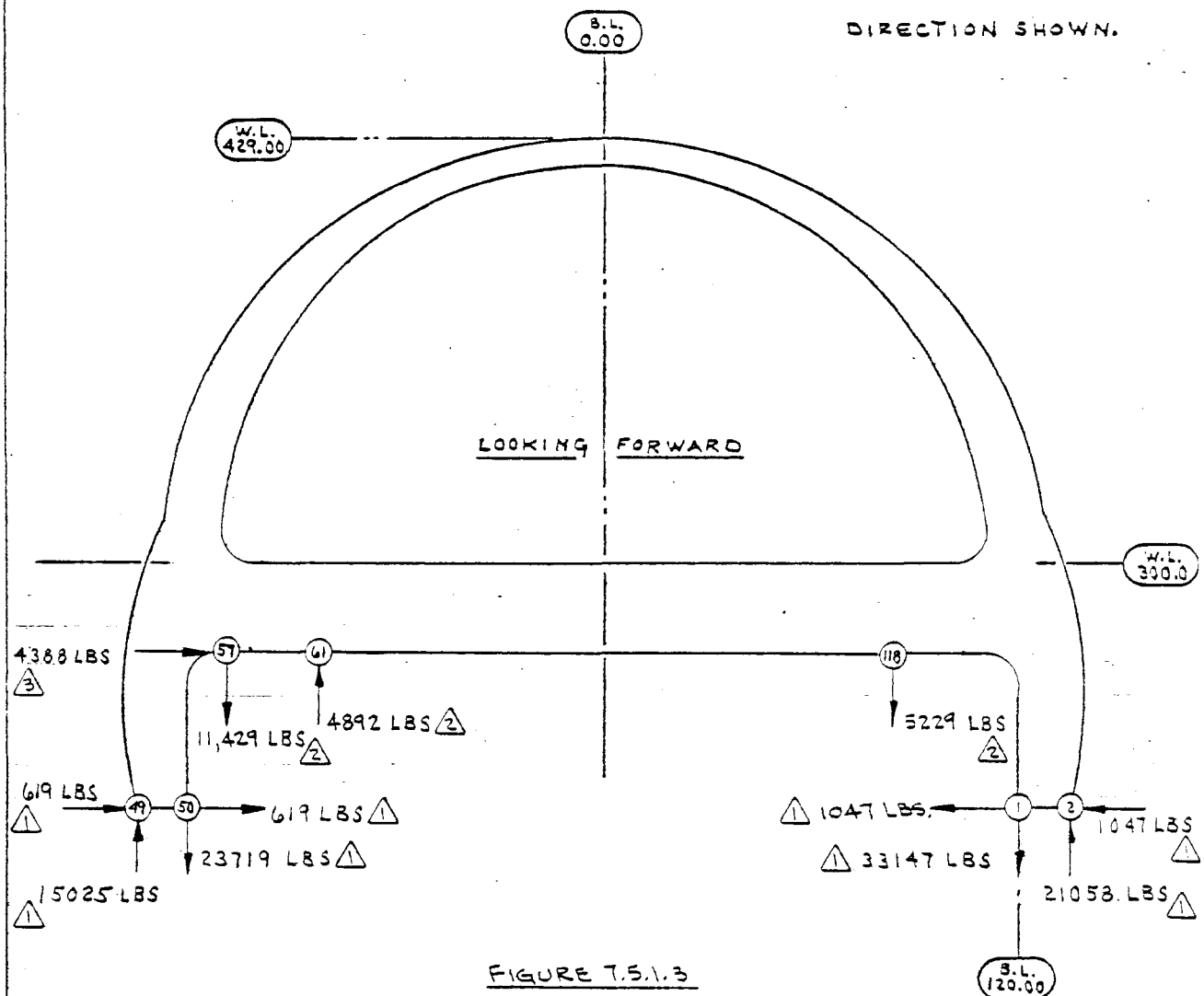
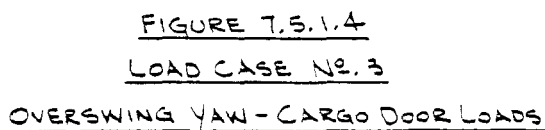


FIGURE 7.5.1.3
LOAD CASE NO. 2
NEGATIVE YAW - CARGO DOOR LOADS

- ① REF. P. T.150.1
- ② REF. P. T.150.3
- ③ REF. 15, P.13.4

REV. (A) KRW 1 AUG. 69

NOTE: ALL LOADS ARE APPLIED
TO THE FRAME IN THE
DIRECTION SHOWN.



-  REF. P. 7.150.1
 REF. P. 7.150.3
 REF. 15, P. 13.4

Prepared	NAME K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I
Checked	F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	
Approved	L. G. Darby	5-20-69		LG1US46-2-2 Report No.	

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S.2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED TO THE FRAME IN THE DIRECTION SHOWN.

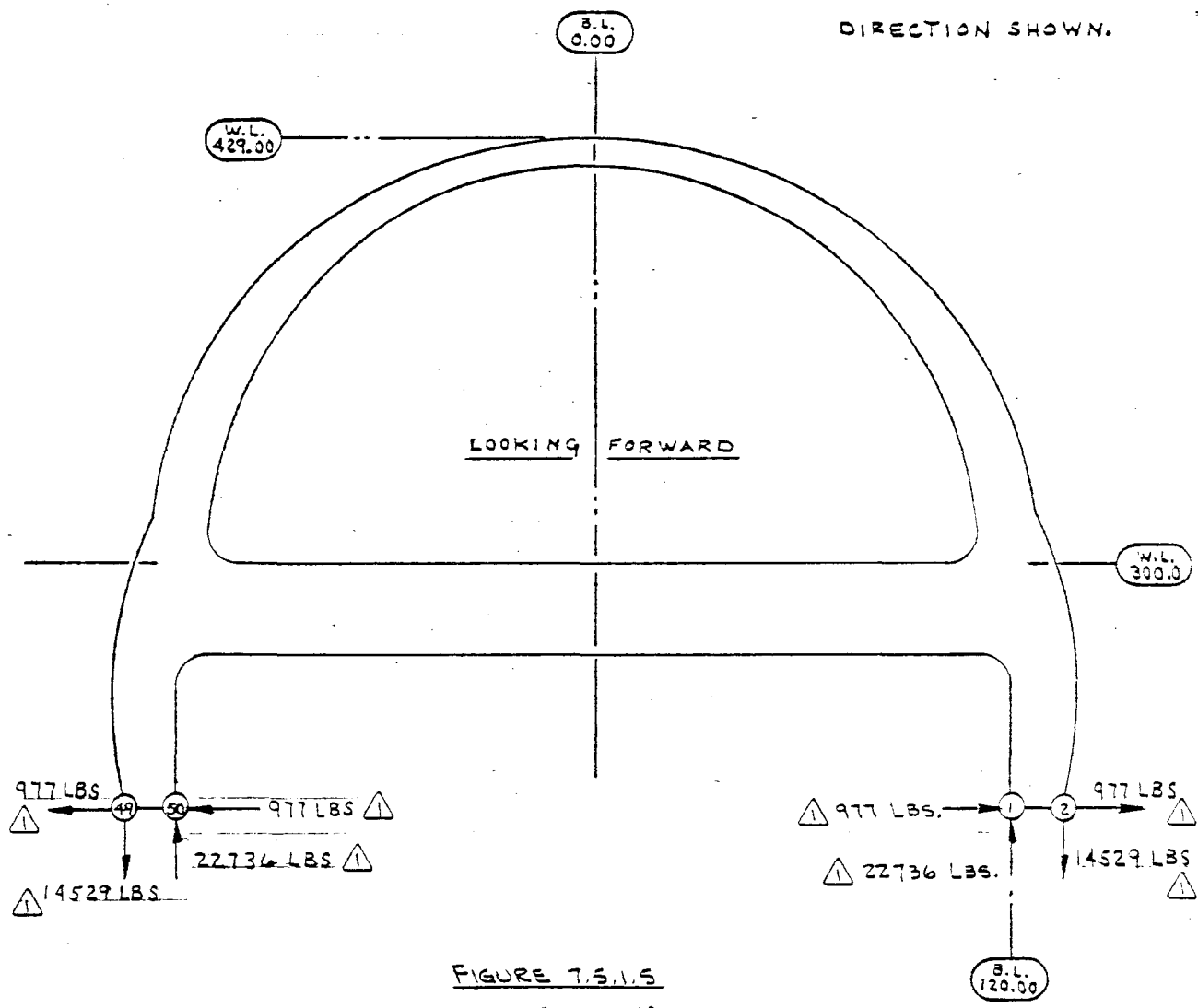


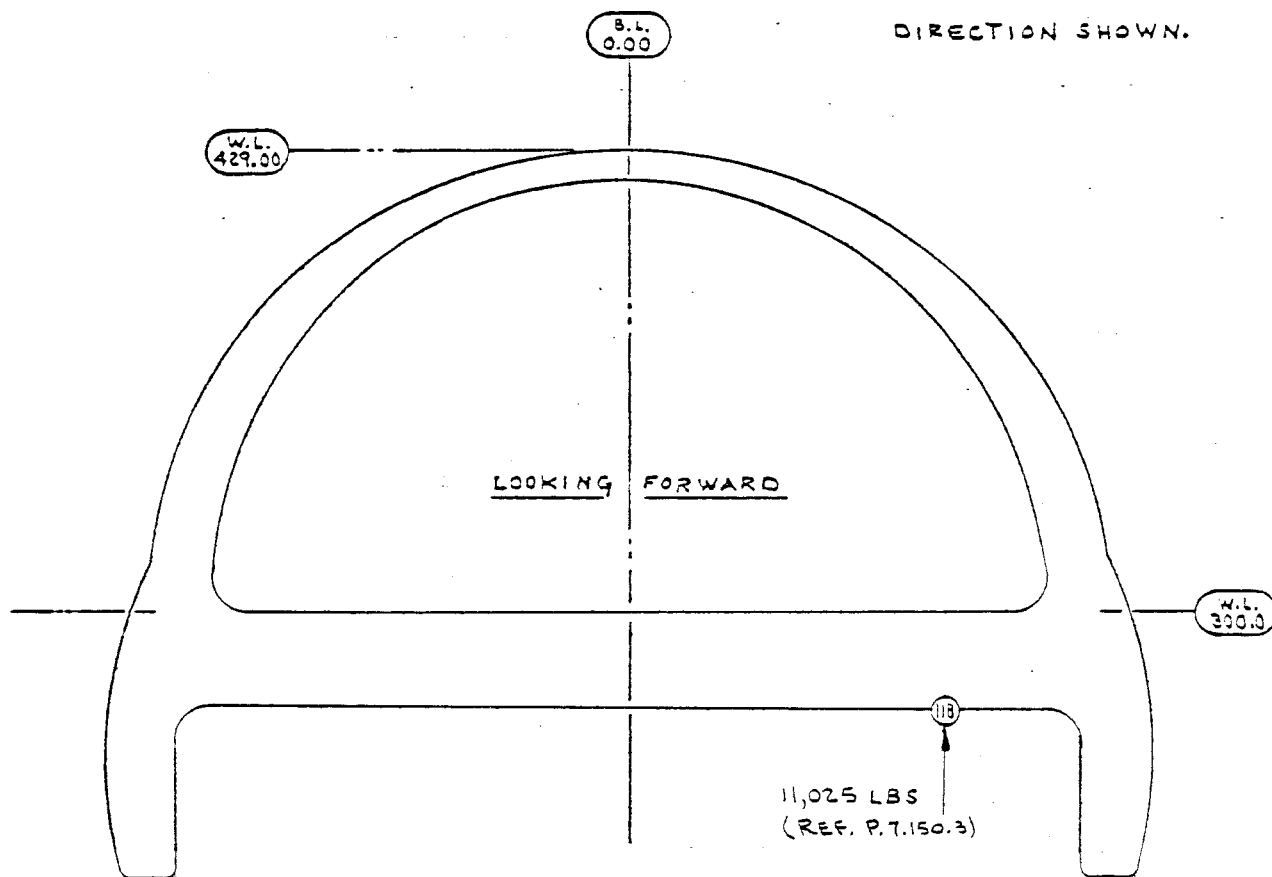
FIGURE 7.5.1.5
LOAD CASE NO. 4
CRUSHING PRESSURE - CARGO DOOR LOADS

Prepared	NAME K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I 7.151.5
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Approved	L. G. Darby	5-20-69		Report No.	LG10546-2-2

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7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED
TO THE FRAME IN THE
DIRECTION SHOWN.



THE CENTER CARGO DOOR MAY JAM IN THE CLOSED POSITION WHEN ATTEMPTING TO LOCK OR UNLOCK THE DOOR. THE JACKSCREW WHICH OPERATES THE DRIVER ARM IS DESIGNED TO REACT ALL THE OUTPUT TORQUE OF THE HYDRAULIC MOTOR ON ONE SIDE ONLY WHEN THE DOOR JAMS. (REF. 15, P. 13.41)

FIGURE 7.5.1.6

LOAD CASE NO. 5

CENTER CARGO DOOR ACTUATOR MOTOR JAM

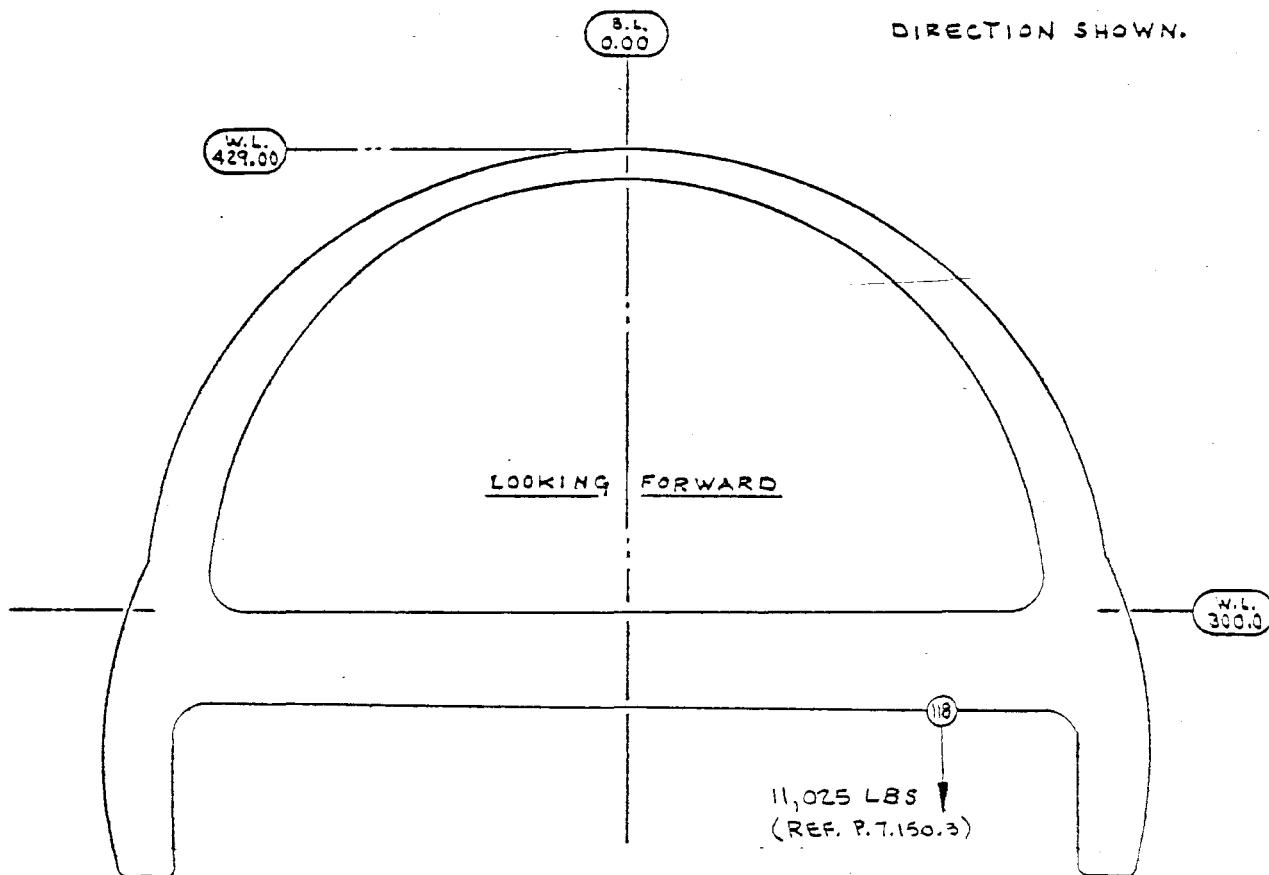
MAX. JACKSCREW TENSION

Prepared	NAME K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PERM. T-51.6
Checked	F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A		
Approved	L. G. Darby	5-20-69		LGTUS46-2-2 Report No.		

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S.2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED
TO THE FRAME IN THE
DIRECTION SHOWN.



THE CENTER CARGO DOOR MAY JAM IN THE CLOSED POSITION WHEN ATTEMPTING TO LOCK OR UNLOCK THE DOOR. THE JACKSCREW WHICH OPERATES THE DRIVER ARM IS DESIGNED TO REACT ALL THE OUTPUT TORQUE OF THE HYDRAULIC MOTOR ON ONE SIDE ONLY WHEN THE DOOR JAMS. (REF. 15, P.13.41)

FIGURE 7.5.1.7

LOAD CASE NO. 6

CENTER CARGO DOOR ACTUATOR MOTOR JAM

MAX. JACKSCREW COMPRESSION

PREPARED	NAME A.D. PINCHOT	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	PERM. 7.152
CHECKED	F.E. SCHREIBER	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	L.G. DARBY	5-20-69		LG10546-2-2 REPORT NO.		

REV. (B) ADP 1-8-71

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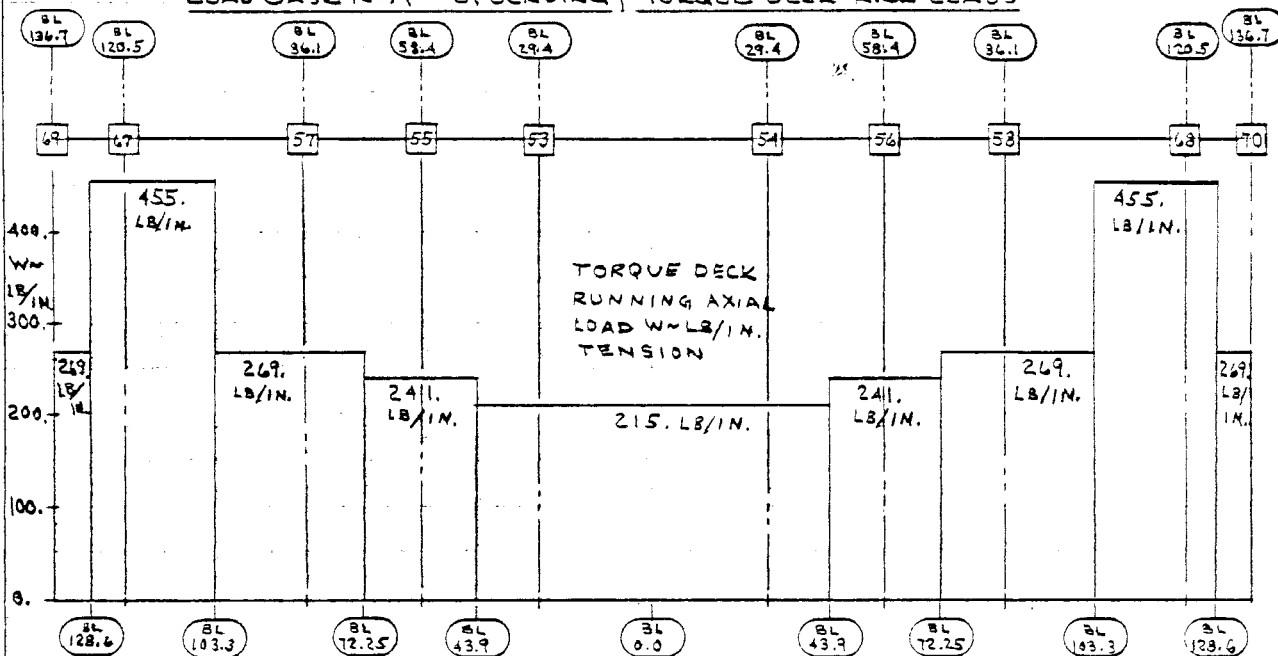
Prepared	K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP. Page	PERM. Vol. I 7.152.1
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REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

CALCULATION OF MAX. TORQUE DECK TENSION LOAD (REF. P. 7.196) FOR

LOAD CASE NO. 7 - UPBENDING, TORQUE DECK KICK LOADS



CRITICAL TENS. LOAD AT STGR 69 = $\frac{(136.7 - 120.5)}{2} \times 0.0375 (10151.) = 2126. \text{ LBS}$
1.45 (REF P 7.196)

CRITICAL TENS. LOAD AT STGR 70 = $\frac{(136.7 - 120.5)}{2} \times 0.0375 (10376.) = 2173. \text{ LBS}$
1.45

(REF P. 7.196)

STGR	CRIT. TEN. LOAD	SPAN	W-LB/IN
69	2126.	8.10	263.
67	11498.	25.30	455. ▴
57	8115.	31.05	262.
55	6420.	28.35	227.
53	8892.	43.90	203.

STGR	CRIT. TEN. LOAD	SPAN	W-LB/IN
54	9428.	43.90	215. ▴
56	6826.	28.35	241. ▴
58	8336.	31.05	269. ▴
68	10077.	25.30	399.
70	2173.	8.10	269. ▴

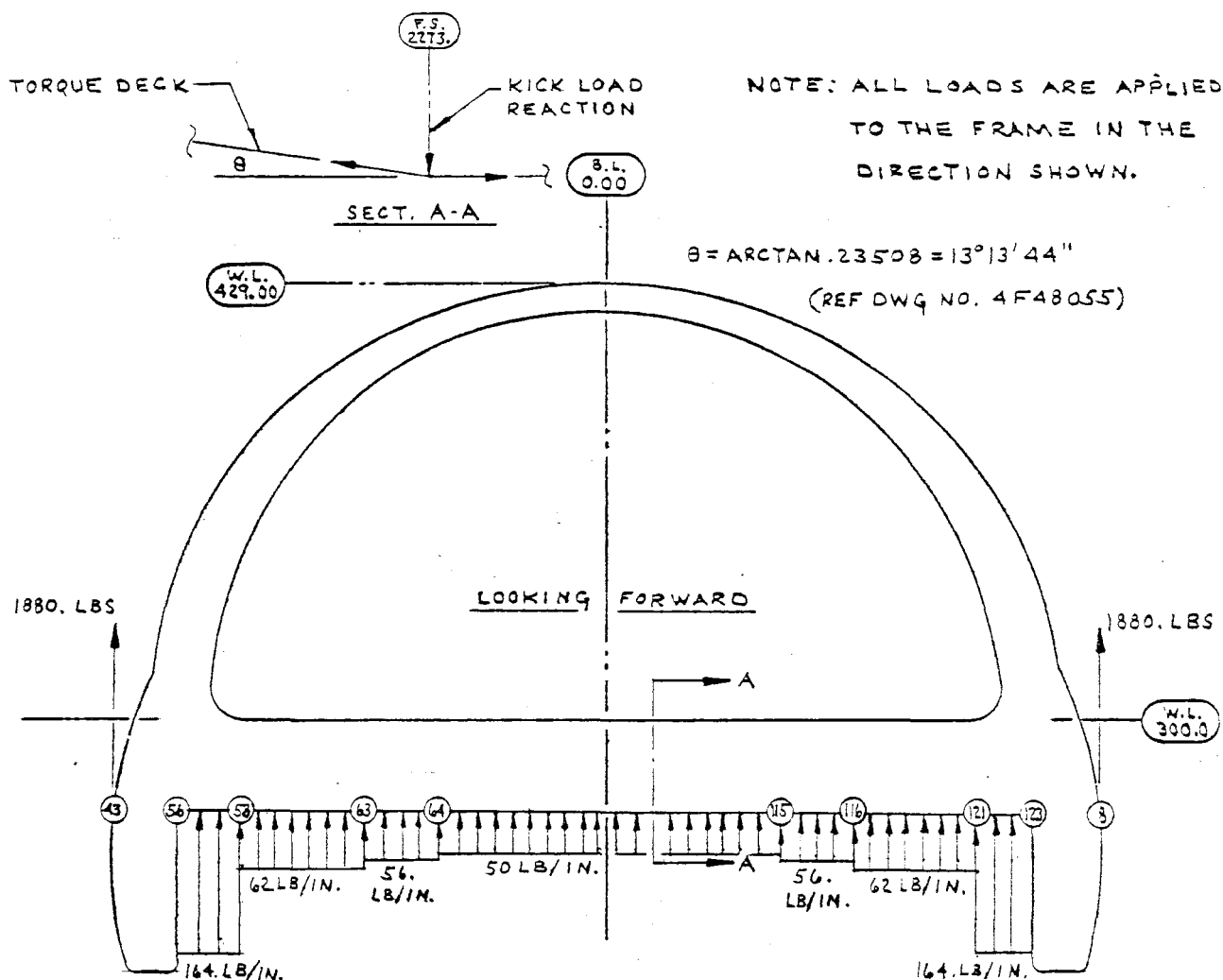
NOTES:

1. A ☐ DENOTES STGR NUMBER.
2. ▴ CONSERVATIVELY ASSUMED MAX. AXIAL LOAD APPLIED TO BOTH SIDES OF AIRCRAFT.
3. THE LOAD REMAINING IN STGRS 69 & 70 IS CARRIED AS AN AXIAL LOAD IN THE LONGERON, THUS:
AT STGR 69 $P_{\text{Long.}} = 10151. - 2126. = 8025. \text{ LBS}$
& AT STGR 70 $P_{\text{Long.}} = 10376. - 2173. = 8203. \text{ LBS}$
4. W IS THE AXIAL RUNNING LOAD IN THE TORQUE DECK.

NAME	DATE	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
Prepared K. R. Watson	5-20-69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.152.2
Checked F. E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved L. G. Darby	5-20-69		LG10S46-2-2	Report No.

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)



G.P. 56 TO 58 & 121 TO 123:

$$\text{CONSERVATIVELY ASSUME } W_{\text{KICK}} = \frac{(2126 + 11498.) \sin \theta}{120.5 - 101.37} = 164. \text{ LB/IN.}$$

G.P. 58 TO 63 & 116 TO 121: $W_{\text{KICK}} = 269. \sin \theta = 62. \text{ LB/IN.}$

G.P. 63 TO 64 & 115 TO 116: $W_{\text{KICK}} = 241 \sin \theta = 56. \text{ LB/IN.}$

G.P. 64 TO 115: $W_{\text{KICK}} = 215 \sin \theta = 50 \text{ LB/IN.}$

G.P. 43 & 3: $P_{\text{KICK}} = 8203 \sin \theta = 1880 \text{ LBS. (FROM LONGERON AXIAL LOAD)}$

FIGURE 7.5.1.3

LOAD CASE NO. 7

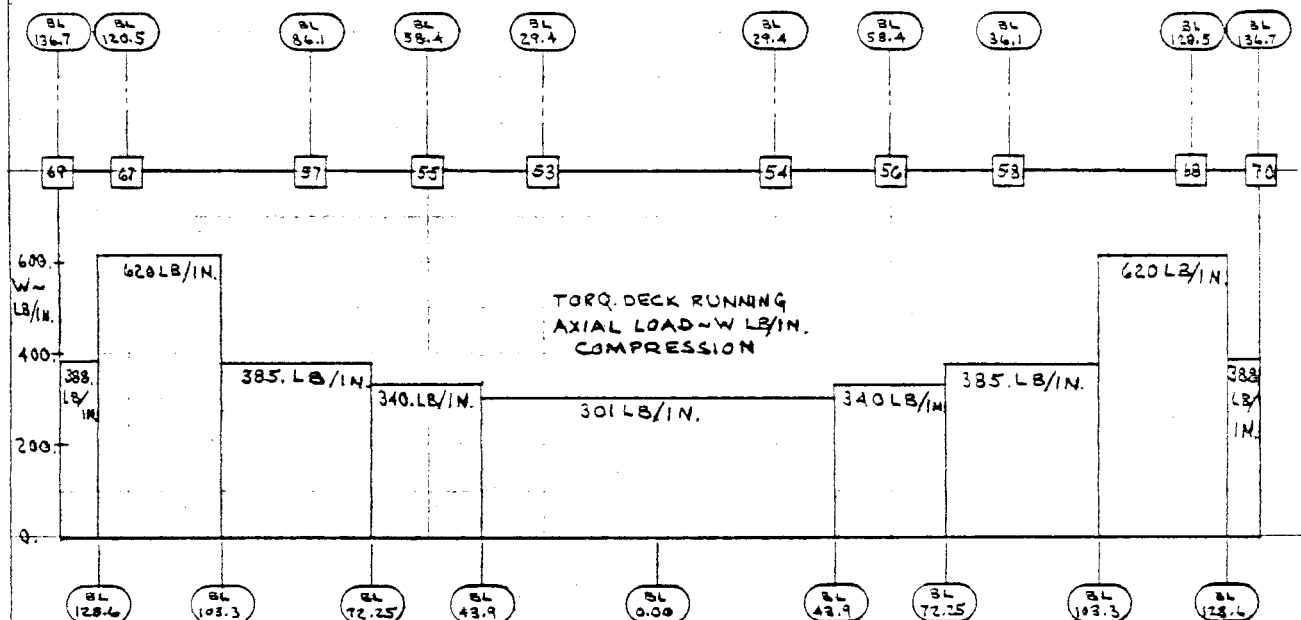
UPBENDING - TORQUE DECK KICK LOADS

NAME	DATE	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
Prepared K. R. Watson	5-20-69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.152.3
Checked F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A
Approved L. G. Darby	5-20-69	AFT FUSELAGE	LG1US46-2-2	Report No.

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7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

CALCULATION OF MAX. TORQUE DECK COMPRESSION LOAD (REF. P. 7.196) FOR
LOAD CASE NO. 8 - DOWNBENDING, TORQUE DECK KICK LOADS



$$\text{CRITICAL COMPRESSION LOAD @ STGR 69} = \frac{8.1 \times .0375}{1.45} (14705) = 3080. \text{ LBS}$$

(REF P 7.196)

$$\text{CRITICAL COMPRESSION LOAD @ STGR 70} = \frac{8.1 \times .0375}{1.45} (15002) = 3142. \text{ LBS}$$

STGR	CRIT. COMP. LOAD	SPAN	W-LB/IN
69	3080.	8.10	381.
67	15666.	25.30	620.2
57	11656.	31.05	376.
55	9223.	28.35	326.
53	12663.	43.90	289.

STGR	CRIT. COMP. LOAD	SPAN	W-LB/IN
70	3142.	3.10	388.2
68	14519.	25.30	574.
58	11939.	31.05	385.2
56	9635.	28.35	340.2
54	13172.	43.90	301.2

(REF P. 7.196)

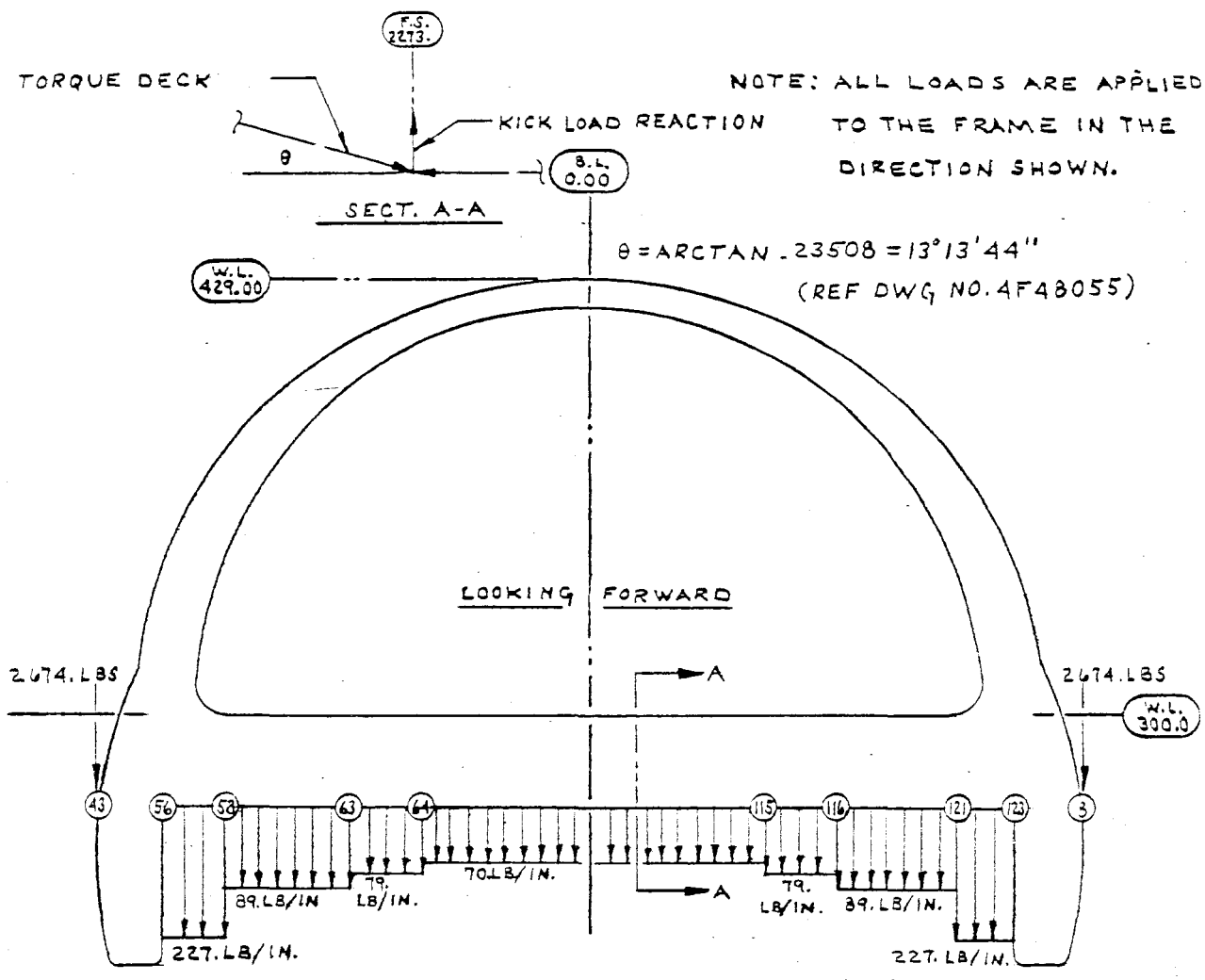
NOTES:

1. A ☐ DENOTES STGR NUMBER.
2. ☒ CONSERVATIVELY ASSUMED MAX. AXIAL LOAD APPLIED TO BOTH SIDES OF AIRCRAFT.
3. THE LOAD REMAINING IN STGRS 69 & 70 IS CARRIED AS AN AXIAL LOAD IN THE LONGERON, THUS:
 AT STGR 69 $P_{LONG} = 14705 - 3080 = 11625. \text{ LBS}$
 & AT STGR 70 $P_{LONG} = 10376 - 2173 = 8203. \text{ LBS}$
4. W IS THE AXIAL RUNNING LOAD IN THE TORQUE DECK

Prepared	K. R. Watson	DATE	5-20-69	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
				A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.152.4
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Approved	L. G. Darby	5-20-69				LG10546-2-2 Report No.

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)



G.P. 56 TO 58 & 121 TO 123: CONSERVATIVELY ASSUME $W_{kick} = \frac{(3142 + 15666.) \sin \theta}{120.5 - 101.37} = 227. \text{ LB/IN.}$

G.P. 58 TO 63 & 116 TO 121: $W_{kick} = 385. \sin \theta = 89. \text{ LB/IN.}$

G.P. 63 TO 64 & 115 TO 116: $W_{kick} = 340 \sin \theta = 79 \text{ LB/IN.}$

G.P. 64 TO 115: $W_{kick} = 301 \sin \theta = 70 \text{ LB/IN.}$

G.P. 43 & 3: $P_{kick} = 11625 \sin \theta = 2674 \text{ LBS. (FROM LONGERON AXIAL LOAD)}$

FIGURE 7.5.1.9

LOAD CASE NO. 3

DOWNBENDING - TORQUE DECK KICK LOADS

Prepared	K. R. Watson	DATE	5-20-69	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
				A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.152.5
Checked	F. E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A
Approved	L. G. Darby	5-20-69				G1US46-2-2 Report No.

REV. (A) KRW. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

NOTE: ALL LOADS ARE APPLIED TO THE FRAME IN THE DIRECTION SHOWN.

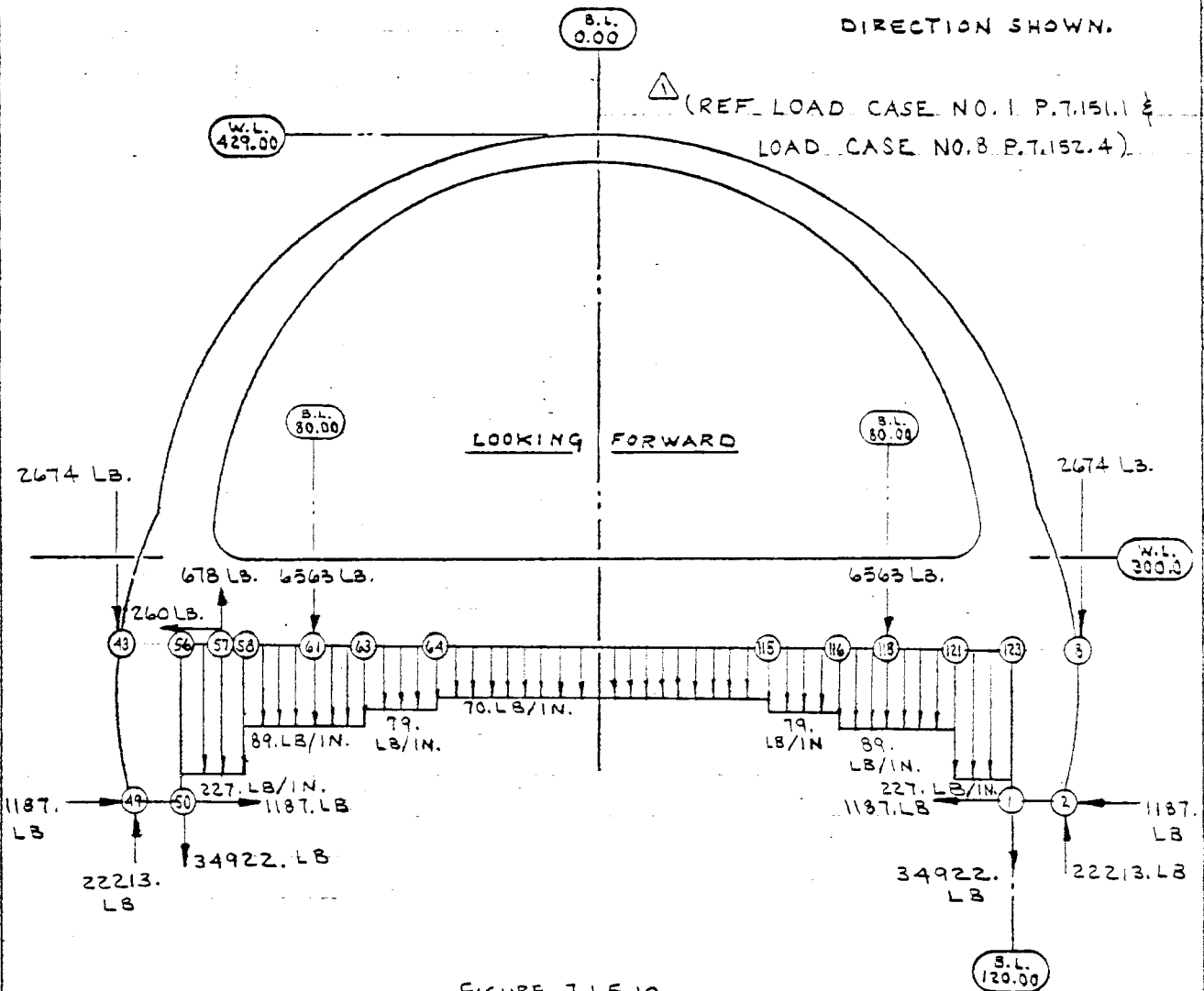


FIGURE 7.1.5.10

LOAD CASE NO. 9

DOWNBENDING TORQUE DECK KICK LOADS
COMBINED WITH MAX. DOOR LOADS

△ LOAD CASES NO. 1 & 8 ARE MODIFIED - MAX. LOADS APPLIED SYM. TO FRAME.

Prepared	NAME K. R. Watson	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. 1	PERM. 7.152.6
Checked	F. E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L. G. Darby	5-20-69		Report No.	LG-46-2-2	

REV. (A) KRW 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

LOAD CASE COMBINATIONS

CASE NO.	CASE TITLE	CASES  COMBINED
10	POSITIVE YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	1 AND 7
11	POSITIVE YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	1 AND 8
12	NEGATIVE YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	2 AND 7
13	NEGATIVE YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	2 AND 8
14	OVERSWING YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	3 AND 7
15	OVERSWING YAW ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	3 AND 8
16	CRUSHING PRESSURE ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	4 AND 7
17	CRUSHING PRESSURE ~ CARGO DOOR LOADS COMBINED WITH TORQUE DECK LOADS	4 AND 8

NOTE:  REFERENCE PAGES 7.151.1 THRU 7.151.6 & 7.152.1 THRU 7.152.5

PREPARED	NAME LD. PINCHOT	DATE 5-20-69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. Vol. I	PERM. 7.153
CHECKED	FE. SCHREIBER	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	LG. DARBY	5-20-69		LGIUS46-2-2 REPORT NO.		

REV. (B) ADP 1-8-71

"PAGE DELETED"

Prepared	NAME A. D. Pinchot	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
					Page	Vol. I 7.153.1
Checked	F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved	L. G. Darby	5-20-69			LG1US46-2-2 Report No.	

REV. (A) A.D.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

TABLE 7.5.1.8
AXIAL ELEMENT LOADS (REF. 17)

△ ELEM. NO.	P _T (MAX.) LBS.	△ CASE NO.	P _C (MAX.) LBS.	△ CASE NO.
1	8924.	16	-13337.	9
2	16559.	17	-25319.	9
3	7729.	17	-10907.	12
4	6248.	17	-8705.	12
5	4960.	17	-6943.	12
6	3808.	17	-5510.	12
7	2730.	17	-4290.	12
8	3882.	17	-4609.	12
9	1949.	17	-3079.	12
10	1826.	5	-4148.	9
11	3969.	12	-5659.	15
12	4071.	12	-7034.	15
13	3602.	12	-7034.	15
14	3382.	12	-6820.	9
15	3597.	16	-8337.	9
16	4363.	16	-9603.	9
17	4952.	16	-10554.	9
18	5385.	16	-11224.	9
19	5805.	16	-11757.	9
20	5821.	16	-11788.	9
21	4992.	16	-9868.	9
22	2937.	16	-5682.	9
23	3426.	9	-1726.	16
24	4973.	9	-2533.	16

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
25	4951.	9	-2522.	16
26	4972.	9	-2533.	16
27	3426.	9	-1727.	16
28	2937.	16	-5627.	9
29	4997.	16	-9778.	9
30	5827.	16	-11684.	9
31	5811.	16	-11653.	9
32	5389.	16	-11128.	9
33	4953.	16	-10464.	9
34	4362.	16	-9527.	9
35	3595.	16	-8231.	9
36	3437.	12	-6789.	9
37	3688.	12	-6549.	15
38	4167.	12	-6549.	15
39	4062.	12	-5221.	17
40	1403.	16	-4073.	9
41	1951.	17	-3974.	12
42	3885.	17	-5815.	12
43	2732.	17	-4610.	12
44	3811.	17	-5694.	10
45	4962.	17	-7220.	10
46	6249.	17	-9098.	10
47	7730.	17	-11452.	10
48	16560.	17	-25318.	11

△ FOR ELEMENT SPECIFICATION, SEE P. 7.147.1 THRU 7.147.3

△ FOR LOAD CASE DEFINITION, SEE P. 7.151.1 THRU 7.151.6 & 7.152.1 THRU 7.152.5

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Approved L. G. Darby	5-20-69			Report No.	LG1US46-2-2

REV. (A) A.D.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

TABLE 7.5.1.8 (CONT'D.)
AXIAL ELEMENT LOADS (REF. 17)

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
49	8923.	16	-13335.	9
50	35713.	10	-23251.	17
51	23995.	10	-16040.	17
52	21591.	10	-14852.	17
53	19598.	10	-13892.	17
54	17881.	10	-13097.	17
55	16341.	10	-12404.	17
56	5113.	12	-5251.	15
57	4161.	15	-2103.	12
58	3034.	15	-4289.	12
59	11772.	15	-6382.	12
60	15350.	15	-3369.	12
61	16331.	15	-3923.	12
62	17186.	15	-9389.	12
63	19011.	15	-10273.	12
64	19872.	15	-10464.	12
65	20615.	9	-10464.	12
66	1871.	17	-1337.	10
67	16122.	16	-31135.	9
68	15466.	16	-30166.	9
69	14120.	16	-28242.	9
70	11996.	16	-25227.	9
71	11101.	16	-23956.	9
72	10116.	16	-22536.	9
73	3937.	16	-19159.	9
74	7643.	16	-15600.	9

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
75	6246.	16	-12115.	11
76	5746.	17	-10325.	13
77	23334.	9	-11593.	16
78	21856.	9	-10871.	16
79	19703.	9	-9789.	16
80	18624.	9	-9350.	16
81	17646.	9	-8932.	16
82	16453.	9	-8389.	16
83	15100.	9	-7748.	16
84	13762.	9	-7057.	16
85	10712.	9	-5591.	16
86	5707.	9	-3041.	16
87	2006.	16	-3940.	9
88	3346.	16	-6430.	9
89	3848.	16	-7363.	9
90	3831.	16	-7342.	9
91	3848.	16	-7375.	9
92	3343.	16	-6474.	9
93	2000.	16	-3998.	9
94	5733.	9	-3043.	16
95	10796.	9	-5589.	16
96	13793.	9	-7052.	16
97	15238.	9	-7742.	16
98	16607.	9	-3352.	16
99	17813.	9	-8924.	16
100	18802.	9	-9341.	16

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared A. D. Pinchot	5-20-69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 7153.3
Checked F.E.Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-51
Approved L. G. Darby	5-20-69			LGTUS48-2-2	Report No.

REV. (A) A.D.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

TABLE 7.5.1.3 (CONT'D.)
AXIAL ELEMENT LOADS (REF. 17)

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
101	19892.	9	-9779.	16
102	22070.	9	-10860.	16
103	23568.	9	-11581.	16
104	5740.	17	-8064.	13
105	6253.	16	-12307.	9
106	7649.	16	-15972.	9
107	10014.	5	-19502.	9
108	12391.	5	-22371.	9
109	12391.	5	-24245.	9
110	11919.	5	-25490.	9
111	14123.	16	-28426.	9
112	15468.	16	-30271.	9
113	1636.	9	-1168.	10
114	19357.	9	-9742.	7
115	18257.	9	-9840.	5
116	15678.	9	-10342.	5
117	14567.	9	-10844.	5
118	13328.	9	-10844.	5
119	10108.	9	-8504.	5
120	6724.	9	-6170.	5
121	3857.	6	-3857.	5
122	2641.	16	-4647.	9
123	15950.	12	-12404.	17
124	17335.	12	-13097.	17
125	18886.	12	-13892.	17
126	20695.	12	-14852.	17

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
127	22885.	12	-16040.	17
128	33898.	12	-23251.	17
129	4362.	12	-4898.	12
130	263.	12	-189.	17
131	322.	12	-230.	17
132	203.	12	-141.	17
133	125.	12	-80.	17
134	2848.	12	-4998.	15
135	858.	9	-486.	16
136	1290.	9	-715.	16
137	1043.	9	-564.	16
138	6137.	17	-6753.	12
139	3706.	9	-1824.	16
140	2473.	16	-5029.	9
141	449.	9	-206.	7
142	387.	9	-445.	16
143	395.	9	-452.	16
144	706.	9	-359.	16
145	1104.	9	-530.	16
146	1915.	9	-946.	16
147	1931.	9	-977.	16
148	1113.	9	-575.	16
149	571.	16	-1142.	9
150	643.	16	-1245.	9
151	510.	16	-976.	9
152	510.	16	-976.	9

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared A. D. Pinchot	5-20-69	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	Vol. I 7.153.4
Checked F.E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A	
Approved L. G. Darby	5-20-69			Report No. LGTUS46-2-2	

REV. (A) A.D.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

TABLE 7.5.1.3 (CONT'D.)
AXIAL ELEMENT LOADS (REF. 17)

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
153	644.	16	-1237.	9
154	573.	16	-1125.	9
155	1102.	9	-575.	16
156	1914.	9	-978.	16
157	1898.	9	-947.	16
158	1094.	9	-530.	16
159	700.	9	-359.	16
160	887.	9	-452.	16
161	379.	9	-445.	16
162	447.	9	-206.	7
163	2475.	16	-4982.	9
164	3669.	9	-1825.	16
165	6143.	17	-7497.	12
166	1040.	9	-565.	16
167	1280.	9	-716.	16
168	848.	9	-487.	16
169	5784.	12	-5827.	15
170	134.	12	-80.	17
171	210.	10	-141.	17
172	334.	10	-230.	17
173	275.	10	-189.	17
174	4581.	10	-5136.	10
175	13600.	13	-6618.	14
176	1256.	8	-899.	10
177	633.	15	-441.	7
178	636.	17	-443.	14

ELEM. NO.	P _T (MAX.) LBS.	CASE NO.	P _C (MAX.) LBS.	CASE NO.
179	9619.	15	-5318.	12
180	591.	3	-411.	10
181	1087.	9	-767.	14
182	1490.	17	-1060.	10
183	1636.	9	-1163.	10
184	1490.	11	-1060.	10
185	1087.	9	-767.	12
186	591.	17	-411.	7
187	11025.	6	-11025.	5
188	636.	11	-443.	12
189	633.	9	-441.	7
190	1256.	3	-899.	12
191	2171.	3	-1569.	14
192	16122.	16	-31135.	9
193	20616.	9	-10326.	7
194	17401.	9	-10097.	16
195	17218.	9	-9764.	16
196	16785.	9	-9312.	16
197	15955.	9	-8633.	16
198	17363.	11	-10097.	16
199	17015.	9	-9768.	16
200	16665.	9	-9318.	16
201	15916.	9	-8642.	16

Prepared	NAME A. D. Pinchot	DATE 5-20-69	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. I	PERM. 7.154
Checked	F.E. Schreiber	5-20-69	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	L. G. Darby	5-20-69				IG1US46-2-2 Report No.	

REV. (A) A.O.P. 1 AUG. 69

7.5.1.3 FRAME LOADS - F.S. 2273 (CONTINUED)

TABLE 7.5.1.9
SHEAR PANEL LOADS (REF. 17)

① PANEL No.	q (MAX.) LB/IN	② CASE No.
1	2940.	9
2	523.	9
3	435.	9
4	372.	9
5	338.	9
6	322.	9
7	398.	12
8	382.	12
9	424.	12
10	562.	12
11	467.	9
12	485.	9
13	149.	9
14	139.	9
15	164.	9
16	185.	9
17	193.	9
18	197.	9
19	228.	9
20	239.	9
21	236.	9
22	109.	9
23	43.	11

PANEL No.	q (MAX.) LB/IN	CASE No.
24	14.	6
25	40.	9
26	109.	9
27	235.	9
28	237.	9
29	226.	9
30	194.	9
31	191.	9
32	183.	9
33	163.	9
34	137.	9
35	148.	9
36	479.	9
37	461.	9
38	627.	12
39	486.	12
40	454.	12
41	484.	12
42	323.	9
43	339.	9
44	372.	9
45	435.	9
46	523.	9

PANEL No.	q (MAX.) LB/IN	CASE No.
47	2939.	9
48	1376.	13
49	614.	15
50	543.	15
51	545.	15
52	522.	15
53	250.	13
54	229.	13
55	189.	13
56	134.	13
57	74.	13
58	75.	13
59	135.	15
60	190.	15
61	229.	15
62	231.	15
63	471.	9
64	495.	9
65	513.	9
66	564.	9
67	1294.	9

① FOR PANEL SPECIFICATION, SEE P. 7.148.1 & 7.148.2

② FOR LOAD CASE DEFINITION, SEE P. 7.151.1 THRU 7.151.6 & 7.152.1 THRU 7.152.5

PREPARED	NAME A.D. PINCHOT	DATE 1 AUG. 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	PERM. 7.155
CHECKED	R.F. WILKINSON	1 AUG. 69	TITLE INTERNAL LOADS ANALYSIS LFT FUSELAGE	MODEL C-52		
APPROVED	L.G. DAZBY	1 AUG. 69		LGIUS 46-2-2 REPORT NO.		

REV. (A) LDP 1 AUG. 69

REV. (B) LDP 1-8-71

"PAGE DELETED"

	NAME	DATE			TEMP.	PERM.
PREPARED	K. Watson	24 Apr 68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I	7.156
CHECKED	F. Schreiber	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	L.G. Darby	24 Apr 68		LGUS 46-2-2 REPORT NO.		

7.5.2 Frame-Fuselage Station 2383.472

7.5.2.1 Discussion

The frame is located in the unpressurized area, aft of the pressure bulkhead, above the sloping longeron box beam.

The primary function of the frame is to react vertical and horizontal (inboard-outboard) kick loads induced by axial loads in the sloping longeron. In addition to its primary function this frame aids general shell stability. The frame is designed to be damage tolerant, (i.e. multi-element cap construction).

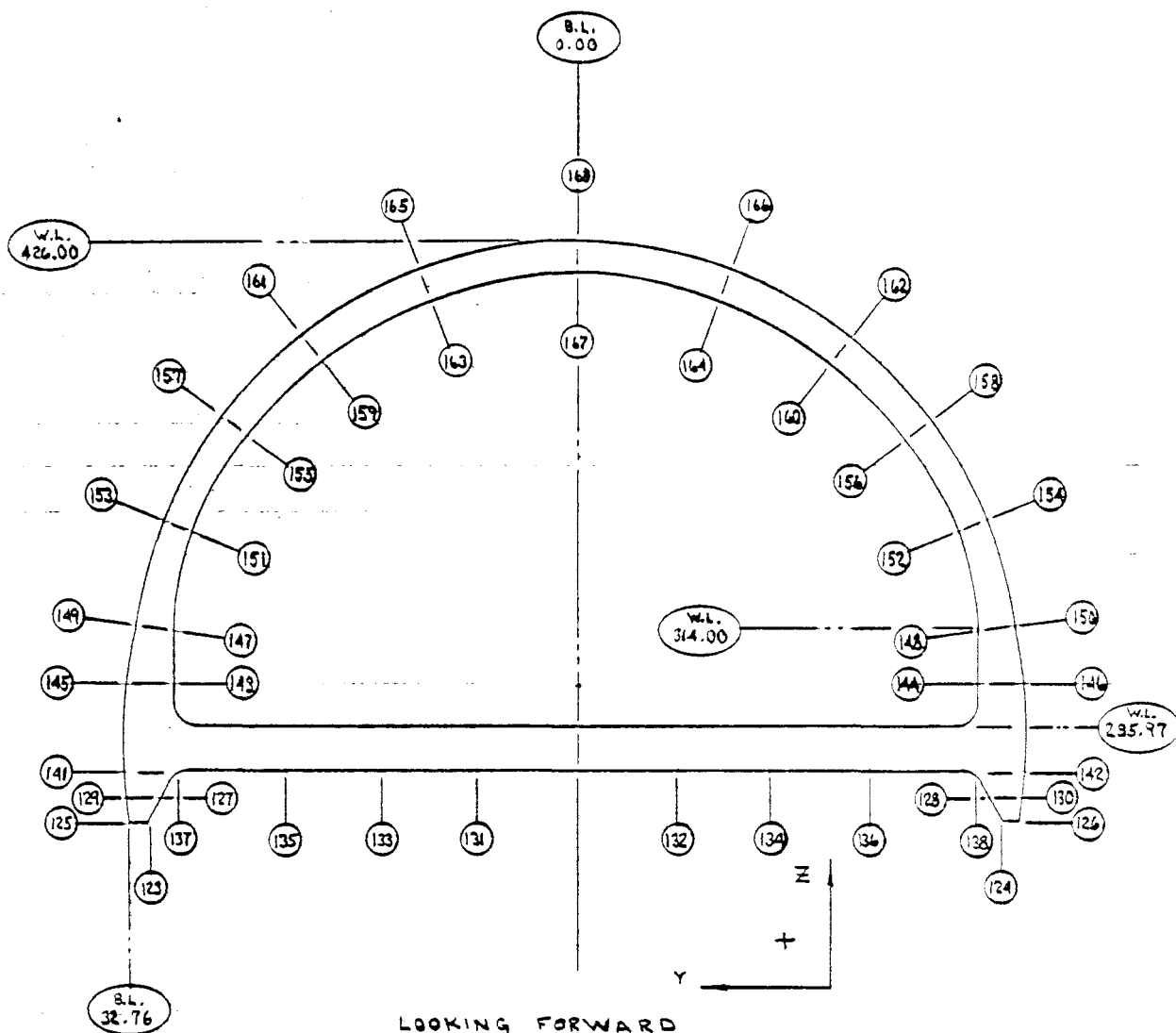
The frame is a "J" section approximately eight inches deep and it is constructed essentially of 7075-T6 aluminum alloy. The cap elements are extruded and are riveted to a chem-milled web. Vertical stiffeners are riveted to the web and cap elements at each stringer location. The cross beam is a "Z" section and is constructed similar to the "J" section described above. It is approximately thirteen inches deep.

Internal load distributions for the frame are obtained from FAMAS, (Reference 8). FAMAS geometry is summarized in Figure 7.5.2.1 and Tables 7.5.2.1, 7.5.2.2, and 7.5.2.3. Critical web shears and axial element loads are summarized in Tables 7.5.2.4 and 7.5.2.6 respectively. The basic applied loads which caused these critical internal loads are identified in Table 7.5.2.5.

Prepared	NAME K. Watson	DATE 24 Apr 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	F. Schreiber	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS APT FUSELAGE	Page	Vol. 17.157
Approved	L.G. Darby	24 Apr 68		Model	C-5A
				Report No.	IG1US 46-4-2

7.5.2 FRAME ~ F.S. 2383.472

7.5.2.2 GEOMETRY



REFERENCE DRAWING NUMBER 4F41030~FRAME

NOTE: ENCIRCLED NUMBERS ARE FAMAS, (REF 3), GRID POINTS.

FIGURE 7.5.2.1 ~ FAMAS MODEL (REF 3, SECTION)

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Prepared K. Watson	24 Apr 68	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page Vol. I	7.158
Checked F. Schreiber	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS WING FUSELAGE		Model C-5A	
Approved L.G. Darby	24 Apr 68			IG1US 46-4-2	Report No.

7.5.2 FRAME ~ F.S. 2383.472

7.5.2.2 GEOMETRY

TABLE 7.5.2.1 ~ FAMES GRIDPOINT COORDINATES (REF 3, SECTION 7.2.1.7)

G.P.	Y-COORD.	Z-COORD.	G.P.	Y-COORD.	Z-COORD.
123	122.630	257.762	124	-122.630	257.762
125	125.850	257.762	126	-125.850	257.762
127	119.090	265.900	128	-119.090	265.900
129	127.150	266.315	130	-127.150	266.315
131	29.396	273.900	132	-29.396	273.900
133	58.380	273.900	134	-58.380	273.900
135	86.104	273.900	136	-86.104	273.900
137	115.550	273.900	138	-115.550	273.900
141	127.370	273.900	142	-127.370	273.900
143	115.550	298.159	144	-115.550	298.159
145	128.111	298.677	146	-128.111	298.677
147	115.550	313.145	148	-115.550	313.145
149	125.850	314.000	150	-125.850	314.000
151	110.271	343.123	152	-110.271	343.123
153	117.688	346.119	154	-117.688	346.119
155	95.402	369.785	156	-95.402	369.785
157	101.791	374.603	158	-101.791	374.603
159	74.198	391.801	160	-74.198	391.801
161	79.205	398.042	162	-79.205	398.042
163	41.236	410.454	164	-41.236	410.454
165	44.005	417.957	166	-44.005	417.957
167	.000	418.000	168	.000	425.776

Prepared	NAME K. Watson	DATE 24 Apr 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. 66.1	PERM. 7.159
Checked	F. Schreiber	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	L.G. Darby	24 Apr 68				LGTUS 46-4-2	
Report No.							

7.5.2 FRAME ~ F.S. 2383.472

7.5.2.2 GEOMETRY

TABLE 7.5.2.2 ~ AXIAL ELEMENT SPECIFICATIONS (REF 3, SECTION 7.3.1.7.3)

ELEM.	GP (1)	GP (2)	AREA	ELEM.	GP (1)	GP (2)	AREA
233	167	168	0.1200	170	131	132	1.0700
171	123	125	1.0000	172	124	126	1.0000
173	123	127	1.3100	174	124	128	1.3100
175	125	129	1.3100	176	126	130	1.3100
177	127	129	0.5000	178	128	130	0.5000
179	127	137	1.3100	180	128	138	1.3100
181	129	141	1.3100	182	130	142	1.3100
183	131	133	1.0700	184	132	134	1.0700
185	133	135	1.0700	186	134	136	1.0700
187	135	137	1.0700	188	136	138	1.0700
191	137	141	1.0700	192	138	142	1.0700
193	137	143	0.6700	194	138	144	0.6700
195	141	145	1.0000	196	142	146	1.0000
197	143	145	0.1200	198	144	146	0.1200
199	143	147	0.5900	200	144	148	0.5900
201	145	149	1.0000	202	146	150	1.0000
203	147	149	0.1200	204	148	150	0.1200
205	147	151	0.5900	206	148	152	0.5900
207	149	153	0.6000	208	150	154	0.6000
209	151	153	0.1200	210	152	154	0.1200
211	151	155	0.5500	212	152	156	0.5500
213	153	157	0.6000	214	154	158	0.6000
215	155	157	0.1200	216	156	158	0.1200
217	155	159	0.5200	218	156	160	0.5200
219	157	161	0.6000	220	158	162	0.6000
221	159	161	0.1200	222	160	162	0.1200
223	159	163	0.5200	224	160	164	0.5200
225	161	165	0.6000	226	162	166	0.6000
227	163	165	0.1200	228	164	166	0.1200
229	163	167	0.5200	230	164	167	0.5200
231	165	168	0.6000	232	166	168	0.6000

Prepared	K. Watson	DATE	24 Apr 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP.	PERM.
Checked	F. Schreiber	DATE	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-5A	
Approved	L.G. Darby	DATE	24 Apr 68			Report No.	LG1US 46-4-2	

7.5.2 FRAME~F.S. 2383.472

7.5.2.2 GEOMETRY

TABLE 7.5.2.3~SHEAR ELEMENT SPECIFICATIONS (REF 8, SECTION 7.3.1.7.6)

ELEM	GP(1)	GP(2)	GP(3)	GP(4)	τ
47	123	125	129	127	.125
49	127	129	141	137	.125
51	137	141	145	143	.085
53	143	145	149	147	.074
55	147	149	153	151	.074
57	151	153	157	155	.063
59	155	157	161	159	.050
61	159	161	165	163	.050
63	163	165	168	167	.050

ELEM	GP(1)	GP(2)	GP(3)	GP(4)	τ
48	124	126	130	128	.125
50	128	130	142	138	.125
52	138	142	146	144	.085
54	144	146	150	148	.074
56	148	150	154	152	.074
58	152	154	158	156	.063
60	156	158	162	160	.050
62	160	162	166	164	.050
64	164	166	168	167	.050

7.5.2.3 LOADS

TABLE 7.5.2.4~SUMMARY OF MAX WEB SHEAR FLOWS

ELEM	+ q	CASE	- q	CASE
47	777.	2	1175.	8
49	240.	2	368.	8
51	238.	8	152.	2
53	121.	8	84.	2
55	114.	7	60.	1
57	116.	2	182.	8
59	68.	2	90.	8
61	108.	14	160.	7
63	201.	14	181.	7

ELEM	+ q	CASE	- q	CASE
48	900.	2	1307.	8
50	293.	2	423.	8
52	287.	8	201.	2
54	140.	8	103.	2
56	114.	7	46.	40
58	111.	2	179.	8
60	61.	2	39.	1
62	107.	14	160.	7
64	199.	14	180.	7

TABLE 7.5.2.5~BASIC LOAD CASES

CASE	DESCRIPTION	REF 10, P.	CASE	DESCRIPTION	REF 10, P.
6	5ARISAT2~AILER.ROLL,HI DRAG	9.4-674	40	SPM194~POS.MANEUV.,HI-DRAG	9.2-458
1	OML001~DYNAM. LAT. GUST	9.3-501	2	1PC242~POS.CHECKED PITCH	9.2-690
14	101513~DYNAM. LANDING	9.6-291	5	1EY107~EQUIL.YAW,RUD.RETURN	9.3-170
7	104018~DYNAM. TAXI	9.6-657	8	SPE201~POS.PITCH ENT,HI-DRAG	9.2-730
			15	OML003~DYNAM. LAT. GUST	9.3-505

Prepared	NAME K. Watson	DATE 24 Apr 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. Vol. I	PER 7.11
Checked	F. Schreiber	24 Apr 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A	
Approved	L.G. Darby	24 Apr 68		Report No.	LGTUS 46-4-2	

7.5.2 FRAME ~ F.S. 2383.472

7.5.2.3 LOADS

TABLE 7.5.2.6 ~ SUMMARY OF MAX AXIAL LOADS

ELEM.	P _r	CASE	P _c	CASE
233	2229.	7	1863.	14
171	3855.	2	5837.	8
173	5746.	8	3831.	2
175	689.	7	751.	1
177	180.	7	139.	1
179	5882.	8	4041.	2
181	14662.	2	21678.	8
183	3571.	8	2409.	2
185	4692.	8	3141.	2
187	6763.	8	4449.	2
191	4284.	8	2753.	2
193	1863.	7	1318.	8
195	12980.	2	19076.	8
197	850.	14	505.	2
199	3391.	2	5187.	8
201	5265.	2	7452.	8
203	144.	2	169.	8
205	3897.	2	5537.	8
207	988.	2	1823.	14
209	825.	2	1098.	8
211	1812.	2	3302.	7
213	1255.	7	1418.	1
215	225.	7	800.	7
217	2671.	14	4558.	7
219	1964.	7	1515.	1
221	420.	8	668.	7
223	2135.	8	1474.	2
225	1414.	2	2038.	8
227	774.	7	562.	14
229	6166.	7	5127.	14
231	2348.	2	3224.	8

ELEM	P _r	CASE	P _c	CASE
170	3198.	8	2195.	2
172	4137.	2	6176.	8
174	6303.	8	4345.	2
176	571.	7	881.	15
178	181.	7	130.	6
180	5971.	8	4030.	2
182	15875.	2	23173.	8
184	3800.	8	2626.	2
186	5057.	8	3477.	2
188	7549.	8	5202.	2
192	4987.	8	3456.	2
194	1853.	7	836.	14
196	14625.	2	20956.	8
198	1044.	8	709.	2
200	3558.	2	5385.	8
202	5662.	2	7991.	8
204	90.	2	120.	3
206	4263.	2	5949.	8
208	822.	5	1882.	14
210	966.	2	125.	8
212	2297.	2	3307.	7
214	1950.	1	1221.	2
216	327.	2	800.	7
218	2530.	14	4546.	7
220	1966.	7	1404.	14
222	349.	14	664.	7
224	2050.	1	1068.	7
226	1034.	2	1736.	14
228	778.	7	590.	14
230	6173.	7	5174.	14
232	2204.	2	3094.	8

	NAME	DATE	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP.	PERM.
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CHECKED	J.P.Kimmel	11-11-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL C-5A		
APPROVED	H.H.Wade	11-11-70		LG1US46-2-2 REPORT NO.		

REV. (B) FES 1-8-71

7.5.3 Frames - Fuselage Stations 2427 - 2516

7.5.3.1 Discussion

Dorsal stability frames from F.S. 2427 through F.S. 2516 redistribute dorsal area loading, forward of the empennage, down into the fuselage.

Frames are made up of typical built-up sheet metal construction.

Internal load distributions for the frames are obtained from FAMAS, (reference 8).

REV. (B) J.P.K 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP AND WEB LOADS

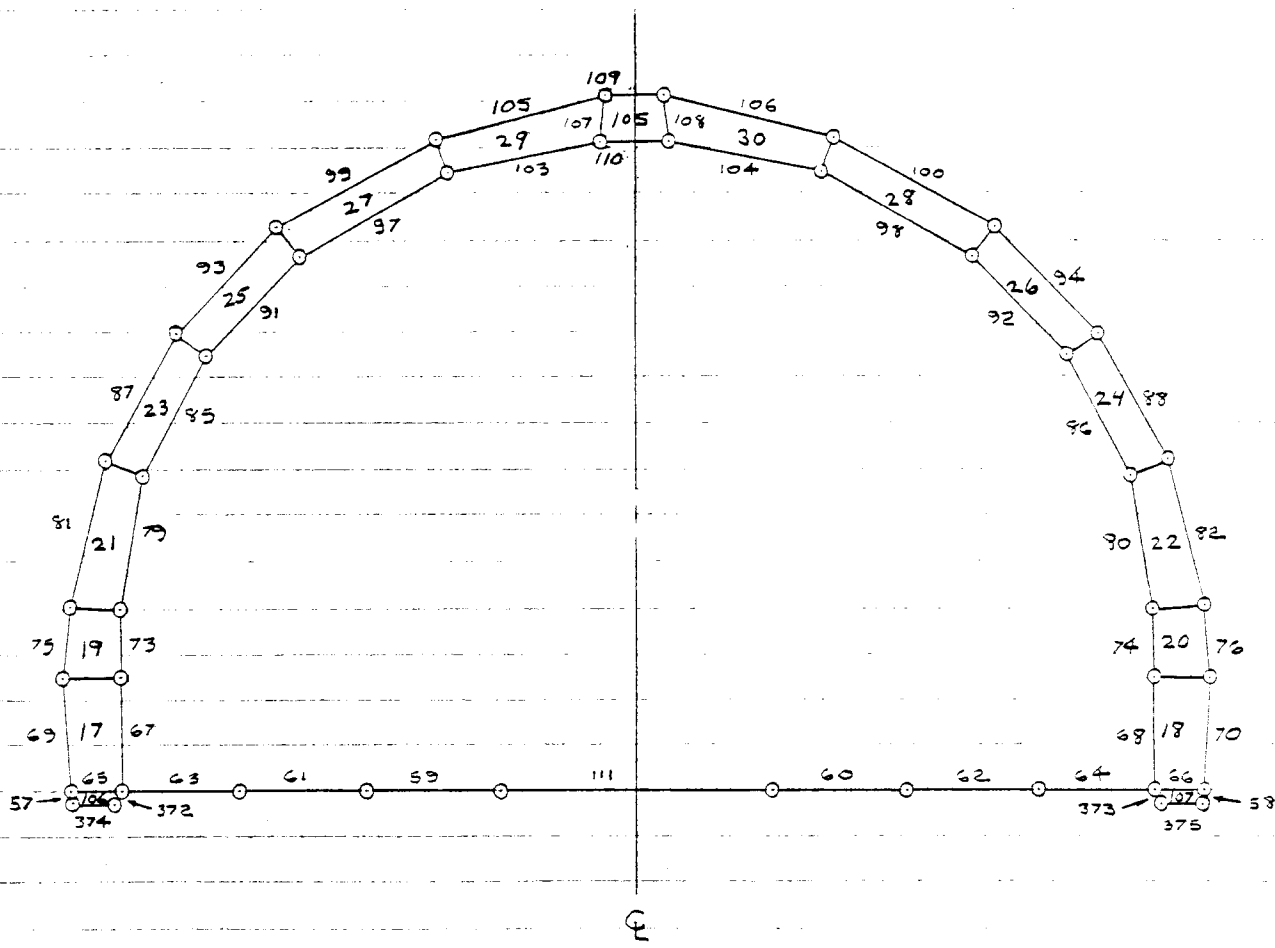


FIGURE 7.5.3.1
F.S. 2427

NUMBERS ARE FRAME BAR AND WEB NO. IN AFT FUSELAGE MOD 6
OF THE FAMES MODEL, REF. 8, VOL. IV, P. 4.60

Prepared	NAME J.E. LEGGETT	DATE 7-7-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL. 2	PERM. 7.167
Checked	F.E. SCHWEIZER	7-7-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model	C-5A	
Approved	H.H. WADE	7-7-70			LSIUS 46-3-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS CONDITION 100400 (I=1.18)

57	.362855 +04	58	.360710 +04	59	.112566 +04
60	.112613 +04	61	.105885 +04	62	.105911 +04
63	.104277 +04	64	.104457 +04	65	-.849854 +03
66	-.844063 +03	67	.461442 +04	68	.469256 +04
69	.394386 +04	70	.390864 +04	73	.336318 +04
74	.333008 +04	75	.915096 +03	76	.892769 +03
79	-.958809 +03	80	-.991517 +03	81	.368121 +03
82	.367518 +03	85	-.708113 +04	86	-.709140 +04
87	.171542 +04	88	.172495 +04	91	-.854445 +04
92	-.852763 +04	93	.188938 +04	94	.189021 +04
97	-.266435 +03	98	-.239939 +04	99	-.578083 +03
100	-.587997 +03	103	.108417 +05	104	.108579 +05
105	.853589 +03	106	.843138 +03	107	.211274 +04
108	.211118 +04	109	.192974 +05	110	.140633 +05
111	.118972 +04	372	.218091 +04	373	.217545 +04
374	.127261 +03	375	.127022 +03		

FRAME WEB LOADS - POUNDS / INCH, CONDITION 100400 (I=1.18)

17	.404187 +02	18	.411352 +02
19	.833561 +02	20	.837682 +02
21	.214932 +03	22	.214639 +03
23	.166733 +03	24	.165657 +03
25	-.728334 +02	26	-.734626 +02
27	-.362799 +03	28	-.362778 +03
29	-.208432 +03	30	-.207919 +03
105	-.511170 +00	106	-.382640 +02
107	-.380622 +02		

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	J.E. LEGGETT	7-8-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	7.65
Checked	F.E. SCHREIBER	7-8-70	TITLE	Model C-52	
Approved	H.H. WADE	7-2-75	INTERNAL LOADS ANALYSIS	LS 105 +6-2-2	
			AFT FUSELAGE	Report No.	

REV. (8) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION SPE201

57	- .133615 + 05	58	- .134625 + 05	59	- .197222 + 03
60	- .329198 + 03	61	- .618506 + 03	62	- .297126 + 03
63	- .169835 + 04	64	- .130908 + 04	65	- .437415 + 04
66	- .438979 + 04	67	- .992292 + 04	68	- .876840 + 04
69	- .194762 + 05	70	- .214973 + 05	73	- .110885 + 05
74	- .112971 + 05	75	- .731610 + 04	76	- .815755 + 04
79	- .777428 + 04	80	- .890139 + 04	81	- .149735 + 04
82	- .156591 + 04	85	- .104000 + 04	86	- .330252 + 03
87	- .285033 + 04	88	- .373367 + 04	91	- .694149 + 04
92	- .593255 + 04	93	- .437934 + 03	94	- .160093 + 04
97	- .992654 + 03	98	- .213762 + 03	99	- .106374 + 04
100	- .284060 + 03	103	- .961654 + 04	104	- .100689 + 05
105	- .810510 + 04	106	- .759095 + 04	107	- .192275 + 04
108	- .190658 + 04	109	- .368133 + 05	110	- .128370 + 05
111	- .567611 + 03	372	- .450927 + 04	373	- .508144 + 04
374	- .198775 + 03	375	- .203105 + 03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION SPE201

17	- .979479 + 02	18	- .184687 + 03
19	- .201942 + 02	20	- .703089 + 02
21	- .219669 + 03	22	- .194831 + 03
23	- .321079 + 03	24	- .329865 + 03
25	- .508757 + 02	26	- .644946 + 02
27	- .344205 + 03	28	- .343326 + 03
29	- .201342 + 03	30	- .185221 + 03
105	- .116456 + 02	106	- .220292 + 03
107	- .199211 + 03		

Prepared	NAME J.E. EGGETT	DATE 7-10-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	F.E. SCHREIBER	7-10-70	TITLE INTERNAL LOADS ANALYSIS		Page	7.166
Approved	H.H. WADE	7-10-70	AFT FUSELAGE		Model C-5A LGUS 46-2-2 Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3. FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION DML001

57	.418018 +04	58	-.105414 +05	59	-.974017 +03
60	.685395 +03	61	-.198630 +04	62	.203816 +04
63	-.308569 +04	64	.363282 +04	65	-.199640 +04
66	.400764 +04	67	.360854 +03	68	-.490506 +04
69	.693185 +04	70	-.165873 +05	73	.327406 +03
74	-.573679 +04	75	.382484 +04	76	-.737291 +04
79	.101821 +04	80	-.502181 +04	81	.213281 +03
82	.629723 +03	85	.176291 +04	86	-.156399 +04
87	-.111213 +04	88	.276652 +04	91	.105995 +04
92	.208237 +04	93	-.444328 +03	94	.911833 +03
97	-.254249 +04	98	.285441 +04	99	.418093 +03
100	-.754453 +03	103	-.461717 +04	104	-.167349 +03
105	-.631808 +03	106	-.314781 +04	107	-.368445 +03
108	-.561719 +03	109	-.890529 +04	110	-.312668 +04
111	-.216835 +03	372	-.831781 +01	373	-.232594 +04
374	.198284 +02	375	-.138350 +03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION DML001

17	.193854 +02	18	.477045 +02
19	-.152162 +02	20	.355548 +02
21	-.330337 +02	22	-.677606 +02
23	-.143005 +02	24	-.142470 +03
25	.582031 +02	26	-.868604 +02
27	.135948 +03	28	.313773 +02
29	-.310970 +02	30	.125800 +03
105	-.116950 +03	106	-.130531 +03
107	.229409 +03		

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	W. D. BALGH	7-15-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	No. 1 7.147
Checked	F. E. SCHREIBER	7-15-70	TITLE	Model C-5A	
Approved	H. H. WADE	7-15-70	INTERNAL LOADS ANALYSIS	LG-15 45-2-2	
			AFT FUSELAGE	Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427 - 2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 1PC242

57	.891060 +04	58	.881599 +04	59	-.961522 +02
60	.704831 +02	61	-.751895 +03	62	-.340389 +03
63	-.137314 +04	64	-.105122 +04	65	-.304416 +04
66	-.297538 +04	67	.652564 +04	68	.517962 +04
69	.128615 +05	70	.147445 +05	73	.722008 +04
74	.734790 +04	75	.497944 +04	76	.574612 +02
79	.530498 +04	80	.644665 +04	81	-.975592 +03
82	-.104680 +04	85	-.592668 +02	86	.140844 +04
87	-.195203 +04	88	-.287162 +04	91	-.336360 +04
92	-.272609 +04	93	-.308593 +03	94	-.145734 +04
97	-.729076 +03	98	.194212 +03	99	.101651 +04
100	.165577 +03	103	.524930 +04	104	.578613 +04
105	.579670 +04	106	.532898 +04	107	.108391 +04
108	.106492 +04	109	.236893 +05	110	.713501 +04
111	.207743 +03	372	.272556 +04	373	.324524 +04
374	.104481 +03				

FRAME WEB LOADS - POUNDS/INCH, CONDITION 1PC242

17	-.716697 +02	18	.880378 +02
19	-.344087 +02	20	.370775 +02
21	-.260562 +02	22	-.228076 +01
23	-.162074 +02	24	-.247899 +02
25	-.174722 +02	26	.466647 +01
27	-.997967 +01	28	.546820 +02
29	.540572 +02	30	-.234068 +02
105	.594871 +02	106	.963663 +02
107	-.781308 +02		

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
	J.P.Kimmel	11-9-70		Page	VOL. 1 7.163
Checked	F.E.Schreiber	11-9-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model	C-5A
Approved	H.H.Wade	11-9-70		Report No.	LG1US46-2-2

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP AND WEB LOADS

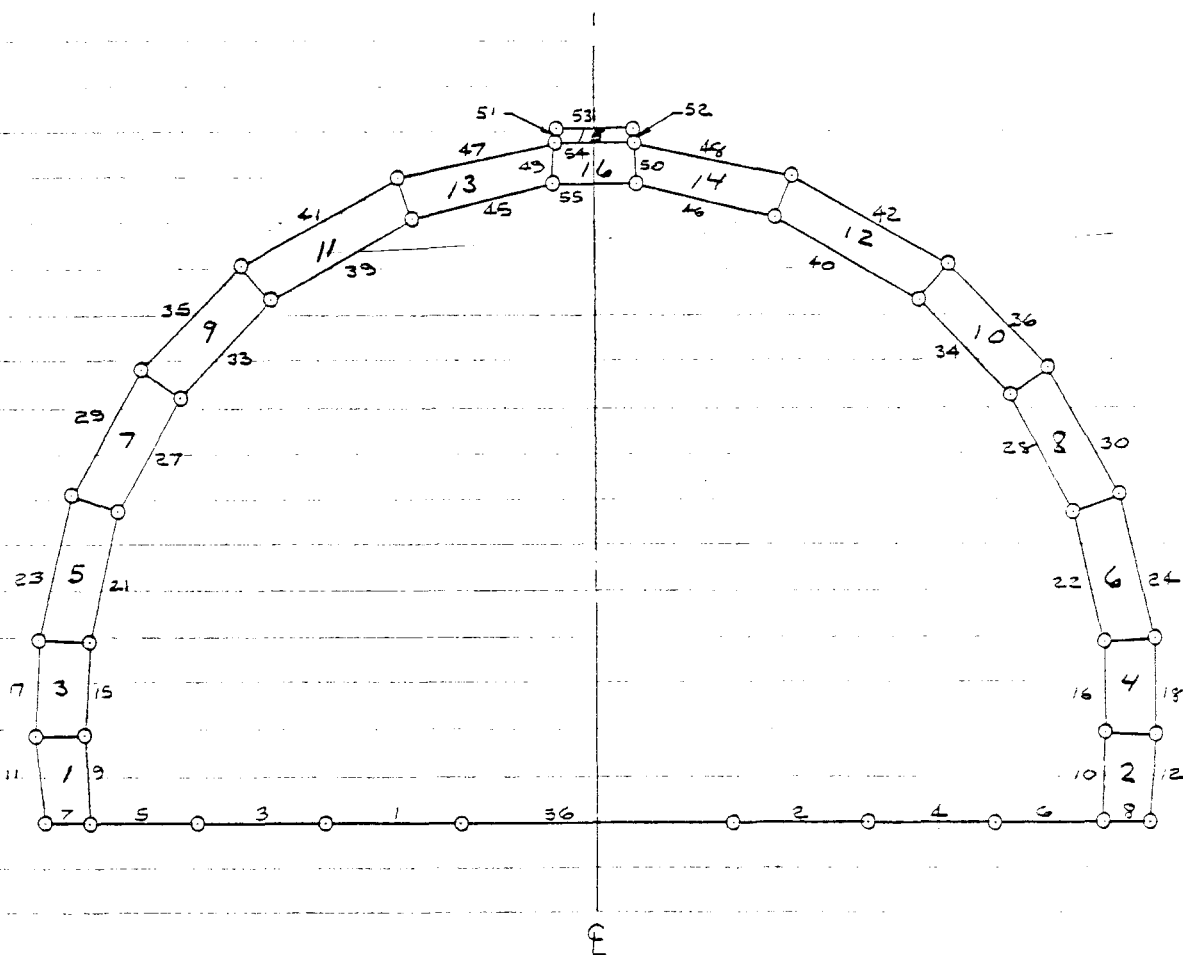


FIGURE 7.5.3.2
F.S.2450

NUMBERS ARE FRAME BAR AND WEB NO. IN AFT FUSELAGE
MOD. 6 OF THE FAMAS MODEL, REF. 8, VOL IV, P. 4.59

Prepared	NAME J. E. LEGGETT	DATE 7-7-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. 7.163	PERM.
Checked	F. E. SCHNEIDER	7-7-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5A		
Approved	H. J. WADE	7-7-70			S103 45-1-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 100400 (I=1.18)

1	.910441 +03	2	.907403 +03	3	.739267 +03
4	.736170 +03	5	-.671033 +03	6	-.672362 +03
7	.550435 +03	8	.549259 +03	9	.474659 +03
10	.472072 +03	11	.186995 +03	12	.183576 +03
15	.562514 +03	16	.556639 +03	17	.142578 +03
18	.136780 +03	21	-.240037 +03	22	-.246284 +03
23	.839237 +02	24	.807118 +02	27	-.179558 +04
28	-.179725 +04	29	.761185 -03	30	.760063 +03
33	-.324041 +04	34	-.323532 +04	35	.166085 +04
36	.166048 +04	39	-.384085 +04	40	-.383298 +04
41	.730323 +03	42	.728754 +03	45	.921669 +03
46	.927530 +03	47	-.101242 +05	48	-.101285 +05
49	.287114 +02	50	.280453 +02	51	-.170191 -04
52	-.170464 +04	53	-.380160 +03	54	-.185844 +05
55	.547169 +04	56	.579428 -03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION 100400 (I=1.18)

1	-.185930 +02	2	-.183911 +02
3	.874939 +01	4	.887497 +01
5	.445842 +02	6	.445193 +02
7	.537223 +02	8	.534955 +02
9	.391387 +02	10	.389330 +02
11	-.884442 +00	12	-.861412 +00
13	-.273903 +03	14	-.273814 +03
15	.282560 +00	16	-.239186 +00

Prepared	NAME J.E. LESGETT	DATE 7-8-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. Vol. 5	PERM. 7-170
Checked	E.S. SCHAEFER	7-8-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	J.H. WADE	7-8-70				LG105 46-2-2 Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2616

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION SPE201

1	- .127788 +04	2	- .141080 +04	3	- .790114 +03
4	- .102273 +04	5	- .218920 +04	6	- .239599 +04
7	- .176510 +04	8	- .163207 +04	9	- .693565 +03
10	- .301931 +03	11	- .271033 +04	12	- .312376 +04
15	- .116216 +04	16	- .136710 +04	17	- .346493 +04
18	- .333491 +04	21	- .132199 +04	22	- .171479 +04
23	- .186388 +04	24	- .147993 +04	27	- .106237 +03
28	- .446950 +03	29	- .162564 +04	30	- .122882 +04
33	- .239403 +04	34	- .212561 +04	35	- .211961 +04
36	- .185383 +04	39	- .466441 +04	40	- .439775 +04
41	- .151439 +04	42	- .132644 +04	45	- .343046 +03
46	- .147249 +02	47	- .104810 +05	48	- .108029 +05
49	- .177374 +03	50	- .309560 +03	51	- .342335 +04
52	- .173768 +04	53	- .549954 +03	54	- .205754 +05
55	- .483946 +04	56	- .122764 +04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION SPE201

1	- .194418 +02	2	- .629556 +02
3	- .265952 +02	4	- .436825 +02
5	- .892682 +01	6	- .882247 +01
7	- .682846 +02	8	- .716118 +02
9	- .920832 +02	10	- .933406 +02
11	- .426743 +02	12	- .417698 +02
13	- .296577 +03	14	- .301092 +03
15	- .995141 +00	16	- .244001 +02

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
	J. E. LEGGETT	7-10-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	VOL I 7.171
Checked	F. E. SCHREIBER	7-10-70	TITLE		Model C-5A	
Approved	H. H. WADE	7-10-70			LG10546-2-2 Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP. LOADS - POUNDS, CONDITION DML001

1 - .3470 00 +03	2 - .378251 +03	3 - .692026 +03
4 .17 8493 +03	5 - .437942 +03	6 .146762 +04
7 .792146 +03	8 - .159257 +04	9 .252691 +03
10 - .4816 01 +03	11 .113165 +04	12 - .244796 +04
15 - .6430 81 +03	16 .493060 +02	17 .254534 +04
18 - .407651 +04	21 - .313772 +03	22 - .389862 +03
23 .165204 +04	24 - .232425 +04	27 .471995 +03
28 - .572256 +03	29 .543298 +03	30 - .103017 +04
33 .775249 +03	34 .339603 +03	35 - .618032 +03
36 - .115503 +03	39 .381895 +03	40 .135449 +04
41 - .747527 +03	42 .163603 +03	45 - .150563 +04
46 .162494 +04	47 .3366 11 +04	48 .182521 +04
49 .604851 +03	50 - .476500 +03	51 .110416 +04
52 - .297023 +03	53 .133451 +03	54 .501053 +04
55 - .116820 +04	56 - .335412 +03	

FRAME WEB LOADS - POUNDS / INCH, CONDITION DML001

1 .833519 +02	2 - .635604 +02
3 .913055 +01	4 .741368 +01
5 - .271640 +02	6 .223873 +02
7 - .222288 +02	8 - .112864 +02
9 .278180 +01	10 - .471943 +02
11 .181426 +02	12 - .400987 +02
13 .891478 +02	14 .574037 +02
15 - .449679 +02	16 - .205049 +03

Prepared	NAME W.D. BAUGH	DATE 7-13-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL I	PERM. 7.172
Checked	P.E. SCHREIBER	7-15-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	H.H. WADE	7-15-70				LG145 46-2-2 Report No.	

REV. (8) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427- 2516

7.5.3.2 FRAME CAP. LOADS - POUNDS, CONDITION 1PC242

1	.741064	+03	2	.877617	+03	3	.368171	+03
4	.629995	+03	5	-.146891	+04	6	-.166137	+04
7	.120638	+04	8	.101693	+04	9	.605553	+03
10	.689993	+02	11	.184837	+04	12	.225569	+04
15	.682477	+03	16	.912291	+03	17	.251661	+04
18	.229807	+04	21	.873545	+03	22	.130093	+04
23	.148246	+04	24	.101649	+04	27	.263153	+03
28	.629322	+03	29	.128437	+04	30	.325961	+03
33	-.127169	+04	34	-.969459	+03	35	.150435	+04
36	.121543	+04	39	-.276536	+04	40	-.244898	+04
41	.110133	+04	42	.928939	+03	45	-.391786	+03
46	-.523263	+01	47	-.588555	+04	48	-.618350	+04
49	-.101625	+03	50	-.225616	+03	51	-.811221	+03
52	-.976365	+03	53	-.346684	+03	54	-.118739	+05
55	.266889	+04	56	.740688	+03			

FRAME WEB LOADS - POUNDS/INCH, CONDITION 1PC242

1	-.731723	+00	2	-.499247	+02
3	-.159093	+02	4	-.345543	+02
5	-.598351	+00	6	-.247026	+00
7	.394621	+02	8	.428970	+02
9	.589674	+02	10	.595755	+02
11	.294050	+02	12	.282060	+02
13	-.169553	+03	14	-.172428	+03
15	-.495712	+01	16	-.243080	+02

Prepared	NAME J.P.Kimmel	DATE 11-9-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. VOL. I	PERM. 7-173
Checked	F.E.Schreiber	11-9-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A		
Approved	H.H.Wade	11-9-70		LGIUS46-2-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP AND WEB LOADS

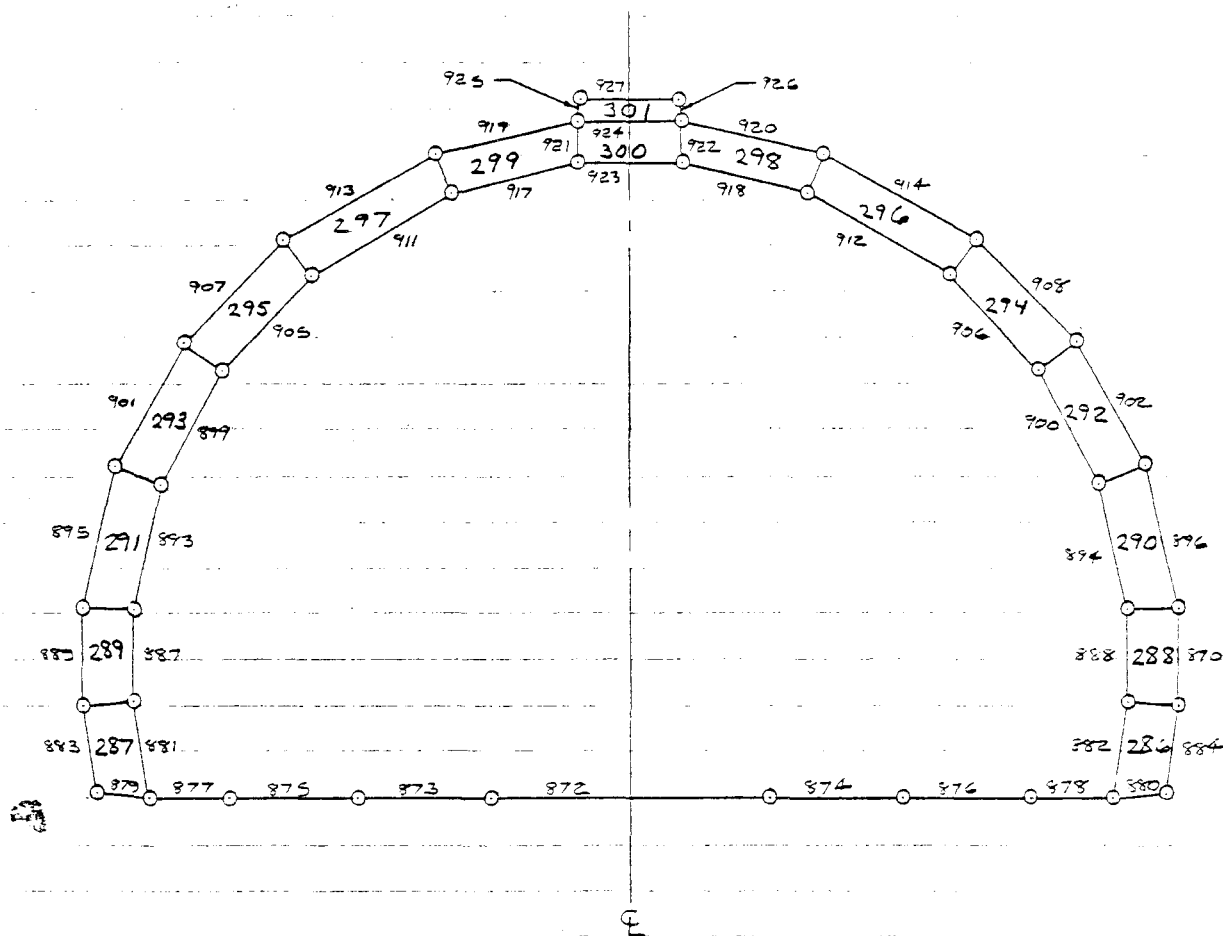


FIGURE 7.5.3.3

F.S. 2472

NUMBERS ARE FRAME BAR AND WEB NO. IN AFT FUSELAGE
MOD. 7 OF THE FAMAS MODEL, REF. 2, VOL. IV, P. 4.83

Prepared	NAME W.E. SEGGET	DATE 7-7-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	TEMP. VOL. II	PERM. 7.174
Checked	F.E. SCHREIBER	7-7-70	TITLE INTERNAL LOADS ANALYSIS	Model	C-5A	
Approved	W.H. WADE	7-7-70	AFT FUSELAGE	Report No.	LS20546-2-2	

REV. (B) J.P.K. 1-8-71

Z.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

Z.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 100400 (I=1.18)

872	.768335 + 03	873	.814312 + 03	874	.819132 + 03
875	.165777 + 04	876	.167002 + 04	877	.571911 + 04
878	.574881 + 04	879	-.668530 + 04	880	-.672743 + 04
881	.605222 + 03	882	.586565 + 03	883	.275291 + 03
884	.299830 + 03	887	.845462 + 03	888	.840589 + 03
889	.862049 + 03	890	.888981 + 03	893	.727647 - 02
894	.838689 + 02	895	.131314 + 04	896	.133113 - 04
899	-.201622 + 04	900	-.199704 + 04	901	.370075 + 04
902	.269962 + 04	905	-.484327 + 04	906	-.482417 + 04
907	.330156 + 04	908	.328476 + 04	911	-.595646 - 04
912	-.593320 + 04	913	-.967614 + 03	914	-.991238 + 03
917	.454663 + 03	918	.484635 + 03	919	-.144197 + 03
920	-.144522 + 05	921	-.216546 + 03	922	-.221711 + 03
923	.626475 + 04	924	-.220261 + 05	925	-.390471 + 04
926	-.391889 + 04	927	.660544 + 03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION 100400 (I=1.18)

286	-.277610 + 02	287	-.271875 + 02
288	.135897 + 01	289	.213637 + 01
290	.479647 + 02	291	.484498 + 02
292	.865414 + 02	293	.865773 + 02
294	.990509 + 02	295	.990171 + 02
296	-.197259 + 02	297	-.194776 + 02
298	-.382658 + 03	299	-.382513 + 03
300	-.138956 + 01	301	.121027 + 01

Prepared	NAME J. E. EGGETT	DATE 7-8-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP. PERM.
Checked	F. S. SCHERER	7-8-70	TITLE INTERNAL LOADS ANALYSIS		Page 1175
Approved	H. H. WADE	7-8-70	AFT FUSELAGE		Model 2-SA LG 108 46-3-2 Report No.

REV. (8) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427 - 2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION SPE201

872	- .103379+04	873	- .114806+04	874	- .128342+04
875	- .313901+04	876	- .339101+04	877	- .120462+05
878	- .122704+05	879	- .117720+05	880	- .133052+05
881	- .731299+03	882	- .271636+03	883	- .173756+04
884	- .220150+04	887	- .117125+04	888	- .138818+04
889	- .358399+04	890	- .358284+04	893	- .127138+04
894	- .174404+04	895	- .311609+04	896	- .281652+04
899	- .300535+03	900	- .565313+02	901	- .345488+04
902	- .300537+04	905	- .392281+04	906	- .369970+04
907	- .359037+04	908	- .314950+04	911	- .708336+04
912	- .667153+04	913	- .500580+03	914	- .101249+04
917	- .184588+04	918	- .108721+04	919	- .148723+05
920	- .157021+05	921	- .619424+03	922	- .787744+03
923	- .454553+04	924	- .237093+05	925	- .430900+04
926	- .470992+04	927	- .828448+03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION SPE201

286	- .599385 +02	287	- .151683 +02
288	- .504600 +02	289	- .275369 +02
290	- .134103 +02	291	- .137012 +02
292	- .954235 +03	293	- .878216 +02
294	- .150145 +03	295	- .149138 +03
296	- .369015 +02	297	- .477057 +02
298	- .397170 +03	299	- .387720 +03
300	- .391867 +02	301	- .331188 +01

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	J.E. FEGGETT	7-10-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	7.176
Checked	R.E. SCHREIBER	7-10-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	
Approved	H.H. WADE	7-10-70		US 46-2-2 Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION DML001

872	- .248222+03	873	- .292237+03	874	- .295633+03
875	- .151708+04	876	- .878476+02	877	- .740845+04
878	- .140139+04	879	- .293519+04	880	- .335912+04
881	- .714243+03	882	- .476882+03	883	- .112556+04
884	- .123143+03	887	- .305251+03	888	- .932423+03
889	- .345606+04	890	- .163759+04	893	- .389547+02
894	- .815058+03	895	- .279904+04	896	- .129768+04
899	- .456823+03	900	- .446676+03	901	- .191229+04
902	- .313903+03	905	- .399349+03	906	- .140158+04
907	- .137358+04	908	- .276074+03	911	- .297681+04
912	- .373344+03	913	- .913554+03	914	- .124509+04
917	- .380057+04	918	- .302917+04	919	- .738882+03
920	- .667663+04	921	- .606169+03	922	- .961883+03
923	- .107065+04	924	- .576699+04	925	- .686929+03
926	- .289458+04	927	- .207316+03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION DML001

286	- .110639+03	287	- .925134+02
288	- .319439+02	289	- .117820+02
290	- .303074+02	291	- .253056+02
292	- .508517+02	293	- .642161+01
294	- .116971+02	295	- .622838+02
296	- .630689+02	297	- .857175+02
298	- .141346+03	299	- .486395+02
300	- .358011+03	301	- .347154+02

NAME		DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	W.D. BAUGH	7-15-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	7.177
Checked	F.E. SCHREIBER	7-15-70	TITLE		Model C-5A	
Approved	H.H. WADE	7-15-70	INTERNAL LOADS ANALYSIS		LG105 46-2-2	
			AFT FUSELAGE		Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427 - 2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 1PC242

372	.616803 +03	373	.660125 +03	374	.810656 +03
375	.190238 +04	376	.216958 +04	377	.754967 +04
378	.771857 +04	379	-.681411 +04	380	-.909824 +04
381	.532699 +03	382	.135956 +02	383	.106044 +04
384	.156345 +04	387	.649800 +03	388	.902611 +03
389	.237312 +04	390	.231974 +04	393	.755333 +03
394	.128831 +04	395	.206629 +04	396	.168207 +04
399	-.470554 +02	399	.349354 +03	391	.216016 +04
392	.162885 +04	395	-.216992 +04	396	-.191257 +04
397	.216672 +04	398	.167521 +04	391	-.424510 +04
392	-.377013 +04	393	-.131279 +03	394	-.670211 +03
397	-.147497 +04	398	-.639786 +03	399	-.835481 +04
392	-.917602 +04	391	-.367759 +03	392	-.573066 +03
393	.237635 +04	394	-.136393 +05	395	-.241342 +04
396	-.283542 +04	397	.489931 +03		

FRAME WEB LOADS - POUNDS/INCH, CONDITION 1PC242

286	-.482335 +02	287	.333355 +02
288	-.397198 +02	289	-.140136 +02
290	.366159 +01	291	.355861 +01
292	.567933 +02	293	.482893 +02
294	.909936 +02	295	.904936 +02
296	.252365 +02	297	.372030 +02
298	-.228113 +03	299	-.219163 +03
300	-.421544 +02	301	-.205204 +01

Prepared	J.P.Kimmel	NAME	DATE	11-9-70	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
					A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	VOL. I 7.173
Checked	F.E.Schreiber	11-9-70	TITLE	INTERNAL LOADS ANALYSIS			
				AFT FUSELAGE			
Approved	H.H.Wade	11-9-70				Model	C-5A
						LGIUS46-2-2	
						Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP AND WEB LOADS

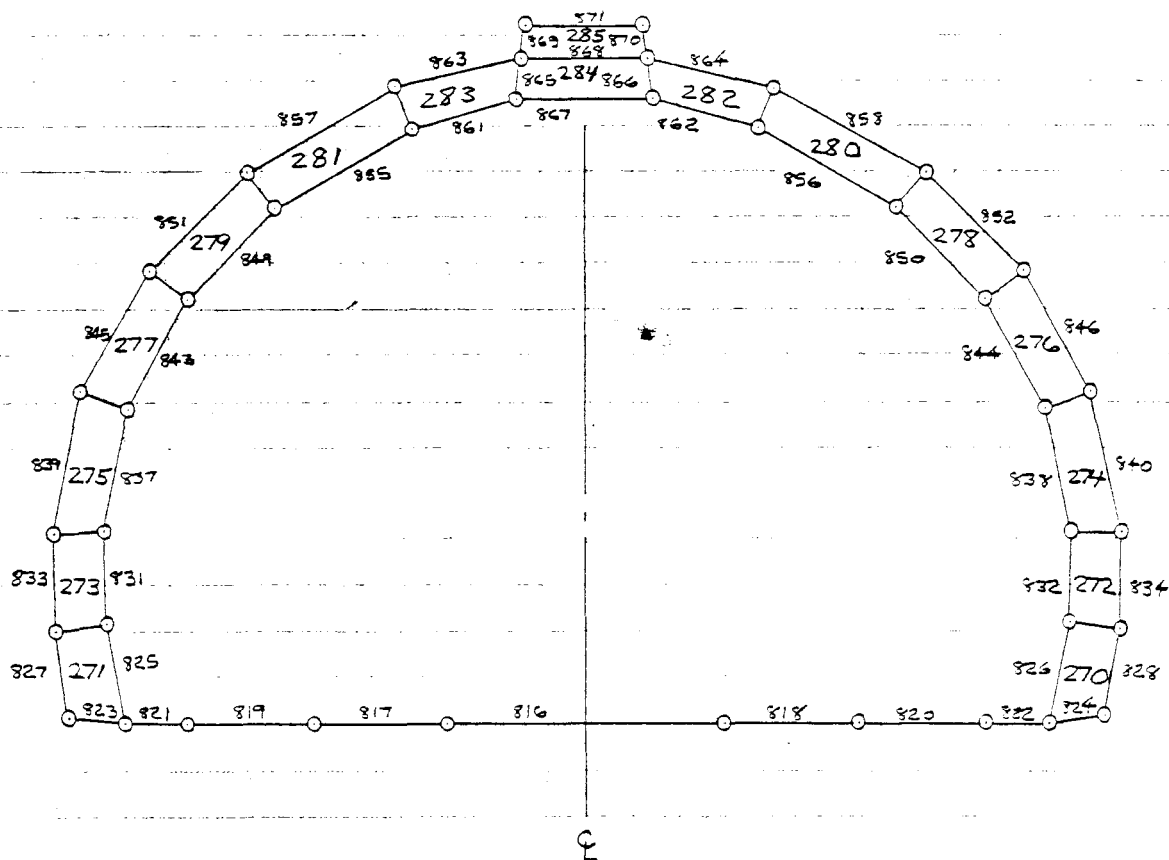


FIGURE 7.5.3.4

F.S. 2494

NUMBERS ARE FRAME BAR AND WEB NO. IN AFT FUSELAGE MOD. 7 OF THE FAMAS MODEL, REF. 8, VOL IV, P. 4.22

Prepared	NAME T. E. LEGGETT	DATE 7-7-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page 1	TEMP. 7.179	PERM. 7.179
Checked	F. E. SCHNEIDER	7-7-70	TITLE INTERNAL LOADS ANALYSIS	Model C-52		
Approved	H. H. WADE	7-7-70	AFT FUSELAGE	LG105 48-2-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-3516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 100400 (± 1.18)

816	.486938+03	817	.332344+03	818	.333393+03
819	-.913789+03	820	-.914696+03	821	-.720757+04
822	-.724410+04	823	-.734540+04	824	-.737596+04
825	.285576+03	826	.271276+03	827	-.321366+03
828	-.301567+03	831	-.302659+02	832	-.369276+02
833	-.119633+03	834	-.100847+03	837	-.714344+03
838	-.705988+03	839	.122992+04	840	.124167+04
843	-.234642+04	844	-.232730+04	845	.332059+04
846	.332135+04	849	-.501579+04	850	-.499229+04
851	.354103+04	852	.352652+04	855	-.510062+04
856	-.607678+04	857	-.119229+04	858	-.121149+04
861	-.429537+02	862	-.182452+02	863	-.142512+03
864	-.142790+03	865	-.476560+03	866	-.477245+03
867	.523436+04	868	-.218329+05	869	-.403218+04
870	-.403921+04	871	.998116+04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION 100400 (± 1.13)

270	.102700 +02	271	.105447 +02
272	.192886 +02	273	.197453 +02
274	.298376 +02	275	.304328 +02
276	.767818 +02	277	.769500 +02
278	.100391 +03	279	.100379 +03
280	-.259286 +02	281	-.258026 +02
282	-.394481 +03	283	-.394576 +03
284	-.809668 +00	285	.532829 +00

Prepared	NAME J.E. LEGGETT	DATE 7-8-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page 2 of 2	TEMP. 7.190	PERM. 7.190
Checked	F.E. SCHOFER	7-8-70	TITLE INTERNAL LOADS ANALYSIS	Model C-5A		
Approved	H.H. WADS	7-8-70	AFT FUSELAGE	LGVS 46-2-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427 - 2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION SPE201

816	- .380415 +03	817	- .224590 +02	818	- .216624 +03
819	.253194 +04	820	.233298 +04	821	.152874 +03
822	.157111 +05	823	.158777 +05	824	.159001 +05
825	- .637123 +02	826	.280727 +03	827	- .374589 +03
828	- .668639 +03	831	.786370 +03	832	.490777 +03
833	- .168874 +04	834	- .145960 +04	837	.463681 +03
838	- .385342 +02	839	- .327917 +04	840	- .384201 +04
843	.883222 +03	844	.562869 +03	846	- .478138 +04
846	- .433504 +04	849	.384332 +04	850	.375384 +04
851	- .399812 +04	852	- .351599 +04	855	.676425 +04
856	.640381 +04	857	.694194 +03	858	.151308 +04
861	.231006 +04	862	.114960 +04	863	.141544 +05
864	.155993 +05	865	.794310 +03	866	.104106 +04
867	- .353089 +04	868	.231638 +05	869	.413472 +04
870	.480887 +04	871	- .118558 +05		

FRAME WEB LOADS - POUNDS/INCH, CONDITION SPE201

270	- .324435 +02	271	- .761736 +02
272	.988077 +01	273	- .987238 +01
274	.280373 +02	275	.290482 +02
276	- .687191 +02	277	- .575763 +02
278	- .143496 +03	279	- .139315 +03
280	- .236575 +02	281	- .415658 +02
282	.402760 +03	283	.369387 +03
284	.567001 +02	285	- .362538 +01

Prepared	NAME J.S. EGGETT	DATE 7-10-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	F.S. SCHREIBER	7-10-70	TITLE INTERNAL LOAD ANALYSIS AFT FUSELAGE		Page	7/81
Approved	U.H. WADE	7-10-70			Model	2-5A
					LG105 46-2-2 Report No.	

REV. (8) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP. LOADS - POUNDS, CONDITION DML001

816	- .858070 +02	817	.474810 +03	818	- .518282 +03
819	.208288 +02	820	- .866562 +03	821	.894320 +04
822	- .127516 +04	823	.102925 +05	824	- .242969 +04
825	- .409417 +03	826	.473420 +03	827	- .191985 +03
828	- .893166 +02	831	.167229 +04	832	- .135129 +04
833	- .238839 +04	834	.156262 +04	837	.126723 +04
838	- .119367 +04	839	- .304452 +04	840	.149377 +04
843	- .288029 +03	844	.570341 +03	845	- .268980 +04
846	.444232 +03	849	- .328561 +03	850	.210520 +04
851	- .189772 +04	852	.703449 +02	855	.255794 +04
856	.602830 +03	857	- .239723 +04	858	.289274 +04
861	.594045 +04	862	- .505871 +04	863	- .244992 +04
864	.964578 +04	865	- .102083 +04	866	.147450 +04
867	- .820751 +03	868	.558525 +04	869	- .215823 +04
870	.432376 +04	871	- .293678 -04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION DML001

270	.139648 +03	271	- .167059 +03
272	.428017 +02	273	- .418187 +02
274	- .425117 +02	275	.583281 +02
276	- .737124 +02	277	.436349 +02
278	- .280716 +02	279	- .412458 +02
280	.104453 +03	281	- .122019 +03
282	.264417 +03	283	- .794852 +02
284	.551053 +03	285	.286399 +02

Prepared	NAME W. D. BAUGH	DATE 7-15-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. V/L = 7.182	PERM.
Checked	F. E. SCHREIBER	7-15-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-5/A		
Approved	H. H. WADE	7-15-70			LG/US 46-Z-2 Report No.		

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427 - 2516

7.5.3.2 FRAME CAP. LOADS - POUNDS, CONDITION 1PC242

816	.20 33 12 +03	817	-.6877 36 +02	818	.15 4442 +03
819	-.16 78 65 +04	820	-.14 29 88 +04	821	-.9553 49 +04
822	-.99 29 38 +04	823	-.10 03 76 +05	824	-.99 45 78 +04
825	.38 31 37 +02	826	-.30 10 82 +03	827	.23 26 26 +03
828	.55 32 97 +03	831	-.60 93 30 +03	832	-.25 35 66 +03
833	.12 37 86 +04	834	.94 07 50 +03	837	-.35 12 78 +03
838	.22 94 66 +03	839	.22 16 61 +04	840	.16 80 07 +04
843	-.39 85 59 +03	844	-.42 23 38 +02	845	.29 95 63 +04
846	.24 60 85 +04	849	-.20 58 76 +04	850	-.19 63 93 +04
851	.24 29 99 +04	852	.18 82 66 +04	855	-.39 96 42 +04
856	-.35 80 35 +04	857	-.15 93 30 +03	858	-.10 64 44 +04
861	-.18 40 46 +04	862	-.51 58 73 +04	863	-.77 19 07 +04
864	-.92 81 34 +04	865	-.43 26 01 +03	866	-.71 01 03 +03
867	.18 02 28 +04	868	-.13 28 18 +05	869	-.22 09 14 +04
870	-.29 61 29 +04	871	.69 46 42 +04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION 1PC242

270	.10 21 59 +02	271	.61 49 92 +02
272	-.13 64 61 +02	273	.90 18 60 +01
274	-.22 03 95 +02	275	-.24 04 83 +02
276	.40 71 39 +02	277	.27 71 27 +02
278	.87 10 28 +02	279	.82 75 08 +02
280	.14 92 59 +02	281	.35 66 94 +02
282	-.23 64 61 +03	283	-.19 90 99 +03
284	-.64 35 88 +02	285	-.48 88 24 +00

Prepared	NAME	DATE	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
	J.P.Kimmel	11-9-70		Page	VOL. II 7.133
Checked	F.E.Shreiber	11-9-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	Model C-5A	
Approved	H.H.Wade	11-9-70		LGLUS46-2-2 Report No.	

REV. (B) J.P.KIMMEL 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP AND WEB LOADS

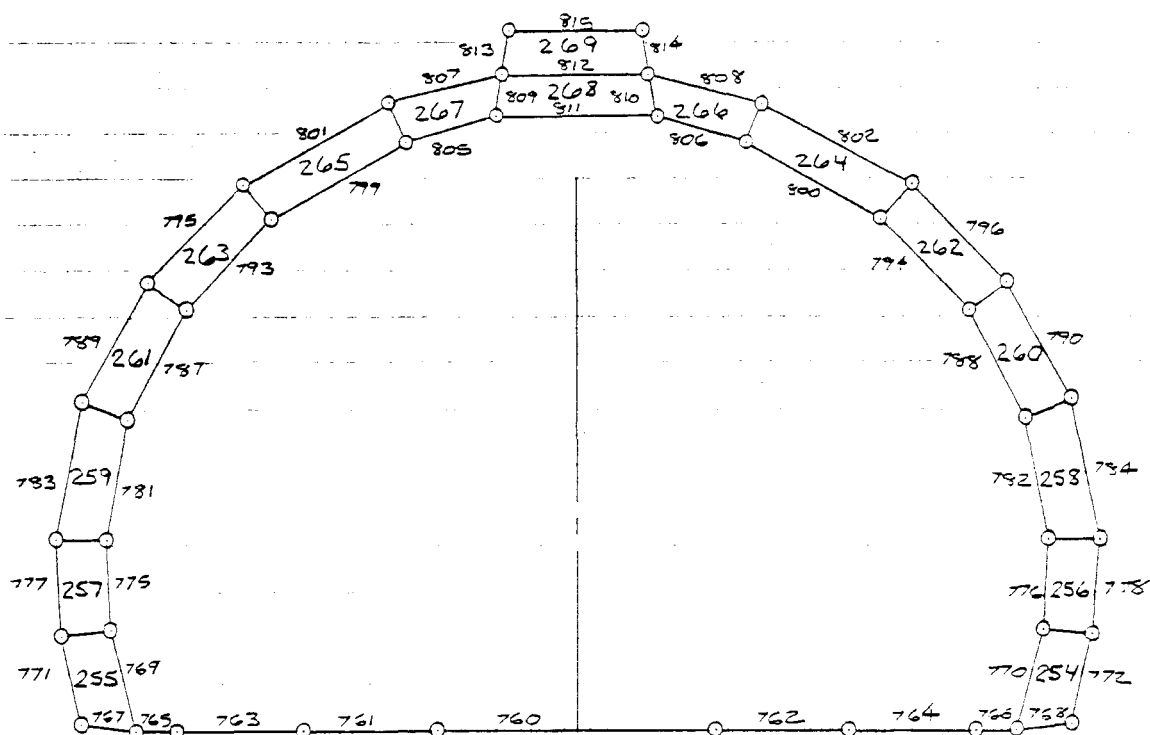


FIGURE 7.5.3.5
F.S. 2516

NUMBERS ARE FRAME BAR AND WEB NO. IN AFT FUSELAGE
MOD. 7 OF THE FAMAS MODEL, REF. 8, VOL IV, P. 4.81

Prepared	NAME J. E. LEGGETT	DATE 7-7-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page Vol. I	TEMP.	PERM. 7.194
Checked	E. E. SCHREIBER	7-7-70	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	H. H. WADE	7-7-70				Report No.	

REV. (8) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 3427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION 100400 (t=1.18)

760 - .610772+02	761 - .197249+03	762 - .194982+03
763 - .995786+03	764 - .998722+03	765 - .171392+04
766 - .172341+04	767 - .199977+04	768 - .200378+04
769 - .654762+02	770 - .535854+02	771 - .617581+02
772 - .443650+02	775 - .662763+03	776 - .666114+03
777 - .252690+02	778 - .706996+01	781 - .134853+04
782 - .133764+04	783 - .951009+03	784 - .962253+03
787 - .242818+04	788 - .240633+04	789 - .337361+04
790 - .337275+04	793 - .469719+04	794 - .464244+04
795 - .381739+04	796 - .380457+04	799 - .599030+04
800 - .597041+04	801 - .212535+04	802 - .213475+04
805 - .399132+03	806 - .388795+03	807 - .143503+05
808 - .143745+05	809 - .751519+03	810 - .748462+03
811 - .462326+04	812 - .204467+05	813 - .413408+04
814 - .413373+04	815 - .505012+04	

FRAME WEB LOADS - POUNDS/INCH, CONDITION 100400 (t=1.18)

254 - .324872 +02	255 - .327808 +02
256 - .366572 +02	257 - .371690 +02
258 - .183615 +02	259 - .189941 +02
260 - .550025 +02	261 - .551660 +02
262 - .991797 +02	263 - .990907 +02
264 - .106378 +02	265 - .107398 +02
266 - .426546 +03	267 - .427140 +03
268 - .814894 -01	269 - .772263 +00

Prepared	NAME J. E. LEGGETT	DATE 7-8-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		TEMP.	PERM.
Checked	F. E. SCHNEIDER	7-8-70	TITLE INTERNAL LOADS ANALYSIS		Page	7.185
Approved	H. H. WADE	7-8-70	AFT FUSELAGE		Model	C-5A
					Report No.	LG10546-2-2

REV. (8) APR. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION SPE201

760	.873824 +03	761	.136524 +04	762	.101308 +04
763	.309203 +04	764	.263866 +04	765	.309808 +04
766	.340316 +04	767	.391501 +04	768	.377350 -04
769	.185537 +03	770	.507485 +03	771	-.120991 +04
772	-.130953 +04	775	.152900 +04	776	.119486 +04
777	-.282528 +04	778	-.231324 +04	781	.127606 +04
782	.717746 +03	783	-.354941 +04	784	-.305232 +04
787	.114466 +04	788	.832812 +03	789	-.486241 +04
790	-.446472 +04	793	.343447 +04	794	.350474 +04
795	-.412021 +04	796	-.356842 +04	799	.624174 +04
800	.598216 +04	801	.135360 +04	802	.277974 +04
805	.256380 +04	806	.931849 +03	807	.133935 +05
808	.157308 +05	809	.914109 +03	810	.125853 -04
811	-.305721 +04	812	.212811 +05	813	.389863 -04
814	.489886 +04	815	.621411 +04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION SPE201

254	-.609982 +02	255	-.105343 +03
256	-.809666 +01	257	-.279722 +02
258	.376515 +02	259	.370538 +02
260	-.463150 +02	261	-.294204 +02
262	-.135258 +03	263	-.126308 +03
264	-.250669 +02	265	-.504543 +02
266	.435874 +03	267	.362737 +03
268	.745965 +02	269	.222552 +01

Prepared	NAME J.E. BIGGETT	DATE 7-10-70	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page	TEMP. VOL. 2	PERM. 7-186
Checked	F.E. SCHREIBER	7-12-70	TITLE INTERNAL (LOAD) ANALYSIS AFT FUSELAGE			Model C-5A	
Approved	H.H. WADE	7-10-70				LG 105 46-2-2 Report No.	

REV. (B) J.P.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP LOADS - POUNDS, CONDITION DML001

760	.225434+03	761	.166298+04	762	-.105641+04
763	.312253+04	764	-.168946+04	765	.989500+03
766	.610675+03	767	.262621+04	768	-.732898+03
769	-.443392+03	770	.621707+03	771	-.112118+04
772	.471484+03	775	.194184+04	776	-.127448+04
777	-.407196+04	778	.274290+04	781	.168795+04
782	-.123711+04	783	-.359505+04	784	.191730+04
787	-.243736+03	788	.654865+03	789	-.250436+04
790	.218103+03	793	-.112097+04	794	.272556+04
795	-.221650+04	796	.367151+03	799	.208035+04
800	.824628+03	801	-.508391+04	802	.602678+04
805	.837341+04	806	-.750394+04	807	-.660902+04
808	.135260+03	809	-.145294+04	810	.197855+04
811	-.702752+03	812	.508170+04	813	-.369355+04
814	.580211+04	815	.150748+04		

FRAME WEB LOADS - POUNDS/INCH, CONDITION DML001

254	.144270 +03	255	-.185565 +03
256	.429848 +02	257	-.505500 +02
258	-.335729 +02	259	.534705 +02
260	-.935456 +02	261	.758316 +02
262	-.468371 +02	263	-.165882 +02
264	.142892 +03	265	-.162264 +03
266	.458189 +03	267	-.269431 +03
268	.731694 +03	269	.121418 +03

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared	W. D. BAUGH	7-15-70	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol I 7.197
Checked	F. E. SCHREIBER	7-15-70	TITLE	Model C-3A	
Approved	H. H. WADE	7-15-70	INTERNAL LOADS ANALYSIS AFT FUSELAGE	LGius 46-2-2 Report No.	

REV. (3) J.R.K. 1-8-71

7.5.3 FRAMES - FUSELAGE STATIONS 2427-2516

7.5.3.2 FRAME CAP. LOADS - POUNDS, CONDITION 1PC242

760	- .589308	+03	761	- .997634	+03	762	- .580290	+03
763	- .211732	+04	764	- .156323	+04	765	- .185478	+04
766	- .218369	+04	767	- .248786	+04	768	- .229512	+04
769	- .602109	+02	770	- .426624	+03	771	- .820812	+03
772	- .913002	+03	775	- .105123	+04	776	- .652529	+03
777	- .207454	+04	778	- .144077	+04	781	- .844349	+03
782	- .200882	+03	783	- .244781	+04	784	- .184104	+04
787	- .562933	+03	788	- .221607	+03	789	- .303138	+04
790	- .255492	+04	793	- .177405	+04	794	- .137129	+04
795	- .250069	+04	796	- .136437	+04	799	- .303065	+04
800	- .332167	+04	801	- .341909	+03	802	- .197766	+04
805	- .210546	+04	806	- .185980	+03	807	- .697099	+04
808	- .956492	+04	809	- .454313	+03	810	- .854255	+03
811	- .154889	+04	812	- .121509	+05	813	- .195177	+04
814	- .309326	+04	815	- .365609	+04			

FRAME WEB LOADS - POUNDS/INCH, CONDITION 1PC242

254	- .269214	+02	255	- .790229	+02
256	- .343675	+01	257	- .193936	+02
258	- .276339	+02	259	- .277649	+02
260	- .294207	+02	261	- .951545	+01
262	- .826753	+02	263	- .217990	+02
264	- .118989	+02	265	- .126384	+03
266	- .265367	+03	267	- .319332	+03
268	- .872506	+02	269	- .412141	+02

PREPARED	P. T. Fletcher	8-22-68	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	Vol. I	7.188
CHECKED	J. Haliday	8-23-68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE	MODEL	C-5A	
APPROVED	L. G. Darby	8-23-68		LG1US46-2-2	REPORT NO.	

7.6 TORQUE DECK

REV. (2) PTF 1 AUG. 69

7.6.1 Aft Torque Deck, F.S. 2100 thru F.S. 2538

For the geometry and description of this structure see the Stress Analysis Report (Reference 15, PG. 7.229).

The internal loads in this area are determined by the FAMAS Computer Program (REF. 8, VOL. 4, REV. A). The honeycomb panels of the deck are represented by a grid of stringers and covers, the stringers are simply the lumped area of the face sheets over the particular distance between adjacent grid points plus any additional area from continuous members such as longerons, edge members or longitudinal beams. The output from FAMAS gives all critical loads due to flight cases and a separate case involving pressurization loads only. (This particular case has different flexibilities in some areas which are subject to internal pressure to allow for the tension capability of the skin shell).

For the area of the torque deck between the pressure bulkhead at F.S. 2100 and F.S. 2273.0 these pressure loads are significant and are included with the flight cases.

A scan of flight cases which give overall maximum tension, compression and shear + corresponding compression loading on the deck with an ultimate pressure case, is given (8.7 x 1.5 P.S.I) and can be added to the flight cases.

The following four cases show up as critical in most areas of the torque deck.

1. 5FE201 Pitch Entry
2. 1PC242 Positive Checked Pitch
3. DML001 Dynamic Lateral Gust
4. 5AR154T2 Aileron Roll

In analyzing the overall stability of the deck when subject to compressive axial loads, it is necessary to consider the change in angles of the panels at F.S. 2273 and F.S. 2538, which will induce a running vertical load onto the corresponding transverse frame beams at these frame stations, as these beams deflect so will the adjacent transverse beams also.

The angle of the sloping torque deck with respect to the horizontal deck (F.S. 2273-2538) is 13° , i.e., $N_z = N_x \tan 13^\circ 13' = 0.235 N_x$ lb/in also at F.S. 2538, $N_z = N_x \tan 4^\circ$ $N_z = 0.070 N_x$ lb/in.

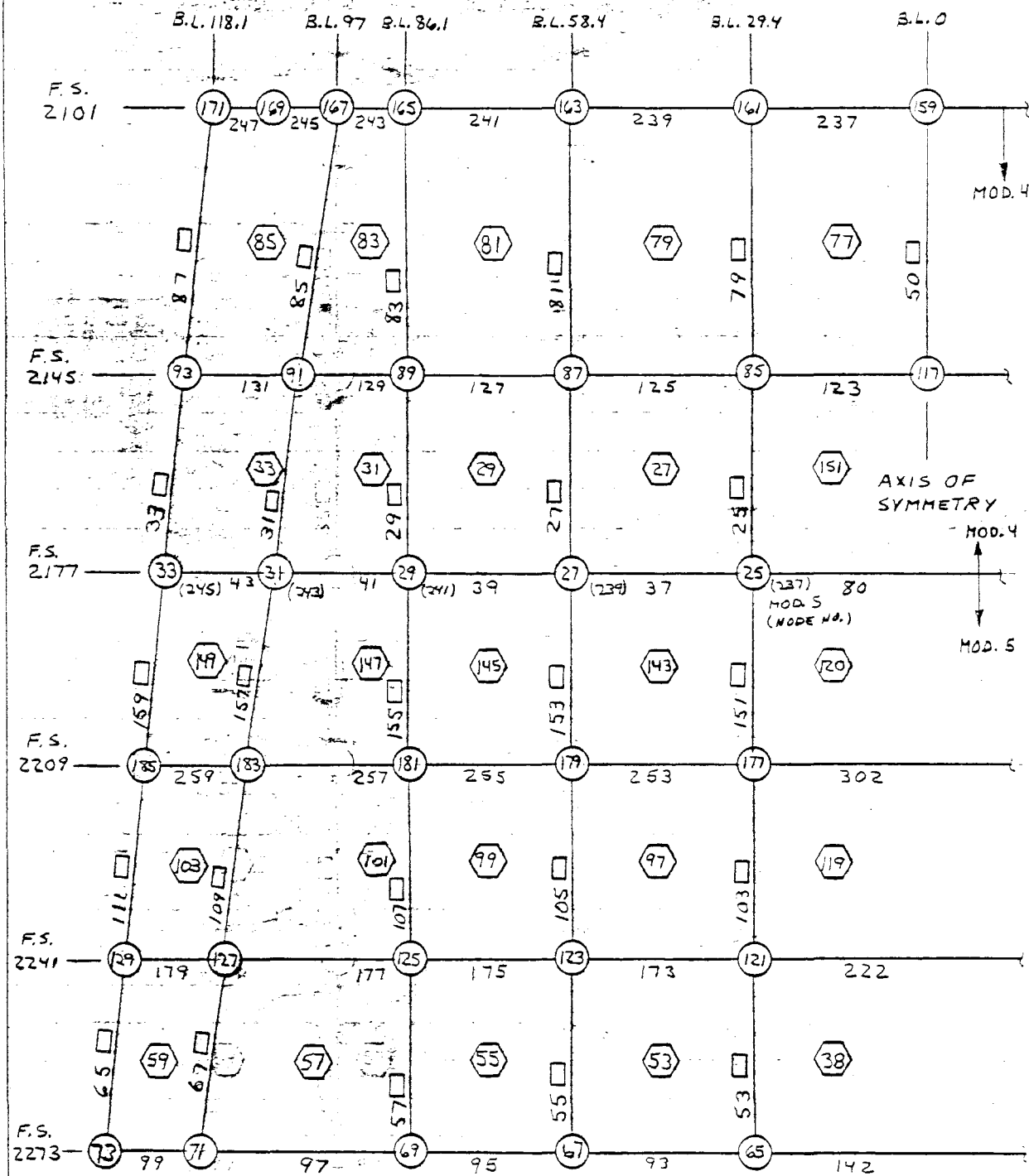
Other loads taken by the aft torque shelf are the aft cargo door actuator screw jack load between F.S. 2317 and F.S. 2339 (Dwg. 4F41245) Ref. Pg. 13.41 and the support load at F.S. 2251 for the driver arm, from Ref. Pg. 13.50 control cables and hydraulic pipes are attached to the beams in places but impose insignificant loads.

The transverse beams at F.S. 2361 thru 2427 are fairly highly loaded with flight conditions because they are close to the sloping longeron intersection with the torque deck longeron where there is a substantial change in the moment of inertia of the fuselage.

The most critical frame is at F.S. 2383, and the bending moments and shear forces for the cross beam are given on pg. 7.160, para. 7.5.2.3; & Pg. 7.161, para. 7.5.2.6.

NAME	DATE	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
Prepared P.T. Fletcher	30 Apr 68	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page	Vol. I 7.139
Checked J. Halliday	30 Apr 68	TITLE	Model C-5A	
Approved L.G. Darby	30 Apr 68	AFT FUSELAGE INTERNAL LOADS ANALYSIS	UGIUS 46-2-2	
			Report No.	

7.6.1 REV. (A) P.T.F. AUG. 69
AFT SLOPING TORQUE SHELF F.S. 2100 - 2273. (REF. 8 - VOL. 4 SECT. 2
 L/H SHOWN, R/H ARE ONE NUMBER HIGHER REV. X)



- NODE OR GRID POINT
- ⬡ PANEL COVER
- STRINGER
- △ 97 FRAME BAR

Prepared	NAME P.T. Fletcher	DATE 30 Apr 68	LOCKHEED-GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	TEMP.	PERM.
Checked	J. Halliday	30 Apr 68		Page	Vol. I
Approved	L.G. Darby	30 Apr 68	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	Model	C-5A
				LGJUS	46-2-2
				Report No.	

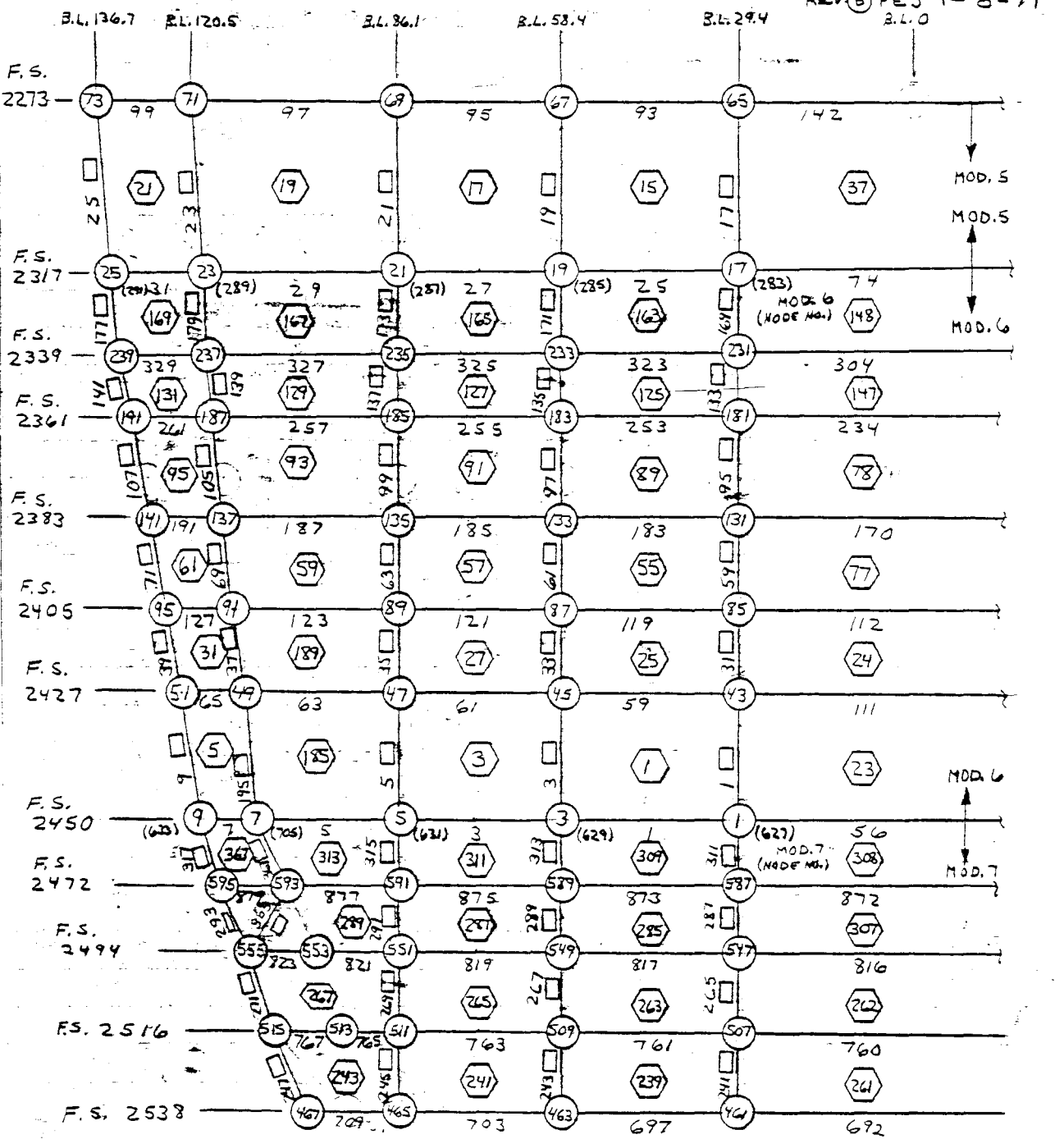
7.6.E

REV. A P.T.F. 1 AUG 69

AFT TORQUE SHELF F.S. 2273 - 2538. (REF. 8 VOL. 4 SECT. 2 REV. X)

L/H SHOWN, R/H ARE ONE NUMBER HIGHER

REV. B FES 1-8-71



○ GRID POINT
 ◡ PANEL COVER
 □ STRINGER
 709 FRAME BAR

NAME	DATE	LOCKHEED-GEORGIA COMPANY		TEMP.	PERM.
Prepared P.T.Fletcher	28 Jun 68	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION		Page Vol. 1	7.191
Checked J. Halliday	28 Jun 68	TITLE INTERNAL LOADS ANALYSIS AFT FUSELAGE		Model C-32	
Approved L. G. Darby	28 Jun 68			LG1US 46-2-2 Report No.	

Rev. (A) PTF 8-1-69

7.6.1.

AFT TORQUE DECK PANEL - COVERS
AND STRINGERS

DERIVED FROM REF. 8 - VOL. 4, SECT. 2 REV. (A).

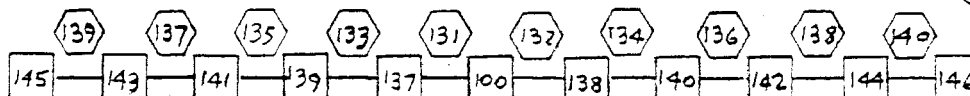
THE CASE CODE NOS AND REF NOS WHICH APPEAR CRITICAL FOR THE TORQUE DECK BETWEEN F.3.2101 AND F.5.2538 ARE LISTED BELOW. THE SHEAR LOAD IN ANY ONE COVER ELEMENT IS HOWEVER NOT NECESSARILY THE HIGHEST. THE THREE CASES CHOSEN ARE THE MAXIMUM TENSION LOAD CASE AND THE TWO CASES COMPRESSION AND SHEAR CRITICAL FOR OVERALL PANEL STABILITY. THE MAX. SHEAR LOAD IN INDIVIDUAL PANELS CAN BE OBTAINED IN THE 'SCAN' OF ALL HIGHEST LOADS. (SEE PAGE 7.209)

CASE CODE	DESCRIPTION	CASE REF. NO.
SPE 201	POSITIVE PITCH ENTRY	48
IPC 242	CHECKED PITCH	17
DML 001	DYNAMIC LATERAL GUST	74
SAR 154T2	AILERON ROLL	40

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	Vol. 1	7.192
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	LGIUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (2) P.T.F. 1 AUG, 69



F.S. 2101-2084 (MOD. 4)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
145	3.30	-6672	17			139	133
143	1.15	-545	17	139	133	137	73
141	1.14	67	17	137	73	135	49
139	1.67	128	17	135	49	133	-73
137	1.72	513	17	133	-73	131	19
100	1.74	434	17				
138	1.72	2423	17	134	206	132	19
140	1.67	2642	17	136	206	134	206
142	1.14	1969	17	138	251	136	206
144	1.15	-274	17	140	263	138	251
146	3.33	-5715	17			140	263

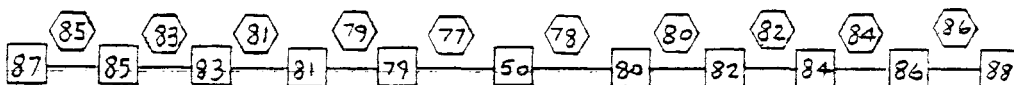
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
145	3.30	15414	48			139	-322
143	1.15	2291	48	139	-322	137	-252
141	1.14	1550	48	137	-252	135	-170
139	1.67	1579	48	135	-170	133	2
137	1.72	1311	48	133	2	131	-15
100	1.74	-258	48				
138	1.72	-124	48	134	-236	132	-15
140	1.67	-378	48	136	-307	134	-236
142	1.14	43	48	138	-414	136	-306
144	1.15	2297	48	140	-438	138	-414
146	3.33	15089	48			140	-438

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/E STRG.	AXIAL LOAD (lbs.)
139	.059	185	40	145	27	143	-511
137	.059	290	40	143	-511	141	2168
135	.059	248	40	141	2168	139	2603
133	.059	408	40	139	2603	137	1527
131	.059	-4	40	137	1528	100	-61
132	.059	-4	40	100	-61	138	-2511
134	.059	-337	40	138	-2511	140	-3329
136	.059	-234	40	140	-3329	142	-2593
138	.059	-271	40	142	-2593	144	-138
140	.059	-208	40	144	-138	146	-956

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	Vol 1	7.193
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	LGLUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2145.776 - 2101.090 (Mod #4)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
87	.69	717	17	-		85	144
85	1.41	1922	17	85	144	83	112
83	1.03	2019	17	83	112	81	49
81	1.23	2223	17	81	49	79	6
79	1.31	1866	17	79	6	77	-90
50	1.32	472	17	77 78		-	
80	1.31	5336	17	80	126	78	71
82	1.23	3864	17	82	176	80	127
84	1.03	3477	17	84	229	82	176
86	1.41	2465	17	86	262	84	229
88	.69	2501	17	-		86	262

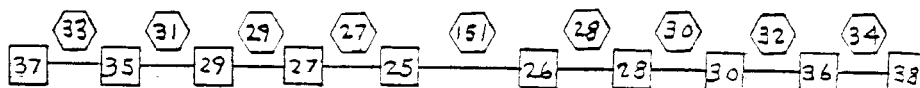
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
87	.69	-463	48	-		85	-226
85	1.41	-1365	48	85	-226	83	-228
83	1.03	-1860	48	83	-228	81	-134
81	1.23	-1709	48	81	-134	79	-54
79	1.31	-1478	48	79	-54	77	74
50	1.32	-272	48	77 78		-	
80	1.31	-4294	48	80	-160	78	-64
82	1.23	-2979	48	82	-246	80	-160
84	1.03	-3021	48	84	-333	82	-246
86	1.41	-1639	48	86	-335	84	-333
88	.69	-1974	48	-		86	-335

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
85	.045	183	40	87	2157	85	100
83	.045	190	40	85	100	83	1618
81	.045	200	40	83	1618	81	1601
79	.045	186	40	81	1621	79	4031
77	.045	235	40	79	4031	50	-105
78	.045	-234	40	50	-105	80	-4362
80	.045	-184	40	80	-4362	82	-2228
82	.045	-195	40	82	-2228	84	-2047
84	.045	-182	40	84	-2047	86	-732
86	.045	-179	40	86	-732	88	-2335

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	TEMP. VOL. 1	REV. 7.194
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	LGLUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2177.577 - 2145.776 (MOD. #4)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
37	.91	-1061	17	-		33	79
35	1.14	4007	17	33	79	31	48
29	.93	3840	17	31	48	29	22
27	1.06	3753	17	29	22	27	-13
25	1.65	5135	17	27	-13	151	81
26	1.65	7230	17	28	122	151	81
28	1.06	5299	17	30	144	28	122
30	.93	5049	17	32	167	30	144
36	1.14	4677	17	34	178	32	167
38	.91	-1802	17	-		34	178
			17				

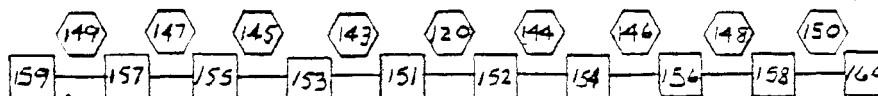
STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
37	.91	2360	48	-		33	-201
35	1.14	-4491	48	33	-201	31	-148
29	.93	-4798	48	31	-148	29	-98
27	1.06	-4496	48	29	-98	27	-36
25	1.65	-6111	48	27	-36	151	-70
26	1.65	-7831	48	28	-154	151	-70
28	1.06	-5747	48	30	-206	28	-154
30	.93	-5778	48	32	-254	30	-206
36	1.14	-4978	48	34	-291	32	-254
38	.91	3134	48	-		34	-291
			48				

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
33	.0375	141	40	37	-1726	35	448
31	.0375	171	40	35	448	29	1346
29	.0375	183	40	29	1346	27	1711
27	.0375	201	40	27	1711	25	2367
151	.0375	-238	40	25	2367	26	-3038
28	.0375	-202	40	26	-3038	28	-2115
30	.0375	-185	40	28	-2115	30	-1644
32	.0375	-184	40	30	-1643	36	-856
34	.0375	-151	40	36	-856	38	1266
			40				

NAME	DATE	LOCKHEED - GEORGIA COMPANY		TEMP.	PERM.
PREPARED	P.T. Fletcher	28 Jun 69	A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL. 1 7.195
CHECKED	R. Wilkinson	28 Jun 69	TITLE	MODEL	C-5A
APPROVED	L.G. Darby	28 Jun 69		IGLUS 46-2-2	REPORT NO.

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) A.T.F. 1 AUG. 69



F.S. 2209 - 2177 (MOD. 5)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
159	1.10	2243	17			149	113
157	1.23	5496	17	149	113	147	53
155	1.01	5002	17	147	53	145	7
153	1.06	4853	17	145	7	143	-26
151	1.65	7124	17	143	-26	120	80
152	1.65	8479	17	144	116	120	80
154	1.06	5967	17	146	136	144	116
156	1.01	5938	17	148	168	146	136
158	1.23	5432	17	150	204	148	168
160	1.10	3375	17			150	204
			17				

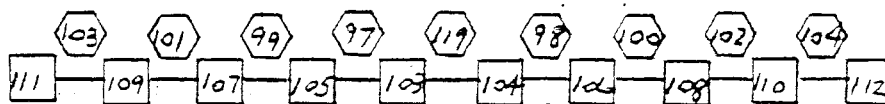
STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
159	1.10	-3124	48			149	-233
157	1.23	-6795	48	149	-233	147	-146
155	1.01	-6722	48	147	-146	145	-78
153	1.06	-6424	48	145	-78	143	-21
151	1.65	-9303	48	143	-21	120	-69
152	1.65	-10427	48	144	-145	120	-69
154	1.06	-7339	48	146	-191	144	-145
156	1.01	-7482	48	148	-248	146	-191
158	1.23	-6676	48	150	-315	148	-248
160	1.10	-4024	48			150	-315

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
149	.029	131	40	159	714	157	-727
147	.0375	168	40	157	-727	155	894
145	.0375	190	40	155	894	153	1138
143	.0375	209	40	153	1138	151	1369
120	.0378	-234	40	151	1369	152	-2044
144	.0375	-210	40	152	-2044	154	-1559
146	.0375	-194	40	154	-1559	156	-1271
148	.0375	-173	40	156	-1271	158	345
150	.029	-134	40	158	345	160	-1832
			40				

NAME	DATE	LOCKHEED - GEORGIA COMPANY		TEMP.	PERM.
PREPARED	P.T. Fletcher	28 Jun 69	A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL 1 7.196
CHECKED	R. Wilkinson	28 Jun 69	TITLE	MODEL C-5A	
APPROVED	L.G. Darby	28 Jun 69	AFT FUSELAGE INTERNAL LOADS ANALYSIS	LG1US 46-2-2 REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

Rev. (A) PTF 1 Aug. 69



F.S. 2241-2209 (MOD. 5)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
111	1.13	11983	17			103	104
109	1.40	7563	17	103	104	101	38
107	1.08	6471	17	101	38	99	-6
105	1.06	5852	17	99	-6	97	-34
103	1.65	8733	17	97	-34	119	78
104	1.65	9606	17	98	111	119	78
106	1.06	6591	17	100	129	98	111
108	1.08	7057	17	102	166	100	129
110	1.40	6624	17	104	202	102	166
112	1.13	11065	17			104	202

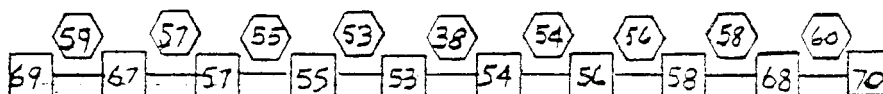
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
111	1.13	-15481	48			103	-220
109	1.40	-9643	48	103	-220	101	-134
107	1.08	-9045	48	101	-134	99	-59
105	1.06	-8158	48	99	-59	97	-9
103	1.65	-12037	48	97	-9	119	-68
104	1.65	-12774	48	98	-137	119	-68
106	1.06	-8780	48	100	-179	98	-137
108	1.08	-9529	48	102	-248	100	-179
110	1.40	-8783	48	104	-310	102	-248
112	1.13	-14889	48			104	-310

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
103	.029	142	40	111	-1262	109	-2022
101	.0375	185	40	109	-2022	107	342
99	.0375	197	40	107	342	105	615
97	.0375	212	40	105	615	103	699
119	.0375	-229	40	103	699	104	-1398
98	.0375	-213	40	104	-1398	106	-1054
100	.0375	-200	40	106	-1054	108	-747
102	.0375	-190	40	108	-747	110	1727
104	.029	-146	40	111	1727	112	2410

NAME	DATE	LOCKHEED - GEORGIA COMPANY	TEMP.	REV.
PREPARED	P.T. Fletcher	28 Jun 69	A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE VOL. 1 7.197
CHECKED	R. Wilkinson	28 Jun 69	TITLE	
APPROVED	L.G. Darby	28 Jun 69	AFT FUSELAGE INTERNAL LOADS ANALYSIS	
			MODEL	C-5A
			IGIUS 46-2-2	
			REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2273 - 2241 (MOD. 5)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
69	1.45	8414	17			59	117
67	1.26	10041	17	59	117	57	33
57	1.16	7979	17	57	33	55	-10
55	1.06	6771	17	55	-10	53	-38
53	1.65	10088	17	53	-38	38	77
54	1.65	10636	17	54	108	38	77
56	1.06	7186	17	56	127	54	108
58	1.16	8714	17	58	155	56	127
68	1.26	8681	17	60	243	58	155
70	1.45	8740	17			60	243

STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
69	1.45	-12074	48			59	-257
67	1.26	-13426	48	59	-257	57	-113
57	1.16	-11281	48	57	-113	55	-50
55	1.06	-9735	48	55	-50	53	-2
53	1.65	-14435	48	53	-2	38	-68
54	1.65	-14927	48	54	-132	38	-68
56	1.06	-10118	48	56	-172	54	-132
58	1.16	-11520	48	58	-226	56	-172
68	1.26	-12255	48	60	-375	58	-226
70	1.45	-12390	48			60	-375

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/E STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
59	.0375	184	40	69	-900	67	-2596
57	.0375	177	40	67	-2596	57	-181
55	.0375	198	40	57	-181	55	168
53	.0375	213	40	55	168	53	248
38	.0375	-225	40	53	248	54	-949
54	.0375	-214	40	54	-949	56	-533
56	.0375	-200	40	56	-533	58	-206
58	.0375	-179	40	58	-206	68	2371
60	.0375	-185	40	68	2371	66	1477

PREPARED	NAME P.T. Fletcher	DATE 28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	EMP. VOL. 1	TECH. 7.193
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		IGIUS 46-2-2 REPORT NO.		

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2317-2273 (MOD. 5)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
25	2.49	33729	17			21	235
23	1.38	13169	17	21	235	19	145
21	1.18	9828	17	19	145	17	25
19	1.06	8465	17	17	85	15	18
17	1.65	12941	17	15	18	37	29
18	1.65	13257	17	16	192	37	29
20	1.06	8814	17	18	254	16	192
22	1.18	9585	17	20	303	18	254
24	1.38	11409	17	22	370	20	303
26	2.49	30868	17			22	

STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
25	2.49	-47708	48			21	-432
23	1.38	-18469	48	21	-432	19	-237
21	1.18	-14064	48	19	-287	17	-193
19	1.06	-12556	48	17	-193	15	-89
17	1.65	-18876	48	15	-29	37	-79
18	1.65	-19204	48	16	-244	37	-79
20	1.06	-12751	48	18	-346	16	-244
22	1.18	-13940	48	20	-430	18	-346
24	1.38	-16999	48	22	-556	20	-430
26	2.49	-45548	48			22	-556

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
21	.0375	196	40	25	-4000	23	-3157
19	.0375	229	40	23	-3157	21	-316
17	.0375	242	40	21	-316	19	-181
15	.0375	247	40	19	-181	17	-51
37	.0375	-251	40	17	-53	18	-628
16	.0375	-246	40	18	-627	20	-199
18	.0375	-240	40	20	-200	22	487
20	.0375	-225	40	22	437	24	2312
22	.0375	-198	40	24	2312	26	5716

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL. 1	7.199
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	IGIUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2339 - 2317 (MOD. 6)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
181	2.46	31600	17			169	179
179	1.44	116205	17	169	179	167	178
173	1.11	12080	17	167	178	165	95
171	1.06	10984	17	165	95	163	18
169	1.65	116460	17	163	18	148	82
170	1.65	116683	17	164	192	148	82
172	1.06	11037	17	166	269	164	192
174	1.11	11631	17	168	356	166	269
180	1.44	14537	17	170	292	169	356
182	2.46	30199	17			170	292

STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
181	2.46	-46383	43			169	-338
179	1.44	-23357	43	169	-338	167	-341
173	1.11	-17566	43	167	-341	165	-210
171	1.06	-16185	43	165	-210	163	-90
169	1.65	-24416	43	163	-90	148	-77
170	1.65	-24692	43	164	-244	148	-77
172	1.06	-16335	43	166	-366	164	-244
174	1.11	-17312	43	168	-503	166	-366
180	1.44	-22056	43	170	-440	168	-503
182	2.46	-45510	43			170	-440

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
169	.031	153	40	181	-2769	179	-2763
167	.0375	259	40	179	-2763	173	-962
165	.0375	243	40	173	-962	171	-244
163	.0375	245	40	171	-244	169	-129
148	.0275	-245	40	169	-129	170	-435
164	.0375	-245	40	170	-435	172	-65
166	.0375	-245	40	172	-65	174	774
168	.0375	-251	40	174	774	180	2524
170	.031	-153	40	180	2524	182	2729

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL. 1	7.200
CHECKED	R. Wilkinson	28 Jun 69	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS	MODEL	C-5A	
APPROVED	L.G. Darby	28 Jun 69		LGUS 46-2-2	REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) R.T.F. 1 AUG. 69



F.S. 2361 - 2339 (MOD. 6)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
141	3.01	46312	17			131	141
139	1.41	20973	17	131	141	129	152
137	1.09	13700	17	129	152	127	86
135	1.06	12598	17	127	86	125	14
133	1.65	18687	17	125	14	147	85
134	1.65	18914	17	126	184	147	85
136	1.06	12683	17	128	259	126	184
138	1.09	13372	17	130	332	128	259
140	1.41	19687	17	132	239	130	332
142	3.01	45468	17			132	239

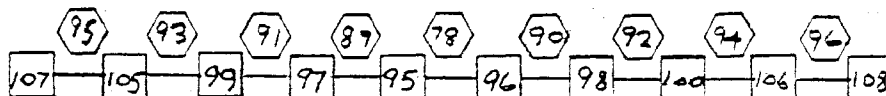
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
141	3.01	-69251	43			131	-275
139	1.41	-30366	43	131	-275	129	-303
137	1.09	-20191	43	129	-303	127	-197
135	1.06	-18781	43	127	-197	125	-82
133	1.65	-28004	43	125	-82	147	-75
134	1.65	-28304	43	126	-233	147	-75
136	1.06	-18987	43	128	-351	126	-233
138	1.09	-20090	43	130	-465	128	-351
140	1.41	-29725	43	132	-363	130	-465
142	3.01	-69170	43			132	-363

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
131	.0305	135	40	141	-1407	139	-1530
129	.0375	257	40	139	-1503	137	-694
127	.0375	245	40	137	-674	135	-169
125	.0375	240	40	135	-169	133	-111
147	.0375	-233	40	133	-111	134	-502
126	.0375	-233	40	134	-502	136	-113
128	.0375	-243	40	136	-113	138	602
130	.0375	-252	40	138	602	140	2615
132	.0305	-147	40	140	2616	142	2653

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	Vol. 1	7.201
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APPROVED	L.G. Darby	28 Jun 69		LGJUS 46-2-2 REPORT NO.		

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2383-2361 (MOD. 6)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
107	3.41	57653	17			95	125
105	2.15	30640	17	95	125	93	100
99	1.06	15003	17	93	100	91	46
97	1.06	13982	17	91	46	89	-9
95	1.65	20523	17	89	-9	78	79
96	1.65	20789	17	90	151	78	79
98	1.06	14133	17	92	209	90	151
100	1.08	14839	17	94	271	92	209
106	2.15	30257	17	96	250	94	271
108	3.41	58853	17			96	250

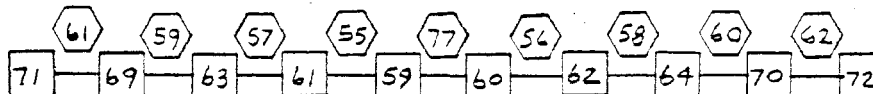
STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
107	3.41	-87470	48			95	-269
105	2.15	-45406	48	95	-269	93	-223
99	1.06	-22355	48	93	-223	91	-134
97	1.06	-21044	48	91	-134	89	-45
95	1.65	-31013	48	89	-45	78	-68
96	1.65	-31261	48	90	-185	78	-68
98	1.06	-21339	48	92	-278	90	-185
100	1.08	-22438	48	94	-375	92	-278
106	2.15	-45641	48	96	-380	94	-375
108	3.41	-89516	48			96	-380

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/E STRG.	AXIAL LOAD (lbs.)	R/E STRG.	AXIAL LOAD (lbs.)
95	.0475	-273	74	107	-24611	105	15304
93	.0375	-269	74	105	15304	99	5401
91	.0375	-251	74	99	-5401	97	2007
89	.0375	-294	74	97	2007	95	-1943
78	.0375	205	74	95	-1943	96	-12015
90	.0375	176	74	96	-12015	98	-11532
92	.0375	148	74	98	-11532	100	-15525
94	.0375	129	74	100	-15525	106	-36034
96	.0475	-96	74	106	-36034	108	-66049

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APPROVED	L.G. Darby	28 Jun 69		LG1US 46-2-2	REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2405 - 2383 (MOD. 6)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
71	3.75	71802	17			61	-78
69	2.23	37766	17	61	-78	59	39
63	1.05	15821	17	59	40	57	19
61	1.06	15072	17	57	19	55	-24
59	1.45	21870	17	55	-24	77	76
60	1.15	22200	17	56	131	77	76
62	1.06	15317	17	58	179	56	131
64	1.05	15834	17	60	207	58	179
70	2.23	38731	17	62	47	60	207
72	3.75	74646	17			62	47

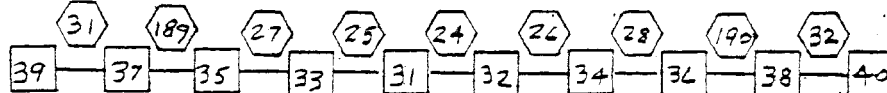
STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTED. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
71	3.75	-109460	48			61	33
69	2.23	-56813	48	61	33	59	-134
63	1.05	-23762	48	59	-134	57	-94
61	1.06	-22823	48	57	-94	55	-22
59	1.45	-33240	48	55	-22	77	-65
60	1.45	-33666	48	56	-157	77	-65
62	1.06	-23218	48	58	-233	56	-157
64	1.05	-24029	48	60	-280	58	-233
70	2.23	-58400	48	62	-73	60	-280
72	3.75	-113331	48			62	-73

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
61	.0615	-386	74	71	26869	69	15459
59	.0375	-302	74	69	15459	63	4831
57	.0375	-270	74	63	4831	61	1573
55	.0375	-249	74	61	1573	59	-2567
77	.0375	222	74	59	-2567	60	-12548
56	.0375	204	74	60	-12548	62	-12029
58	.0375	188	74	62	-12029	64	-15747
60	.0375	197	74	64	-15747	70	-41970
62	.0615	354	74	70	-41970	72	-79160

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL 1	7.203
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APPROVED	L.G. Darby	28 Jun 69		LG1US 46-2-2 REPORT NO.		

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) D.T.F. 1 AUG. 69



F.S. 2427 - 2405 (MOD. 6)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
39	4.03	90953	17			31	-165
37	2.31	42239	17	31	-165	189	67
35	1.00	16622	17	189	67	27	16
33	1.06	16115	17	27	16	25	-35
31	1.65	22888	17	25	-35	24	74
32	1.65	23308	17	26	119	24	74
34	1.06	14667	17	28	174	26	119
36	1.00	15031	17	190	230	28	174
38	2.31	39337	17	32	-17	190	230
40	4.03	83555	17			32	17

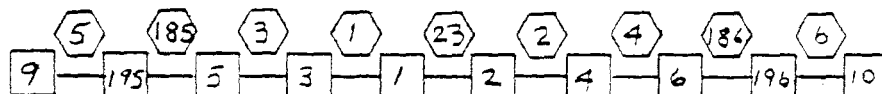
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
39	4.03	-138963	48			31	146
37	2.31	-64846	48	31	146	189	-172
35	1.00	-25159	48	189	-172	27	-88
33	1.06	-24576	48	27	-88	25	-6
31	1.65	-35023	48	25	-6	24	-64
32	1.65	-35540	48	26	-138	24	-64
34	1.06	-25086	48	28	-226	26	-138
36	1.00	-25569	48	190	-315	28	-226
38	2.31	-68126	48	32	-8	190	-315
40	4.03	-143742	48			32	-8

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
31	.098	-516	74	39	32160	37	11999
189	.0375	-279	74	37	11999	35	4453
27	.0375	-279	74	35	4453	33	1175
25	.0375	-265	74	33	1175	31	-3157
24	.0375	240	74	31	-3157	32	-12826
26	.0375	229	74	32	-12886	34	-12529
28	.0375	201	74	34	-12529	36	-16127
190	.0375	159	74	36	-16127	38	-43012
32	.098	530	74	38	-43012	40	-98617

PREPARED	NAME P.T. Fletcher	DATE 28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	TEMP. Vol. 1	RECD. 7.204
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APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	LGIUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. 2 P.T.F. 1 AUG. 69



F.S. 2450-2427 (MOD. 6)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
9	3.55	93807	17			5	-235
195	2.03	42947	17	5	-235	185	43
5	.96	17873	17	185	43	3	-13
3	1.06	17177	17	3	-13	1	-53
1	1.65	23642	17	1	-53	23	76
2	1.65	24189	17	2	25	23	76
4	1.06	17715	17	4	158	2	106
6	.96	18234	17	186	229	4	158
196	2.03	46330	17	6	-151	186	229
10	3.55	79840	17			6	-151

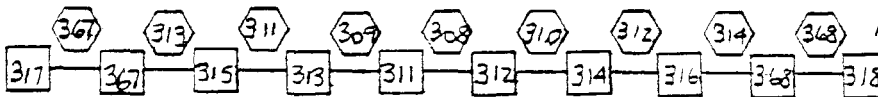
STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
9	3.55	-144270	48			5	283
195	2.03	-66496	48	5	283	185	-150
5	.96	-27154	48	185	-150	3	-54
3	1.06	-26283	48	3	-54	1	17
1	1.65	-36327	48	1	17	23	-65
2	1.65	-36950	48	2	-119	23	-65
4	1.06	-26978	48	4	-201	2	-119
6	.96	-27794	48	186	-313	4	-201
196	2.03	-70305	48	6	173	186	-313
10	3.55	-149633	48			6	173

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
5	.0375	-422	74	9	29085	195	9256
185	.0375	-363	74	195	9256	5	4030
3	.0375	-327	74	5	4030	3	5754
1	.0375	-288	74	3	5754	1	-3827
23	.0375	254	74	1	-3827	2	-12935
2	.0375	262	74	2	-12935	4	-12935
4	.0375	263	74	4	-12935	6	-16778
186	.0375	248	74	6	-16778	196	-41396
6	.0375	520	74	196	-41396	10	-98330

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.		PAGE	VOL. 1	7.205
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APPROVED	L.G. Darby	28 Jun 69			LGUS 46-2-2	REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69
REV. B FES 1-8-71



F.S. 2472-2450 (MOD. 7)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
317	2.57	80480	1			367	-541
367	2.03	36633	1	367	-541	313	-93
315	.84	18119	1	313	-93	311	-38
313	1.06	17839	1	311	-38	309	-47
311	1.62	24340	1	309	-47	308	78
312	1.62	25018	1	310	117	308	78
314	1.06	18581	1	312	139	310	117
316	.84	18911	1	314	114	312	139
368	2.03	38010	1	368	-451	314	114
318	2.57	86525	1			368	-451

STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
317	2.57	-129747	28			367	811
367	2.03	-58457	28	367	811	313	60
315	.84	-29073	28	313	60	311	-11
313	1.06	-28737	28	311	-11	309	12
311	1.62	-39171	28	309	12	308	-68
312	1.62	-39843	28	310	-131	308	-68
314	1.06	-29485	28	312	-164	310	-131
316	.84	-29924	28	314	-119	312	-164
368	2.03	-60032	28	368	739	314	-119
318	2.57	-135849	28			368	739

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
367	.109	756	19	317	-74316	367	-35934
313	.0375	450	19	367	-35934	315	-15709
311	.0375	318	19	315	-15709	313	-12709
309	.0375	276	19	313	-12709	311	-13488
308	.0375	-267	19	311	-13488	312	-6000
310	.0375	-305	19	312	-6000	314	-1663
312	.0375	-360	19	314	-1663	316	1130
314	.0375	-462	19	316	1130	368	6599
368	.109	-369	19	368	6599	318	8612

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRENGTH	AXIAL LOAD (lbs.)	R/H STRENGTH	AXIAL LOAD (lbs.)
289	.0375	41	19	365	-33313	291	-16406
287	.0375	235	19	291	-16406	289	-12633
285	.0375	271	19	289	-12633	287	-13350
307	.0375	-268	19	287	-13350	288	-6842
286	.0375	-306	19	288	-6842	290	-2698
288	.0375	-344	19	290	-2698	292	230
290	.0375	-323	19	292	230	344	7864

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	Vol. 1	7.207
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APPROVED	L.G. Darby	28 Jun 69		REPORT NO.	IGIUS 46-2-2	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2516 - 2494 (MOD. 7)

STRG. NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
271	3.40	75803	1			267	-237
269	.90	20449	1	267	-237	265	-58
267	1.07	19768	1	265	-58	263	-53
265	1.68	25882	1	263	-53	262	83
266	1.68	26904	1	264	118	262	83
268	1.07	21071	1	266	122	264	118
270	.90	21566	1	268	-48	266	122
272	3.40	84643	1			268	-48

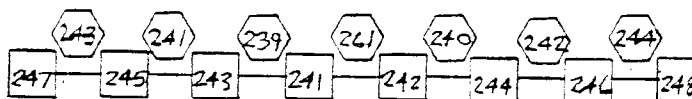
STRG. NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
271	3.40	-125027	28			267	294
269	.90	-33145	28	267	294	265	21
267	1.07	-32166	28	265	21	263	20
265	1.68	-41891	28	263	20	262	-72
266	1.68	-42856	28	264	-128	262	-72
268	1.07	-33328	28	266	-135	264	-128
270	.90	-34265	28	268	133	266	-135
272	3.40	-133402	28			268	133

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/H STRG.	AXIAL LOAD (lbs.)	R/H STRG.	AXIAL LOAD (lbs.)
267	.0375	574	19	271	-56624	269	-15912
265	.0375	334	19	269	-15912	267	-12496
263	.0375	287	19	267	-12496	265	-13278
262	.0375	-279	19	265	-13228	266	-7639
264	.0375	-313	19	266	-7639	268	-3657
266	.0375	-361	19	268	-3657	270	-693
268	.0375	-463	19	270	-693	272	-6932

PREPARED	P.T. Fletcher	28 Jun 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORP.	PAGE	VOL. 1	7,208
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APPROVED	L.G. Darby	28 Jun 69		LGUS 46-2-2	REPORT NO.	

7.6.1 Critical Flight Cases - Torque Shelf

REV. (A) P.T.F. 1 AUG. 69



F.S. 2538 - 2516 (MOD.7)

STRG NO.	AREA (in ²)	CRIT. TENS. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
247	3.25	64423	1			243	-238
245	.81	17164	1	243	-238	241	-120
243	1.06	19163	1	241	-120	239	-71
241	1.67	26397	1	239	-71	241	88
242	1.67	27520	1	240	108	241	88
244	1.06	20602	1	242	62	240	108
246	.81	18458	1	244	-49	242	62
248	3.25	73237	1			244	-49

STRG NO.	AREA (in ²)	CRIT. COMP. LOAD (lbs.)	CASE	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
247	3.25	-108018	28			243	300
245	.81	-28106	28	243	300	241	119
243	1.06	-31330	28	241	119	239	45
241	1.67	-42813	28	239	45	241	-76
242	1.67	-43858	28	240	-110	241	-76
244	1.06	-32661	28	242	-37	240	-110
246	.81	-29335	28	244	139	242	-37
248	3.25	-116777	28			244	139

COVER NO.	THICK (ins)	CRIT. SHEAR LOAD (lb/in)	CASE	L/R STRG.	AXIAL LOAD (lbs.)	R/L STRG.	AXIAL LOAD (lbs.)
243	.0375	538	19	247	-39685	245	-11713
241	.0375	399	19	245	-11713	243	-11168
239	.0375	326	19	243	-11168	241	-12938
241	.0375	-308	19	241	-12938	242	-3388
240	.0375	-342	19	242	-8388	244	-4581
242	.0375	-377	19	244	-4581	246	-2406
244	.0375	-427	19	246	-2406	248	-15383

NAME	DATE	LOCKHEED-GEORGIA COMPANY	TEMP.	PERM.
Prepared P.T. Fletcher	15 Aug 68	A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	Page Vol. I	7,209
Checked J. Halliday	15 Aug 68	TITLE	Model C-5A	
Approved L.G. Darby	15 Aug 68	AFT FUSELAGE INTERNAL LOADS ANALYSIS	LGUS 46-2-2	Report No.

REV. (A) P.T.F. 1 AUG. 69.

7.6.1

AFT TORQUE DECK PANEL & COVERS

& STRINGERS (SCAN.) DERIVED FROM REF. 3 - VOL. 4
SECT. 2 REV.(A)

CASE CODE	DESCRIPTION	CASE REF. NO.
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5 P E 2 0 1	POSITIVE PITCH ENTRY	48
1 P C 2 4 2	NEGATIVE PITCH CHECK	17
D M L 0 0 1	DYNAMIC LATERAL GUST	74
5 A R 1 5 4 T 2	AILERON ROLL	40
1 0 1 5 0 0 . 1 3	DYNAMIC LANDING	63
5 A R 1 5 4 T 3	AILERON ROLL (HI-DRAG)	41
5 E M 1 1 4	EQUILIBRIUM YAW	8
1 E Y 1 0 7	NEG. MANEUVER (HI-DRAG)	44
5 N M 0 6 4	DYNAMIC LATERAL GUST	76
D M L 0 0 3	DYNAMIC TAXI	60
1 P C 2 4 2		
1 0 4 0 0 1 . 1 8		
1 P C 2 4 2		
5 E M 1 1 4		

PREPARED	NAME A.D. PINCHOT	DATE 1 AUG. 69	LOCKHEED - GEORGIA COMPANY A DIVISION OF LOCKHEED AIRCRAFT CORPORATION	PAGE	TEMP. VOL. I	PERM. 
CHECKED	R.F. WILKINSON	1 AUG. 69	TITLE INTERNAL LOADS ANALYSIS LFT FUSELAGE	MODEL C-5A		
APPROVED	LG. DARBY	1 AUG. 69		LG10S46-2-2 REPORT NO.		

REV. (2) ADP 1 AUG. 69

REV. (3) ADP 1-8-71

 PAGES 7.210 - 7.217 "NOT USED"

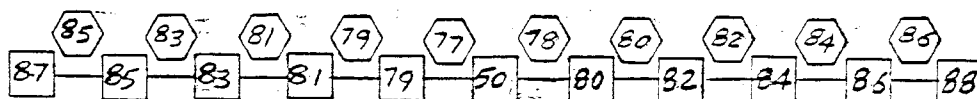
7.218 - 7.234 "DELETED"

7.235 - 7.244 "NOT USED"

NAME		DATE	LOCKHEED - GEORGIA COMPANY		TEMP.	PERM.
PREPARED	P.T. Fletcher	28 Jun 68	A DIVISION OF LOCKHEED AIRCRAFT CORP.		PAGE	Vol. I 7.245
CHECKED	J. Halliday	28 Jun 68	TITLE		C-5A	
APPROVED	L.G. Darby	28 Jun 68	AFT FUSELAGE INTERNAL LOADS ANALYSIS		MODEL	
					LGIUS 46-2-2	
					REPORT NO.	

7.6.1 Aft Torque Deck, Ult. Press (8.7 x 1.5) PSI

Stringers and Adj. Covers



F.S. 2145-2101 (MOD. 4)

STRG. NO.	AREA (in ²)	TENSION LOAD (lbs.)	COVER THICK	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
87	.67	2050				85	482
85	1.41	-6309	.045	85	482	83	372
83	1.03	-6176	.045	83	372	81	230
81	1.23	-8406	.045	81	230	79	150
79	1.31	-12887	.045	79	150	77	357
50	1.32	-16345	.045	77/78	357/354		
80	1.31	-12884	.045	80	147	78	354
82	1.23	-8431	.045	82	227	80	147
84	1.03	-6171	.045	84	368	82	227
86	1.41	-6349	.045	86	481	84	368
88	.67	2042				86	481



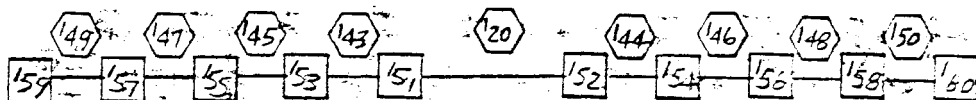
F.S. 217.7-2145 (MOD. 4)

STRG. NO.	AREA (in ²)	TENSION LOAD (lbs.)	COVER THICK	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
37	.91	-4796				33	294
35	1.14	-5111	.038	33	294	31	183
29	.93	-3050	.038	31	188	29	197
27	1.06	-6198	.038	29	197	27	175
25	1.65	-14789	.038	27	175	151	-1.5
26	1.65	-14804	.038	28	171	151	-1.5
28	1.06	-6217	.038	30	193	28	171
30	.93	-3087	.038	32	184	30	193
36	1.14	-5133	.038	34	289	32	184
38	.91	-4841				34	289

NAME		DATE	LOCKHEED - GEORGIA COMPANY		TEMP.	PERM.
PREPARED	P.T. Fletcher	28 Jun 68	A DIVISION OF LOCKHEED AIRCRAFT CORP.		PAGE	Vol. I 7.246
CHECKED	J. Halliday	28 Jun 68	TITLE AFT FUSELAGE INTERNAL LOADS ANALYSIS		MODEL	C-5A
APPROVED	L.G. Darby	28 Jun 68			LGIUS	46-2-2
					REPORT NO.	

7.6.1 Aft Torque Deck, Ult. Press (8.7 x 1.5) PSI

Stringers and Adj. Covers



F.S. 2209-2177 (MOD. 5)

STRG. NO.	AREA (in ²)	TENSION LOAD (lbs.)	COVER THICK	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
159	1.10	4115				149	191
157	1.23	-3235	.029	149	191	147	183
155	1.01	-2865	.038	147	183	145	163
153	1.06	-5035	.038	145	163	143	114
151	1.65	-10119	.038	143	114	120	-1.5
152	1.65	-10139	.038	144	111	120	-1.5
154	1.06	-5063	.038	146	160	144	111
156	1.01	-2915	.038	148	180	146	160
158	1.23	-3278	.029	150	187	148	180
160	1.10	371				150	187



F.S. 2241-2209 (MOD. 5)

STRG. NO.	AREA (in ²)	TENSION LOAD (lbs.)	COVER THICK	OUTBD. COVER	SHEAR LOAD (lb/in)	INBD. COVER	SHEAR LOAD (lb/in)
111	1.13	174				103	143
109	1.40	-3100	.029	103	143	101	143
107	1.08	-2293	.038	101	143	99	128
105	1.06	-3564	.038	99	128	97	87
103	1.65	-6882	.038	97	87	119	-1.5
104	1.65	-6904	.038	98	84	119	-1.5
106	1.06	-3597	.038	100	125	98	84
108	1.08	-2346	.038	102	139	100	125
110	1.40	-3150	.029	104	140	102	139
112	1.13	60				104	140

[illegible]